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Afterbody/Nozzle Pressure Distributions of a Twin-Tail Twin-Engine Fighter With Axisymmetric Nozzles at Mach Numbers From 0.6 to 1.2

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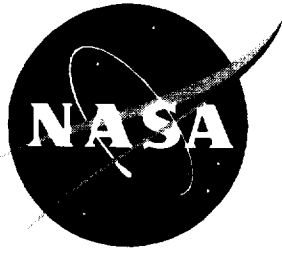
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Summary

Distributions of static pressure coefficient over the afterbody and axisymmetric nozzles of a generic, twin-tail twin-engine fighter aircraft were obtained in the Langley 16-Foot Transonic Tunnel. Three vertical-tail longitudinal positions and two horizontal-tail longitudinal positions were tested for a total of six aft-end configurations. Static pressure coefficients were obtained at Mach numbers of 0.6, 0.8, 0.9, and 1.2, angles of attack of 0°, 4°, and 8°, and nozzle pressure ratios ranging from jet-off to 8. Pressure measurements were made at approximately 150 locations on the afterbody, interfairing, and nozzles. Additionally, limited computational modeling of the flow field about this model was performed using a three-dimensional Reynolds-averaged Navier-Stokes code.

The results of this investigation indicate that the influence of the vertical and horizontal tails is not limited to the vicinity of the tail-afterbody juncture. The pressure distribution affecting the aft-end drag is influenced more by the position of the vertical tails than by the position of the horizontal tails. Forward positioning of the vertical tails reduces the aft-end drag by moving the pressure deficit between the tails to a surface with a more shallow boattail angle and less closure area.

At supersonic free-stream conditions, aft vertical- or horizontal-tail positions produce low static pressures on the nozzle surfaces, and this results in higher aft-end drag attributable to rapid aft-end area distribution closure.

Transonic effects strong enough to produce a local shock wave between the vertical tails are evident at a Mach number of 0.9. Increasing the angle of attack can have the effect of producing transonic tail-interference effects at lower free-stream Mach numbers than at an angle of attack of 0°.

Predictions of surface pressure coefficient from computational modeling of this geometry showed excellent agreement with experimental results with the exception of the recovery pressure level downstream of the upper surface transonic shock at a Mach number of 0.9 and vertical-tail-position effects on pressure coefficients on the windward side of the aircraft.

Introduction

Investigations of generic fighter configurations have shown that the relative locations and placement of the empennage surface on the afterbody have a significant effect on aft-end drag (refs. 1–6). These investigations have covered single-engine configurations (refs. 1–4) and twin-engine configurations with both single and twin vertical tails (refs. 5 and 6). Studies of tail-

interference effects are of significant value because aft-end drag (i.e., 35 percent aft of the aircraft) has been shown to contribute up to 50 percent of the total aircraft drag at transonic conditions (ref. 7).

A series of wind-tunnel tests were previously conducted on an isolated fuselage-nozzle combination with propulsion simulation to investigate empennage effects on aft-end drag of a single-engine configuration (refs. 1–3). In these tests, total aft-end drag in the transonic speed range as measured by the force balance was reduced by staggering the vertical and horizontal tails longitudinally. The extent of the tail-interference effects could not be determined, however, without detailed pressure measurements in the vicinity of the tails. Therefore, additional tests were conducted to acquire approximately 300 pressure measurements distributed around the afterbody and nozzles (ref. 4). These pressure distributions showed that tail-interference effects extended around the entire afterbody and were not confined to the regions close to the tail surfaces. An additional benefit of having the mapping of the pressure distribution was the utility of the data for validation of computational fluid dynamics (CFD) codes. The permutations in aft-end geometry can be extensive, and using reliable CFD tools to determine the drag characteristics of these afterbodies can be an efficient means of evaluating the configurations. For CFD code validation, the calculations must be compared with experimentally determined data to determine the reliability of the code for such applications. Reference 4 includes such comparisons for two types of theoretical prediction codes. Additionally, the model from reference 4 was used as a baseline configuration in a CFD study in reference 8 to reduce total aft-end drag by afterbody reshaping. Results from this study were used to fabricate new wind-tunnel models for validation.

Wind-tunnel studies of a generic twin-engine fighter configuration were also conducted with emphasis on aft-end drag characteristics. As in the single-engine studies, the force balance was the primary source of data (refs. 5 and 6). The variables included single and twin vertical tails, the relative positioning of vertical and horizontal tails, vertical-tail toe angle (the pointing angle in the horizontal plane), vertical-tail cant angle (the tilt angle from the vertical plane), and vertical-tail camber. Force balance data indicated that significant reductions in aft-end drag were attainable through tailoring of the aft-end geometry. Again, for purposes of determining the extent of tail interference around the afterbody and providing a database for CFD code validation, the model used in these investigations was reinstrumented with pressure orifices and a limited number of configurations were retested. The pressure distributions measured in this test are presented in this report.

The model was tested in the Langley 16-Foot Transonic Tunnel, and the Mach number range of the test was 0.6 to 1.2. Propulsion was simulated by exhausting high-pressure room-temperature air through twin, axisymmetric nozzles with a geometry representing a dry power setting (no afterburning). The nozzle pressure ratio (NPR) range for an angle of attack of 0° was jet-off to 8. The model was also tested at angles of attack of 4° and 8° , with the NPR set at both jet-off and design (fully expanded) conditions. Twin, uncambered vertical tails were tested at toe and cant angles of 0° in three longitudinal positions: forward, middle, and aft. Horizontal tails at incidence angles of 0° were tested in the corresponding positions of middle and aft, resulting in a total of six geometric configurations. Approximately 150 static pressure measurements were acquired on the surface at each test condition.

Additionally, limited computational modeling of the flow field about this model was performed using PAB3D, a three-dimensional (3D) Reynolds-averaged Navier-Stokes code developed at the Langley Research Center. Comparisons of computed surface pressure coefficients with experimental results are presented for two configurations at transonic free-stream conditions.

Symbols

C_D	total aft-end drag coefficient, referenced to wing area of support system
C_L	aft-end lift coefficient, referenced to wing area of support system
C_p	pressure coefficient, $(p_s - p_\infty) / q_\infty$
C_p^*	critical pressure coefficient for which local flow is sonic
$C_{p, \text{un}}$	estimated uncertainty in pressure coefficient
ΔC_p	pressure coefficient with horizontal tail in mid position subtracted from pressure coefficient with horizontal tail in aft position
c	chord length, in.
d_{max}	maximum nozzle diameter at nozzle connect station, 3.88 in.
L	length of model from nose to nozzle exit, 68.80 in.
M	Mach number
NPR	nozzle pressure ratio, $p_{t,j} / p_\infty$
p_s	surface static pressure, psi
$p_{t,j}$	jet total pressure, psi
p_∞	free-stream static pressure, psi
q_∞	free-stream dynamic pressure, psi
r	nozzle radius, in.

r_{max}	maximum nozzle radius at nozzle connect station, $d_{\text{max}}/2$
t	airfoil thickness, in.
x	axial distance from model nose, in.
x/L_{vt}	x/L of vertical-tail root leading edge
α	forebody angle of attack, deg
θ	meridian angle on nacelles or nozzles measured outboard from centerline on top surface, deg
Λ_{le}	sweep angle at leading edge, deg

Abbreviations:

BL	buttock line
CFD	computational fluid dynamics
FS	fuselage station
fwd	forward
HTAIL	horizontal-tail position
mid	middle
TS	tunnel station
VTAIL	vertical-tail position
WL	waterline

Apparatus and Methods

Wind Tunnel

The test was conducted in the Langley 16-Foot Transonic Tunnel, which is a closed-circuit, atmospheric wind tunnel with an octagonal, slotted test section. (See sketch in fig. 1.) The Mach number range of the tunnel is 0.1 to 1.3 and is continuously variable. Supersonic Mach numbers are achieved by removing air from a plenum chamber surrounding the slotted test section, thus allowing more tunnel air to expand into the chamber. Reynolds number per foot, which is a function of Mach number and atmospheric conditions, ranges from approximately 3×10^6 at $M = 0.6$ to 4×10^6 at $M = 1.2$. A complete description of the tunnel and its subsystems is provided in reference 9.

Support System

The support system used for this model is the bifurcated wingtip support system. A photograph is shown in figure 2 of both the forward part of the support system and the model installed in the tunnel. The support system consists of a pair of support booms extending forward from the main tunnel strut and attaching to the wingtips of the model. The support booms are capable of internally housing instrumentation cables and tubing and

passing high-pressure air to the model for propulsion simulation.

This support system, which was designed to provide minimal support interference to the model while allowing a more realistic aerodynamic flow field to develop over the afterbody (the area of interest), was superior to a blade-mounted support system. Although no support-interference tests have been conducted using this system, support-interference results on a similar wingtip support system (ref. 10) indicated that relatively low levels of interference exist compared with that of other propulsion support systems. The wingtip support system also provided the ability to test models at high angles of attack by allowing the wing incidence angle to be varied with respect to the support booms, although this capability was not utilized in the present study. The model was placed at angles of attack by rotation of the main support strut to which the wingtip support system was attached.

Model

The model represented a generic fighter design with faired-over inlets and a twin-engine exhaust system with axisymmetric nozzles. A sketch of the model with the major components labeled is shown in figure 3(a). The wing was mounted high on the fuselage with a leading-edge sweep of 45° , a taper ratio of 0.4, an aspect ratio of 2.32, and a cranked trailing edge. The airfoil was symmetrical, and the thickness ratios from the wing-fuselage junction at BL 5.00 to BL 11.00 ($t/c = 0.066$ to 0.077 , respectively) were typical of fighter aircraft. From BL 11.00 to the support booms, wing thickness ratio increased from 0.077 to 0.100 . (The thickness was held constant while the chord was reduced.) This constant thickness provided structural support for the model and permitted the transfer of instrumentation cables and high-pressure air from the booms to the model propulsion system. The force balance used in earlier tests of this model was replaced by a dummy balance (a solid 2-in-diameter steel rod).

The center fuselage was essentially rectangular in cross section, with a constant width and height of 10 in. and 5 in., respectively. The corners were rounded to a radius of 1.00 in. The maximum cross-sectional area of the fuselage was 49.14 in^2 , and the forebody was vertically and horizontally symmetric (i.e., no canopy) and had faired-over inlets, as is typical for exhaust-flow simulation models. The afterbody represented a smooth blend from the rectangular fuselage cross section (FS 47.26) to the axisymmetric nozzle connect station (FS 65.21). The afterbody lines were chosen to be typical of current fighter designs with closely spaced engine nacelles centered at $\text{BL} \pm 2.17$.

Propulsion was simulated by routing high-pressure room-temperature air (540°R entering the nozzles) through the support system to the model and allowing it to exhaust through the nozzles. A sketch showing the internal details of the model propulsion system is shown in figure 3(b). A choke plate, flow straightener, four total pressure probes, and a total temperature probe were located in ducting immediately upstream of each nozzle. The system was designed with symmetrical flow paths to provide equal flow rates to the two nozzles. The nozzles used in this test represented a dry power setting with an expansion ratio (nozzle exit area divided by throat area) of 1.16 and a throat area of 2.69 in^2 for each nozzle. A sketch of the nozzle geometry is shown in figure 3(c).

The model was tested with twin, uncambered vertical tails with chord planes positioned at $\text{BL} \pm 4.00$ on the afterbody. (See geometric details of vertical tails in fig. 3(d).) The leading-edge sweep was 36.52° , the taper ratio was 0.375, and the aspect ratio was 1.514. The tails were tested in three longitudinal positions, referred to as fwd (forward), mid (middle), and aft (rearward), corresponding to root leading-edge positions of FS 50.00, FS 53.81, and FS 57.31, respectively. Fillers were contoured for each tail location to fair the tails into the afterbody. The tails in the aft position, however, were not faired to the nozzle, and this resulted in a gap between the nozzles and the tails.

The horizontal tails were tested at 0° incidence with the chord plane positioned at $\text{WL} 0.0$. The geometric details of the horizontal tails are shown in figure 3(e). The leading-edge sweep was 50° , the taper ratio was 0.175, and the aspect ratio was 2.564. The horizontal tails were tested in two longitudinal positions corresponding to the middle and aft positions of the vertical tails (root leading edges at FS 53.81 and FS 57.31, respectively). Fillers were also used to contour the horizontal tails to the afterbody at each tail location. The longitudinal positions of the vertical and horizontal tails tested are shown in figure 3(f). The three vertical-tail positions and the two horizontal-tail positions result in a total of six configurations tested, and these were the same locations as those tested in reference 5.

Instrumentation

Static pressures were measured on the external surfaces of the afterbody, nozzles, and engine interfairing, and the locations of the measurement stations are listed in table 1. Approximately 150 pressure orifices were distributed around the model. Because of the density of the pressure measurements desired, symmetry was assumed in the flow field about the right and left halves of the model, and pressure orifices were distributed on both halves of the model to accommodate the internal

plumbing. The orifices were routed to electronic pressure-scanning modules mounted in the fuselage with a range from 0 to 30 psia.

Angle of attack was measured by an accelerometer located in the forebody, and a correction of 0.1° was made for the tunnel upflow angle based on upflow measurements of similar models in the tunnel. Deflections of the dummy balance under load would result in a difference in afterbody angle of attack as compared with the value measured in the forebody. A deflection was estimated from the lift and pitching-moment characteristics of this model published in reference 5 at the worst case condition of $\alpha = 8^\circ$ at $M = 1.2$ with the horizontal tails in the aft position. Under these conditions, the afterbody was estimated to deflect 0.15° , thereby reducing the afterbody angle of attack. No corrections for afterbody deflections, however, were applied to the angles of attack tabulated in this report.

Test Conditions, Transition Control, and Data Uncertainty

Data were taken at Mach numbers of 0.6, 0.8, 0.9, and 1.2. At each Mach number, the NPR was varied between jet-off and 8 at an angle of attack of 0° . Data at angles of attack of 4° and 8° were also taken at each Mach number with both the jet off and $\text{NPR} = 3.4$ (near the nozzle design pressure ratio of 3.54, the condition of fully expanded flow).

To ensure proper boundary-layer transition, 0.1-in-wide strips of No. 120 silicon carbide grit, sparsely distributed in a lacquer film, were applied to the model forebody section, the upper and lower surfaces of the wings and horizontal tails, and the left and right sides of the vertical tails, as recommended in reference 11. The forebody strip was placed in a ring 1 in. aft of the nose tip, the wing strips were placed 1 in. aft of the leading edge measured in the streamwise direction, and the strips for the horizontal and vertical tails were placed 1/2 in. aft of the leading edge.

An analysis was performed to determine the uncertainty of the pressure coefficient data. The dominant adverse effect on the data accuracy is the temperature dependency of the electronic pressure-scanning modules. As the modules heat up during a tunnel run, the calibrations tend to lose their precision; therefore, to minimize this effect, the modules are recalibrated with the tunnel running at each new Mach number. Because an uncertainty exists in the values used to calculate the pressure coefficient, including the temperature effect on the modules, the uncertainty of the pressure coefficient has been estimated for each Mach number tested, and the results are presented in the following chart:

M	$C_{p, \text{un}}$
0.6	± 0.016
0.8	± 0.011
0.9	± 0.010
1.2	± 0.008

The uncertainty of the angle-of-attack measurement is estimated to be $\pm 0.05^\circ$ and is limited primarily by the knowledge of the tunnel upflow angle. The bias associated with deflections of the afterbody under load were discussed in the "Instrumentation" section of this report.

Data Reduction and Presentation of Results

Each data point represents the average of 5 sec of data taken at a rate of 10 samples/sec. This average value was used to compute the steady-state results. All the data presented in this report represent the pressure coefficient distribution over the afterbody and nozzles of the model, or the change in this coefficient as the tail positions are moved. A detailed description of the data reduction procedures is presented in reference 12.

A complete listing of the data from this investigation is included in tables 2-7. Each of these six tables corresponds to an aft-end empennage configuration. Free-stream static and dynamic pressures are included in the data tables for the purpose of CFD modeling of this experiment. To illustrate certain trends in the data and the effect of various parameters, selected data have been plotted to aid in the discussion of results. Force data on the same configurations were presented by Leavitt in reference 5 and will be referred to in the discussion.

Two basic formats for plotting data were used: a rectangular format and a polar format. The rectangular plots show the longitudinal pressure distributions along the model and require five pages of plots to present all the orifice rows. The polar plots show the circumferential pressure distribution around one of the nozzles at five stations along the length of the nozzle. Although the measurements were distributed between the two nozzles, they were combined in the polar plots onto the left nozzle for the presentation of data. Therefore, the horizontal tail is on the left-hand side of each polar plot and the inter-fairing is on the right-hand side.

To aid in locating the longitudinal positions of the vertical and horizontal tails relative to the scale used for plotting data in this report, the tails have been superimposed on sample grids in figure 4. The angular positions of the pressure orifice rows are illustrated in figure 5. The rows of orifices on the model were not placed as shown but, assuming flow symmetry about the centerline vertical plane, were distributed between the nacelles to best use the space in the model.

Discussion of Results

Effect of Tail Position at $M = 0.8$, $\alpha = 0^\circ$, and $NPR = 3.4$

Afterbody and nozzle pressure distributions for each of the tail-position combinations at $M = 0.8$, $\alpha = 0^\circ$, and $NPR = 3.4$ (near design) are shown in figures 6 and 7. Polar plots of the pressure distributions around the nozzles are shown in figure 8 for the same conditions. The angle θ represents the circumferential angle about the nozzle axis measured outboard from the top position on the nacelle.

As shown in figures 6(a) and 7(a), a characteristic pressure deficit forms at $\theta = 0^\circ$ as the airstream is accelerated between the vertical tails. The location of the root leading edge corresponding to the vertical-tail position is included in the key. Moving the tails aft shifts the pressure deficit aft. Although pressure data on the upper surface of the interfairing (on the aircraft centerline) is limited, a pressure deficit in this region becomes evident at $x/L = 0.898$ as the vertical tails are moved aft (figs. 6(e) and 7(e)). The surface boattail angles on the nacelles and interfairing increase as x/L increases, and this results in an increase in aft-projected area, particularly for the interfairing. Shifting the pressure deficit aft should increase the total aft-end drag. This effect is confirmed by the drag data published by Leavitt on the same configuration (fig. 36(a) in ref. 5). At $M = 0.8$ and $\alpha = 0.1^\circ$, total aft-end drag coefficient increased by 0.0010 (a 14-percent increase) as the vertical tails were moved from the forward position to the aft position. Lift and drag coefficients in reference 5 were referenced to the wing area of the support system.

The effect of vertical- and horizontal-tail positions on afterbody and nozzle pressure distributions is not limited to the tail-afterbody juncture regions but extends to all circumferential positions around the model. For instance, as shown in figure 6(c), shifting the vertical tails aft affects the pressure distributions for a range of θ from 180° to 240° , positions which are on the opposite side of the nacelle from the vertical tail. A similar result was reported by Henderson and Burley on a single-engine afterbody (ref. 4) in which the influence of tail position was found to extend completely around the afterbody.

As mentioned earlier, the most predominant effect of shifting the position of the vertical tails aft is to shift the pressure deficit between the tails aft ($270^\circ \leq \theta \leq 45^\circ$). As shown in figures 6(c), 6(d), 7(c), and 7(d), shifting the vertical tails from the forward position to the mid position also reduces the pressure coefficient on the inboard surface of the nozzle between the nacelles ($225^\circ \leq \theta \leq 315^\circ$ at $x/L > 0.948$). However, the effect

of shifting the vertical tails from the mid to the aft position is generally a reduction in the pressure coefficients on the afterbody rather than on the nozzles.

Immediately upstream of the nozzle connect station ($x/L = 0.948$), the flow around the afterbody passes through an expansion characterized by a decrease in surface static pressure coefficient. The subsequent pressure recovery on the nozzle ($x/L > 0.948$) provides pressure coefficients over most of the nozzle surface that are greater than zero, a result indicating that the nozzle external surface (neglecting skin friction) is producing negative drag (i.e., thrust). In fact, Leavitt in reference 5 determined that nozzle drag was negative for all configurations operating at subsonic speeds and typical jet pressure ratios except for the configuration with both vertical and horizontal tails in the aft position. In that case, as seen in figure 7(a) ($\theta = 45^\circ$) and figure 7(b) ($\theta = 90^\circ$) in the vicinity of the vertical tail, the expansion region extended farther aft on the nozzles such that the net load from the expansion and recovery on the nozzle was a positive drag. A similar trend for the configuration with aft vertical tails and mid horizontal tails is shown in figures 6(a) ($\theta = 45^\circ$) and 6(b) ($\theta = 90^\circ$). However, the overall pressure coefficient levels on the nozzles is greater with the result that positive pressure coefficients prevail and cause negative nozzle drag.

Flow expansion, which results in negative pressure coefficients on the model surface, increases the aft-end drag if acting over an aft-facing area. For the current model, the expansion is most severe at $\theta = 225^\circ$ and 240° (see fig. 6(c)), which corresponds to the region on the inboard underside of the model near the interfairing. (See fig. 5.) The combined boattailing of the interfairing and nacelle in this location results in a significant acceleration of the flow and, therefore, lower pressures.

Polar plots showing the circumferential pressure distribution around the nozzle are shown in figure 8. The data are presented as distributions around the left-hand nozzle such that the interfairing is on the right and the horizontal tail is on the left. The plots show two distinct features, a high-pressure region above the interfairing at $x/L = 0.951$ and a pressure reduction in the upper left quadrant of the nozzle when the vertical tail is moved to the aft position. The pressure reduction, which is a result of the expansion region extending farther aft on the nozzle, was discussed previously. The pressurization near the interfairing is probably a pressure recovery from the strong expansion which occurs between the vertical tails ahead of the nozzle. Figures 6(e) and 7(e) show pressure coefficients on the top and bottom interfairing surfaces, and a pressure recovery on the top surface between the aft two pressure taps is evident for the mid and aft positions of the vertical tail.

The effect of moving the horizontal tails from the mid position to the aft position at $M = 0.8$, $\alpha = 0^\circ$, and $NPR = 3.4$ is summarized in figure 9. The data are presented as ΔC_p , which is the aft-position C_p minus the mid-position C_p . Therefore, a positive ΔC_p represents a beneficial effect of moving the tails aft. Note that calculations of ΔC_p were made only for pressure orifices that were not covered by the tails in either position. As can be seen, most effects are small. The most significant effects are seen in the regions near the horizontal tails ($45^\circ \leq \theta \leq 135^\circ$). An area of negative ΔC_p can be seen over the nozzle in the vicinity of the horizontal tail ($45^\circ \leq \theta \leq 135^\circ$ at $x/L \geq 0.925$), a result indicating lower static pressures in this region. Corresponding positive values of ΔC_p can be seen farther forward ($0.8 \leq x/L \leq 0.9$), a result indicating that low static pressure, or accelerated flow, follows the horizontal tail as it is moved. In general, however, the influence of the horizontal-tail position on afterbody pressure distributions affecting aft-end drag is not as strong as that of the vertical-tail position.

Effect of Tail Position at $M = 1.2$, $\alpha = 0^\circ$, and $NPR = 3.4$

Afterbody and nozzle pressure distributions at $M = 1.2$, $\alpha = 0^\circ$, and $NPR = 3.4$ (near design) are shown in figures 10 and 11, and polar plots of the nozzle pressure distributions are shown in figure 12 at the same conditions.

The pressure distribution for the supersonic flow field is different in character from the subsonic flow field in many respects. The influence of the shocks emanating from the leading and trailing edges of the tails can be seen in the pressure coefficient data near the tail surfaces at $\theta = 0^\circ$, 45° , and 90° (the horizontal-tail leading-edge shock occurred only at $\theta = 90^\circ$) as a sharp increase in pressure coefficient. (For example, see the circular data symbols in fig. 10(a) at $\theta = 0^\circ$, $x/L = 0.738$, and $x/L = 0.898$.) Pressure distributions upstream of the leading-edge shocks are independent of tail position, as expected in a supersonic flow field. Pressure coefficients on the lower side of the model ($135^\circ \leq \theta \leq 255^\circ$) show almost no effect of tail position up to the nozzle ($x/L = 0.948$). The flow at the nozzle recovers pressure through a shock, and tail-position effects can be seen beyond this point. Flow separation downstream of the shock on the nozzle is often evident, particularly for the configuration with mid horizontal and vertical tails (the square data symbols), which has a relatively flat pressure distribution over the last three or four pressure taps. This configuration has both the horizontal- and vertical-tail closures occurring over a small range of x/L as compared with the other two configurations. (See fig. 4.)

The flow over the afterbody expands to a lower pressure at $M = 1.2$ than at $M = 0.8$, which should result in higher total aft-end drag. Additionally, the pressure recovery at the nozzle does not generally result in any pressurized regions ($C_p > 0$) that would reduce afterbody/nozzle drag, as was the case at subsonic free-stream conditions. Leavitt (ref. 5) measured a significantly higher aft-end drag at supersonic conditions than at subsonic conditions, as further support of the pressure data findings. Additionally, he determined that moving the vertical tails from the forward position to the mid position (with the horizontal tails in the mid or aft positions) at $M = 1.2$ had no effect on total aft-end drag. Moving the vertical tails to the aft position, however, resulted in a total aft-end drag increase. This result is consistent with the pressure data in figures 10 and 11 in which the pressure distributions for the forward and mid vertical-tail positions typically cross each other at some point along the nozzle ($x/L > 0.948$), thus resulting in a net-equivalent contribution to drag. The pressure distribution for the aft vertical-tail position, however, is generally at much lower pressures along the nozzle than for the other tail positions. This results in a higher drag, which is attributable to the area distribution of the model. Moving the vertical tail aft modifies the aircraft area distribution such that the closure at the aft end is more rapid and causes a larger wave drag.

Polar plots of the nozzle pressure distributions (plotted for the left nozzle) are shown in figure 12 for $M = 1.2$, $\alpha = 0^\circ$, and $NPR = 3.4$. An asymmetric flow field over the nozzles is evident. Disturbances tend to be concentrated inboard immediately downstream of the interfairing trailing edge ($x/L = 0.951$ and 0.963). Low pressures in this region are seen for the vertical tail in the aft position. Pressures in this region also decrease as the horizontal tail is moved aft. The complex pressure distributions are the result of an interaction of nozzle pressure-recovery shocks, nozzle flow separation, vertical-tail trailing-edge shocks, and interfairing trailing-edge shocks. Although sufficient pressure data are not available for confirmation, separation effects may also be present on the interfairing.

The effect of moving the horizontal tail aft at $M = 1.2$ is summarized in figure 13. The change in pressure coefficient (ΔC_p) as the tail is moved aft has been plotted for each vertical-tail position. Regions of favorable effects are generally located between $0.875 \leq x/L \leq 0.950$. The most adverse effects are generally limited to regions aft of the nozzle-connect station ($x/L \geq 0.948$) on the inboard surface of the nozzle ($255^\circ \leq \theta \leq 315^\circ$). Strong expansions accompanied by strong recovery shocks in this region can be seen relative to the mid horizontal-tail location. (See fig. 13(d).) As noted earlier when discussing moving the vertical tails

aft, moving the horizontal tails aft increases the cross-sectional area near the rear of the aircraft, thus increasing the rate of closure which adversely affects the wave drag of the aircraft at supersonic speeds.

Effect of Nozzle Pressure Ratio

The effect of varying nozzle pressure ratio is shown in figures 14–17 for two configurations identified by Leavitt (ref. 5) as having low and high aft-end drag. The first configuration (horizontal tails in the mid position and vertical tails in the forward position (mid-fwd)) was determined to have low total aft-end drag. The second configuration (both horizontal and vertical tails in the aft position (aft-aft)) was determined to have high total aft-end drag. Figures 14 and 15 show pressure distributions at $M = 0.8$ for the mid-fwd and aft-aft configurations, respectively, and figures 16 and 17 show similar data at $M = 1.2$. The general effect of increasing NPR at $M = 0.8$ (figs. 14 and 15) is to pressurize the afterbody and nozzle surfaces. Leavitt determined that for both tail position configurations (see figs. 18(f) and 18(l) in ref. 5), the effects of increasing nozzle pressure ratio on total aft-end drag were a large decrease in drag with initial jet operation, a very slight increase in drag as NPR was increased to fully expanded jet conditions (design NPR = 3.54), and a moderate decrease in drag as NPR was increased through the underexpanded jet range (NPR greater than design).

The pressure distributions support the measured drag data. The largest beneficial effect on afterbody/nozzle pressure distributions is the initial operation of the jet (NPR = 2). Increases in pressure coefficient are significant around the nozzles and extend as far forward as $x/L = 0.854$. (See fig. 14(c) at $\theta = 180^\circ$.) In other words, the influence of the NPR extends over the aft 15 percent of the aircraft. Increasing the NPR from 2 to 4 has no noticeable effect on pressure distributions, and this corresponds to the minimal change in drag reported by Leavitt. Increasing the NPR to values greater than 4 results in further increases in static pressure coefficient on the afterbody and nozzles, thus corresponding to the reduction in drag measured by Leavitt. The drag reduction with initial jet operation is attributable to base bleeding in the nozzle exit region. The slight drag increase as NPR is increased to the fully expanded flow condition is the result of increased entrainment of external flow by the nozzle flow, and this entrainment results in higher flow velocities (lower pressure and higher skin friction) over the external surfaces. The decrease in drag as the jet becomes underexpanded is attributable to plume effects as the plume becomes large, and this results in a compression near the nozzle exit and pressurization of the external nozzle surfaces upstream of the plume.

The effect of NPR at $M = 1.2$ is shown in figures 16 and 17. The effects of NPR at supersonic free-stream conditions are generally limited to the nozzle surface only (the last 5 percent of the aircraft) and do not extend forward to the afterbody. The lowest pressures on the nozzles generally correspond to either the jet-off condition (circular symbols) or the nearly fully expanded condition (diamond symbols). This behavior is consistent with the drag data measured by Leavitt in which the drags at these two conditions were essentially equal in magnitude and had the highest values throughout the NPR range tested. Increases in NPR above design (NPR > 3.54) result in external surface pressurization on the nozzle and a reduction in aft-end drag.

Effect of Mach Number

Pressure distributions for the same two tail configurations (mid-fwd and aft-aft) are plotted in figures 18 and 19, respectively, for Mach numbers of 0.6, 0.8, 0.9, and 1.2. The data trends at the subsonic ($M \leq 0.8$) and supersonic ($M = 1.2$) free-stream Mach numbers have previously been discussed. The pressure coefficient distributions at $M = 0.6$ and 0.8 are essentially the same. However, transonic effects between the vertical tails ($\theta = 0^\circ$) produce a greater reduction in pressure coefficient at $M = 0.8$ than at $M = 0.6$. This region would be expected to show the first transonic effects on the model because the flow is accelerated both by the fuselage and the vertical tails. For the mid-fwd configuration shown in figure 18, the effect of the transonic behavior on drag is minimal because the aft-facing area in this region is small. However, an increase in afterbody lift and nose-down aircraft pitching moment would be expected. The pressure distribution confirms the trends in the force data presented by Leavitt (in fig. 36(a) in ref. 5), which showed an increase in lift coefficient of approximately 0.002 and a very small increase in drag coefficient as Mach number was increased from 0.6 to 0.8. Pitching-moment characteristics are confirmed in figure 7(f) of reference 5.

For the aft-aft configuration shown in figure 19, the accelerated region is located in an area of higher boattail angle, and it would therefore be expected to result in a larger drag increase than the mid-fwd configuration as Mach number is increased from 0.6 to 0.8. The force data in reference 5 indicate an increase in drag for the aft-aft configuration over the mid-fwd configuration (compare figs. 36(a) and 37(a) in ref. 5); the drag coefficient is increased approximately 0.0005 as Mach number is increased from 0.6 to 0.8. Lift coefficient, however, which would be expected to increase with this change in Mach number, decreases by approximately 0.002. This unexpected result indicates that the increase in lift from the transonic flow acceleration between the vertical tails

is offset by a decrease in lift elsewhere on the afterbody. For a range of θ from 90° to 270° , lower static pressures result in a loss of lift and an increase in nose-up pitching moment. Reductions in static pressure can be seen at $\theta = 135^\circ$ (for values of x/L between 0.869 and 0.974) and at $\theta = 180^\circ$, 225° , and 240° (for values of x/L between 0.898 and 0.951). An additional decrease in lift can be seen at $\theta = 0^\circ$ (in this case an increase in static pressure at x/L between 0.781 and 0.854).

A more significant transonic effect is seen in figures 18 and 19 at $M = 0.9$. At $\theta = 0^\circ$, the flow is accelerated to supersonic speeds between the vertical tails ($C_p^* = -0.188$ for $M = 0.9$). In both tail arrangements, the critical pressure ratio (C_p^*), an indication of sonic velocity, is reached near the station of maximum vertical-tail thickness. (Compare figs. 18(a) and 19(a) with fig. 4.) For the mid-fwd configuration, a shock is formed approximately at the station of the vertical-tail trailing edge, and this results in subsonic velocities and also in pressure coefficients aft of this location that are generally greater than those at lower Mach numbers. In reference 5 for the mid-fwd configuration, Leavitt reported an increase in C_D of approximately 0.0010 as Mach number was increased from 0.8 to 0.9 and roughly equivalent lift coefficients. For the aft-aft configuration, C_D increased by 0.0005 and C_L decreased by 0.004. As can be seen in figure 19(a), the pressure coefficients at $\theta = 0^\circ$ for $M = 0.9$ are greater than those at $M = 0.8$ over most of the upper surface of the afterbody, thus confirming the lower lift reported by Leavitt.

Polar plots of the nozzle pressure distributions are shown in figure 20 for the same configurations (mid-fwd and aft-aft) at the same free-stream conditions. Mach number effects at subsonic speeds are concentrated in the region near the interfairing ($x/L = 0.951$, inboard or right side of plot) and may indicate the beginning of flow separation from the nozzle. The high-drag nature of supersonic flow relative to subsonic flow can also be seen, as shown by the low pressure coefficients around the nozzles at $M = 1.2$. The lack of pressure recovery to positive values of C_p also contributes to the large difference between subsonic and supersonic drag.

Effect of Angle of Attack

The effect of increasing angle of attack on the pressure distributions of the mid-fwd tail configuration is shown in figure 21 at $M = 0.8$ and $NPR = 3.4$, and similar data for the aft-aft tail configuration are shown in figure 22. At these conditions, the dominant effects of increasing angle of attack are to reduce the pressure on the leeward side of the model as far aft as $x/L = 0.9$ and to increase the pressure on the windward side of the

model as far aft as $x/L = 0.95$. For the case of the mid-fwd configuration (fig. 21(a)), the flow between the vertical tails ($\theta = 0^\circ$) is accelerated to sonic conditions at $\alpha = 4^\circ$ ($C_p^* = -0.435$ at $M = 0.8$) and to supersonic conditions at $\alpha = 8^\circ$. At this condition, a shock forms aft of the pressure orifice at $x/L = 0.825$ to recompress the flow. The significant reduction in leeward pressure coefficients seen at $\theta = 0^\circ$ (as compared with the increase in pressure on the windward side) as angle of attack is increased from 0° to 8° results in an increase in total aft-end drag coefficient of 0.0095 at $M = 0.8$, as reported by Leavitt.

The pressure coefficients for the aft-aft configuration are not as strongly affected by angle of attack (fig. 22(a)) as for the mid-fwd configuration. A relatively constant reduction in static pressure at $\theta = 0^\circ$ is indicated as far aft as $x/L = 0.900$ as angle of attack is increased. The region between the vertical tails does not exhibit the supersonic flow acceleration seen for the mid-fwd configuration, thereby indicating a smaller effect of angle of attack on aft-end drag for the aft-aft configuration. In reference 5, Leavitt reported a total aft-end drag increase of approximately 0.0062 at $M = 0.8$ as angle of attack was increased from 0° to 8° , which is a smaller drag increase than was measured for the mid-fwd configuration, thus confirming the results indicated by the pressure distributions.

In the study of the afterbody/nozzle aerodynamic characteristics of this model (ref. 5), Leavitt found only one apparent trend in total aft-end drag with the model at positive angle of attack. At $\alpha = 8^\circ$ and $M > 0.6$, the aft vertical-tail location provided a significant reduction in aft-end drag over the forward and mid locations with horizontal tails in the mid or aft positions. The pressure distributions for the mid horizontal-tail location at $M = 0.9$, $\alpha = 8^\circ$, and $NPR = 3.4$ are presented in figure 23. As the vertical tails are moved aft, the static pressure coefficient on the nozzles is reduced and the pressure deficit between the vertical tails is moved aft with the tails, both being factors that would typically result in a drag increase. The pressure deficit, however, is reduced in severity, both in the magnitude of pressure reduction and in the longitudinal extent of the deficit. Additionally, the amount of closure area over which the low pressure acts is reduced. The sum effect could be not only a reduction in drag but also a reduction in lift as well, confirmed by the force data presented by Leavitt. The smaller pressure deficit of the aft vertical-tail position is a result of the influence of the final closure of the model, which imposes a pressure recovery on the accelerating flow between the tails. This result probably accounts for the trend identified by Leavitt.

Comparisons With Computational Fluid Dynamics Solutions

Computational fluid dynamics solutions for the surface pressures about this model were obtained by using the PAB3D (version 10) code developed in the former Propulsion Aerodynamics Branch at the Langley Research Center. PAB3D is a multiblock/multizone state-of-the-art code that solves the three-dimensional, Reynolds-averaged Navier-Stokes equations with either a time-dependent or space-marching technique. The code, described in reference 13, was developed to study the integration of the propulsion system and the airframe, but it is currently applicable to general aerodynamic problems. In this application, the time-dependent technique was applied and a combination of turbulence models was used. The Baldwin-Lomax algebraic turbulence scheme was used in the upstream blocks containing the nose and the faired inlet. The remaining blocks used the Jones-Launder k - ϵ turbulence model. In addition, laminar viscosity was used in the first few grid planes of the internal nozzle block. Eleven blocks in all were used to model the aircraft in half-plane symmetry about the longitudinal axis.

Grids were generated for two empennage arrangements: vertical tails in the mid position and vertical tails in the forward position. In both cases, the horizontal tails were in the mid position. The grids for this model contained approximately 1.5 million grid points for each empennage arrangement and were generated using GRIDGEN (version 8 in ref. 14). The computational model geometry showing the forward vertical-tail position is shown in figure 24.

Surface pressure coefficients were compared for experimental results and computational predictions, as shown in figure 25. Test cases corresponded to the vertical tails in the mid and forward positions. The test condition was $M = 0.9$, $\alpha = 0^\circ$, and $NPR = 3.4$. Overall, the computations closely matched the experimental results, and certain flow features were accurately predicted by the computations. An example was the expansion of the flow between the vertical tails ($\theta = 0^\circ$) as the flow accelerated to supersonic speeds and the location of the recompression shock. Other examples were the stagnation effects near the horizontal-tail leading edge ($\theta = 90^\circ$) and similar stagnation effects near the leading edge of the vertical tail at $\theta = 0^\circ$ and 45° ($x/L = 0.72$ and 0.78 for the forward and mid positions, respectively). The compression at $\theta = 0^\circ$ indicated that the influence of the stagnation region of the leading edge of the vertical tail extended inboard on the fuselage, probably reducing the afterbody lift and drag.

Certain differences also exist between the computed and experimental results. An examination of the pressure

coefficient data at $\theta = 135^\circ$, 180° , and 225° indicates that the computations did not predict the effects of vertical-tail position on the windward side of the aircraft that were seen in the experimental results. Also, the pressure coefficient level immediately downstream of the shock between the vertical tails for the forward vertical-tail position was predicted low ($\theta = 285^\circ$, 315° , and 0°). In reference 13, Abdol-Hamid indicated that pressure recovery levels downstream of shocks were sensitive to the applied turbulence model.

Conclusions

The distributions of static pressure coefficient over the afterbody and nozzles of a generic, twin-tail twin-engine fighter aircraft were obtained in the Langley 16-Foot Transonic Tunnel. Three vertical-tail longitudinal positions and two horizontal-tail longitudinal positions were tested for a total of six aft-end configurations. Static pressure coefficients were obtained at Mach numbers (M) of 0.6, 0.8, 0.9, and 1.2, angles of attack (α) of 0° , 4° , and 8° , and nozzle pressure ratios ranging from jet-off to 8. Pressure measurements were made at approximately 150 locations on the afterbody, interfairing, and nozzles. Limited computational modeling of the pressure distribution about this model was performed. Based on the discussion of results, the following conclusions are presented:

1. The pressure data support the force and moment data published previously for this model.
2. The influence of the vertical- and horizontal-tail positions is evident over nearly all regions of the afterbody and nozzles, not just in the vicinity of the tail-afterbody juncture.
3. The pressure distribution affecting the aft-end drag is influenced more by the position of the vertical tails than by the position of the horizontal tails. A region of accelerated flow and reduced pressure forms at subsonic speeds between the vertical tails. At $\alpha = 0^\circ$, placing the vertical tails forward on the afterbody reduces the aft-facing projected area of this low-pressure region between the vertical tails and therefore reduces the aft-end drag. At positive angles of attack, placing the vertical tails in an aft position produces lower aft-end drag because the pressure deficit between the vertical tails is reduced by the aft-end pressure recovery.
4. At supersonic free-stream conditions, aft vertical- or horizontal-tail positions produce low static pressures on the nozzle surfaces and thus cause higher aft-end drag.
5. The effects of Mach number at and below $M = 0.8$ at $\alpha = 0^\circ$ are small, with the exception of the region between the vertical tails where the flow is highly

accelerated. The pressure reduction in this region has the effect of increasing afterbody lift for the forward vertical-tail configuration and increasing drag for the aft vertical-tail configuration. Transonic effects strong enough to produce a local shock wave in this region are evident at $M = 0.9$.

6. The effects of nozzle pressure ratio in subsonic flow extend over the last 15 percent of the aircraft, but in supersonic flow the effects are limited to the nozzle surface, or to the last 5 percent of the aircraft.

7. Predictions of surface pressure coefficient from computational modeling of this geometry show excellent agreement with experimental results. Exceptions are the recovery pressure level downstream of the upper surface transonic shock at $M = 0.9$ and the effects of vertical-tail position on pressure coefficients on the windward side of the aircraft.

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Table 1. Locations of Pressure Orifices on Afterbody and Nozzles

(a) Measurement stations for afterbody pressure orifices

x/L	Meridian angle, θ , of—												Interfairing	
	0°	45°	90°	135°	180°	225°	240°	255°	270°	285°	300°	315°	Top	Bottom
0.709	x	x	x	x	x								x	x
.723	x	x	x	x	x								x	x
.738	x	x	x	x	x									x
.752	x	x	x	x	x									x
.767	x	x	x	x	x									x
.781	x	x	x	x	x									x
.796	x		x	x	x									
.810	x			x	x									
.825	x		x	x	x									
.840	x			x	x									
.854	x				x									
.869	x			x										x
.883	x	x		x	x	x						x		x
.898	x	x		x	x	x	x				x	x	x	
.912	x	x		x	x	x	x				x	x		
.927	x	x		x	x	x	x			x	x	x		
.941	x	x		x	x	x	x	x		x	x	x	x	x

(b) Measurement stations for nozzle pressure orifices

x/L	Meridian angle, θ , of—											
	0°	45°	90°	135°	180°	225°	240°	255°	270°	285°	300°	315°
0.951	x	x	x	x	x	x	x		x	x	x	x
.963	x	x	x	x		x	x	x	x	x	x	x
.974	x	x	x	x	x	x	x	x	x	x	x	x
.986			x	x	x	x	x	x	x	x	x	x
.997		x	x	x		x		x		x	x	x

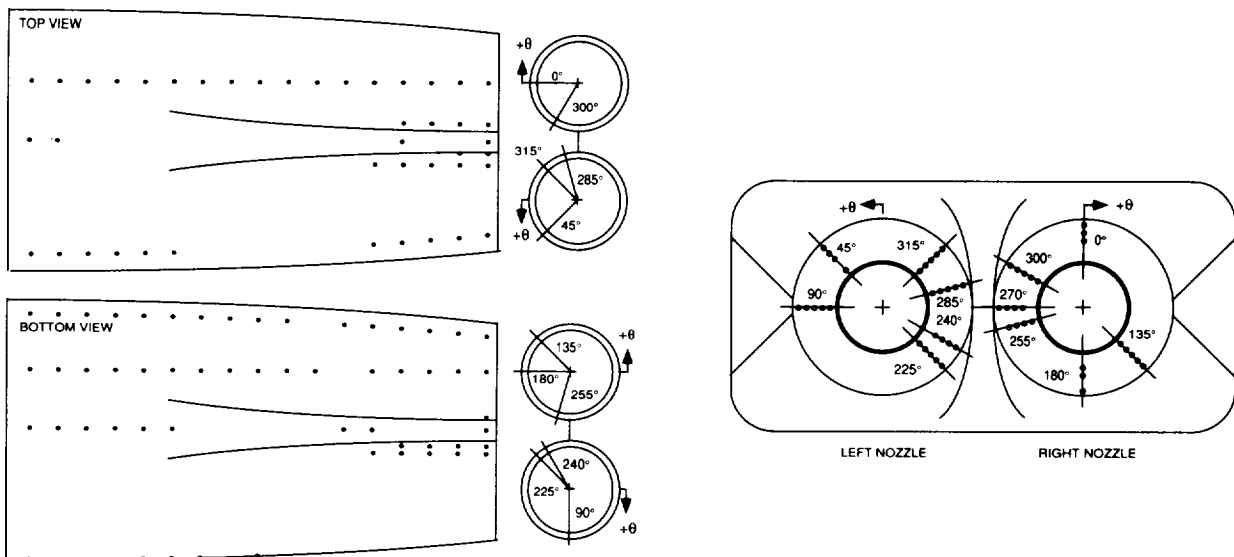


Table 2. Static Pressure Coefficients of Afterbody Surface With Horizontal Tail in Mid Position and Vertical Tail in Forward Position

(a) $\theta = 0^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
0.799	1.506	0.02	4.305	9.641	0.001	0.006	-0.034	-0.111	-0.163	-0.263	-0.331	-0.330	-0.295	-0.227	-0.153
0.799	1.999	0.03	4.307	9.639	0.003	0.007	-0.032	-0.109	-0.162	-0.262	-0.331	-0.331	-0.294	-0.227	-0.152
0.798	2.988	0.02	4.298	9.652	0.002	0.008	-0.032	-0.109	-0.161	-0.260	-0.328	-0.329	-0.294	-0.225	-0.151
0.800	3.459	0.02	4.316	9.625	0.004	0.010	-0.030	-0.109	-0.162	-0.260	-0.331	-0.333	-0.296	-0.227	-0.153
0.800	3.992	0.03	4.316	9.627	0.002	0.009	-0.031	-0.109	-0.162	-0.262	-0.330	-0.333	-0.297	-0.228	-0.154
0.799	6.023	-0.03	4.307	9.639	0.002	0.010	-0.031	-0.110	-0.162	-0.261	-0.329	-0.330	-0.293	-0.225	-0.152
0.800	5.969	-0.03	4.312	9.632	0.002	0.010	-0.031	-0.110	-0.162	-0.261	-0.327	-0.328	-0.292	-0.224	-0.151
0.799	7.962	-0.02	4.309	9.638	0.002	0.011	-0.030	-0.110	-0.161	-0.258	-0.324	-0.324	-0.289	-0.221	-0.150
0.799	3.455	8.01	4.310	9.642	-0.069	-0.061	-0.116	-0.215	-0.275	-0.397	-0.503	-0.576	-0.582	-0.315	-0.203
0.806	3.432	4.01	4.351	9.578	-0.032	-0.024	-0.074	-0.166	-0.225	-0.336	-0.430	-0.467	-0.380	-0.267	-0.182
0.801	3.445	4.01	4.322	9.622	-0.032	-0.023	-0.075	-0.167	-0.225	-0.335	-0.423	-0.440	-0.362	-0.264	-0.182
0.898	1.504	0.00	4.916	8.710	0.031	0.060	0.045	-0.018	-0.066	-0.167	-0.258	-0.321	-0.397	-0.478	-0.536
0.898	1.995	0.01	4.916	8.709	0.032	0.062	0.046	-0.017	-0.065	-0.165	-0.256	-0.319	-0.396	-0.477	-0.536
0.899	2.999	0.02	4.919	8.703	0.032	0.062	0.046	-0.016	-0.064	-0.165	-0.255	-0.318	-0.394	-0.476	-0.543
0.899	3.462	0.03	4.919	8.703	0.032	0.062	0.046	-0.016	-0.064	-0.164	-0.255	-0.318	-0.394	-0.475	-0.544
0.899	3.986	-0.01	4.923	8.696	0.029	0.061	0.046	-0.017	-0.065	-0.164	-0.255	-0.318	-0.393	-0.475	-0.544
0.899	5.982	0.01	4.923	8.697	0.032	0.063	0.047	-0.016	-0.064	-0.163	-0.253	-0.316	-0.392	-0.473	-0.541
0.899	7.953	0.02	4.923	8.698	0.035	0.066	0.048	-0.016	-0.064	-0.162	-0.252	-0.315	-0.391	-0.472	-0.525
0.900	3.461	8.03	4.926	8.694	-0.034	-0.009	-0.037	-0.111	-0.162	-0.262	-0.359	-0.428	-0.501	-0.580	-0.653
0.901	3.427	4.00	4.935	8.677	0.014	0.040	0.012	-0.059	-0.109	-0.206	-0.297	-0.364	-0.435	-0.518	-0.572
0.601	1.492	0.00	2.915	11.523	-0.008	-0.015	-0.058	-0.112	-0.134	-0.185	-0.208	-0.193	-0.172	-0.144	-0.116
0.602	2.004	0.00	2.921	11.517	-0.008	-0.013	-0.056	-0.111	-0.134	-0.185	-0.209	-0.194	-0.174	-0.147	-0.119
0.601	2.996	0.01	2.916	11.523	-0.010	-0.016	-0.060	-0.116	-0.138	-0.189	-0.212	-0.196	-0.177	-0.149	-0.122
0.601	3.441	-0.01	2.916	11.522	-0.010	-0.016	-0.059	-0.115	-0.137	-0.189	-0.213	-0.197	-0.177	-0.150	-0.121
0.601	3.980	-0.01	2.916	11.523	-0.010	-0.015	-0.058	-0.114	-0.136	-0.188	-0.212	-0.196	-0.177	-0.149	-0.121
0.601	5.968	0.01	2.914	11.526	-0.012	-0.018	-0.061	-0.117	-0.139	-0.191	-0.213	-0.198	-0.178	-0.150	-0.121
0.601	7.933	0.02	2.913	11.524	-0.009	-0.016	-0.058	-0.114	-0.136	-0.189	-0.211	-0.195	-0.173	-0.146	-0.118
0.600	3.503	8.01	2.906	11.536	-0.081	-0.100	-0.158	-0.226	-0.248	-0.304	-0.322	-0.293	-0.260	-0.218	-0.174
0.602	3.448	3.99	2.919	11.520	-0.047	-0.061	-0.113	-0.172	-0.195	-0.247	-0.265	-0.244	-0.217	-0.182	-0.146
1.201	1.469	0.09	6.130	6.066	-0.047	-0.051	0.052	0.061	0.070	0.044	0.007	-0.025	-0.071	-0.122	-0.179
1.202	1.962	0.09	6.129	6.065	-0.048	-0.050	0.051	0.059	0.069	0.042	0.006	-0.025	-0.071	-0.121	-0.179
1.201	2.928	0.10	6.129	6.068	-0.048	-0.051	0.051	0.059	0.068	0.042	0.006	-0.026	-0.072	-0.122	-0.180
1.201	3.960	0.10	6.128	6.067	-0.047	-0.051	0.052	0.061	0.070	0.043	0.007	-0.025	-0.071	-0.122	-0.180
1.201	3.960	0.11	6.128	6.068	-0.048	-0.051	0.050	0.057	0.068	0.042	0.006	-0.025	-0.071	-0.121	-0.179
1.201	5.879	0.02	6.128	6.067	-0.050	-0.054	0.049	0.059	0.069	0.041	0.004	-0.027	-0.074	-0.124	-0.182
1.202	7.827	0.03	6.130	6.061	-0.051	-0.054	0.049	0.058	0.067	0.040	0.003	-0.027	-0.073	-0.123	-0.181
1.199	3.394	8.04	6.125	6.084	-0.197	-0.207	-0.124	-0.128	-0.152	-0.169	-0.185	-0.207	-0.240	-0.271	-0.327
1.202	3.409	3.98	6.130	6.060	-0.129	-0.136	-0.047	-0.054	-0.055	-0.079	-0.111	-0.132	-0.164	-0.209	-0.269

Table 2. Continued

(a) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.869	0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	
0.799	1.506	0.02	4.305	9.641	-0.081	-0.053	-0.051	-0.060	-0.064	-0.057	-0.109	0.036	0.127	
0.799	1.999	0.03	4.307	9.639	-0.081	-0.053	-0.052	-0.060	-0.064	-0.059	-0.113	0.032	0.123	
0.798	2.988	0.02	4.298	9.652	-0.081	-0.053	-0.052	-0.060	-0.066	-0.062	-0.117	0.029	0.119	
0.800	3.459	0.02	4.316	9.625	-0.082	-0.053	-0.052	-0.060	-0.066	-0.062	-0.118	0.030	0.120	
0.800	3.992	0.03	4.316	9.627	-0.083	-0.054	-0.053	-0.059	-0.065	-0.062	-0.117	0.030	0.122	
0.799	6.023	-0.03	4.307	9.639	-0.082	-0.052	-0.051	-0.054	-0.059	-0.054	-0.107	0.042	0.136	
0.800	5.969	-0.03	4.312	9.632	-0.082	-0.051	-0.050	-0.053	-0.059	-0.055	-0.106	0.043	0.137	
0.799	7.962	-0.02	4.309	9.638	-0.081	-0.049	-0.047	-0.046	-0.050	-0.043	-0.091	0.059	0.153	
0.799	3.455	8.01	4.310	9.642	-0.128	-0.088	-0.081	-0.071	-0.073	-0.066	-0.105	0.036	0.120	
0.806	3.432	4.01	4.351	9.578	-0.110	-0.072	-0.068	-0.061	-0.067	-0.064	-0.109	0.034	0.118	
0.801	3.445	4.01	4.322	9.622	-0.110	-0.072	-0.069	-0.061	-0.067	-0.063	-0.110	0.033	0.117	
0.898	1.504	0.00	4.916	8.710	-0.103	-0.024	-0.017	-0.028	-0.030	-0.022	-0.066	0.071	0.148	
0.898	1.995	0.01	4.916	8.709	-0.101	-0.024	-0.018	-0.029	-0.033	-0.026	-0.071	0.067	0.144	
0.899	2.999	0.02	4.919	8.703	-0.109	-0.028	-0.022	-0.033	-0.038	-0.032	-0.079	0.061	0.139	
0.899	3.462	0.03	4.919	8.703	-0.110	-0.028	-0.022	-0.033	-0.038	-0.032	-0.080	0.061	0.139	
0.899	3.986	-0.01	4.923	8.696	-0.111	-0.027	-0.021	-0.031	-0.036	-0.030	-0.079	0.062	0.141	
0.899	5.982	0.01	4.923	8.697	-0.108	-0.025	-0.018	-0.026	-0.030	-0.023	-0.067	0.073	0.153	
0.899	7.953	0.02	4.923	8.698	-0.098	-0.020	-0.012	-0.018	-0.020	-0.010	-0.049	0.090	0.168	
0.900	3.461	8.03	4.926	8.694	-0.217	-0.085	-0.050	-0.036	-0.030	-0.019	-0.035	0.071	0.140	
0.901	3.427	4.00	4.935	8.677	-0.139	-0.044	-0.031	-0.028	-0.030	-0.022	-0.055	0.068	0.135	
0.601	1.492	0.00	2.915	11.523	-0.070	-0.048	-0.047	-0.050	-0.056	-0.054	-0.099	0.027	0.110	
0.602	2.004	0.00	2.921	11.517	-0.075	-0.051	-0.051	-0.054	-0.062	-0.062	-0.111	0.016	0.096	
0.601	2.996	0.01	2.916	11.523	-0.077	-0.054	-0.054	-0.057	-0.065	-0.066	-0.117	0.010	0.089	
0.601	3.441	-0.01	2.916	11.522	-0.078	-0.054	-0.054	-0.056	-0.064	-0.067	-0.118	0.010	0.090	
0.601	3.980	-0.01	2.916	11.523	-0.078	-0.054	-0.054	-0.057	-0.065	-0.066	-0.117	0.010	0.089	
0.601	5.968	0.01	2.914	11.526	-0.078	-0.053	-0.052	-0.054	-0.061	-0.061	-0.109	0.020	0.104	
0.601	7.933	0.02	2.913	11.524	-0.073	-0.049	-0.047	-0.050	-0.055	-0.052	-0.096	0.034	0.120	
0.600	3.503	8.01	2.906	11.536	-0.120	-0.091	-0.082	-0.082	-0.081	-0.074	-0.112	0.012	0.087	
0.602	3.448	3.99	2.919	11.520	-0.096	-0.071	-0.065	-0.067	-0.070	-0.067	-0.106	0.013	0.089	
1.201	1.469	0.09	6.130	6.066	-0.243	-0.276	-0.198	-0.204	-0.251	-0.285	-0.255	-0.325	-0.147	
1.202	1.962	0.09	6.129	6.065	-0.243	-0.276	-0.198	-0.203	-0.252	-0.286	-0.257	-0.326	-0.217	
1.201	2.928	0.10	6.129	6.068	-0.244	-0.276	-0.198	-0.203	-0.252	-0.285	-0.256	-0.326	-0.248	
1.201	3.396	0.10	6.128	6.067	-0.244	-0.277	-0.197	-0.201	-0.251	-0.286	-0.258	-0.326	-0.223	
1.201	3.960	0.11	6.128	6.068	-0.244	-0.276	-0.197	-0.209	-0.258	-0.290	-0.261	-0.333	-0.210	
1.201	5.879	0.02	6.128	6.067	-0.246	-0.280	-0.200	-0.209	-0.258	-0.292	-0.264	-0.331	-0.093	
1.202	7.827	0.03	6.130	6.061	-0.245	-0.280	-0.201	-0.210	-0.258	-0.292	-0.264	-0.331	-0.093	
1.199	3.394	8.04	6.125	6.084	-0.379	-0.402	-0.304	-0.321	-0.378	-0.416	-0.403	-0.386	-0.188	
1.202	3.409	3.98	6.130	6.060	-0.318	-0.355	-0.271	-0.249	-0.283	-0.324	-0.347	-0.379	-0.148	

Table 2. Continued

(b) $\theta = 45^\circ$

		α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—															
M	NPR				0.709	0.723	0.738	0.752	0.767	0.781	0.883	0.912	0.941	0.951	0.963	0.974	0.997			
0.799	1.506	0.02	4.305	9.641	0.043	0.102	-0.006	-0.039	-0.038	-0.043	-0.030	-0.059	-0.002	-0.020	0.053	0.119	0.177			
0.799	1.999	0.03	4.307	9.639	0.050	0.110	0.000	-0.034	-0.034	-0.040	-0.028	-0.059	-0.003	-0.022	0.050	0.116	0.164			
0.798	2.988	0.02	4.298	9.652	0.049	0.108	0.001	-0.032	-0.031	-0.038	-0.028	-0.058	-0.004	-0.024	0.049	0.115	0.163			
0.800	3.459	0.02	4.316	9.625	0.046	0.106	-0.001	-0.033	-0.033	-0.039	-0.028	-0.058	-0.005	-0.024	0.050	0.116	0.163			
0.800	3.992	0.03	4.316	9.627	0.045	0.106	-0.001	-0.034	-0.035	-0.041	-0.027	-0.059	-0.005	-0.023	0.051	0.117	0.167			
0.799	6.023	-0.03	4.307	9.639	0.046	0.106	-0.001	-0.033	-0.034	-0.040	-0.025	-0.054	0.000	-0.014	0.063	0.133	0.185			
0.800	5.969	-0.03	4.312	9.632	0.046	0.106	0.000	-0.032	-0.033	-0.039	-0.023	-0.053	0.001	-0.013	0.064	0.133	0.185			
0.799	7.962	-0.02	4.309	9.638	0.046	0.106	0.000	-0.031	-0.032	-0.038	-0.018	-0.047	0.009	-0.002	0.078	0.149	0.196			
0.799	3.455	8.01	4.310	9.642	-0.013	0.054	-0.030	-0.064	-0.071	-0.087	-0.061	-0.077	-0.017	-0.011	0.058	0.119	0.156			
0.806	3.432	4.01	4.351	9.578	0.030	0.099	0.010	-0.026	-0.034	-0.046	-0.039	-0.064	-0.010	-0.020	0.055	0.120	0.128			
0.801	3.445	4.01	4.322	9.622	0.031	0.099	0.011	-0.024	-0.031	-0.044	-0.039	-0.064	-0.010	-0.020	0.056	0.121	0.129			
0.898	1.504	0.00	4.916	8.710	0.066	0.132	0.010	-0.029	-0.024	-0.028	-0.020	-0.042	0.024	0.006	0.078	0.142	0.161			
0.898	1.995	0.01	4.916	8.709	0.069	0.135	0.011	-0.028	-0.024	-0.028	-0.021	-0.043	0.021	0.003	0.076	0.139	0.153			
0.899	2.999	0.02	4.919	8.703	0.066	0.131	0.008	-0.031	-0.027	-0.031	-0.023	-0.046	0.019	-0.001	0.073	0.136	0.153			
0.899	3.462	0.03	4.919	8.703	0.065	0.130	0.007	-0.032	-0.028	-0.031	-0.023	-0.045	0.018	-0.001	0.073	0.137	0.155			
0.899	3.986	-0.01	4.923	8.696	0.065	0.131	0.010	-0.028	-0.024	-0.028	-0.021	-0.045	0.019	0.001	0.076	0.140	0.159			
0.899	5.982	0.01	4.923	8.697	0.066	0.132	0.009	-0.030	-0.025	-0.029	-0.019	-0.041	0.025	0.010	0.086	0.152	0.172			
0.899	7.953	0.02	4.923	8.698	0.064	0.130	0.010	-0.028	-0.025	-0.028	-0.021	-0.041	0.023	0.023	0.101	0.169	0.182			
0.900	3.461	8.03	4.926	8.694	-0.009	0.056	-0.055	-0.110	-0.115	-0.117	-0.072	-0.051	0.019	0.022	0.087	0.144	0.156			
0.901	3.427	4.00	4.935	8.677	0.052	0.120	0.009	-0.042	-0.048	-0.054	-0.030	-0.044	0.021	0.011	0.082	0.141	0.136			
0.601	1.492	0.00	2.915	11.523	0.043	0.090	-0.002	-0.027	-0.031	-0.036	-0.014	-0.045	0.000	-0.018	0.049	0.109	0.162			
0.602	2.004	0.00	2.921	11.517	0.038	0.087	-0.005	-0.031	-0.034	-0.038	-0.018	-0.049	-0.007	-0.026	0.039	0.097	0.131			
0.601	2.996	0.01	2.916	11.523	0.038	0.086	-0.006	-0.032	-0.034	-0.039	-0.021	-0.051	-0.010	-0.030	0.035	0.092	0.122			
0.601	3.441	-0.01	2.916	11.522	0.039	0.088	-0.004	-0.033	-0.033	-0.038	-0.020	-0.051	-0.009	-0.031	0.034	0.092	0.125			
0.601	3.980	-0.01	2.916	11.523	0.037	0.085	-0.008	-0.033	-0.036	-0.040	-0.021	-0.052	-0.009	-0.030	0.035	0.092	0.132			
0.601	5.968	0.01	2.914	11.526	0.039	0.086	-0.007	-0.033	-0.034	-0.038	-0.021	-0.050	-0.004	-0.025	0.042	0.103	0.158			
0.601	7.933	0.02	2.913	11.524	0.040	0.088	-0.005	-0.032	-0.034	-0.037	-0.018	-0.045	0.003	-0.017	0.052	0.116	0.177			
0.600	3.503	8.01	2.906	11.536	-0.002	0.059	-0.006	-0.036	-0.045	-0.062	-0.067	-0.079	-0.021	-0.031	0.029	0.085	0.129			
0.602	3.448	3.99	2.919	11.520	0.032	0.090	0.013	-0.017	-0.024	-0.036	-0.039	-0.062	-0.011	-0.032	0.031	0.088	0.125			
1.201	1.469	0.09	6.130	6.066	-0.047	0.107	0.095	0.002	-0.047	-0.108	-0.107	-0.151	-0.226	-0.275	-0.386	-0.166	0.043			
1.202	1.962	0.09	6.129	6.065	-0.048	0.107	0.094	0.002	-0.047	-0.108	-0.107	-0.151	-0.226	-0.275	-0.386	-0.211	0.028			
1.201	2.928	0.10	6.129	6.068	-0.048	0.108	0.095	0.002	-0.048	-0.108	-0.107	-0.151	-0.227	-0.275	-0.386	-0.226	0.018			
1.201	3.396	0.10	6.128	6.067	-0.048	0.107	0.096	0.003	-0.047	-0.108	-0.104	-0.150	-0.226	-0.275	-0.386	-0.229	0.017			
1.201	3.960	0.11	6.128	6.068	-0.049	0.107	0.093	0.002	-0.048	-0.108	-0.106	-0.151	-0.226	-0.275	-0.386	-0.230	0.019			
1.201	5.879	0.02	6.128	6.067	-0.051	0.104	0.090	-0.006	-0.053	-0.114	-0.112	-0.156	-0.231	-0.281	-0.396	-0.211	0.034			
1.202	7.827	0.03	6.130	6.061	-0.051	0.103	0.088	-0.006	-0.053	-0.113	-0.118	-0.157	-0.231	-0.281	-0.396	-0.222	0.055			
1.199	3.394	8.04	6.125	6.084	-0.170	0.002	0.040	0.028	0.024	0.034	-0.247	-0.347	-0.372	-0.388	-0.404	-0.185	-0.044			
1.202	3.409	3.98	6.130	6.060	-0.111	0.052	0.069	0.009	-0.042	-0.089	-0.214	-0.196	-0.264	-0.301	-0.414	-0.222	0.012			

Table 2. Continued

(c) $\theta = 90^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.951	0.963	0.974	0.986	0.997
0.799	1.506	0.02	4.305	9.641	0.016	0.023	0.003	-0.041	0.015	0.164	0.032	0.079	0.129	0.169	0.205
0.799	1.999	0.03	4.307	9.639	0.020	0.027	0.007	-0.037	0.018	0.168	0.030	0.077	0.126	0.164	0.196
0.798	2.988	0.02	4.298	9.652	0.020	0.028	0.007	-0.038	0.018	0.168	0.030	0.075	0.122	0.159	0.189
0.800	3.459	0.02	4.316	9.625	0.020	0.028	0.006	-0.039	0.017	0.168	0.030	0.075	0.123	0.160	0.189
0.800	3.992	0.03	4.316	9.627	0.019	0.028	0.005	-0.040	0.016	0.166	0.032	0.076	0.124	0.161	0.194
0.799	6.023	-0.03	4.307	9.639	0.019	0.028	0.005	-0.041	0.016	0.167	0.041	0.087	0.137	0.180	0.221
0.800	5.969	-0.03	4.312	9.632	0.019	0.029	0.005	-0.042	0.016	0.168	0.042	0.087	0.137	0.179	0.220
0.799	7.962	-0.02	4.309	9.638	0.019	0.029	0.003	-0.042	0.016	0.168	0.054	0.100	0.152	0.198	0.245
0.799	3.455	8.01	4.310	9.642	0.010	0.008	-0.027	-0.097	-0.016	0.228	0.040	0.080	0.124	0.159	0.192
0.806	3.432	4.01	4.351	9.578	0.024	0.030	0.000	-0.052	0.014	0.199	0.037	0.075	0.119	0.154	0.183
0.801	3.445	4.01	4.322	9.622	0.026	0.031	0.001	-0.051	0.015	0.201	0.038	0.075	0.119	0.155	0.183
0.898	1.504	0.00	4.916	8.710	0.026	0.043	0.024	-0.025	0.029	0.173	0.052	0.099	0.148	0.184	0.216
0.898	1.995	0.01	4.916	8.709	0.028	0.046	0.026	-0.023	0.030	0.174	0.051	0.096	0.144	0.180	0.208
0.899	2.999	0.02	4.919	8.703	0.025	0.043	0.023	-0.026	0.028	0.172	0.048	0.093	0.141	0.176	0.201
0.899	3.462	0.03	4.919	8.703	0.025	0.043	0.022	-0.027	0.027	0.171	0.049	0.094	0.142	0.177	0.202
0.899	3.986	-0.01	4.923	8.696	0.026	0.044	0.023	-0.026	0.029	0.172	0.051	0.096	0.144	0.180	0.208
0.899	5.982	0.01	4.923	8.697	0.026	0.045	0.023	-0.026	0.029	0.172	0.060	0.106	0.156	0.196	0.234
0.899	7.953	0.02	4.923	8.698	0.027	0.046	0.024	-0.026	0.029	0.172	0.072	0.119	0.171	0.216	0.259
0.900	3.461	8.03	4.926	8.694	-0.018	-0.015	-0.057	-0.158	-0.059	0.204	0.066	0.108	0.151	0.182	0.209
0.901	3.427	4.00	4.935	8.677	0.014	0.029	-0.002	-0.068	0.003	0.199	0.060	0.098	0.140	0.172	0.196
0.601	1.492	0.00	2.915	11.523	0.018	0.019	0.003	-0.032	0.019	0.158	0.031	0.073	0.119	0.156	0.190
0.602	2.004	0.00	2.921	11.517	0.015	0.017	0.000	-0.035	0.016	0.156	0.022	0.064	0.107	0.142	0.169
0.601	2.996	0.01	2.916	11.523	0.014	0.015	-0.001	-0.034	0.015	0.156	0.018	0.059	0.101	0.135	0.158
0.601	3.441	-0.01	2.916	11.522	0.015	0.016	0.001	-0.034	0.016	0.157	0.017	0.058	0.100	0.135	0.158
0.601	3.980	-0.01	2.916	11.523	0.014	0.014	-0.002	-0.035	0.014	0.154	0.018	0.059	0.101	0.136	0.161
0.601	5.968	0.01	2.914	11.526	0.014	0.014	0.000	-0.033	0.016	0.156	0.023	0.066	0.113	0.152	0.189
0.601	7.933	0.02	2.913	11.524	0.015	0.015	0.001	-0.031	0.018	0.156	0.031	0.077	0.126	0.171	0.214
0.600	3.503	8.01	2.906	11.536	0.032	0.019	0.003	-0.043	0.017	0.225	0.016	0.058	0.101	0.134	0.155
0.602	3.448	3.99	2.919	11.520	0.032	0.025	0.013	-0.020	0.029	0.179	0.014	0.057	0.100	0.134	0.154
1.201	1.469	0.09	6.130	6.066	-0.066	-0.003	0.055	-0.030	-0.063	0.123	-0.220	-0.264	-0.190	-0.036	0.032
1.202	1.962	0.09	6.129	6.065	-0.066	-0.004	0.055	-0.031	-0.064	0.124	-0.219	-0.265	-0.221	-0.078	0.014
1.201	2.928	0.10	6.129	6.068	-0.065	-0.002	0.055	-0.032	-0.065	0.123	-0.218	-0.264	-0.229	-0.108	-0.001
1.201	3.396	0.10	6.128	6.067	-0.066	-0.002	0.055	-0.031	-0.064	0.124	-0.218	-0.265	-0.231	-0.112	-0.001
1.201	3.960	0.11	6.128	6.068	-0.065	-0.002	0.054	-0.032	-0.065	0.123	-0.218	-0.265	-0.231	-0.109	0.003
1.201	5.879	0.02	6.128	6.067	-0.073	-0.011	0.051	-0.034	-0.068	0.119	-0.229	-0.274	-0.213	-0.053	0.014
1.202	7.827	0.03	6.130	6.061	-0.074	-0.011	0.050	-0.034	-0.068	0.118	-0.229	-0.271	-0.126	-0.012	0.037
1.199	3.394	8.04	6.125	6.084	0.099	0.063	0.015	-0.040	-0.038	0.284	-0.262	-0.300	-0.202	-0.128	-0.075
1.202	3.409	3.98	6.130	6.060	0.035	0.014	0.030	-0.047	-0.083	0.173	-0.215	-0.268	-0.231	-0.122	-0.009

Table 2. Continued

(d) $\theta = 135^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.869
0.799	1.506	0.02	4.305	9.641	-0.027	-0.033	-0.048	-0.040	-0.028	-0.016	-0.020	-0.038	-0.042	-0.056	-0.079
0.799	1.999	0.03	4.307	9.639	-0.026	-0.032	-0.047	-0.039	-0.026	-0.015	-0.018	-0.036	-0.040	-0.055	-0.078
0.798	2.988	0.02	4.298	9.652	-0.025	-0.031	-0.047	-0.037	-0.026	-0.015	-0.018	-0.035	-0.039	-0.054	-0.077
0.800	3.459	0.02	4.316	9.625	-0.025	-0.031	-0.047	-0.037	-0.025	-0.016	-0.018	-0.037	-0.040	-0.055	-0.079
0.800	3.992	0.03	4.316	9.627	-0.026	-0.032	-0.047	-0.036	-0.026	-0.017	-0.018	-0.036	-0.039	-0.054	-0.079
0.799	6.023	-0.03	4.307	9.639	-0.027	-0.033	-0.048	-0.035	-0.027	-0.018	-0.019	-0.036	-0.039	-0.053	-0.078
0.800	5.969	-0.03	4.312	9.632	-0.027	-0.032	-0.047	-0.035	-0.026	-0.018	-0.019	-0.036	-0.039	-0.053	-0.078
0.799	7.962	-0.02	4.309	9.638	-0.027	-0.033	-0.048	-0.034	-0.026	-0.019	-0.018	-0.036	-0.038	-0.052	-0.077
0.799	3.455	8.01	4.310	9.642	-0.013	-0.031	-0.056	-0.044	-0.038	-0.021	-0.001	-0.004	0.006	-0.003	-0.033
0.806	3.432	4.01	4.351	9.578	-0.011	-0.022	-0.041	-0.024	-0.020	-0.010	0.000	-0.015	-0.012	-0.029	-0.061
0.801	3.445	4.01	4.322	9.622	-0.010	-0.021	-0.040	-0.023	-0.020	-0.010	0.000	-0.013	-0.015	-0.027	-0.061
0.898	1.504	0.00	4.916	8.710	-0.031	-0.026	-0.038	-0.032	-0.017	-0.005	-0.011	-0.033	-0.041	-0.060	-0.088
0.898	1.995	0.01	4.916	8.709	-0.028	-0.024	-0.036	-0.029	-0.014	-0.002	-0.008	-0.029	-0.038	-0.057	-0.088
0.899	2.999	0.02	4.919	8.703	-0.031	-0.026	-0.038	-0.031	-0.015	-0.004	-0.010	-0.033	-0.041	-0.060	-0.091
0.899	3.462	0.03	4.919	8.703	-0.030	-0.025	-0.037	-0.030	-0.015	-0.004	-0.010	-0.032	-0.040	-0.060	-0.090
0.899	3.986	-0.01	4.923	8.696	-0.033	-0.026	-0.037	-0.028	-0.014	-0.003	-0.008	-0.030	-0.038	-0.058	-0.089
0.899	5.982	0.01	4.923	8.697	-0.030	-0.024	-0.036	-0.027	-0.014	-0.004	-0.009	-0.031	-0.039	-0.058	-0.089
0.899	7.953	0.02	4.923	8.698	-0.029	-0.024	-0.035	-0.025	-0.012	-0.003	-0.007	-0.029	-0.037	-0.056	-0.086
0.900	3.461	8.03	4.926	8.694	-0.052	-0.069	-0.100	-0.099	-0.083	-0.056	-0.036	-0.037	-0.025	-0.032	-0.050
0.901	3.427	4.00	4.935	8.677	-0.037	-0.040	-0.058	-0.044	-0.033	-0.019	-0.013	-0.029	-0.031	-0.046	-0.077
0.601	1.492	0.00	2.915	11.523	-0.018	-0.030	-0.044	-0.034	-0.023	-0.017	-0.016	-0.030	-0.030	-0.041	-0.057
0.602	2.004	0.00	2.921	11.517	-0.020	-0.032	-0.046	-0.035	-0.026	-0.019	-0.018	-0.033	-0.032	-0.042	-0.061
0.601	2.996	0.01	2.916	11.523	-0.020	-0.033	-0.047	-0.035	-0.027	-0.022	-0.020	-0.034	-0.033	-0.043	-0.063
0.601	3.441	-0.01	2.916	11.522	-0.020	-0.032	-0.046	-0.035	-0.028	-0.021	-0.020	-0.035	-0.033	-0.043	-0.064
0.601	3.980	-0.01	2.916	11.523	-0.020	-0.033	-0.047	-0.036	-0.028	-0.022	-0.020	-0.034	-0.033	-0.043	-0.064
0.601	5.968	0.01	2.914	11.526	-0.020	-0.032	-0.046	-0.036	-0.028	-0.021	-0.020	-0.035	-0.032	-0.041	-0.061
0.601	7.933	0.02	2.913	11.524	-0.019	-0.032	-0.045	-0.036	-0.026	-0.019	-0.018	-0.033	-0.032	-0.041	-0.061
0.600	3.503	8.01	2.906	11.536	0.015	-0.005	-0.025	-0.023	-0.010	0.007	0.019	0.016	0.026	0.019	-0.008
0.602	3.448	3.99	2.919	11.520	0.008	-0.009	-0.026	-0.021	-0.009	0.003	0.007	-0.005	-0.001	-0.013	-0.038
1.201	1.469	0.09	6.130	6.066	-0.063	-0.074	-0.071	-0.056	-0.060	-0.077	-0.054	-0.071	-0.087	-0.111	-0.146
1.202	1.962	0.09	6.129	6.065	-0.062	-0.074	-0.072	-0.056	-0.060	-0.077	-0.054	-0.072	-0.087	-0.111	-0.146
1.201	2.928	0.10	6.129	6.068	-0.062	-0.073	-0.071	-0.056	-0.061	-0.078	-0.054	-0.071	-0.086	-0.111	-0.146
1.201	3.966	0.10	6.128	6.067	-0.062	-0.074	-0.072	-0.055	-0.060	-0.077	-0.054	-0.071	-0.086	-0.111	-0.146
1.201	3.960	0.11	6.128	6.068	-0.061	-0.073	-0.071	-0.055	-0.060	-0.077	-0.054	-0.071	-0.086	-0.111	-0.146
1.201	5.879	0.02	6.128	6.067	-0.070	-0.082	-0.078	-0.064	-0.066	-0.081	-0.060	-0.079	-0.095	-0.120	-0.154
1.202	7.827	0.03	6.130	6.061	-0.071	-0.083	-0.080	-0.066	-0.066	-0.081	-0.061	-0.080	-0.095	-0.120	-0.155
1.199	3.394	8.04	6.125	6.084	0.145	0.101	0.056	0.003	-0.025	-0.028	0.054	0.053	0.046	0.015	-0.046
1.202	3.409	3.98	6.130	6.060	0.054	0.038	0.003	-0.027	-0.037	-0.051	-0.013	-0.020	-0.028	-0.051	-0.106

Table 2. Continued

(d) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—										
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
0.799	1.506	0.02	4.305	9.641	-0.085	-0.090	-0.075	-0.053	-0.007	-0.031	0.048	0.115	0.161	0.185	
0.799	1.999	0.03	4.307	9.639	-0.083	-0.089	-0.074	-0.052	-0.007	-0.033	0.043	0.109	0.155	0.173	
0.798	2.988	0.02	4.298	9.652	-0.084	-0.088	-0.074	-0.052	-0.008	-0.034	0.041	0.108	0.153	0.169	
0.800	3.459	0.02	4.316	9.625	-0.085	-0.091	-0.075	-0.054	-0.009	-0.034	0.041	0.108	0.154	0.170	
0.800	3.992	0.03	4.316	9.627	-0.086	-0.091	-0.076	-0.055	-0.010	-0.034	0.041	0.108	0.156	0.173	
0.799	6.023	-0.03	4.307	9.639	-0.084	-0.087	-0.073	-0.049	-0.005	-0.025	0.049	0.119	0.173	0.194	
0.800	5.969	-0.03	4.312	9.632	-0.084	-0.087	-0.072	-0.048	-0.003	-0.024	0.049	0.120	0.173	0.195	
0.799	7.962	-0.02	4.309	9.638	-0.081	-0.083	-0.068	-0.043	0.004	-0.013	0.060	0.135	0.193	0.214	
0.799	3.455	8.01	4.310	9.642	-0.043	-0.051	-0.047	-0.032	0.001	-0.026	0.039	0.109	0.163	0.169	
0.806	3.432	4.01	4.351	9.578	-0.073	-0.077	-0.070	-0.049	-0.012	-0.027	0.031	0.099	0.151	0.151	
0.801	3.445	4.01	4.322	9.622	-0.071	-0.075	-0.068	-0.047	-0.012	-0.026	0.031	0.099	0.151	0.152	
0.898	1.504	0.00	4.916	8.710	-0.102	-0.111	-0.086	-0.051	0.005	-0.012	0.068	0.133	0.177	0.187	
0.898	1.995	0.01	4.916	8.709	-0.103	-0.112	-0.088	-0.053	0.004	-0.016	0.063	0.128	0.170	0.177	
0.899	2.999	0.02	4.919	8.703	-0.105	-0.114	-0.090	-0.055	0.001	-0.018	0.060	0.126	0.168	0.173	
0.899	3.462	0.03	4.919	8.703	-0.104	-0.115	-0.090	-0.054	0.001	-0.018	0.061	0.126	0.169	0.174	
0.899	3.986	0.01	4.923	8.696	-0.104	-0.114	-0.090	-0.054	0.002	-0.017	0.061	0.127	0.172	0.179	
0.899	5.982	0.01	4.923	8.697	-0.102	-0.111	-0.086	-0.049	0.007	-0.008	0.070	0.138	0.187	0.199	
0.899	7.953	0.02	4.923	8.698	-0.098	-0.105	-0.080	-0.042	0.016	0.003	0.082	0.152	0.205	0.218	
0.900	3.461	8.03	4.926	8.694	-0.062	-0.070	-0.058	-0.034	0.011	-0.010	0.063	0.133	0.183	0.180	
0.901	3.427	4.00	4.935	8.677	-0.092	-0.101	-0.085	-0.052	-0.003	-0.011	0.052	0.118	0.165	0.161	
0.601	1.492	0.00	2.915	11.523	-0.058	-0.059	-0.049	-0.037	-0.001	-0.028	0.041	0.103	0.150	0.171	
0.602	2.004	0.00	2.921	11.517	-0.061	-0.063	-0.054	-0.042	-0.008	-0.037	0.029	0.090	0.134	0.145	
0.601	2.996	0.01	2.916	11.523	-0.064	-0.066	-0.057	-0.044	-0.011	-0.043	0.023	0.083	0.127	0.135	
0.601	3.441	-0.01	2.916	11.522	-0.064	-0.067	-0.057	-0.045	-0.011	-0.042	0.023	0.083	0.127	0.137	
0.601	3.980	-0.01	2.916	11.523	-0.065	-0.065	-0.057	-0.044	-0.012	-0.043	0.023	0.084	0.128	0.142	
0.601	5.968	0.01	2.914	11.526	-0.063	-0.063	-0.054	-0.042	-0.007	-0.036	0.032	0.096	0.144	0.168	
0.601	7.933	0.02	2.913	11.524	-0.060	-0.061	-0.051	-0.037	-0.001	-0.028	0.044	0.110	0.162	0.191	
0.600	3.503	8.01	2.906	11.536	-0.013	-0.023	-0.022	-0.018	0.009	-0.029	0.038	0.093	0.132	0.141	
0.602	3.448	3.99	2.919	11.520	-0.041	-0.048	-0.042	-0.035	-0.006	-0.032	0.031	0.086	0.124	0.138	
1.201	1.469	0.09	6.130	6.066	-0.140	-0.123	-0.154	-0.192	-0.200	-0.258	-0.339	-0.146	-0.019	0.026	
1.202	1.962	0.09	6.129	6.065	-0.141	-0.123	-0.153	-0.192	-0.200	-0.258	-0.340	-0.227	-0.052	0.016	
1.201	2.928	0.10	6.129	6.068	-0.140	-0.122	-0.154	-0.192	-0.200	-0.260	-0.342	-0.251	-0.091	0.004	
1.201	3.396	0.10	6.128	6.067	-0.140	-0.122	-0.154	-0.192	-0.200	-0.259	-0.341	-0.250	-0.095	0.004	
1.201	3.960	0.11	6.128	6.068	-0.139	-0.122	-0.154	-0.192	-0.200	-0.259	-0.342	-0.247	-0.091	0.003	
1.201	5.879	0.02	6.128	6.067	-0.146	-0.129	-0.160	-0.199	-0.206	-0.268	-0.348	-0.224	-0.055	0.017	
1.202	7.827	0.03	6.130	6.061	-0.148	-0.130	-0.160	-0.199	-0.207	-0.266	-0.344	-0.110	-0.016	0.035	
1.199	3.394	8.04	6.125	6.084	-0.057	-0.097	-0.145	-0.164	-0.158	-0.215	-0.324	-0.267	-0.183	-0.072	
1.202	3.409	3.98	6.130	6.060	-0.103	-0.098	-0.124	-0.156	-0.155	-0.226	-0.306	-0.245	-0.145	-0.008	

Table 2. Continued

(e) $\theta = 180^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854	
0.799	1.506	0.02	4.305	9.641	-0.036	-0.050	-0.051	-0.046	-0.043	-0.043	-0.051	-0.061	-0.063	-0.066	-0.071	-0.074
0.799	1.999	0.03	4.307	9.639	-0.034	-0.047	-0.049	-0.043	-0.041	-0.041	-0.049	-0.059	-0.061	-0.064	-0.069	-0.072
0.798	2.988	0.02	4.298	9.652	-0.033	-0.047	-0.048	-0.043	-0.039	-0.039	-0.048	-0.059	-0.059	-0.063	-0.068	-0.071
0.800	3.459	0.02	4.316	9.625	-0.034	-0.047	-0.049	-0.044	-0.041	-0.041	-0.049	-0.061	-0.060	-0.064	-0.069	-0.073
0.800	3.992	0.03	4.316	9.627	-0.034	-0.046	-0.049	-0.044	-0.040	-0.040	-0.050	-0.062	-0.061	-0.065	-0.070	-0.074
0.799	6.023	-0.03	4.307	9.639	-0.036	-0.045	-0.049	-0.044	-0.039	-0.039	-0.049	-0.061	-0.060	-0.062	-0.068	-0.073
0.800	5.969	-0.03	4.312	9.632	-0.036	-0.044	-0.048	-0.044	-0.040	-0.040	-0.049	-0.062	-0.060	-0.062	-0.068	-0.072
0.799	7.962	-0.02	4.309	9.638	-0.036	-0.044	-0.049	-0.045	-0.040	-0.038	-0.048	-0.062	-0.058	-0.061	-0.066	-0.071
0.799	3.455	8.01	4.310	9.642	0.006	-0.010	-0.023	-0.025	-0.020	-0.030	-0.043	-0.033	-0.033	-0.031	-0.035	-0.040
0.806	3.432	4.01	4.351	9.578	0.010	-0.021	-0.031	-0.030	-0.023	-0.035	-0.051	-0.044	-0.042	-0.045	-0.053	-0.061
0.801	3.445	4.01	4.322	9.622	-0.009	-0.018	-0.029	-0.028	-0.022	-0.033	-0.050	-0.042	-0.044	-0.045	-0.051	-0.059
0.898	1.504	0.00	4.916	8.710	-0.044	-0.051	-0.047	-0.039	-0.035	-0.043	-0.055	-0.059	-0.065	-0.065	-0.073	-0.080
0.898	1.995	0.01	4.916	8.709	-0.040	-0.047	-0.044	-0.036	-0.031	-0.040	-0.052	-0.057	-0.063	-0.066	-0.074	-0.082
0.899	2.999	0.02	4.919	8.703	-0.044	-0.050	-0.046	-0.038	-0.033	-0.043	-0.055	-0.060	-0.066	-0.066	-0.074	-0.082
0.899	3.462	0.03	4.919	8.703	-0.042	-0.049	-0.045	-0.038	-0.033	-0.043	-0.056	-0.060	-0.066	-0.066	-0.074	-0.082
0.899	3.986	-0.01	4.923	8.696	-0.045	-0.048	-0.044	-0.037	-0.031	-0.040	-0.054	-0.058	-0.064	-0.064	-0.073	-0.081
0.899	5.982	0.01	4.923	8.697	-0.043	-0.047	-0.044	-0.037	-0.031	-0.041	-0.055	-0.058	-0.064	-0.064	-0.073	-0.081
0.899	7.953	0.02	4.923	8.698	-0.044	-0.045	-0.043	-0.035	-0.029	-0.039	-0.053	-0.056	-0.061	-0.061	-0.070	-0.078
0.900	3.461	8.03	4.926	8.694	-0.025	-0.052	-0.068	-0.069	-0.065	-0.074	-0.082	-0.070	-0.062	-0.062	-0.063	-0.063
0.901	3.427	4.00	4.935	8.677	-0.036	-0.047	-0.052	-0.047	-0.039	-0.049	-0.064	-0.059	-0.060	-0.068	-0.076	-0.076
0.601	1.492	0.00	2.915	11.523	-0.027	-0.039	-0.042	-0.037	-0.031	-0.040	-0.052	-0.050	-0.050	-0.053	-0.055	-0.055
0.602	2.004	0.00	2.921	11.517	-0.030	-0.041	-0.044	-0.040	-0.033	-0.042	-0.055	-0.052	-0.053	-0.057	-0.058	-0.058
0.601	2.996	0.01	2.916	11.523	-0.030	-0.039	-0.045	-0.040	-0.034	-0.043	-0.056	-0.053	-0.054	-0.058	-0.061	-0.061
0.601	3.441	-0.01	2.916	11.522	-0.031	-0.040	-0.044	-0.040	-0.033	-0.044	-0.057	-0.054	-0.054	-0.058	-0.061	-0.061
0.601	3.980	-0.01	2.916	11.523	-0.031	-0.040	-0.044	-0.041	-0.035	-0.043	-0.057	-0.053	-0.052	-0.057	-0.059	-0.059
0.601	5.968	0.01	2.914	11.526	-0.030	-0.039	-0.044	-0.040	-0.034	-0.043	-0.055	-0.052	-0.052	-0.056	-0.057	-0.057
0.601	7.933	0.02	2.913	11.524	-0.029	-0.041	-0.044	-0.040	-0.034	-0.043	-0.054	-0.052	-0.052	-0.056	-0.057	-0.057
0.600	3.503	8.01	2.906	11.536	0.029	0.011	0.001	0.005	0.007	-0.002	-0.013	-0.009	-0.008	-0.013	-0.013	-0.013
0.602	3.448	3.99	2.919	11.520	0.005	-0.011	-0.017	-0.013	-0.011	-0.019	-0.031	-0.030	-0.030	-0.035	-0.036	-0.036
1.201	1.469	0.09	6.130	6.066	-0.019	-0.052	-0.078	-0.079	-0.076	-0.076	-0.094	-0.113	-0.125	-0.133	-0.148	-0.148
1.202	1.962	0.09	6.129	6.065	-0.019	-0.052	-0.078	-0.081	-0.077	-0.076	-0.094	-0.112	-0.124	-0.133	-0.149	-0.149
1.201	2.928	0.10	6.129	6.068	-0.019	-0.051	-0.078	-0.081	-0.077	-0.076	-0.094	-0.112	-0.124	-0.133	-0.149	-0.149
1.201	3.996	0.10	6.128	6.067	-0.018	-0.051	-0.078	-0.081	-0.076	-0.075	-0.094	-0.112	-0.124	-0.133	-0.149	-0.150
1.201	3.960	0.11	6.128	6.068	-0.018	-0.051	-0.077	-0.079	-0.075	-0.075	-0.094	-0.121	-0.133	-0.140	-0.155	-0.155
1.201	5.879	0.02	6.130	6.067	-0.026	-0.060	-0.086	-0.088	-0.084	-0.084	-0.100	-0.121	-0.132	-0.139	-0.154	-0.154
1.202	7.827	0.03	6.125	6.061	-0.026	-0.060	-0.086	-0.089	-0.085	-0.084	-0.100	-0.121	-0.132	-0.140	-0.154	-0.154
1.199	3.394	8.04	6.130	6.084	0.187	0.140	0.103	0.080	0.055	0.025	-0.021	-0.042	-0.042	-0.043	-0.049	-0.049
1.202	3.409	3.98	6.130	6.060	0.077	0.051	0.030	0.022	0.011	-0.009	-0.046	-0.074	-0.091	-0.105	-0.122	-0.122

Table 2. Continued

(e) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.974	0.986		
0.799	1.506	0.02	4.305	9.641	-0.077	-0.074	-0.077	-0.071	-0.077	-0.102	0.120	0.168		
0.799	1.999	0.03	4.307	9.639	-0.075	-0.073	-0.077	-0.072	-0.078	-0.107	0.114	0.161		
0.798	2.988	0.02	4.298	9.652	-0.074	-0.072	-0.077	-0.072	-0.080	-0.108	0.112	0.158		
0.800	3.459	0.02	4.316	9.625	-0.076	-0.073	-0.079	-0.074	-0.080	-0.108	0.112	0.159		
0.800	3.992	0.03	4.316	9.627	-0.074	-0.073	-0.079	-0.074	-0.079	-0.108	0.114	0.161		
0.799	6.023	-0.03	4.307	9.639	-0.069	-0.069	-0.074	-0.069	-0.070	-0.096	0.126	0.175		
0.800	5.969	-0.03	4.312	9.632	-0.069	-0.068	-0.074	-0.068	-0.069	-0.095	0.127	0.175		
0.799	7.962	-0.02	4.309	9.638	-0.064	-0.064	-0.068	-0.061	-0.059	-0.080	0.142	0.192		
0.799	3.455	8.01	4.310	9.642	-0.033	-0.041	-0.057	-0.064	-0.076	-0.125	0.118	0.162		
0.806	3.432	4.01	4.351	9.578	-0.053	-0.060	-0.072	-0.075	-0.080	-0.120	0.110	0.150		
0.801	3.445	4.01	4.322	9.622	-0.051	-0.059	-0.071	-0.074	-0.079	-0.119	0.111	0.151		
0.898	1.504	0.00	4.916	8.710	-0.088	-0.085	-0.083	-0.068	-0.063	-0.088	0.138	0.180		
0.898	1.995	0.01	4.916	8.709	-0.089	-0.086	-0.085	-0.070	-0.066	-0.092	0.133	0.173		
0.899	2.999	0.02	4.919	8.703	-0.090	-0.088	-0.088	-0.072	-0.069	-0.096	0.131	0.171		
0.899	3.462	0.03	4.919	8.703	-0.090	-0.088	-0.088	-0.073	-0.069	-0.096	0.131	0.172		
0.899	3.986	-0.01	4.923	8.696	-0.088	-0.087	-0.087	-0.072	-0.068	-0.094	0.133	0.174		
0.899	5.982	0.01	4.923	8.697	-0.086	-0.084	-0.083	-0.068	-0.061	-0.084	0.143	0.186		
0.899	7.953	0.02	4.923	8.698	-0.080	-0.077	-0.076	-0.060	-0.050	-0.070	0.157	0.201		
0.900	3.461	8.03	4.926	8.694	-0.053	-0.056	-0.066	-0.064	-0.070	-0.121	0.144	0.184		
0.901	3.427	4.00	4.935	8.677	-0.073	-0.078	-0.085	-0.078	-0.077	-0.115	0.129	0.163		
0.601	1.492	0.00	2.915	11.523	-0.052	-0.051	-0.058	-0.056	-0.065	-0.090	0.107	0.153		
0.602	2.004	0.00	2.921	11.517	-0.056	-0.056	-0.064	-0.064	-0.073	-0.101	0.094	0.137		
0.601	2.996	0.01	2.916	11.523	-0.057	-0.059	-0.067	-0.067	-0.077	-0.108	0.088	0.130		
0.601	3.441	-0.01	2.916	11.522	-0.057	-0.059	-0.066	-0.067	-0.076	-0.107	0.088	0.130		
0.601	3.980	-0.01	2.916	11.523	-0.058	-0.060	-0.067	-0.068	-0.077	-0.108	0.088	0.131		
0.601	5.968	0.01	2.914	11.526	-0.057	-0.057	-0.064	-0.063	-0.071	-0.099	0.100	0.147		
0.601	7.933	0.02	2.913	11.524	-0.055	-0.055	-0.059	-0.057	-0.065	-0.088	0.115	0.166		
0.600	3.503	8.01	2.906	11.536	-0.022	-0.026	-0.039	-0.046	-0.071	-0.113	0.096	0.139		
0.602	3.448	3.99	2.919	11.520	-0.044	-0.045	-0.055	-0.057	-0.077	-0.113	0.090	0.130		
1.201	1.469	0.09	6.130	6.066	-0.167	-0.166	-0.178	-0.192	-0.214	-0.240	-0.127	-0.022		
1.202	1.962	0.09	6.129	6.065	-0.167	-0.166	-0.179	-0.192	-0.213	-0.240	-0.223	-0.039		
1.201	2.928	0.10	6.129	6.068	-0.167	-0.166	-0.179	-0.192	-0.213	-0.240	-0.264	-0.063		
1.201	3.396	0.10	6.128	6.067	-0.166	-0.166	-0.179	-0.193	-0.213	-0.239	-0.255	-0.065		
1.201	3.960	0.11	6.128	6.068	-0.166	-0.166	-0.178	-0.192	-0.213	-0.240	-0.225	-0.063		
1.201	5.879	0.02	6.128	6.067	-0.178	-0.174	-0.186	-0.199	-0.223	-0.252	-0.222	-0.043		
1.202	7.827	0.03	6.130	6.061	-0.177	-0.174	-0.186	-0.200	-0.224	-0.251	-0.088	-0.017		
1.199	3.394	8.04	6.125	6.084	-0.049	-0.060	-0.101	-0.146	-0.203	-0.206	-0.324	-0.180		
1.202	3.409	3.98	6.130	6.060	-0.134	-0.133	-0.144	-0.153	-0.179	-0.202	-0.278	-0.094		

Table 2. Continued

(f) $\theta = 225^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.833	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
0.799	1.506	0.02	4.305	9.641	-0.085	-0.065	-0.046	-0.048	-0.111	-0.148	0.073	0.143	0.180	0.201	
0.799	1.999	0.03	4.307	9.639	-0.084	-0.064	-0.046	-0.051	-0.117	-0.160	0.066	0.139	0.177	0.195	
0.798	2.988	0.02	4.298	9.652	-0.083	-0.064	-0.047	-0.053	-0.122	-0.166	0.063	0.139	0.177	0.192	
0.800	3.459	0.02	4.316	9.625	-0.083	-0.064	-0.048	-0.053	-0.121	-0.165	0.064	0.138	0.176	0.192	
0.800	3.992	0.03	4.316	9.627	-0.082	-0.064	-0.047	-0.052	-0.121	-0.163	0.066	0.139	0.177	0.193	
0.799	6.023	-0.03	4.307	9.639	-0.077	-0.058	-0.040	-0.042	-0.106	-0.141	0.079	0.148	0.184	0.205	
0.800	5.969	-0.03	4.312	9.632	-0.076	-0.058	-0.040	-0.041	-0.106	-0.140	0.081	0.149	0.185	0.205	
0.799	7.962	-0.02	4.309	9.638	-0.070	-0.051	-0.030	-0.028	-0.087	-0.115	0.098	0.157	0.190	0.215	
0.799	3.455	8.01	4.310	9.642	-0.049	-0.037	-0.026	-0.039	-0.139	-0.217	0.054	0.139	0.180	0.217	
0.806	3.432	4.01	4.351	9.578	-0.065	-0.053	-0.038	-0.046	-0.133	-0.193	0.059	0.132	0.165	0.188	
0.801	3.445	4.01	4.322	9.622	-0.062	-0.051	-0.036	-0.044	-0.132	-0.194	0.059	0.133	0.165	0.168	
0.898	1.504	0.00	4.916	8.710	-0.092	-0.069	-0.042	-0.034	-0.086	-0.108	0.099	0.140	0.166	0.191	
0.898	1.995	0.01	4.916	8.709	-0.093	-0.070	-0.044	-0.037	-0.091	-0.115	0.096	0.139	0.166	0.188	
0.899	2.999	0.02	4.919	8.703	-0.094	-0.072	-0.048	-0.041	-0.098	-0.125	0.091	0.138	0.165	0.185	
0.899	3.462	0.03	4.919	8.703	-0.094	-0.072	-0.048	-0.041	-0.098	-0.125	0.091	0.138	0.166	0.186	
0.899	3.986	-0.01	4.923	8.696	-0.093	-0.072	-0.047	-0.041	-0.098	-0.124	0.093	0.141	0.169	0.189	
0.899	5.982	0.01	4.923	8.697	-0.089	-0.068	-0.043	-0.034	-0.088	-0.111	0.101	0.145	0.172	0.198	
0.899	7.953	0.02	4.923	8.698	-0.082	-0.060	-0.034	-0.024	-0.075	-0.093	0.110	0.152	0.180	0.210	
0.900	3.461	8.03	4.926	8.694	-0.066	-0.049	-0.029	-0.034	-0.123	-0.194	0.087	0.156	0.189	0.201	
0.901	3.427	4.00	4.935	8.677	-0.083	-0.066	-0.044	-0.045	-0.125	-0.183	0.077	0.136	0.162	0.168	
0.601	1.492	0.00	2.915	11.523	-0.063	-0.048	-0.032	-0.041	-0.108	-0.146	0.057	0.139	0.183	0.196	
0.602	2.004	0.00	2.921	11.517	-0.068	-0.053	-0.039	-0.050	-0.120	-0.162	0.042	0.127	0.169	0.174	
0.601	2.996	0.01	2.916	11.523	-0.070	-0.056	-0.041	-0.052	-0.125	-0.168	0.036	0.121	0.165	0.167	
0.601	3.441	-0.01	2.916	11.522	-0.070	-0.055	-0.041	-0.052	-0.125	-0.167	0.036	0.122	0.166	0.169	
0.601	3.980	-0.01	2.916	11.523	-0.072	-0.057	-0.042	-0.053	-0.125	-0.167	0.037	0.123	0.167	0.172	
0.601	5.968	0.01	2.914	11.526	-0.070	-0.054	-0.037	-0.047	-0.113	-0.152	0.049	0.134	0.181	0.194	
0.601	7.933	0.02	2.913	11.524	-0.066	-0.049	-0.031	-0.039	-0.100	-0.134	0.065	0.146	0.193	0.213	
0.600	3.503	8.01	2.906	11.536	-0.046	-0.031	-0.019	-0.043	-0.133	-0.198	0.026	0.122	0.175	0.177	
0.602	3.448	3.99	2.919	11.520	-0.063	-0.047	-0.032	-0.050	-0.129	-0.185	0.028	0.116	0.162	0.167	
1.201	1.469	0.09	6.130	6.066	-0.195	-0.195	-0.171	-0.132	-0.191	-0.309	-0.216	-0.087	-0.044	-0.001	
1.202	1.962	0.09	6.129	6.065	-0.195	-0.195	-0.171	-0.133	-0.191	-0.309	-0.325	-0.111	-0.055	-0.011	
1.201	2.928	0.10	6.129	6.068	-0.195	-0.195	-0.171	-0.132	-0.190	-0.309	-0.381	-0.131	-0.065	-0.021	
1.201	3.396	0.10	6.128	6.067	-0.194	-0.195	-0.170	-0.131	-0.190	-0.310	-0.404	-0.139	-0.062	-0.014	
1.201	3.960	0.11	6.128	6.068	-0.194	-0.195	-0.171	-0.131	-0.190	-0.309	-0.432	-0.148	-0.051	0.001	
1.201	5.879	0.02	6.128	6.067	-0.206	-0.205	-0.180	-0.141	-0.196	-0.315	-0.383	-0.123	-0.057	0.006	
1.202	7.827	0.03	6.130	6.061	-0.205	-0.205	-0.181	-0.144	-0.199	-0.316	-0.313	-0.082	-0.014	-0.032	
1.199	3.394	8.04	6.125	6.084	-0.065	-0.063	-0.051	-0.057	-0.170	-0.299	-0.446	-0.373	-0.178	-0.090	
1.202	3.409	3.98	6.130	6.060	-0.155	-0.151	-0.124	-0.099	-0.171	-0.289	-0.419	-0.155	-0.067	-0.016	

Table 2. Continued

(g) $\theta = 240^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986
0.799	1.506	0.02	4.305	9.641	-0.067	-0.034	-0.031	-0.070	-0.172	0.095	0.137	0.172
0.799	1.999	0.03	4.307	9.639	-0.067	-0.035	-0.034	-0.077	-0.188	0.088	0.135	0.173
0.798	2.988	0.02	4.298	9.652	-0.068	-0.036	-0.037	-0.083	-0.198	0.086	0.136	0.175
0.800	3.459	0.02	4.316	9.625	-0.070	-0.037	-0.037	-0.082	-0.192	0.085	0.133	0.173
0.800	3.992	0.03	4.316	9.627	-0.071	-0.035	-0.035	-0.081	-0.189	0.086	0.133	0.174
0.799	6.023	-0.03	4.307	9.639	-0.066	-0.028	-0.024	-0.064	-0.161	0.096	0.137	0.176
0.800	5.969	-0.03	4.312	9.632	-0.066	-0.027	-0.024	-0.063	-0.158	0.096	0.137	0.177
0.799	7.962	-0.02	4.309	9.638	-0.060	-0.016	-0.008	-0.040	-0.120	0.108	0.139	0.178
0.799	3.455	8.01	4.310	9.642	-0.055	-0.015	-0.025	-0.095	-0.251	0.073	0.130	0.177
0.806	3.432	4.01	4.351	9.578	-0.069	-0.025	-0.029	-0.085	-0.214	0.073	0.114	0.157
0.801	3.445	4.01	4.322	9.622	-0.067	-0.024	-0.028	-0.085	-0.215	0.073	0.115	0.158
0.898	1.504	0.00	4.916	8.710	-0.069	-0.028	-0.013	-0.035	-0.099	0.102	0.120	0.151
0.898	1.995	0.01	4.916	8.709	-0.071	-0.030	-0.016	-0.041	-0.108	0.101	0.120	0.153
0.899	2.999	0.02	4.919	8.703	-0.075	-0.033	-0.021	-0.049	-0.119	0.097	0.120	0.154
0.899	3.462	0.03	4.919	8.703	-0.075	-0.033	-0.021	-0.049	-0.119	0.097	0.120	0.154
0.899	3.986	-0.01	4.923	8.696	-0.076	-0.033	-0.020	-0.048	-0.118	0.097	0.122	0.157
0.899	5.982	0.01	4.923	8.697	-0.072	-0.027	-0.013	-0.037	-0.101	0.102	0.124	0.157
0.899	7.953	0.02	4.923	8.698	-0.065	-0.018	-0.002	-0.023	-0.081	0.109	0.127	0.160
0.900	3.461	8.03	4.926	8.694	-0.060	-0.017	-0.016	-0.069	-0.194	0.096	0.135	0.177
0.901	3.427	4.00	4.935	8.677	-0.079	-0.031	-0.026	-0.072	-0.181	0.080	0.106	0.140
0.601	1.492	0.00	2.915	11.523	-0.051	-0.023	-0.025	-0.071	-0.191	0.081	0.149	0.183
0.602	2.004	0.00	2.921	11.517	-0.058	-0.030	-0.034	-0.084	-0.211	0.068	0.138	0.174
0.601	2.996	0.01	2.916	11.523	-0.061	-0.033	-0.038	-0.089	-0.221	0.063	0.134	0.170
0.601	3.441	-0.01	2.916	11.522	-0.060	-0.032	-0.037	-0.088	-0.219	0.063	0.133	0.170
0.601	3.980	-0.01	2.916	11.523	-0.061	-0.034	-0.039	-0.089	-0.219	0.064	0.135	0.171
0.601	5.968	0.01	2.914	11.526	-0.056	-0.030	-0.032	-0.077	-0.198	0.076	0.144	0.179
0.601	7.933	0.02	2.913	11.524	-0.049	-0.023	-0.022	-0.062	-0.175	0.091	0.153	0.188
0.600	3.503	8.01	2.906	11.536	-0.034	-0.017	-0.030	-0.099	-0.275	0.059	0.143	0.182
0.602	3.448	3.99	2.919	11.520	-0.047	-0.028	-0.035	-0.093	-0.251	0.060	0.131	0.166
1.201	1.469	0.09	6.130	6.066	-0.202	-0.148	-0.113	-0.154	-0.412	-0.171	-0.093	-0.042
1.202	1.962	0.09	6.129	6.065	-0.202	-0.148	-0.114	-0.155	-0.412	-0.236	-0.122	-0.059
1.201	2.928	0.10	6.129	6.068	-0.203	-0.147	-0.113	-0.154	-0.411	-0.296	-0.140	-0.073
1.201	3.396	0.10	6.128	6.067	-0.203	-0.147	-0.112	-0.153	-0.412	-0.355	-0.149	-0.074
1.201	3.960	0.11	6.128	6.068	-0.203	-0.147	-0.112	-0.153	-0.412	-0.447	-0.157	-0.070
1.201	5.879	0.02	6.128	6.067	-0.209	-0.158	-0.122	-0.161	-0.421	-0.254	-0.132	-0.062
1.202	7.827	0.03	6.130	6.061	-0.210	-0.159	-0.124	-0.164	-0.421	-0.204	-0.095	-0.023
1.199	3.394	8.04	6.125	6.084	-0.074	-0.037	-0.034	-0.115	-0.405	-0.555	-0.270	-0.230
1.202	3.409	3.98	6.130	6.060	-0.161	-0.104	-0.075	-0.126	-0.398	-0.344	-0.168	-0.104

Table 2. Continued

(h) $\theta = 255^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.941	0.963	0.974	0.986	0.997
0.799	1.506	0.02	4.305	9.641	-0.042	0.090	0.125	0.168	0.205
0.799	1.999	0.03	4.307	9.639	-0.051	0.087	0.123	0.168	0.204
0.798	2.988	0.02	4.298	9.652	-0.058	0.088	0.126	0.171	0.203
0.800	3.459	0.02	4.316	9.625	-0.057	0.086	0.125	0.170	0.203
0.800	3.992	0.03	4.316	9.627	-0.055	0.086	0.123	0.169	0.203
0.799	6.023	-0.03	4.307	9.639	-0.035	0.092	0.124	0.172	0.209
0.800	5.969	-0.03	4.312	9.632	-0.035	0.094	0.126	0.174	0.210
0.799	7.962	-0.02	4.309	9.638	-0.008	0.102	0.128	0.174	0.214
0.799	3.455	8.01	4.310	9.642	-0.071	0.074	0.111	0.169	0.206
0.806	3.432	4.01	4.351	9.578	-0.055	0.072	0.099	0.150	0.184
0.801	3.445	4.01	4.322	9.622	-0.054	0.073	0.101	0.152	0.186
0.898	1.504	0.00	4.916	8.710	0.003	0.093	0.113	0.145	0.182
0.898	1.995	0.01	4.916	8.709	-0.004	0.091	0.114	0.147	0.184
0.899	2.999	0.02	4.919	8.703	-0.013	0.089	0.112	0.149	0.185
0.899	3.462	0.03	4.919	8.703	-0.013	0.089	0.112	0.149	0.185
0.899	3.986	-0.01	4.923	8.696	-0.013	0.089	0.114	0.151	0.189
0.899	5.982	0.01	4.923	8.697	-0.001	0.093	0.114	0.152	0.191
0.899	7.953	0.02	4.923	8.698	0.015	0.098	0.117	0.152	0.193
0.900	3.461	8.03	4.926	8.694	-0.034	0.081	0.115	0.168	0.207
0.901	3.427	4.00	4.935	8.677	-0.035	0.061	0.082	0.126	0.163
0.601	1.492	0.00	2.915	11.523	-0.052	0.094	0.141	0.181	0.204
0.602	2.004	0.00	2.921	11.517	-0.067	0.082	0.131	0.171	0.188
0.601	2.996	0.01	2.916	11.523	-0.072	0.076	0.126	0.166	0.181
0.601	3.441	-0.01	2.916	11.522	-0.071	0.076	0.127	0.166	0.182
0.601	3.980	-0.01	2.916	11.523	-0.071	0.076	0.127	0.166	0.183
0.601	5.968	0.01	2.914	11.526	-0.059	0.084	0.134	0.174	0.197
0.601	7.933	0.02	2.913	11.524	-0.043	0.099	0.142	0.182	0.211
0.600	3.503	8.01	2.906	11.536	-0.086	0.078	0.139	0.177	0.189
0.602	3.448	3.99	2.919	11.520	-0.075	0.074	0.128	0.162	0.177
1.201	1.469	0.09	6.130	6.066	-0.159	-0.219	-0.079	-0.004	0.038
1.202	1.962	0.09	6.129	6.065	-0.160	-0.245	-0.113	-0.033	0.014
1.201	2.928	0.10	6.128	6.068	-0.160	-0.262	-0.138	-0.053	-0.003
1.201	3.996	0.11	6.128	6.067	-0.160	-0.245	-0.145	-0.066	-0.013
1.201	3.960	0.11	6.128	6.068	-0.160	-0.207	-0.151	-0.088	-0.029
1.201	5.879	0.02	6.128	6.067	-0.166	-0.253	-0.126	-0.047	0.011
1.202	7.827	0.03	6.130	6.061	-0.169	-0.188	-0.090	-0.023	0.031
1.199	3.394	8.04	6.125	6.084	-0.120	-0.260	-0.210	-0.177	-0.137
1.202	3.409	3.98	6.130	6.060	-0.135	-0.248	-0.155	-0.078	-0.017

Table 2. Continued

(i) $\theta = 270^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—			
					0.951	0.963	0.974	0.986
0.799	1.506	0.02	4.305	9.641	0.032	0.088	0.125	0.168
0.799	1.999	0.03	4.307	9.639	0.020	0.085	0.123	0.169
0.798	2.988	0.02	4.298	9.652	0.009	0.086	0.127	0.171
0.800	3.459	0.02	4.316	9.625	0.012	0.085	0.125	0.170
0.800	3.992	0.03	4.316	9.627	0.016	0.085	0.123	0.170
0.799	6.023	-0.03	4.307	9.639	0.041	0.092	0.124	0.171
0.800	5.969	-0.03	4.312	9.632	0.040	0.094	0.126	0.172
0.799	7.962	-0.02	4.309	9.638	0.070	0.103	0.128	0.171
0.799	3.455	8.01	4.310	9.642	0.020	0.073	0.111	0.167
0.806	3.432	4.01	4.351	9.578	0.034	0.075	0.097	0.147
0.801	3.445	4.01	4.322	9.622	0.032	0.077	0.098	0.149
0.898	1.504	0.00	4.916	8.710	0.077	0.093	0.112	0.143
0.898	1.995	0.01	4.916	8.709	0.073	0.091	0.113	0.145
0.899	2.999	0.02	4.919	8.703	0.065	0.088	0.112	0.147
0.899	3.462	0.03	4.919	8.703	0.071	0.088	0.112	0.147
0.899	3.986	-0.01	4.923	8.696	0.065	0.088	0.113	0.149
0.899	5.982	0.01	4.923	8.697	0.077	0.094	0.113	0.149
0.899	7.953	0.02	4.923	8.698	0.088	0.100	0.116	0.149
0.900	3.461	8.03	4.926	8.694	0.057	0.084	0.114	0.166
0.901	3.427	4.00	4.935	8.677	0.054	0.064	0.079	0.120
0.601	1.492	0.00	2.915	11.523	-0.005	0.095	0.143	0.184
0.602	2.004	0.00	2.921	11.517	-0.021	0.082	0.134	0.175
0.601	2.996	0.01	2.916	11.523	-0.022	0.078	0.128	0.171
0.601	3.441	-0.01	2.916	11.522	-0.020	0.077	0.129	0.171
0.601	3.980	-0.01	2.916	11.523	-0.019	0.078	0.128	0.171
0.601	5.968	0.01	2.914	11.526	-0.017	0.087	0.135	0.177
0.601	7.933	0.02	2.913	11.524	0.007	0.100	0.143	0.185
0.600	3.503	8.01	2.906	11.536	-0.043	0.079	0.140	0.183
0.602	3.448	3.99	2.919	11.520	-0.033	0.073	0.127	0.166
1.201	1.469	0.09	6.130	6.066	-0.335	-0.148	-0.074	-0.006
1.202	1.962	0.09	6.129	6.065	-0.346	-0.177	-0.107	-0.033
1.201	2.928	0.10	6.129	6.068	-0.350	-0.199	-0.127	-0.049
1.201	3.396	0.10	6.128	6.067	-0.349	-0.198	-0.139	-0.061
1.201	3.960	0.11	6.128	6.068	-0.349	-0.191	-0.158	-0.081
1.201	5.879	0.02	6.128	6.067	-0.357	-0.187	-0.120	-0.049
1.202	7.827	0.03	6.130	6.061	-0.322	-0.138	-0.095	-0.030
1.199	3.394	8.04	6.125	6.084	-0.362	-0.216	-0.222	-0.178
1.202	3.409	3.98	6.130	6.060	-0.349	-0.173	-0.134	-0.081

Table 2. Continued

(j) $\theta = 285^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.927	0.941	0.951	0.963	0.974	0.986	0.997	
0.799	1.506	0.02	4.305	9.641	-0.016	-0.034	-0.015	0.088	0.124	0.166	0.208	
0.799	1.999	0.03	4.307	9.639	-0.022	-0.045	-0.032	0.084	0.123	0.166	0.207	
0.798	2.988	0.02	4.298	9.652	-0.027	-0.055	-0.039	0.083	0.123	0.168	0.209	
0.800	3.459	0.02	4.316	9.625	-0.026	-0.052	-0.035	0.081	0.120	0.165	0.208	
0.800	3.992	0.03	4.316	9.627	-0.025	-0.050	-0.031	0.082	0.119	0.164	0.209	
0.799	6.023	-0.03	4.307	9.639	-0.012	-0.029	-0.008	0.093	0.123	0.163	0.213	
0.800	5.969	-0.03	4.312	9.632	-0.013	-0.030	-0.007	0.093	0.123	0.164	0.214	
0.799	7.962	-0.02	4.309	9.638	0.007	0.003	0.031	0.104	0.125	0.162	0.218	
0.799	3.455	8.01	4.310	9.642	-0.029	-0.044	-0.027	0.072	0.110	0.158	0.218	
0.806	3.432	4.01	4.351	9.578	-0.016	-0.019	0.012	0.067	0.093	0.135	0.195	
0.801	3.445	4.01	4.322	9.622	-0.018	-0.021	0.012	0.067	0.093	0.135	0.197	
0.898	1.504	0.00	4.916	8.710	0.043	0.054	0.065	0.094	0.111	0.143	0.187	
0.898	1.995	0.01	4.916	8.709	0.037	0.044	0.055	0.090	0.112	0.146	0.189	
0.899	2.999	0.02	4.919	8.703	0.026	0.029	0.044	0.086	0.111	0.146	0.191	
0.899	3.462	0.03	4.919	8.703	0.026	0.030	0.045	0.086	0.110	0.146	0.191	
0.899	3.986	-0.01	4.923	8.696	0.027	0.030	0.045	0.087	0.111	0.148	0.194	
0.899	5.982	0.01	4.923	8.697	0.038	0.046	0.060	0.094	0.112	0.146	0.195	
0.899	7.953	0.02	4.923	8.698	0.056	0.068	0.080	0.102	0.114	0.146	0.197	
0.900	3.461	8.03	4.926	8.694	0.007	0.030	0.047	0.084	0.111	0.157	0.217	
0.901	3.427	4.00	4.935	8.677	0.015	0.031	0.051	0.066	0.076	0.112	0.175	
0.601	1.492	0.00	2.915	11.523	-0.027	-0.059	-0.063	0.092	0.138	0.179	0.212	
0.602	2.004	0.00	2.921	11.517	-0.036	-0.073	-0.083	0.080	0.130	0.169	0.196	
0.601	2.996	0.01	2.916	11.523	-0.041	-0.079	-0.091	0.076	0.125	0.165	0.191	
0.601	3.441	-0.01	2.916	11.522	-0.041	-0.078	-0.091	0.076	0.125	0.164	0.191	
0.601	3.980	-0.01	2.916	11.523	-0.041	-0.078	-0.090	0.077	0.125	0.165	0.193	
0.601	5.968	0.01	2.914	11.526	-0.034	-0.066	-0.071	0.087	0.133	0.175	0.207	
0.601	7.933	0.02	2.913	11.524	-0.021	-0.045	-0.045	0.098	0.141	0.183	0.220	
0.600	3.503	8.01	2.906	11.536	-0.054	-0.092	-0.108	0.077	0.136	0.180	0.201	
0.602	3.448	3.99	2.919	11.520	-0.043	-0.068	-0.068	0.072	0.117	0.162	0.189	
1.201	1.469	0.09	6.130	6.066	-0.134	-0.152	-0.417	-0.150	-0.117	-0.055	0.013	
1.202	1.962	0.09	6.129	6.065	-0.135	-0.153	-0.424	-0.192	-0.134	-0.066	0.002	
1.201	2.928	0.10	6.129	6.068	-0.134	-0.153	-0.425	-0.221	-0.150	-0.075	-0.004	
1.201	3.396	0.10	6.128	6.067	-0.133	-0.152	-0.426	-0.241	-0.147	-0.067	0.003	
1.201	3.960	0.11	6.128	6.068	-0.133	-0.152	-0.426	-0.276	-0.133	-0.046	0.019	
1.201	5.879	0.02	6.128	6.067	-0.136	-0.155	-0.432	-0.201	-0.144	-0.076	0.000	
1.202	7.827	0.03	6.130	6.061	-0.139	-0.156	-0.427	-0.154	-0.082	-0.026	0.033	
1.199	3.394	8.04	6.125	6.084	-0.268	-0.239	-0.461	-0.342	-0.202	-0.133	-0.072	
1.202	3.409	3.98	6.130	6.060	-0.241	-0.227	-0.435	-0.183	-0.123	-0.075	-0.020	

Table 2. Continued

(k) $\theta = 300^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
0.799	1.506	0.02	4.305	9.641	-0.063	-0.028	-0.026	-0.054	0.098	0.132	0.169	0.204	0.175	
0.799	1.999	0.03	4.307	9.639	-0.064	-0.031	-0.030	-0.062	0.095	0.132	0.171	0.203	0.162	
0.798	2.988	0.02	4.298	9.652	-0.066	-0.034	-0.035	-0.070	0.093	0.136	0.173	0.200	0.162	
0.800	3.459	0.02	4.316	9.625	-0.066	-0.032	-0.035	-0.069	0.092	0.134	0.173	0.199	0.164	
0.800	3.992	0.03	4.316	9.627	-0.066	-0.032	-0.035	-0.067	0.093	0.134	0.173	0.198	0.169	
0.799	6.023	-0.03	4.307	9.639	-0.061	-0.024	-0.025	-0.050	0.103	0.138	0.178	0.205	0.187	
0.800	5.969	-0.03	4.312	9.632	-0.061	-0.024	-0.025	-0.051	0.105	0.139	0.180	0.206	0.189	
0.799	7.962	-0.02	4.309	9.638	-0.051	-0.011	-0.007	-0.023	0.115	0.142	0.180	0.208	0.202	
0.799	3.455	8.01	4.310	9.642	-0.083	-0.041	-0.041	-0.069	0.088	0.129	0.178	0.197	0.164	
0.806	3.432	4.01	4.351	9.578	-0.070	-0.032	-0.031	-0.044	0.084	0.117	0.160	0.174	0.138	
0.801	3.445	4.01	4.322	9.622	-0.072	-0.032	-0.032	-0.045	0.082	0.119	0.162	0.176	0.139	
0.898	1.504	0.00	4.916	8.710	-0.015	0.018	0.032	0.026	0.099	0.117	0.144	0.181	0.160	
0.898	1.995	0.01	4.916	8.709	-0.018	0.013	0.025	0.015	0.098	0.118	0.148	0.185	0.151	
0.899	2.999	0.02	4.919	8.703	-0.026	0.005	0.014	0.001	0.097	0.119	0.150	0.186	0.152	
0.899	3.462	0.03	4.919	8.703	-0.026	0.004	0.015	0.001	0.097	0.119	0.151	0.185	0.154	
0.899	3.986	-0.01	4.923	8.696	-0.025	0.006	0.016	0.002	0.099	0.122	0.154	0.187	0.159	
0.899	5.982	0.01	4.923	8.697	-0.018	0.014	0.027	0.018	0.101	0.121	0.153	0.189	0.173	
0.899	7.953	0.02	4.923	8.698	-0.004	0.030	0.045	0.042	0.104	0.123	0.152	0.189	0.184	
0.900	3.461	8.03	4.926	8.694	-0.086	-0.034	0.001	0.013	0.087	0.124	0.171	0.201	0.160	
0.901	3.427	4.00	4.935	8.677	-0.038	-0.006	0.012	0.024	0.069	0.094	0.134	0.162	0.142	
0.601	1.492	0.00	2.915	11.523	-0.063	-0.030	-0.033	-0.066	0.090	0.149	0.182	0.202	0.160	
0.602	2.004	0.00	2.921	11.517	-0.069	-0.036	-0.041	-0.079	0.075	0.139	0.172	0.184	0.127	
0.601	2.996	0.01	2.916	11.523	-0.071	-0.039	-0.046	-0.083	0.069	0.135	0.167	0.175	0.121	
0.601	3.441	-0.01	2.916	11.522	-0.072	-0.038	-0.046	-0.084	0.069	0.135	0.167	0.175	0.121	
0.601	3.980	-0.01	2.916	11.523	-0.071	-0.038	-0.046	-0.083	0.070	0.135	0.167	0.178	0.127	
0.601	5.968	0.01	2.914	11.526	-0.068	-0.034	-0.039	-0.073	0.082	0.142	0.176	0.195	0.152	
0.601	7.933	0.02	2.913	11.524	-0.060	-0.026	-0.028	-0.056	0.099	0.151	0.183	0.212	0.171	
0.600	3.503	8.01	2.906	11.536	-0.098	-0.058	-0.059	-0.102	0.069	0.140	0.171	0.186	0.119	
0.602	3.448	3.99	2.919	11.520	-0.080	-0.046	-0.047	-0.077	0.065	0.127	0.157	0.176	0.118	
1.201	1.469	0.09	6.130	6.066	-0.366	-0.347	-0.140	-0.121	-0.226	-0.085	-0.021	0.028	0.045	
1.202	1.962	0.09	6.129	6.065	-0.366	-0.347	-0.141	-0.122	-0.290	-0.112	-0.048	0.003	0.030	
1.201	2.928	0.10	6.129	6.068	-0.367	-0.347	-0.141	-0.122	-0.353	-0.131	-0.066	-0.015	0.018	
1.201	3.396	0.10	6.128	6.067	-0.366	-0.347	-0.140	-0.121	-0.309	-0.135	-0.071	-0.023	0.019	
1.201	3.960	0.11	6.128	6.068	-0.367	-0.346	-0.140	-0.121	-0.207	-0.136	-0.081	-0.037	0.021	
1.201	5.879	0.02	6.128	6.067	-0.369	-0.350	-0.142	-0.124	-0.253	-0.121	-0.063	-0.004	0.031	
1.202	7.827	0.03	6.130	6.061	-0.368	-0.352	-0.146	-0.127	-0.152	-0.092	-0.038	0.020	0.053	
1.199	3.394	8.04	6.125	6.084	-0.446	-0.452	-0.307	-0.215	-0.218	-0.201	-0.188	-0.134	-0.035	
1.202	3.409	3.98	6.130	6.060	-0.409	-0.417	-0.264	-0.204	-0.182	-0.124	-0.079	-0.036	0.015	

Table 2. Continued

(1) $\theta = 315^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997		
0.799	1.506	0.02	4.305	9.641	-0.076	-0.053	-0.031	-0.035	-0.073	-0.111	0.077	0.143	0.180	0.202		
0.799	1.999	0.03	4.307	9.639	-0.076	-0.054	-0.033	-0.039	-0.079	-0.123	0.070	0.140	0.174	0.197		
0.798	2.988	0.02	4.298	9.652	-0.077	-0.054	-0.035	-0.042	-0.084	-0.129	0.067	0.139	0.172	0.193		
0.800	3.459	0.02	4.316	9.625	-0.077	-0.053	-0.033	-0.041	-0.082	-0.127	0.067	0.139	0.171	0.192		
0.800	3.992	0.03	4.316	9.627	-0.077	-0.051	-0.032	-0.039	-0.080	-0.124	0.069	0.139	0.172	0.192		
0.799	6.023	-0.03	4.307	9.639	-0.074	-0.045	-0.025	-0.030	-0.066	-0.103	0.083	0.148	0.180	0.205		
0.800	5.969	-0.03	4.312	9.632	-0.073	-0.044	-0.025	-0.030	-0.066	-0.103	0.084	0.149	0.180	0.204		
0.799	7.962	-0.02	4.309	9.638	-0.066	-0.034	-0.013	-0.015	-0.044	-0.072	0.104	0.156	0.185	0.213		
0.799	3.455	8.01	4.310	9.642	-0.100	-0.060	-0.040	-0.044	-0.088	-0.140	0.070	0.124	0.163	0.188		
0.806	3.432	4.01	4.351	9.578	-0.085	-0.047	-0.030	-0.034	-0.062	-0.086	0.074	0.124	0.139	0.165		
0.801	3.445	4.01	4.322	9.622	-0.086	-0.047	-0.031	-0.034	-0.062	-0.086	0.074	0.124	0.140	0.166		
0.898	1.504	0.00	4.916	8.710	-0.050	-0.013	0.012	0.013	-0.009	-0.028	0.110	0.132	0.161	0.184		
0.898	1.995	0.01	4.916	8.709	-0.050	-0.016	0.008	0.008	-0.017	-0.041	0.106	0.134	0.159	0.184		
0.899	2.999	0.02	4.919	8.703	-0.056	-0.021	0.001	-0.001	-0.029	-0.058	0.101	0.136	0.162	0.185		
0.899	3.462	0.03	4.919	8.703	-0.057	-0.021	0.001	-0.001	-0.028	-0.056	0.101	0.135	0.161	0.184		
0.899	3.986	-0.01	4.923	8.696	-0.055	-0.019	0.003	0.001	-0.027	-0.056	0.102	0.137	0.163	0.186		
0.899	5.982	0.01	4.923	8.697	-0.051	-0.012	0.010	0.011	-0.012	-0.034	0.111	0.138	0.168	0.192		
0.899	7.953	0.02	4.923	8.698	-0.041	0.000	0.024	0.026	0.009	-0.022	0.101	0.137	0.169	0.196		
0.900	3.461	8.03	4.926	8.694	-0.136	-0.066	-0.024	-0.005	-0.015	-0.022	0.101	0.140	0.165	0.195		
0.901	3.427	4.00	4.935	8.677	-0.073	-0.024	-0.003	0.004	0.004	0.035	0.074	0.096	0.126	0.154		
0.601	1.492	0.00	2.915	11.523	-0.068	-0.048	-0.028	-0.038	-0.075	-0.055	0.059	0.138	0.175	0.198		
0.602	2.004	0.00	2.921	11.517	-0.072	-0.052	-0.032	-0.045	-0.086	-0.069	0.045	0.126	0.156	0.178		
0.601	2.996	0.01	2.916	11.523	-0.075	-0.053	-0.035	-0.048	-0.090	-0.074	0.040	0.120	0.149	0.169		
0.601	3.441	-0.01	2.916	11.522	-0.074	-0.053	-0.035	-0.047	-0.090	-0.074	0.040	0.121	0.150	0.170		
0.601	3.980	-0.01	2.916	11.523	-0.075	-0.054	-0.035	-0.048	-0.090	-0.074	0.041	0.121	0.153	0.174		
0.601	5.968	0.01	2.914	11.526	-0.073	-0.052	-0.032	-0.043	-0.082	-0.061	0.054	0.133	0.171	0.196		
0.601	7.933	0.02	2.913	11.524	-0.068	-0.046	-0.024	-0.032	-0.068	-0.044	0.070	0.145	0.186	0.215		
0.600	3.503	8.01	2.906	11.536	-0.111	-0.085	-0.057	-0.065	-0.113	-0.095	0.037	0.125	0.160	0.181		
0.602	3.448	3.99	2.919	11.520	-0.090	-0.070	-0.046	-0.054	-0.089	-0.063	0.047	0.118	0.151	0.170		
1.201	1.469	0.09	6.130	6.066	-0.295	-0.324	-0.346	-0.161	-0.134	-0.271	-0.193	-0.087	-0.026	0.000		
1.202	1.962	0.09	6.129	6.065	-0.296	-0.324	-0.347	-0.161	-0.134	-0.270	-0.300	-0.110	-0.039	-0.011		
1.201	2.928	0.10	6.129	6.068	-0.296	-0.324	-0.347	-0.159	-0.133	-0.271	-0.359	-0.128	-0.048	-0.018		
1.201	3.966	0.10	6.128	6.067	-0.296	-0.323	-0.347	-0.157	-0.132	-0.270	-0.384	-0.135	-0.048	-0.012		
1.201	3.960	0.11	6.128	6.068	-0.296	-0.324	-0.347	-0.157	-0.132	-0.270	-0.420	-0.140	-0.042	0.000		
1.201	5.879	0.02	6.128	6.067	-0.300	-0.332	-0.352	-0.162	-0.139	-0.277	-0.326	-0.123	-0.038	-0.005		
1.202	7.827	0.03	6.130	6.061	-0.299	-0.331	-0.353	-0.167	-0.141	-0.277	-0.231	-0.081	-0.007	0.024		
1.199	3.394	8.04	6.125	6.084	-0.385	-0.405	-0.437	-0.380	-0.250	-0.353	-0.484	-0.196	-0.117	-0.097		
1.202	3.409	3.98	6.130	6.060	-0.343	-0.370	-0.407	-0.305	-0.215	-0.308	-0.254	-0.109	-0.055	-0.030		

Table 2. Continued

(m) Top interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—				
					0.709	0.723	0.898	0.941	
0.799	1.506	0.02	4.305	9.641	-0.024	-0.023	-0.062	-0.014	
0.799	1.999	0.03	4.307	9.639	-0.022	-0.021	-0.064	-0.026	
0.798	2.988	0.02	4.298	9.652	-0.024	-0.021	-0.066	-0.036	
0.800	3.459	0.02	4.316	9.625	-0.023	-0.020	-0.066	-0.032	
0.800	3.992	0.03	4.316	9.627	-0.024	-0.020	-0.066	-0.029	
0.799	6.023	-0.03	4.307	9.639	-0.024	-0.019	-0.061	-0.007	
0.800	5.969	-0.03	4.312	9.632	-0.024	-0.019	-0.061	-0.008	
0.799	7.962	-0.02	4.309	9.638	-0.024	-0.017	-0.051	0.026	
0.799	3.455	8.01	4.310	9.642	-0.081	-0.066	-0.084	-0.020	
0.806	3.432	4.01	4.351	9.578	-0.057	-0.047	-0.072	0.002	
0.801	3.445	4.01	4.322	9.622	-0.056	-0.047	-0.072	0.000	
0.898	1.504	0.00	4.916	8.710	0.007	0.029	-0.018	0.062	
0.898	1.995	0.01	4.916	8.709	0.009	0.029	-0.022	0.055	
0.899	2.999	0.02	4.919	8.703	0.010	0.030	-0.029	0.042	
0.899	3.462	0.03	4.919	8.703	0.010	0.030	-0.029	0.042	
0.899	3.986	-0.01	4.923	8.696	0.005	0.028	-0.028	0.043	
0.899	5.982	0.01	4.923	8.697	0.008	0.031	-0.021	0.057	
0.899	7.953	0.02	4.923	8.698	0.011	0.034	-0.008	0.075	
0.900	3.461	8.03	4.926	8.694	-0.034	-0.003	-0.093	0.036	
0.901	3.427	4.00	4.935	8.677	-0.005	0.015	-0.045	0.035	
0.601	1.492	0.00	2.915	11.523	-0.032	-0.039	-0.061	-0.042	
0.602	2.004	0.00	2.921	11.517	-0.035	-0.040	-0.066	-0.056	
0.601	2.996	0.01	2.916	11.523	-0.037	-0.042	-0.070	-0.062	
0.601	3.441	-0.01	2.916	11.522	-0.037	-0.041	-0.069	-0.062	
0.601	3.980	-0.01	2.916	11.523	-0.037	-0.041	-0.070	-0.061	
0.601	5.968	0.01	2.914	11.526	-0.039	-0.044	-0.066	-0.048	
0.601	7.933	0.02	2.913	11.524	-0.036	-0.041	-0.060	-0.028	
0.600	3.503	8.01	2.906	11.536	-0.096	-0.106	-0.095	-0.071	
0.602	3.448	3.99	2.919	11.520	-0.067	-0.077	-0.078	-0.051	
1.201	1.469	0.09	6.130	6.066	-0.033	-0.033	-0.353	-0.167	
1.202	1.962	0.09	6.129	6.065	-0.034	-0.033	-0.354	-0.168	
1.201	2.928	0.10	6.129	6.068	-0.035	-0.034	-0.354	-0.167	
1.201	3.396	0.10	6.128	6.067	-0.035	-0.033	-0.354	-0.167	
1.201	3.960	0.11	6.128	6.068	-0.034	-0.033	-0.354	-0.167	
1.201	5.879	0.02	6.128	6.067	-0.036	-0.036	-0.358	-0.171	
1.202	7.827	0.03	6.130	6.061	-0.036	-0.036	-0.357	-0.172	
1.199	3.394	8.04	6.125	6.084	-0.174	-0.163	-0.435	-0.243	
1.202	3.409	3.98	6.130	6.060	-0.118	-0.117	-0.399	-0.233	

Table 2. Concluded

(n) Bottom interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of--									
					0.709	0.723	0.738	0.752	0.767	0.781	0.869	0.883	0.941	
0.799	1.506	0.02	4.305	9.641	-0.039	-0.051	-0.053	-0.054	-0.063	-0.085	-0.089	-0.082	-0.030	
0.799	1.999	0.03	4.307	9.639	-0.036	-0.048	-0.051	-0.051	-0.061	-0.083	-0.087	-0.081	-0.040	
0.798	2.988	0.02	4.298	9.652	-0.035	-0.047	-0.049	-0.049	-0.060	-0.082	-0.086	-0.081	-0.048	
0.800	3.459	0.02	4.316	9.625	-0.035	-0.047	-0.050	-0.049	-0.061	-0.084	-0.088	-0.082	-0.046	
0.800	3.992	0.03	4.316	9.627	-0.034	-0.046	-0.049	-0.049	-0.062	-0.085	-0.087	-0.082	-0.044	
0.799	6.023	-0.03	4.307	9.639	-0.034	-0.045	-0.048	-0.047	-0.061	-0.084	-0.084	-0.077	-0.023	
0.800	5.969	-0.03	4.312	9.632	-0.034	-0.045	-0.048	-0.047	-0.062	-0.084	-0.083	-0.077	-0.023	
0.799	7.962	-0.02	4.309	9.638	-0.033	-0.043	-0.046	-0.046	-0.061	-0.084	-0.079	-0.071	0.006	
0.799	3.455	8.01	4.310	9.642	0.017	-0.004	-0.015	-0.019	-0.042	-0.067	-0.055	-0.053	-0.058	
0.806	3.432	4.01	4.351	9.578	-0.003	-0.019	-0.027	-0.028	-0.049	-0.074	-0.072	-0.070	-0.039	
0.801	3.445	4.01	4.322	9.622	-0.001	-0.017	-0.025	-0.026	-0.047	-0.073	-0.070	-0.067	-0.039	
0.898	1.504	0.00	4.916	8.710	-0.050	-0.053	-0.050	-0.046	-0.053	-0.076	-0.096	-0.089	0.020	
0.898	1.995	0.01	4.916	8.709	-0.045	-0.049	-0.046	-0.043	-0.051	-0.074	-0.098	-0.090	0.013	
0.899	2.999	0.02	4.919	8.703	-0.049	-0.052	-0.048	-0.044	-0.053	-0.077	-0.098	-0.092	0.003	
0.899	3.462	0.03	4.919	8.703	-0.048	-0.051	-0.048	-0.045	-0.054	-0.077	-0.099	-0.092	0.003	
0.899	3.986	-0.01	4.923	8.696	-0.048	-0.049	-0.046	-0.042	-0.052	-0.075	-0.098	-0.091	0.003	
0.899	5.982	0.01	4.923	8.697	-0.046	-0.049	-0.046	-0.043	-0.052	-0.076	-0.095	-0.088	0.015	
0.899	7.953	0.02	4.923	8.698	-0.046	-0.047	-0.044	-0.040	-0.050	-0.075	-0.090	-0.082	0.031	
0.900	3.461	8.03	4.926	8.694	-0.017	-0.045	-0.060	-0.065	-0.082	-0.110	-0.075	-0.068	-0.017	
0.901	3.427	4.00	4.935	8.677	-0.032	-0.044	-0.048	-0.046	-0.061	-0.087	-0.089	-0.085	-0.017	
0.601	1.492	0.00	2.915	11.523	-0.027	-0.040	-0.044	-0.044	-0.056	-0.075	-0.068	-0.062	-0.045	
0.602	2.004	0.00	2.921	11.517	-0.028	-0.041	-0.045	-0.046	-0.060	-0.078	-0.072	-0.067	-0.060	
0.601	2.996	0.01	2.916	11.523	-0.029	-0.043	-0.046	-0.048	-0.061	-0.079	-0.074	-0.069	-0.064	
0.601	3.441	-0.01	2.916	11.522	-0.029	-0.042	-0.045	-0.046	-0.060	-0.079	-0.073	-0.068	-0.065	
0.601	3.980	-0.01	2.916	11.523	-0.030	-0.044	-0.046	-0.048	-0.062	-0.079	-0.075	-0.069	-0.065	
0.601	5.968	0.01	2.914	11.526	-0.030	-0.042	-0.046	-0.048	-0.060	-0.077	-0.072	-0.067	-0.052	
0.601	7.933	0.02	2.913	11.524	-0.030	-0.043	-0.047	-0.048	-0.060	-0.077	-0.071	-0.064	-0.034	
0.600	3.503	8.01	2.906	11.536	0.029	0.012	0.003	-0.001	-0.013	-0.035	-0.041	-0.039	-0.082	
0.602	3.448	3.99	2.919	11.520	0.003	-0.013	-0.020	-0.022	-0.034	-0.053	-0.061	-0.057	-0.068	
1.201	1.469	0.09	6.130	6.066	-0.021	-0.046	-0.070	-0.084	-0.094	-0.095	-0.178	-0.197	-0.167	
1.202	1.962	0.09	6.129	6.065	-0.021	-0.046	-0.070	-0.084	-0.095	-0.096	-0.177	-0.197	-0.168	
1.201	2.928	0.10	6.129	6.068	-0.021	-0.046	-0.070	-0.084	-0.095	-0.096	-0.177	-0.197	-0.168	
1.201	3.396	0.10	6.128	6.067	-0.021	-0.046	-0.070	-0.084	-0.095	-0.096	-0.177	-0.197	-0.168	
1.201	3.960	0.11	6.128	6.068	-0.020	-0.045	-0.069	-0.083	-0.094	-0.096	-0.177	-0.197	-0.167	
1.201	5.879	0.02	6.128	6.067	-0.029	-0.055	-0.078	-0.093	-0.101	-0.103	-0.185	-0.207	-0.175	
1.202	7.827	0.03	6.130	6.061	-0.027	-0.054	-0.078	-0.092	-0.101	-0.103	-0.185	-0.206	-0.176	
1.199	3.394	8.04	6.125	6.084	0.179	0.146	0.113	0.088	0.059	0.022	-0.061	-0.069	-0.130	
1.202	3.409	3.98	6.130	6.060	0.065	0.049	0.033	0.027	0.011	-0.018	-0.147	-0.160	-0.146	

Table 3. Static Pressure Coefficients of Afterbody Surface With Horizontal Tail in Mid Position and Vertical Tail in Mid Position

(a) $\theta = 0^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854	
1.202	1.501	0.05	6.148	6.079	-0.052	-0.075	-0.079	-0.074	-0.045	-0.026	0.022	0.007	-0.014	-0.037	-0.087	
1.199	1.971	0.01	6.142	6.099	-0.052	-0.075	-0.080	-0.074	-0.045	-0.026	0.022	0.006	-0.015	-0.038	-0.088	
1.200	2.948	0.02	6.142	6.098	-0.054	-0.075	-0.081	-0.076	-0.047	-0.027	0.021	0.005	-0.016	-0.039	-0.091	
1.200	3.392	0.03	6.143	6.094	-0.054	-0.074	-0.080	-0.077	-0.047	-0.027	0.021	0.006	-0.016	-0.039	-0.091	
1.198	3.919	0.01	6.140	6.109	-0.055	-0.075	-0.081	-0.078	-0.048	-0.027	0.021	0.005	-0.016	-0.040	-0.092	
1.200	5.871	0.02	6.143	6.092	-0.055	-0.074	-0.080	-0.079	-0.049	-0.029	0.020	0.004	-0.016	-0.039	-0.092	
1.200	7.788	0.04	6.143	6.093	-0.057	-0.074	-0.081	-0.080	-0.051	-0.029	0.020	0.004	-0.017	-0.040	-0.092	
1.199	3.394	8.02	6.141	6.100	-0.201	-0.218	-0.213	-0.193	-0.146	-0.105	-0.105	-0.133	-0.160	-0.161	-0.201	
1.199	3.449	8.03	6.141	6.100	-0.201	-0.219	-0.213	-0.194	-0.146	-0.104	-0.105	-0.135	-0.161	-0.162	-0.202	
1.198	3.395	4.02	6.138	6.109	-0.137	-0.154	-0.151	-0.140	-0.108	-0.084	-0.069	-0.089	-0.120	-0.139	-0.178	
0.901	1.476	0.02	4.963	8.729	-0.029	-0.013	0.002	0.021	0.056	0.046	-0.010	-0.077	-0.153	-0.232	-0.327	
0.901	1.964	0.02	4.958	8.735	-0.033	-0.016	0.000	0.018	0.054	0.045	-0.012	-0.078	-0.154	-0.233	-0.329	
0.902	2.959	0.04	4.967	8.723	-0.033	-0.017	-0.001	0.018	0.054	0.045	-0.012	-0.078	-0.154	-0.232	-0.328	
0.901	3.403	0.00	4.963	8.727	-0.031	-0.016	0.000	0.019	0.054	0.044	-0.012	-0.079	-0.155	-0.233	-0.328	
0.900	3.910	0.00	4.956	8.739	-0.031	-0.015	0.000	0.019	0.054	0.045	-0.012	-0.078	-0.155	-0.234	-0.330	
0.901	5.937	0.01	4.960	8.732	-0.032	-0.017	-0.001	0.019	0.054	0.045	-0.013	-0.079	-0.155	-0.234	-0.329	
0.900	7.806	0.03	4.957	8.737	-0.030	-0.016	0.001	0.020	0.056	0.045	-0.012	-0.078	-0.154	-0.233	-0.328	
0.898	3.430	8.03	4.945	8.759	-0.107	-0.104	-0.093	-0.071	-0.030	-0.039	-0.099	-0.173	-0.252	-0.330	-0.422	
0.898	3.418	3.99	4.946	8.758	-0.041	-0.035	-0.024	-0.007	0.024	0.011	-0.052	-0.123	-0.199	-0.278	-0.371	
0.801	1.489	0.02	4.351	9.689	-0.033	-0.031	-0.025	-0.016	0.007	-0.021	-0.098	-0.171	-0.253	-0.328	-0.388	
0.801	1.965	0.04	4.354	9.684	-0.032	-0.029	-0.024	-0.016	0.008	-0.021	-0.097	-0.172	-0.255	-0.331	-0.394	
0.803	2.944	0.01	4.362	9.673	-0.032	-0.031	-0.025	-0.016	0.008	-0.021	-0.097	-0.172	-0.254	-0.331	-0.394	
0.802	3.402	0.02	4.361	9.676	-0.032	-0.030	-0.024	-0.016	0.007	-0.021	-0.097	-0.173	-0.256	-0.332	-0.396	
0.802	3.914	0.03	4.358	9.679	-0.032	-0.030	-0.023	-0.015	0.009	-0.020	-0.098	-0.172	-0.254	-0.330	-0.389	
0.802	5.862	0.03	4.358	9.679	-0.033	-0.030	-0.024	-0.015	0.009	-0.020	-0.098	-0.172	-0.254	-0.330	-0.389	
0.803	7.807	0.00	4.368	9.665	-0.031	-0.029	-0.023	-0.013	0.010	-0.018	-0.095	-0.168	-0.249	-0.323	-0.382	
0.799	3.389	8.04	4.342	9.706	-0.099	-0.096	-0.087	-0.076	-0.053	-0.093	-0.186	-0.275	-0.368	-0.462	-0.563	
0.800	3.383	4.00	4.345	9.698	-0.057	-0.055	-0.046	-0.036	-0.018	-0.056	-0.146	-0.228	-0.315	-0.400	-0.471	
0.599	1.500	0.00	2.918	11.631	-0.026	-0.034	-0.033	-0.030	-0.013	-0.054	-0.116	-0.157	-0.196	-0.222	-0.239	
0.599	1.975	0.00	2.917	11.632	-0.025	-0.034	-0.032	-0.029	-0.012	-0.053	-0.117	-0.157	-0.197	-0.224	-0.240	
0.598	2.926	0.00	2.915	11.636	-0.025	-0.033	-0.032	-0.030	-0.013	-0.055	-0.119	-0.159	-0.199	-0.225	-0.242	
0.599	3.372	0.01	2.919	11.629	-0.022	-0.031	-0.030	-0.026	-0.010	-0.051	-0.114	-0.155	-0.195	-0.222	-0.238	
0.598	3.922	0.01	2.916	11.634	-0.025	-0.035	-0.033	-0.029	-0.012	-0.054	-0.117	-0.157	-0.198	-0.224	-0.241	
0.599	5.826	0.03	2.921	11.626	-0.025	-0.035	-0.033	-0.029	-0.013	-0.054	-0.118	-0.157	-0.197	-0.223	-0.239	
0.599	7.770	-0.01	2.919	11.629	-0.024	-0.033	-0.032	-0.028	-0.011	-0.052	-0.115	-0.154	-0.194	-0.219	-0.234	
0.599	3.411	8.01	2.918	11.633	-0.081	-0.088	-0.085	-0.084	-0.070	-0.127	-0.211	-0.260	-0.298	-0.319	-0.323	
0.600	3.469	3.99	2.929	11.617	-0.049	-0.056	-0.055	-0.051	-0.041	-0.091	-0.167	-0.211	-0.248	-0.270	-0.280	

Table 3. Continued

(a) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.869	0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	
1.202	1.501	0.05	6.148	6.079	-0.133	-0.175	-0.221	-0.279	-0.346	-0.391	-0.275	-0.366	-0.141	
1.199	1.971	0.01	6.142	6.099	-0.135	-0.175	-0.223	-0.279	-0.347	-0.393	-0.278	-0.371	-0.194	
1.200	2.948	0.02	6.142	6.098	-0.138	-0.178	-0.225	-0.279	-0.348	-0.395	-0.281	-0.373	-0.242	
1.200	3.392	0.03	6.143	6.094	-0.139	-0.177	-0.225	-0.278	-0.348	-0.396	-0.281	-0.373	-0.245	
1.198	3.919	0.01	6.140	6.109	-0.141	-0.179	-0.227	-0.280	-0.350	-0.398	-0.284	-0.375	-0.232	
1.200	5.871	0.02	6.143	6.092	-0.141	-0.179	-0.227	-0.278	-0.348	-0.398	-0.285	-0.374	-0.184	
1.200	7.788	0.04	6.143	6.093	-0.143	-0.179	-0.228	-0.278	-0.349	-0.399	-0.286	-0.204	-0.139	
1.199	3.394	8.02	6.141	6.100	-0.248	-0.275	-0.321	-0.378	-0.445	-0.491	-0.407	-0.246	-0.245	
1.199	3.449	8.03	6.141	6.100	-0.250	-0.275	-0.322	-0.378	-0.445	-0.492	-0.404	-0.244	-0.242	
1.198	3.395	4.02	6.138	6.109	-0.230	-0.256	-0.295	-0.347	-0.416	-0.444	-0.306	-0.164	-0.162	
0.901	1.476	0.02	4.963	8.729	-0.413	-0.462	-0.409	-0.135	-0.049	-0.002	-0.005	0.083	0.128	
0.901	1.964	0.02	4.958	8.735	-0.415	-0.469	-0.425	-0.148	-0.055	-0.006	-0.012	0.078	0.124	
0.902	2.959	0.04	4.967	8.723	-0.415	-0.476	-0.509	-0.160	-0.052	-0.012	-0.026	0.074	0.124	
0.901	3.403	0.00	4.963	8.727	-0.414	-0.476	-0.501	-0.158	-0.052	-0.011	-0.025	0.075	0.123	
0.900	3.910	0.00	4.956	8.739	-0.414	-0.481	-0.496	-0.157	-0.054	-0.012	-0.024	0.074	0.123	
0.901	5.937	0.01	4.960	8.732	-0.411	-0.479	-0.461	-0.143	-0.049	-0.005	-0.013	0.083	0.131	
0.900	7.806	0.03	4.957	8.737	-0.411	-0.475	-0.419	-0.123	-0.038	0.003	-0.004	0.094	0.140	
0.898	3.430	8.03	4.945	8.759	-0.503	-0.544	-0.342	-0.154	-0.075	-0.015	0.014	0.065	0.102	
0.898	3.418	3.99	4.946	8.758	-0.452	-0.490	-0.330	-0.125	-0.055	-0.020	-0.021	0.055	0.091	
0.801	1.489	0.02	4.351	9.689	-0.356	-0.285	-0.201	-0.126	-0.069	-0.044	-0.085	0.045	0.122	
0.801	1.965	0.04	4.354	9.684	-0.363	-0.290	-0.206	-0.129	-0.073	-0.049	-0.092	0.039	0.115	
0.803	2.944	0.01	4.362	9.673	-0.368	-0.294	-0.209	-0.132	-0.076	-0.053	-0.097	0.035	0.109	
0.802	3.402	0.02	4.361	9.676	-0.365	-0.292	-0.207	-0.131	-0.075	-0.052	-0.097	0.034	0.110	
0.802	3.914	0.03	4.358	9.679	-0.365	-0.292	-0.207	-0.131	-0.075	-0.051	-0.095	0.036	0.111	
0.802	5.862	0.03	4.358	9.679	-0.357	-0.286	-0.201	-0.126	-0.069	-0.043	-0.084	0.047	0.124	
0.803	7.807	0.00	4.368	9.665	-0.348	-0.278	-0.193	-0.118	-0.060	-0.032	-0.066	0.064	0.140	
0.799	3.389	8.04	4.342	9.706	-0.524	-0.326	-0.216	-0.140	-0.080	-0.046	-0.061	0.048	0.106	
0.800	3.383	4.00	4.345	9.698	-0.405	-0.304	-0.208	-0.133	-0.074	-0.045	-0.071	0.043	0.109	
0.599	1.500	0.00	2.918	11.631	-0.218	-0.193	-0.152	-0.117	-0.078	-0.057	-0.086	0.034	0.113	
0.599	1.975	0.00	2.917	11.632	-0.219	-0.195	-0.155	-0.121	-0.082	-0.062	-0.090	0.029	0.105	
0.598	2.926	0.00	2.915	11.636	-0.221	-0.198	-0.157	-0.123	-0.085	-0.066	-0.091	0.028	0.103	
0.599	3.372	0.01	2.919	11.629	-0.218	-0.194	-0.155	-0.122	-0.083	-0.063	-0.090	0.030	0.105	
0.598	3.922	0.01	2.916	11.634	-0.219	-0.196	-0.156	-0.123	-0.084	-0.064	-0.088	0.031	0.107	
0.599	5.826	0.03	2.921	11.626	-0.218	-0.193	-0.154	-0.121	-0.080	-0.059	-0.082	0.039	0.118	
0.599	7.770	-0.01	2.919	11.629	-0.212	-0.188	-0.148	-0.114	-0.073	-0.050	-0.070	0.053	0.133	
0.599	3.411	8.01	2.918	11.633	-0.288	-0.250	-0.197	-0.151	-0.101	-0.071	-0.086	0.029	0.100	
0.600	3.469	3.99	2.929	11.617	-0.252	-0.222	-0.175	-0.134	-0.090	-0.064	-0.086	0.028	0.101	

Table 3. Continued

(b) $\theta = 45^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—													
					0.709	0.723	0.738	0.752	0.767	0.781	0.912	0.941	0.951	0.963	0.974	0.997		
1.202	1.501	0.05	6.148	6.079	-0.050	0.001	0.011	-0.029	-0.059	0.085	-0.379	-0.186	-0.233	-0.290	-0.113	0.004		
1.199	1.971	0.01	6.142	6.099	-0.051	0.001	0.012	-0.029	-0.060	0.083	-0.379	-0.186	-0.233	-0.300	-0.130	-0.016		
1.200	2.948	0.02	6.142	6.098	-0.052	-0.001	0.011	-0.029	-0.060	0.084	-0.381	-0.188	-0.235	-0.308	-0.147	-0.031		
1.200	3.392	0.03	6.143	6.094	-0.052	-0.001	0.011	-0.029	-0.059	0.084	-0.381	-0.188	-0.235	-0.310	-0.151	-0.027		
1.198	3.919	0.01	6.140	6.109	-0.054	-0.002	0.011	-0.029	-0.061	0.083	-0.381	-0.189	-0.236	-0.311	-0.153	-0.018		
1.200	5.871	0.02	6.143	6.092	-0.054	-0.002	0.010	-0.029	-0.059	0.083	-0.381	-0.190	-0.237	-0.314	-0.162	0.013		
1.200	7.788	0.04	6.143	6.093	-0.055	-0.003	0.010	-0.028	-0.059	0.083	-0.380	-0.190	-0.238	-0.315	-0.155	0.030		
1.199	3.394	8.02	6.141	6.100	-0.177	-0.085	-0.085	-0.099	-0.108	0.139	-0.509	-0.351	-0.376	-0.396	-0.199	-0.056		
1.199	3.449	8.03	6.141	6.100	-0.177	-0.086	-0.085	-0.099	-0.108	0.138	-0.510	-0.351	-0.377	-0.393	-0.200	-0.055		
1.198	3.395	4.02	6.138	6.109	-0.119	-0.045	-0.048	-0.076	-0.109	0.057	-0.445	-0.244	-0.287	-0.343	-0.201	-0.039		
0.901	1.476	0.02	4.963	8.729	0.011	0.035	0.028	0.040	0.083	0.173	-0.226	0.062	0.058	0.090	0.136	0.137		
0.901	1.964	0.02	4.958	8.735	0.008	0.033	0.027	0.038	0.081	0.170	-0.234	0.057	0.054	0.087	0.133	0.129		
0.902	2.959	0.04	4.967	8.723	0.006	0.031	0.026	0.037	0.082	0.171	-0.245	0.053	0.049	0.081	0.126	0.125		
0.901	3.403	0.00	4.963	8.727	0.007	0.033	0.026	0.038	0.082	0.170	-0.246	0.052	0.049	0.080	0.127	0.128		
0.900	3.910	0.00	4.956	8.739	0.010	0.034	0.027	0.038	0.082	0.172	-0.242	0.055	0.057	0.089	0.136	0.146		
0.901	5.937	0.01	4.960	8.732	0.011	0.035	0.027	0.038	0.083	0.172	-0.233	0.061	0.057	0.089	0.147	0.155		
0.900	7.806	0.03	4.957	8.737	0.011	0.035	0.027	0.038	0.083	0.172	-0.221	0.069	0.064	0.098	0.147	0.155		
0.898	3.430	8.03	4.945	8.759	-0.065	-0.051	-0.074	-0.067	-0.014	0.082	-0.223	0.027	0.050	0.078	0.116	0.121		
0.898	3.418	3.99	4.946	8.758	0.004	0.019	0.001	0.008	0.053	0.151	-0.208	0.049	0.044	0.068	0.108	0.106		
0.801	1.489	0.02	4.351	9.689	0.004	0.017	0.005	0.011	0.048	0.139	-0.136	0.024	0.010	0.061	0.117	0.153		
0.801	1.965	0.04	4.354	9.684	0.001	0.015	0.003	0.009	0.046	0.138	-0.140	0.020	0.005	0.055	0.111	0.139		
0.803	2.944	0.01	4.362	9.673	0.003	0.017	0.004	0.010	0.047	0.141	-0.141	0.019	0.002	0.051	0.107	0.136		
0.802	3.402	0.02	4.361	9.676	0.002	0.015	0.003	0.009	0.046	0.139	-0.141	0.018	0.002	0.052	0.107	0.138		
0.802	3.914	0.03	4.358	9.679	0.001	0.016	0.003	0.010	0.046	0.139	-0.134	0.025	0.010	0.061	0.118	0.143		
0.802	5.862	0.03	4.358	9.679	0.001	0.016	0.003	0.010	0.046	0.139	-0.128	0.033	0.020	0.073	0.132	0.170		
0.803	7.807	0.00	4.368	9.665	0.002	0.017	0.004	0.010	0.047	0.140	-0.168	0.019	0.017	0.054	0.102	0.130		
0.799	3.389	8.04	4.342	9.706	-0.051	-0.037	-0.054	-0.050	-0.013	0.092	-0.148	0.021	0.007	0.050	0.102	0.126		
0.800	3.383	4.00	4.345	9.698	-0.004	0.009	-0.008	-0.005	0.032	0.134	-0.148	0.021	0.007	0.050	0.102	0.126		
0.599	1.500	0.00	2.918	11.631	0.009	0.016	0.002	0.006	0.037	0.121	-0.096	0.020	-0.002	0.047	0.102	0.155		
0.599	1.975	0.00	2.917	11.632	0.010	0.016	0.002	0.006	0.037	0.121	-0.100	0.016	-0.008	0.041	0.093	0.127		
0.598	2.926	0.00	2.915	11.636	0.008	0.014	0.000	0.005	0.036	0.120	-0.103	0.013	-0.011	0.036	0.089	0.123		
0.599	3.372	0.01	2.919	11.629	0.011	0.017	0.003	0.007	0.037	0.122	-0.100	0.015	-0.010	0.037	0.091	0.128		
0.598	3.922	0.01	2.916	11.634	0.011	0.017	0.003	0.007	0.037	0.120	-0.102	0.015	-0.011	0.037	0.091	0.134		
0.599	5.826	0.03	2.921	11.626	0.012	0.018	0.004	0.007	0.037	0.122	-0.099	0.019	-0.004	0.045	0.101	0.159		
0.599	7.770	-0.01	2.919	11.629	0.011	0.018	0.004	0.008	0.039	0.123	-0.093	0.024	0.004	0.056	0.115	0.175		
0.599	3.411	8.01	2.918	11.633	-0.028	-0.020	-0.035	-0.033	-0.005	0.099	-0.141	-0.002	-0.011	0.036	0.089	0.126		
0.600	3.469	3.99	2.929	11.617	0.003	-0.009	-0.007	-0.003	0.024	0.120	-0.122	0.008	-0.010	0.037	0.092	0.127		

Table 3. Continued

(c) $\theta = 90^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.951	0.963	0.974	0.986	0.997
1.202	1.501	0.05	6.148	6.079	-0.055	-0.009	0.032	-0.061	-0.101	0.111	-0.202	-0.222	-0.144	-0.075	-0.014
1.199	1.971	0.01	6.142	6.099	-0.057	-0.008	0.031	-0.064	-0.102	0.110	-0.201	-0.225	-0.157	-0.093	-0.035
1.200	2.948	0.02	6.142	6.098	-0.056	-0.008	0.030	-0.065	-0.103	0.109	-0.201	-0.229	-0.168	-0.108	-0.049
1.200	3.392	0.03	6.143	6.094	-0.057	-0.008	0.030	-0.065	-0.103	0.109	-0.201	-0.229	-0.171	-0.109	-0.047
1.198	3.919	0.01	6.140	6.109	-0.058	-0.008	0.028	-0.068	-0.104	0.107	-0.201	-0.230	-0.172	-0.108	-0.037
1.200	5.871	0.02	6.143	6.092	-0.058	-0.010	0.027	-0.067	-0.104	0.108	-0.201	-0.232	-0.173	-0.083	-0.011
1.200	7.788	0.04	6.143	6.093	-0.059	-0.010	0.027	-0.067	-0.104	0.107	-0.200	-0.231	-0.142	-0.021	-0.028
1.199	3.394	8.02	6.141	6.100	0.097	0.063	0.004	-0.074	-0.099	0.223	-0.257	-0.264	-0.172	-0.113	-0.054
1.199	3.449	8.03	6.141	6.100	0.097	0.063	0.003	-0.075	-0.099	0.223	-0.257	-0.265	-0.172	-0.114	-0.053
1.198	3.395	4.02	6.138	6.109	0.033	0.009	0.011	-0.079	-0.131	0.133	-0.210	-0.250	-0.191	-0.124	-0.048
0.901	1.476	0.02	4.963	8.729	0.014	0.034	0.025	-0.004	0.060	0.187	0.055	0.093	0.139	0.172	0.200
0.901	1.964	0.02	4.958	8.735	0.012	0.032	0.024	-0.005	0.058	0.185	0.051	0.090	0.135	0.168	0.192
0.902	2.959	0.04	4.967	8.723	0.010	0.030	0.023	-0.006	0.058	0.184	0.046	0.084	0.129	0.161	0.181
0.901	3.403	0.00	4.963	8.727	0.012	0.031	0.024	-0.005	0.059	0.186	0.046	0.084	0.130	0.162	0.183
0.900	3.910	0.00	4.956	8.739	0.012	0.032	0.025	-0.005	0.058	0.186	0.045	0.085	0.131	0.164	0.186
0.901	5.937	0.01	4.960	8.732	0.012	0.031	0.025	-0.003	0.060	0.187	0.061	0.105	0.142	0.180	0.210
0.900	7.806	0.03	4.957	8.737	0.013	0.031	0.026	-0.003	0.060	0.187	0.061	0.105	0.155	0.196	0.231
0.898	3.430	8.03	4.945	8.759	-0.043	-0.046	-0.071	-0.148	-0.034	0.218	0.044	0.085	0.126	0.156	0.174
0.898	3.418	3.99	4.946	8.758	0.002	0.011	0.000	-0.043	0.034	0.217	0.039	0.078	0.118	0.146	0.164
0.801	1.489	0.02	4.351	9.689	0.010	0.018	0.005	-0.028	0.034	0.171	0.040	0.081	0.128	0.164	0.194
0.801	1.965	0.04	4.354	9.684	0.009	0.015	0.003	-0.030	0.033	0.170	0.035	0.075	0.122	0.156	0.183
0.803	2.944	0.01	4.362	9.673	0.009	0.016	0.003	-0.029	0.033	0.170	0.032	0.072	0.118	0.151	0.174
0.802	3.402	0.02	4.361	9.676	0.010	0.016	0.004	-0.028	0.034	0.171	0.032	0.072	0.118	0.151	0.174
0.802	3.914	0.03	4.358	9.679	0.009	0.015	0.004	-0.029	0.033	0.171	0.032	0.073	0.120	0.153	0.177
0.802	5.862	0.03	4.358	9.679	0.009	0.015	0.004	-0.028	0.034	0.171	0.039	0.082	0.130	0.168	0.198
0.803	7.807	0.00	4.368	9.665	0.009	0.016	0.005	-0.027	0.034	0.172	0.048	0.094	0.145	0.187	0.218
0.799	3.389	8.04	4.342	9.706	-0.003	-0.012	-0.031	-0.088	-0.003	0.232	0.028	0.070	0.115	0.146	0.166
0.800	3.383	4.00	4.345	9.698	0.016	0.016	0.004	-0.033	0.034	0.203	0.027	0.068	0.113	0.145	0.165
0.599	1.500	0.00	2.918	11.631	0.015	0.015	0.007	-0.021	0.033	0.172	0.030	0.073	0.118	0.153	0.180
0.599	1.975	0.00	2.917	11.632	0.015	0.014	0.006	-0.021	0.033	0.172	0.025	0.067	0.110	0.142	0.163
0.598	2.926	0.00	2.915	11.636	0.015	0.015	0.006	-0.021	0.033	0.172	0.025	0.067	0.110	0.142	0.163
0.599	3.372	0.01	2.919	11.629	0.017	0.016	0.008	-0.020	0.034	0.171	0.021	0.063	0.105	0.135	0.152
0.598	3.922	0.01	2.916	11.634	0.015	0.015	0.008	-0.020	0.033	0.174	0.022	0.064	0.107	0.137	0.154
0.599	5.826	0.03	2.921	11.626	0.015	0.015	0.007	-0.020	0.033	0.173	0.022	0.064	0.106	0.138	0.157
0.599	7.770	-0.01	2.919	11.629	0.015	0.015	0.007	-0.021	0.033	0.173	0.027	0.072	0.118	0.153	0.182
0.599	3.411	8.01	2.918	11.633	0.028	0.016	0.008	-0.020	0.034	0.174	0.035	0.082	0.130	0.171	0.203
0.599	3.469	3.99	2.929	11.617	0.030	0.025	0.013	-0.016	0.040	0.190	0.023	0.062	0.105	0.135	0.153
												0.063	0.106	0.136	0.154

Table 3. Continued

(d) $\theta = 135^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.869
1.202	1.501	0.05	6.148	6.079	-0.066	-0.077	-0.077	-0.065	-0.073	-0.099	-0.071	-0.077	-0.086	-0.110	-0.151
1.199	1.971	0.01	6.142	6.099	-0.067	-0.078	-0.079	-0.065	-0.075	-0.103	-0.073	-0.079	-0.089	-0.113	-0.153
1.200	2.948	0.02	6.142	6.098	-0.067	-0.079	-0.080	-0.064	-0.076	-0.105	-0.074	-0.080	-0.090	-0.114	-0.155
1.200	3.392	0.03	6.143	6.094	-0.067	-0.079	-0.079	-0.064	-0.076	-0.105	-0.074	-0.079	-0.089	-0.113	-0.155
1.198	3.919	0.01	6.140	6.109	-0.069	-0.080	-0.081	-0.064	-0.078	-0.107	-0.074	-0.081	-0.090	-0.115	-0.157
1.200	5.871	0.02	6.143	6.092	-0.070	-0.082	-0.082	-0.064	-0.077	-0.106	-0.074	-0.081	-0.090	-0.115	-0.158
1.200	7.788	0.04	6.143	6.093	-0.071	-0.082	-0.081	-0.063	-0.077	-0.106	-0.074	-0.081	-0.090	-0.114	-0.159
1.199	3.394	8.02	6.141	6.100	0.144	0.099	0.053	0.001	-0.037	-0.067	0.007	0.030	0.040	0.013	-0.051
1.199	3.449	8.03	6.141	6.100	0.144	0.099	0.053	0.000	-0.038	-0.067	0.007	0.031	0.041	0.014	-0.051
1.198	3.395	4.02	6.138	6.109	0.056	0.034	-0.003	-0.033	-0.052	-0.080	-0.049	-0.045	-0.040	-0.061	-0.114
0.901	1.476	0.02	4.963	8.729	-0.036	-0.029	-0.036	-0.024	-0.007	0.003	-0.007	-0.036	-0.050	-0.072	-0.110
0.901	1.964	0.02	4.958	8.735	-0.037	-0.031	-0.038	-0.026	-0.009	0.001	-0.010	-0.038	-0.051	-0.072	-0.111
0.902	2.959	0.04	4.967	8.723	-0.039	-0.032	-0.038	-0.026	-0.009	0.002	-0.010	-0.039	-0.052	-0.074	-0.113
0.901	3.403	0.00	4.963	8.727	-0.037	-0.030	-0.037	-0.026	-0.009	0.003	-0.009	-0.037	-0.050	-0.072	-0.112
0.900	3.910	0.00	4.956	8.739	-0.036	-0.030	-0.036	-0.026	-0.008	0.004	-0.009	-0.038	-0.050	-0.072	-0.110
0.901	5.937	0.01	4.960	8.732	-0.036	-0.030	-0.036	-0.026	-0.008	0.004	-0.009	-0.037	-0.050	-0.072	-0.112
0.900	7.806	0.03	4.957	8.737	-0.035	-0.029	-0.036	-0.026	-0.007	0.005	-0.008	-0.037	-0.050	-0.071	-0.109
0.898	3.430	8.03	4.945	8.759	-0.062	-0.083	-0.111	-0.115	-0.085	-0.049	-0.035	-0.040	-0.033	-0.042	-0.064
0.898	3.418	3.99	4.946	8.758	-0.039	-0.044	-0.058	-0.051	-0.027	-0.005	-0.010	-0.031	-0.039	-0.057	-0.091
0.801	1.489	0.02	4.351	9.689	-0.030	-0.037	-0.050	-0.040	-0.024	-0.014	-0.019	-0.040	-0.046	-0.062	-0.087
0.801	1.965	0.04	4.354	9.684	-0.030	-0.037	-0.051	-0.040	-0.025	-0.015	-0.021	-0.041	-0.047	-0.063	-0.088
0.803	2.944	0.01	4.362	9.673	-0.030	-0.038	-0.051	-0.041	-0.025	-0.014	-0.021	-0.041	-0.047	-0.063	-0.089
0.802	3.402	0.02	4.361	9.676	-0.030	-0.037	-0.051	-0.041	-0.025	-0.014	-0.021	-0.041	-0.047	-0.063	-0.088
0.802	3.914	0.03	4.358	9.679	-0.029	-0.036	-0.049	-0.041	-0.024	-0.013	-0.020	-0.042	-0.047	-0.063	-0.088
0.802	5.862	0.03	4.368	9.665	-0.031	-0.037	-0.050	-0.041	-0.024	-0.013	-0.020	-0.041	-0.047	-0.062	-0.087
0.803	7.807	0.00	4.342	9.706	-0.019	-0.039	-0.063	-0.062	-0.040	-0.015	-0.003	-0.006	-0.002	-0.013	-0.037
0.799	3.389	8.04	4.345	9.698	-0.014	-0.026	-0.043	-0.038	-0.017	0.001	0.000	-0.016	-0.021	-0.038	-0.064
0.800	3.383	4.00	4.345	9.698	-0.014	-0.026	-0.043	-0.038	-0.017	0.001	0.000	-0.016	-0.021	-0.038	-0.064
0.599	1.500	0.00	2.918	11.631	-0.019	-0.030	-0.044	-0.039	-0.021	-0.011	-0.015	-0.030	-0.032	-0.046	-0.057
0.599	1.975	0.00	2.917	11.632	-0.019	-0.030	-0.044	-0.039	-0.021	-0.011	-0.015	-0.030	-0.032	-0.046	-0.057
0.598	2.926	0.00	2.915	11.636	-0.020	-0.031	-0.044	-0.040	-0.022	-0.012	-0.016	-0.032	-0.034	-0.048	-0.060
0.598	3.372	0.01	2.919	11.629	-0.017	-0.029	-0.042	-0.038	-0.020	-0.010	-0.014	-0.030	-0.032	-0.046	-0.058
0.598	3.922	0.01	2.916	11.634	-0.019	-0.031	-0.044	-0.040	-0.022	-0.011	-0.015	-0.030	-0.034	-0.047	-0.059
0.599	5.826	0.03	2.921	11.626	-0.020	-0.031	-0.044	-0.040	-0.021	-0.011	-0.015	-0.030	-0.033	-0.047	-0.058
0.599	7.770	-0.01	2.919	11.629	-0.020	-0.031	-0.044	-0.040	-0.021	-0.010	-0.015	-0.030	-0.033	-0.047	-0.057
0.599	3.411	8.01	2.918	11.633	0.008	-0.010	-0.031	-0.031	-0.009	0.010	0.021	0.019	0.025	0.013	-0.003
0.600	3.469	3.99	2.929	11.617	0.005	-0.010	-0.027	-0.022	-0.005	0.009	0.012	0.000	0.000	0.017	-0.034

Table 3. Continued

(d) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.202	1.501	0.05	6.148	6.079	-0.146	-0.126	-0.156	-0.192	-0.201	-0.261	-0.331	-0.212	-0.067	0.003
1.199	1.971	0.01	6.142	6.099	-0.142	-0.125	-0.158	-0.195	-0.203	-0.262	-0.334	-0.233	-0.121	-0.026
1.200	2.948	0.02	6.142	6.098	-0.145	-0.126	-0.160	-0.195	-0.205	-0.263	-0.338	-0.240	-0.148	-0.050
1.200	3.392	0.03	6.143	6.094	-0.145	-0.126	-0.159	-0.195	-0.205	-0.264	-0.340	-0.242	-0.148	-0.049
1.198	3.919	0.01	6.140	6.109	-0.143	-0.127	-0.162	-0.197	-0.207	-0.265	-0.341	-0.240	-0.139	-0.040
1.200	5.871	0.02	6.143	6.092	-0.147	-0.127	-0.161	-0.196	-0.207	-0.264	-0.342	-0.228	-0.083	-0.004
1.200	7.788	0.04	6.143	6.093	-0.147	-0.128	-0.162	-0.197	-0.207	-0.266	-0.344	-0.183	-0.039	-0.025
1.199	3.394	8.02	6.141	6.100	-0.065	-0.104	-0.152	-0.168	-0.161	-0.224	-0.334	-0.238	-0.154	-0.088
1.199	3.449	8.03	6.141	6.100	-0.066	-0.105	-0.152	-0.168	-0.161	-0.223	-0.333	-0.235	-0.151	-0.086
1.198	3.395	4.02	6.138	6.109	-0.109	-0.104	-0.130	-0.161	-0.162	-0.232	-0.311	-0.222	-0.133	-0.063
0.901	1.476	0.02	4.963	8.729	-0.127	-0.139	-0.110	-0.072	-0.013	-0.034	0.049	0.115	0.157	0.171
0.901	1.964	0.02	4.958	8.735	-0.128	-0.139	-0.111	-0.074	-0.015	-0.037	0.046	0.110	0.152	0.162
0.902	2.959	0.04	4.967	8.723	-0.131	-0.143	-0.114	-0.077	-0.017	-0.039	0.044	0.108	0.148	0.154
0.901	3.403	0.00	4.963	8.727	-0.129	-0.141	-0.112	-0.075	-0.015	-0.039	0.045	0.109	0.148	0.156
0.900	3.910	0.00	4.956	8.739	-0.128	-0.140	-0.111	-0.075	-0.015	-0.039	0.046	0.110	0.150	0.160
0.901	5.937	0.01	4.960	8.732	-0.126	-0.138	-0.108	-0.070	-0.010	-0.032	0.055	0.121	0.165	0.183
0.900	7.806	0.03	4.957	8.737	-0.123	-0.134	-0.104	-0.065	-0.003	-0.020	0.069	0.136	0.182	0.205
0.898	3.430	8.03	4.945	8.759	-0.077	-0.091	-0.078	-0.058	-0.012	-0.041	0.044	0.108	0.148	0.149
0.898	3.418	3.99	4.946	8.758	-0.108	-0.122	-0.102	-0.074	-0.023	-0.038	0.037	0.096	0.131	0.133
0.801	1.489	0.02	4.351	9.689	-0.090	-0.094	-0.077	-0.056	-0.009	-0.036	0.046	0.110	0.154	0.173
0.801	1.965	0.04	4.354	9.684	-0.092	-0.097	-0.080	-0.059	-0.012	-0.041	0.041	0.103	0.146	0.161
0.803	2.944	0.01	4.362	9.673	-0.093	-0.098	-0.081	-0.061	-0.014	-0.044	0.039	0.100	0.142	0.154
0.802	3.402	0.02	4.361	9.676	-0.093	-0.098	-0.081	-0.061	-0.014	-0.044	0.039	0.100	0.142	0.155
0.802	3.914	0.03	4.358	9.679	-0.093	-0.098	-0.081	-0.060	-0.013	-0.043	0.040	0.102	0.144	0.159
0.802	5.862	0.03	4.358	9.679	-0.090	-0.095	-0.077	-0.056	-0.007	-0.034	0.050	0.114	0.159	0.181
0.803	7.807	0.00	4.368	9.665	-0.089	-0.092	-0.073	-0.051	0.000	-0.024	0.063	0.129	0.176	0.203
0.799	3.389	8.04	4.342	9.706	-0.046	-0.058	-0.051	-0.040	-0.004	-0.039	0.041	0.101	0.140	0.147
0.800	3.383	4.00	4.345	9.698	-0.072	-0.082	-0.069	-0.054	-0.013	-0.035	0.040	0.096	0.131	0.143
0.599	1.500	0.00	2.918	11.631	-0.058	-0.062	-0.049	-0.039	0.001	-0.021	0.056	0.113	0.152	0.173
0.599	1.975	0.00	2.917	11.632	-0.059	-0.064	-0.052	-0.043	-0.004	-0.024	0.052	0.108	0.142	0.156
0.598	2.926	0.00	2.915	11.636	-0.060	-0.066	-0.053	-0.044	-0.006	-0.025	0.051	0.106	0.140	0.149
0.599	3.372	0.01	2.919	11.629	-0.058	-0.064	-0.052	-0.043	-0.004	-0.023	0.053	0.108	0.142	0.153
0.598	3.922	0.01	2.916	11.634	-0.059	-0.065	-0.052	-0.044	-0.005	-0.022	0.055	0.110	0.144	0.158
0.599	5.826	0.03	2.921	11.626	-0.058	-0.064	-0.051	-0.041	-0.001	-0.017	0.061	0.120	0.158	0.181
0.599	7.770	-0.01	2.919	11.629	-0.057	-0.062	-0.047	-0.037	0.004	-0.009	0.072	0.133	0.176	0.204
0.599	3.411	8.01	2.918	11.633	-0.011	-0.024	-0.019	-0.019	0.011	-0.014	0.057	0.110	0.143	0.148
0.600	3.469	3.99	2.929	11.617	-0.039	-0.048	-0.039	-0.034	0.000	-0.018	0.049	0.101	0.134	0.144

Table 3. Continued

(e) $\theta = 180^\circ$

M	NPR	α, deg	q _∞ , psi	p _∞ , psi	C _p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
1.202	1.501	0.05	6.148	6.079	-0.021	-0.054	-0.080	-0.084	-0.081	-0.083	-0.104	-0.123	-0.135	-0.140	-0.156
1.199	1.971	0.01	6.142	6.099	-0.024	-0.055	-0.082	-0.087	-0.082	-0.084	-0.106	-0.125	-0.137	-0.143	-0.159
1.200	2.948	0.02	6.142	6.098	-0.025	-0.055	-0.083	-0.087	-0.082	-0.085	-0.108	-0.126	-0.137	-0.144	-0.161
1.200	3.392	0.03	6.143	6.094	-0.025	-0.054	-0.083	-0.088	-0.082	-0.085	-0.108	-0.125	-0.136	-0.143	-0.161
1.198	3.919	0.01	6.140	6.109	-0.027	-0.054	-0.083	-0.089	-0.083	-0.085	-0.109	-0.126	-0.138	-0.145	-0.163
1.200	5.871	0.02	6.143	6.092	-0.026	-0.053	-0.083	-0.088	-0.082	-0.085	-0.110	-0.126	-0.137	-0.144	-0.162
1.200	7.788	0.04	6.143	6.093	-0.027	-0.051	-0.083	-0.089	-0.082	-0.085	-0.111	-0.126	-0.136	-0.143	-0.163
1.199	3.394	8.02	6.141	6.100	0.183	0.138	0.103	0.078	0.053	0.021	-0.031	-0.061	-0.066	-0.063	-0.065
1.199	3.449	8.03	6.141	6.100	0.183	0.138	0.102	0.078	0.053	0.021	-0.031	-0.061	-0.065	-0.062	-0.065
1.198	3.395	4.02	6.138	6.109	0.074	0.050	0.028	0.018	0.007	-0.017	-0.061	-0.090	-0.110	-0.125	-0.140
0.901	1.476	0.02	4.963	8.729	-0.049	-0.051	-0.047	-0.037	-0.031	-0.041	-0.056	-0.062	-0.072	-0.085	-0.095
0.901	1.964	0.02	4.958	8.735	-0.051	-0.053	-0.049	-0.039	-0.032	-0.042	-0.057	-0.063	-0.073	-0.086	-0.097
0.902	2.959	0.04	4.967	8.723	-0.053	-0.054	-0.049	-0.039	-0.033	-0.043	-0.059	-0.065	-0.075	-0.088	-0.099
0.901	3.403	0.00	4.963	8.727	-0.050	-0.052	-0.048	-0.039	-0.032	-0.042	-0.057	-0.064	-0.074	-0.087	-0.098
0.900	3.910	0.00	4.956	8.739	-0.049	-0.052	-0.048	-0.038	-0.031	-0.041	-0.056	-0.063	-0.073	-0.086	-0.097
0.901	5.937	0.01	4.960	8.732	-0.050	-0.052	-0.047	-0.037	-0.032	-0.041	-0.055	-0.062	-0.072	-0.085	-0.096
0.900	7.806	0.03	4.957	8.737	-0.049	-0.052	-0.047	-0.037	-0.032	-0.041	-0.055	-0.062	-0.072	-0.085	-0.094
0.898	3.430	8.03	4.945	8.759	-0.032	-0.067	-0.078	-0.076	-0.071	-0.078	-0.081	-0.074	-0.070	-0.071	-0.070
0.898	3.418	3.99	4.946	8.758	-0.034	-0.054	-0.055	-0.047	-0.042	-0.050	-0.060	-0.063	-0.068	-0.077	-0.083
0.801	1.489	0.02	4.351	9.689	-0.037	-0.053	-0.053	-0.045	-0.041	-0.050	-0.062	-0.065	-0.068	-0.075	-0.078
0.801	1.965	0.04	4.354	9.684	-0.038	-0.053	-0.055	-0.047	-0.042	-0.052	-0.063	-0.066	-0.070	-0.076	-0.080
0.803	2.944	0.01	4.362	9.673	-0.038	-0.054	-0.054	-0.047	-0.042	-0.052	-0.063	-0.066	-0.070	-0.077	-0.081
0.802	3.402	0.02	4.361	9.676	-0.037	-0.053	-0.054	-0.047	-0.043	-0.052	-0.063	-0.066	-0.071	-0.077	-0.081
0.802	3.914	0.03	4.358	9.679	-0.037	-0.053	-0.053	-0.046	-0.043	-0.052	-0.063	-0.066	-0.071	-0.077	-0.081
0.802	5.862	0.03	4.358	9.679	-0.037	-0.053	-0.053	-0.046	-0.042	-0.051	-0.062	-0.065	-0.070	-0.076	-0.080
0.803	7.807	0.00	4.368	9.665	-0.038	-0.054	-0.054	-0.046	-0.043	-0.052	-0.061	-0.066	-0.069	-0.075	-0.078
0.799	3.389	8.04	4.342	9.706	0.003	-0.026	-0.033	-0.029	-0.028	-0.036	-0.041	-0.040	-0.040	-0.041	-0.042
0.800	3.383	4.00	4.345	9.698	-0.010	-0.034	-0.036	-0.030	-0.029	-0.037	-0.045	-0.048	-0.052	-0.056	-0.059
0.599	1.500	0.00	2.918	11.631	-0.028	-0.045	-0.044	-0.038	-0.034	-0.042	-0.050	-0.053	-0.056	-0.054	-0.053
0.599	1.975	0.00	2.917	11.632	-0.027	-0.046	-0.045	-0.038	-0.035	-0.043	-0.051	-0.054	-0.057	-0.055	-0.054
0.598	2.926	0.00	2.915	11.636	-0.028	-0.046	-0.045	-0.038	-0.035	-0.043	-0.051	-0.054	-0.057	-0.055	-0.055
0.599	3.372	0.01	2.919	11.629	-0.026	-0.045	-0.044	-0.038	-0.034	-0.042	-0.049	-0.053	-0.056	-0.054	-0.052
0.598	3.922	0.01	2.916	11.634	-0.028	-0.047	-0.045	-0.038	-0.035	-0.043	-0.051	-0.054	-0.057	-0.054	-0.053
0.599	5.826	0.03	2.921	11.626	-0.027	-0.046	-0.044	-0.037	-0.035	-0.043	-0.050	-0.053	-0.056	-0.053	-0.052
0.599	7.770	-0.01	2.919	11.629	-0.027	-0.047	-0.044	-0.037	-0.035	-0.043	-0.050	-0.053	-0.056	-0.053	-0.051
0.599	3.411	8.01	2.918	11.633	0.026	0.001	-0.002	-0.038	-0.035	-0.043	-0.050	-0.053	-0.056	-0.053	-0.051
0.600	3.469	3.99	2.929	11.617	0.005	-0.017	-0.017	-0.011	-0.010	-0.018	-0.027	-0.029	-0.033	-0.031	-0.031

Table 3. Continued

(e) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—								
					0.883	0.898	0.912	0.927	0.941	0.951	0.974	0.986	
1.202	1.501	0.05	6.148	6.079	-0.171	-0.168	-0.181	-0.192	-0.217	-0.241	-0.244	-0.048	
1.199	1.971	0.01	6.142	6.099	-0.171	-0.169	-0.180	-0.193	-0.213	-0.237	-0.287	-0.104	
1.200	2.948	0.02	6.142	6.098	-0.171	-0.169	-0.180	-0.195	-0.215	-0.240	-0.295	-0.138	
1.200	3.392	0.03	6.143	6.094	-0.170	-0.168	-0.180	-0.196	-0.216	-0.241	-0.296	-0.137	
1.198	3.919	0.01	6.140	6.109	-0.171	-0.169	-0.180	-0.197	-0.215	-0.240	-0.295	-0.130	
1.200	5.871	0.02	6.143	6.092	-0.169	-0.169	-0.181	-0.198	-0.218	-0.243	-0.281	-0.076	
1.200	7.788	0.04	6.143	6.093	-0.168	-0.169	-0.182	-0.199	-0.218	-0.244	-0.200	-0.047	
1.199	3.394	8.02	6.141	6.100	-0.056	-0.063	-0.103	-0.145	-0.201	-0.212	-0.328	-0.195	
1.199	3.449	8.03	6.141	6.100	-0.056	-0.064	-0.103	-0.146	-0.201	-0.211	-0.326	-0.192	
1.198	3.395	4.02	6.138	6.109	-0.142	-0.137	-0.145	-0.154	-0.179	-0.206	-0.264	-0.136	
0.901	1.476	0.02	4.963	8.729	-0.109	-0.110	-0.109	-0.094	-0.098	-0.119	0.114	0.157	
0.901	1.964	0.02	4.958	8.735	-0.110	-0.111	-0.110	-0.096	-0.101	-0.120	0.110	0.151	
0.902	2.959	0.04	4.967	8.723	-0.112	-0.114	-0.113	-0.098	-0.103	-0.126	0.106	0.147	
0.901	3.403	0.00	4.963	8.727	-0.111	-0.112	-0.111	-0.096	-0.102	-0.124	0.107	0.148	
0.900	3.910	0.00	4.956	8.739	-0.112	-0.112	-0.111	-0.096	-0.103	-0.124	0.107	0.149	
0.901	5.937	0.01	4.960	8.732	-0.109	-0.110	-0.108	-0.092	-0.097	-0.116	0.117	0.161	
0.900	7.806	0.03	4.957	8.737	-0.108	-0.107	-0.104	-0.086	-0.090	-0.104	0.131	0.177	
0.898	3.430	8.03	4.945	8.759	-0.078	-0.080	-0.090	-0.088	-0.118	-0.171	0.106	0.146	
0.898	3.418	3.99	4.946	8.758	-0.100	-0.102	-0.108	-0.101	-0.123	-0.161	0.093	0.129	
0.801	1.489	0.02	4.351	9.689	-0.082	-0.079	-0.082	-0.076	-0.089	-0.106	0.110	0.156	
0.801	1.965	0.04	4.354	9.684	-0.085	-0.082	-0.085	-0.080	-0.094	-0.112	0.104	0.148	
0.803	2.944	0.01	4.362	9.673	-0.086	-0.084	-0.088	-0.081	-0.097	-0.115	0.100	0.144	
0.802	3.402	0.02	4.361	9.676	-0.086	-0.083	-0.087	-0.081	-0.097	-0.115	0.101	0.144	
0.802	3.914	0.03	4.358	9.679	-0.086	-0.084	-0.087	-0.081	-0.096	-0.115	0.102	0.146	
0.802	5.862	0.03	4.358	9.679	-0.084	-0.081	-0.083	-0.076	-0.090	-0.105	0.113	0.161	
0.803	7.807	0.00	4.368	9.665	-0.083	-0.078	-0.078	-0.069	-0.081	-0.092	0.127	0.178	
0.799	3.389	8.04	4.342	9.706	-0.055	-0.055	-0.067	-0.071	-0.104	-0.144	0.097	0.141	
0.800	3.383	4.00	4.345	9.698	-0.073	-0.071	-0.079	-0.077	-0.104	-0.132	0.094	0.135	
0.599	1.500	0.00	2.918	11.631	-0.062	-0.055	-0.061	-0.056	-0.078	-0.085	0.110	0.157	
0.599	1.975	0.00	2.917	11.632	-0.065	-0.058	-0.064	-0.060	-0.085	-0.092	0.102	0.147	
0.598	2.926	0.00	2.915	11.636	-0.067	-0.059	-0.066	-0.062	-0.087	-0.091	0.101	0.144	
0.599	3.372	0.01	2.919	11.629	-0.065	-0.057	-0.063	-0.060	-0.085	-0.090	0.102	0.146	
0.598	3.922	0.01	2.916	11.634	-0.065	-0.057	-0.064	-0.061	-0.086	-0.089	0.104	0.148	
0.599	5.826	0.03	2.921	11.626	-0.064	-0.055	-0.061	-0.057	-0.080	-0.083	0.114	0.163	
0.599	7.770	-0.01	2.919	11.629	-0.061	-0.052	-0.057	-0.051	-0.072	-0.071	0.128	0.181	
0.599	3.411	8.01	2.918	11.633	-0.028	-0.024	-0.038	-0.043	-0.079	-0.101	0.104	0.148	
0.600	3.469	3.99	2.929	11.617	-0.047	-0.042	-0.053	-0.054	-0.084	-0.101	0.098	0.140	

Table 3. Continued

(f) $\theta = 225^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.202	1.501	0.05	6.148	6.079	-0.197	-0.197	-0.174	-0.135	-0.196	-0.310	-0.189	-0.139	-0.119	-0.082
1.199	1.971	0.01	6.142	6.099	-0.197	-0.197	-0.172	-0.129	-0.192	-0.311	-0.242	-0.154	-0.133	-0.096
1.200	2.948	0.02	6.142	6.098	-0.197	-0.198	-0.173	-0.131	-0.194	-0.313	-0.287	-0.166	-0.146	-0.112
1.200	3.392	0.03	6.143	6.094	-0.195	-0.198	-0.174	-0.132	-0.195	-0.314	-0.288	-0.166	-0.148	-0.115
1.198	3.919	0.01	6.140	6.109	-0.197	-0.199	-0.173	-0.129	-0.193	-0.315	-0.293	-0.167	-0.147	-0.115
1.200	5.871	0.02	6.143	6.092	-0.194	-0.198	-0.175	-0.134	-0.199	-0.317	-0.334	-0.172	-0.130	-0.082
1.200	7.788	0.04	6.143	6.093	-0.193	-0.198	-0.176	-0.134	-0.199	-0.317	-0.413	-0.116	-0.039	0.001
1.199	3.394	8.02	6.141	6.100	-0.071	-0.065	-0.053	-0.051	-0.175	-0.301	-0.446	-0.305	-0.206	-0.123
1.199	3.449	8.03	6.141	6.100	-0.071	-0.065	-0.053	-0.051	-0.175	-0.302	-0.446	-0.304	-0.206	-0.122
1.198	3.395	4.02	6.138	6.109	-0.165	-0.160	-0.131	-0.093	-0.171	-0.295	-0.424	-0.198	-0.124	-0.072
0.901	1.476	0.02	4.963	8.729	-0.116	-0.095	-0.071	-0.072	-0.143	-0.176	0.057	0.123	0.158	0.180
0.901	1.964	0.02	4.958	8.735	-0.118	-0.097	-0.073	-0.073	-0.145	-0.178	0.057	0.123	0.158	0.176
0.902	2.959	0.04	4.967	8.723	-0.119	-0.099	-0.076	-0.077	-0.151	-0.189	0.049	0.117	0.152	0.169
0.901	3.403	0.00	4.963	8.727	-0.118	-0.097	-0.074	-0.075	-0.149	-0.185	0.051	0.118	0.155	0.171
0.900	3.910	0.00	4.956	8.739	-0.119	-0.099	-0.075	-0.076	-0.149	-0.184	0.051	0.118	0.154	0.173
0.901	5.937	0.01	4.960	8.732	-0.117	-0.096	-0.071	-0.073	-0.143	-0.176	0.055	0.124	0.162	0.187
0.900	7.806	0.03	4.957	8.737	-0.115	-0.091	-0.067	-0.068	-0.136	-0.167	0.061	0.130	0.169	0.199
0.898	3.430	8.03	4.945	8.759	-0.098	-0.077	-0.061	-0.079	-0.189	-0.299	0.022	0.109	0.151	0.167
0.898	3.418	3.99	4.946	8.758	-0.116	-0.096	-0.061	-0.091	-0.193	-0.282	0.011	0.093	0.129	0.140
0.801	1.489	0.02	4.351	9.689	-0.090	-0.069	-0.049	-0.053	-0.122	-0.153	0.060	0.126	0.162	0.187
0.801	1.965	0.04	4.354	9.684	-0.093	-0.073	-0.053	-0.059	-0.131	-0.165	0.051	0.120	0.157	0.178
0.803	2.944	0.01	4.362	9.673	-0.095	-0.074	-0.055	-0.061	-0.135	-0.170	0.048	0.118	0.155	0.174
0.802	3.402	0.02	4.361	9.676	-0.094	-0.074	-0.056	-0.061	-0.135	-0.171	0.047	0.118	0.155	0.174
0.802	3.914	0.03	4.358	9.679	-0.094	-0.074	-0.056	-0.061	-0.133	-0.169	0.048	0.119	0.157	0.177
0.802	5.862	0.03	4.358	9.679	-0.092	-0.071	-0.051	-0.055	-0.123	-0.152	0.059	0.126	0.163	0.190
0.803	7.807	0.00	4.368	9.665	-0.089	-0.065	-0.044	-0.046	-0.108	-0.132	0.072	0.134	0.171	0.205
0.799	3.389	8.04	4.342	9.706	-0.075	-0.055	-0.043	-0.063	-0.167	-0.245	0.015	0.102	0.148	0.171
0.800	3.383	4.00	4.345	9.698	-0.088	-0.067	-0.052	-0.065	-0.154	-0.212	0.026	0.102	0.140	0.161
0.599	1.500	0.00	2.918	11.631	-0.073	-0.053	-0.039	-0.049	-0.119	-0.152	0.044	0.127	0.170	0.192
0.599	1.975	0.00	2.917	11.632	-0.076	-0.057	-0.044	-0.056	-0.129	-0.167	0.032	0.116	0.158	0.174
0.598	2.926	0.00	2.915	11.636	-0.078	-0.059	-0.046	-0.058	-0.132	-0.171	0.028	0.113	0.155	0.168
0.598	3.372	0.01	2.919	11.629	-0.077	-0.057	-0.044	-0.057	-0.131	-0.168	0.030	0.114	0.155	0.171
0.598	3.922	0.01	2.916	11.634	-0.077	-0.056	-0.043	-0.057	-0.130	-0.169	0.030	0.115	0.157	0.175
0.599	5.826	0.03	2.921	11.626	-0.076	-0.054	-0.040	-0.051	-0.121	-0.154	0.042	0.126	0.168	0.195
0.599	7.770	-0.01	2.919	11.629	-0.071	-0.049	-0.034	-0.041	-0.107	-0.136	0.057	0.136	0.179	0.214
0.599	3.411	8.01	2.918	11.633	-0.047	-0.031	-0.023	-0.046	-0.142	-0.202	0.019	0.115	0.160	0.175
0.600	3.469	3.99	2.929	11.617	-0.064	-0.045	-0.035	-0.053	-0.139	-0.188	0.023	0.109	0.150	0.167

Table 3. Continued

(g) $\theta = 240^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986
1.202	1.501	0.05	6.148	6.079	-0.205	-0.150	-0.116	-0.156	-0.411	-0.156	-0.164	-0.134
1.199	1.971	0.01	6.142	6.099	-0.207	-0.147	-0.111	-0.151	-0.412	-0.179	-0.187	-0.155
1.200	2.948	0.02	6.142	6.098	-0.209	-0.148	-0.112	-0.153	-0.413	-0.198	-0.203	-0.171
1.200	3.392	0.03	6.143	6.094	-0.211	-0.148	-0.113	-0.155	-0.412	-0.200	-0.205	-0.173
1.198	3.919	0.01	6.140	6.109	-0.212	-0.147	-0.111	-0.152	-0.413	-0.202	-0.205	-0.173
1.200	5.871	0.02	6.143	6.092	-0.213	-0.150	-0.116	-0.158	-0.415	-0.218	-0.211	-0.158
1.200	7.788	0.04	6.143	6.093	-0.214	-0.150	-0.116	-0.158	-0.414	-0.273	-0.157	-0.054
1.199	3.394	8.02	6.141	6.100	-0.075	-0.034	-0.032	-0.115	-0.404	-0.534	-0.271	-0.254
1.199	3.449	8.03	6.141	6.100	-0.075	-0.035	-0.032	-0.115	-0.404	-0.534	-0.271	-0.253
1.198	3.395	4.02	6.138	6.109	-0.170	-0.104	-0.075	-0.126	-0.401	-0.334	-0.216	-0.156
0.901	1.476	0.02	4.963	8.729	-0.099	-0.060	-0.054	-0.095	-0.195	0.067	0.103	0.140
0.901	1.964	0.02	4.958	8.735	-0.101	-0.062	-0.056	-0.096	-0.195	0.067	0.104	0.142
0.902	2.959	0.04	4.967	8.723	-0.103	-0.065	-0.060	-0.104	-0.208	0.061	0.099	0.138
0.901	3.403	0.00	4.963	8.727	-0.100	-0.064	-0.058	-0.101	-0.206	0.063	0.102	0.140
0.900	3.910	0.00	4.956	8.739	-0.100	-0.065	-0.059	-0.101	-0.207	0.063	0.101	0.139
0.901	5.937	0.01	4.960	8.732	-0.097	-0.061	-0.055	-0.096	-0.202	0.067	0.106	0.144
0.900	7.806	0.03	4.957	8.737	-0.092	-0.057	-0.049	-0.089	-0.193	0.072	0.109	0.148
0.898	3.430	8.03	4.945	8.759	-0.081	-0.056	-0.065	-0.143	-0.340	0.041	0.094	0.135
0.898	3.418	3.99	4.946	8.758	-0.098	-0.071	-0.076	-0.148	-0.333	0.027	0.072	0.105
0.801	1.489	0.02	4.351	9.689	-0.073	-0.038	-0.034	-0.070	-0.174	0.078	0.114	0.153
0.801	1.965	0.04	4.354	9.684	-0.076	-0.043	-0.040	-0.080	-0.192	0.071	0.110	0.150
0.803	2.944	0.01	4.362	9.673	-0.078	-0.045	-0.043	-0.085	-0.197	0.068	0.109	0.149
0.802	3.402	0.02	4.361	9.676	-0.077	-0.045	-0.043	-0.085	-0.198	0.069	0.109	0.148
0.802	3.914	0.03	4.358	9.679	-0.077	-0.045	-0.043	-0.084	-0.197	0.069	0.111	0.150
0.802	5.862	0.03	4.358	9.679	-0.073	-0.040	-0.036	-0.072	-0.176	0.078	0.114	0.153
0.803	7.807	0.00	4.368	9.665	-0.066	-0.033	-0.025	-0.055	-0.148	0.090	0.119	0.156
0.799	3.389	8.04	4.342	9.706	-0.060	-0.037	-0.048	-0.119	-0.302	0.042	0.099	0.144
0.800	3.383	4.00	4.345	9.698	-0.069	-0.043	-0.048	-0.102	-0.257	0.053	0.094	0.131
0.599	1.500	0.00	2.918	11.631	-0.051	-0.030	-0.034	-0.081	-0.208	0.075	0.138	0.175
0.599	1.975	0.00	2.917	11.632	-0.055	-0.036	-0.042	-0.090	-0.228	0.064	0.130	0.167
0.598	2.926	0.00	2.915	11.636	-0.056	-0.038	-0.044	-0.094	-0.234	0.060	0.128	0.164
0.599	3.372	0.01	2.919	11.629	-0.055	-0.036	-0.042	-0.092	-0.231	0.062	0.130	0.165
0.598	3.922	0.01	2.916	11.634	-0.055	-0.036	-0.041	-0.092	-0.231	0.063	0.130	0.167
0.599	5.826	0.03	2.921	11.626	-0.052	-0.032	-0.035	-0.081	-0.211	0.074	0.137	0.174
0.599	7.770	-0.01	2.919	11.629	-0.045	-0.025	-0.024	-0.066	-0.182	0.088	0.143	0.179
0.599	3.411	8.01	2.918	11.633	-0.032	-0.018	-0.033	-0.106	-0.282	0.054	0.133	0.176
0.600	3.469	3.99	2.929	11.617	-0.046	-0.028	-0.037	-0.097	-0.258	0.057	0.122	0.161

Table 3. Continued

(h) $\theta = 255^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.941	0.963	0.974	0.986	0.997
1.202	1.501	0.05	6.148	6.079	-0.160	-0.183	-0.115	-0.080	-0.024
1.199	1.971	0.01	6.142	6.099	-0.158	-0.211	-0.137	-0.103	-0.047
1.200	2.948	0.02	6.142	6.098	-0.160	-0.235	-0.153	-0.120	-0.066
1.200	3.392	0.03	6.143	6.094	-0.162	-0.237	-0.154	-0.120	-0.065
1.198	3.919	0.01	6.140	6.109	-0.160	-0.239	-0.154	-0.118	-0.057
1.200	5.871	0.02	6.143	6.092	-0.165	-0.258	-0.155	-0.093	-0.015
1.200	7.788	0.04	6.143	6.093	-0.165	-0.206	-0.178	-0.123	-0.050
1.199	3.394	8.02	6.141	6.100	-0.120	-0.255	-0.256	-0.252	-0.219
1.199	3.449	8.03	6.141	6.100	-0.120	-0.253	-0.255	-0.249	-0.217
1.198	3.395	4.02	6.138	6.109	-0.135	-0.185	-0.196	-0.194	-0.156
0.901	1.476	0.02	4.963	8.729	-0.072	0.044	0.078	0.125	0.165
0.901	1.964	0.02	4.958	8.735	-0.072	0.045	0.081	0.129	0.167
0.902	2.959	0.04	4.967	8.723	-0.082	0.039	0.076	0.124	0.161
0.901	3.403	0.00	4.963	8.727	-0.078	0.041	0.077	0.125	0.161
0.900	3.910	0.00	4.956	8.739	-0.079	0.041	0.078	0.125	0.163
0.901	5.937	0.01	4.960	8.732	-0.074	0.044	0.080	0.128	0.169
0.900	7.806	0.03	4.957	8.737	-0.069	0.047	0.083	0.129	0.173
0.898	3.430	8.03	4.945	8.759	-0.130	0.013	0.063	0.118	0.157
0.898	3.418	3.99	4.946	8.758	-0.133	-0.008	0.035	0.076	0.111
0.801	1.489	0.02	4.351	9.689	-0.041	0.068	0.100	0.145	0.184
0.801	1.965	0.04	4.354	9.684	-0.052	0.063	0.096	0.144	0.180
0.803	2.944	0.01	4.362	9.673	-0.058	0.062	0.097	0.143	0.177
0.802	3.402	0.02	4.361	9.676	-0.058	0.060	0.096	0.143	0.178
0.802	3.914	0.03	4.358	9.679	-0.056	0.061	0.096	0.143	0.178
0.802	5.862	0.03	4.358	9.679	-0.043	0.067	0.101	0.145	0.184
0.803	7.807	0.00	4.368	9.665	-0.024	0.077	0.106	0.148	0.189
0.799	3.389	8.04	4.342	9.706	-0.097	0.028	0.070	0.127	0.167
0.800	3.383	4.00	4.345	9.698	-0.074	0.037	0.069	0.115	0.152
0.599	1.500	0.00	2.918	11.631	-0.061	0.095	0.142	0.179	0.205
0.599	1.975	0.00	2.917	11.632	-0.073	0.088	0.139	0.174	0.192
0.598	2.926	0.00	2.915	11.636	-0.077	0.088	0.141	0.174	0.191
0.599	3.372	0.01	2.919	11.629	-0.075	0.090	0.143	0.176	0.193
0.598	3.922	0.01	2.916	11.634	-0.075	0.092	0.143	0.177	0.197
0.599	5.826	0.03	2.921	11.626	-0.061	0.099	0.146	0.182	0.209
0.599	7.770	-0.01	2.919	11.629	-0.043	0.107	0.147	0.186	0.220
0.599	3.411	8.01	2.918	11.633	-0.092	0.084	0.141	0.180	0.197
0.600	3.469	3.99	2.929	11.617	-0.078	0.079	0.130	0.164	0.183

Table 3. Continued

(i) $\theta = 270^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.951	0.963	0.974	0.986	
1.202	1.501	0.05	6.148	6.079	-0.261	-0.118	-0.107	-0.092	
1.199	1.971	0.01	6.142	6.099	-0.295	-0.137	-0.126	-0.112	
1.200	2.948	0.02	6.142	6.098	-0.318	-0.152	-0.142	-0.129	
1.200	3.392	0.03	6.143	6.094	-0.320	-0.153	-0.143	-0.130	
1.198	3.919	0.01	6.140	6.109	-0.322	-0.154	-0.143	-0.129	
1.200	5.871	0.02	6.143	6.092	-0.340	-0.165	-0.150	-0.115	
1.200	7.788	0.04	6.143	6.093	-0.253	-0.146	-0.171	-0.150	
1.199	3.394	8.02	6.141	6.100	-0.330	-0.190	-0.238	-0.253	
1.199	3.449	8.03	6.141	6.100	-0.328	-0.188	-0.237	-0.250	
1.198	3.395	4.02	6.138	6.109	-0.214	-0.150	-0.186	-0.195	
0.901	1.476	0.02	4.963	8.729	0.003	0.036	0.075	0.121	
0.901	1.964	0.02	4.958	8.735	0.005	0.040	0.080	0.127	
0.902	2.959	0.04	4.967	8.723	-0.005	0.032	0.075	0.122	
0.901	3.403	0.00	4.963	8.727	-0.003	0.035	0.077	0.123	
0.900	3.910	0.00	4.956	8.739	-0.004	0.034	0.078	0.124	
0.901	5.937	0.01	4.960	8.732	-0.001	0.036	0.077	0.124	
0.900	7.806	0.03	4.957	8.737	0.002	0.036	0.076	0.122	
0.898	3.430	8.03	4.945	8.759	-0.057	-0.005	0.054	0.115	
0.898	3.418	3.99	4.946	8.758	-0.065	-0.039	0.002	0.058	
0.801	1.489	0.02	4.351	9.689	0.036	0.064	0.099	0.141	
0.801	1.965	0.04	4.354	9.684	0.027	0.058	0.095	0.140	
0.803	2.944	0.01	4.362	9.673	0.018	0.057	0.096	0.140	
0.802	3.402	0.02	4.361	9.676	0.019	0.055	0.094	0.140	
0.802	3.914	0.03	4.358	9.679	0.021	0.054	0.094	0.140	
0.802	5.862	0.03	4.358	9.679	0.034	0.062	0.099	0.142	
0.803	7.807	0.00	4.368	9.665	0.050	0.072	0.105	0.145	
0.799	3.389	8.04	4.342	9.706	-0.013	0.016	0.070	0.126	
0.800	3.383	4.00	4.345	9.698	0.007	0.030	0.067	0.111	
0.599	1.500	0.00	2.918	11.631	-0.003	0.088	0.143	0.182	
0.599	1.975	0.00	2.917	11.632	-0.017	0.082	0.142	0.179	
0.598	2.926	0.00	2.915	11.636	-0.019	0.082	0.142	0.179	
0.599	3.372	0.01	2.919	11.629	-0.017	0.084	0.144	0.181	
0.598	3.922	0.01	2.916	11.634	-0.015	0.084	0.145	0.182	
0.599	5.826	0.03	2.921	11.626	-0.001	0.091	0.147	0.185	
0.599	7.770	-0.01	2.919	11.629	0.025	0.098	0.149	0.188	
0.599	3.411	8.01	2.918	11.633	-0.033	0.076	0.143	0.184	
0.600	3.469	3.99	2.929	11.617	-0.017	0.072	0.129	0.167	

Table 3. Continued

(j) $\theta = 285^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.501	0.05	6.148	6.079	-0.199	-0.264	-0.159	-0.106	-0.134	-0.144	-0.111	
1.199	1.971	0.01	6.142	6.099	-0.200	-0.267	-0.189	-0.124	-0.154	-0.166	-0.132	
1.200	2.948	0.02	6.142	6.098	-0.202	-0.270	-0.215	-0.138	-0.171	-0.185	-0.146	
1.200	3.392	0.03	6.143	6.094	-0.202	-0.270	-0.218	-0.139	-0.173	-0.188	-0.148	
1.198	3.919	0.01	6.140	6.109	-0.204	-0.272	-0.220	-0.139	-0.174	-0.189	-0.149	
1.200	5.871	0.02	6.143	6.092	-0.203	-0.272	-0.242	-0.152	-0.185	-0.194	-0.138	
1.200	7.788	0.04	6.143	6.093	-0.204	-0.272	-0.452	-0.163	-0.148	-0.111	-0.009	
1.199	3.394	8.02	6.141	6.100	-0.258	-0.308	-0.513	-0.268	-0.209	-0.201	-0.117	
1.199	3.449	8.03	6.141	6.100	-0.258	-0.309	-0.514	-0.267	-0.209	-0.200	-0.116	
1.198	3.395	4.02	6.138	6.109	-0.265	-0.314	-0.445	-0.189	-0.163	-0.147	-0.072	
0.901	1.476	0.02	4.963	8.729	-0.048	-0.030	0.006	0.040	0.074	0.124	0.181	
0.901	1.964	0.02	4.958	8.735	-0.052	-0.030	0.008	0.044	0.080	0.130	0.184	
0.902	2.959	0.04	4.967	8.723	-0.065	-0.041	-0.002	0.038	0.075	0.126	0.180	
0.901	3.403	0.00	4.963	8.727	-0.062	-0.038	0.002	0.041	0.080	0.130	0.181	
0.900	3.910	0.00	4.956	8.739	-0.062	-0.039	0.001	0.040	0.079	0.130	0.181	
0.901	5.937	0.01	4.960	8.732	-0.055	-0.037	0.003	0.041	0.079	0.131	0.182	
0.900	7.806	0.03	4.957	8.737	-0.046	-0.033	0.006	0.040	0.077	0.129	0.182	
0.898	3.430	8.03	4.945	8.759	-0.121	-0.103	-0.054	0.003	0.059	0.125	0.177	
0.898	3.418	3.99	4.946	8.758	-0.077	-0.080	-0.069	-0.039	0.002	0.064	0.125	
0.801	1.489	0.02	4.351	9.689	-0.021	-0.008	0.016	0.065	0.096	0.141	0.191	
0.801	1.965	0.04	4.354	9.684	-0.033	-0.023	-0.001	0.058	0.092	0.139	0.187	
0.803	2.944	0.01	4.362	9.673	-0.039	-0.031	-0.007	0.056	0.092	0.139	0.187	
0.802	3.402	0.02	4.361	9.676	-0.038	-0.031	-0.009	0.056	0.092	0.139	0.186	
0.802	3.914	0.03	4.358	9.679	-0.037	-0.029	-0.008	0.057	0.093	0.139	0.186	
0.802	5.862	0.03	4.358	9.679	-0.024	-0.011	0.011	0.064	0.098	0.142	0.190	
0.803	7.807	0.00	4.368	9.665	-0.001	0.017	0.036	0.075	0.105	0.145	0.193	
0.799	3.389	8.04	4.342	9.706	-0.042	-0.025	-0.012	0.020	0.068	0.131	0.190	
0.800	3.383	4.00	4.345	9.698	-0.020	-0.005	0.006	0.031	0.065	0.115	0.167	
0.599	1.500	0.00	2.918	11.631	-0.053	-0.068	-0.063	0.079	0.127	0.173	0.203	
0.599	1.975	0.00	2.917	11.632	-0.061	-0.081	-0.081	0.069	0.120	0.166	0.190	
0.598	2.926	0.00	2.915	11.636	-0.064	-0.086	-0.088	0.066	0.120	0.164	0.186	
0.599	3.372	0.01	2.919	11.629	-0.062	-0.083	-0.085	0.067	0.121	0.165	0.188	
0.598	3.922	0.01	2.916	11.634	-0.063	-0.083	-0.085	0.067	0.121	0.165	0.187	
0.599	5.826	0.03	2.921	11.626	-0.053	-0.068	-0.064	0.076	0.127	0.172	0.199	
0.599	7.770	-0.01	2.919	11.629	-0.038	-0.045	-0.032	0.085	0.132	0.176	0.208	
0.599	3.411	8.01	2.918	11.633	-0.075	-0.095	-0.099	0.065	0.125	0.175	0.198	
0.600	3.469	3.99	2.929	11.617	-0.062	-0.070	-0.056	0.056	0.108	0.158	0.186	

Table 3. Continued

(k) $\theta = 300^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.501	0.05	6.148	6.079	-0.235	-0.245	-0.209	-0.251	-0.284	-0.102	-0.067	-0.019	0.008	
1.199	1.971	0.01	6.142	6.099	-0.236	-0.245	-0.210	-0.253	-0.323	-0.117	-0.084	-0.041	-0.010	
1.200	2.948	0.02	6.142	6.098	-0.239	-0.246	-0.213	-0.255	-0.342	-0.133	-0.098	-0.061	-0.023	
1.200	3.392	0.03	6.143	6.094	-0.239	-0.246	-0.213	-0.255	-0.344	-0.134	-0.098	-0.060	-0.018	
1.198	3.919	0.01	6.140	6.109	-0.241	-0.248	-0.215	-0.257	-0.344	-0.133	-0.095	-0.054	-0.008	
1.200	5.871	0.02	6.143	6.092	-0.240	-0.247	-0.216	-0.257	-0.315	-0.144	-0.079	-0.016	0.024	
1.200	7.788	0.04	6.143	6.093	-0.240	-0.246	-0.217	-0.258	-0.156	-0.151	-0.124	-0.081	0.042	
1.199	3.394	8.02	6.141	6.100	-0.357	-0.339	-0.273	-0.293	-0.194	-0.224	-0.233	-0.214	-0.052	
1.199	3.449	8.03	6.141	6.100	-0.357	-0.339	-0.273	-0.294	-0.192	-0.222	-0.231	-0.213	-0.049	
1.198	3.395	4.02	6.138	6.109	-0.320	-0.313	-0.274	-0.310	-0.145	-0.166	-0.171	-0.156	-0.032	
0.901	1.476	0.02	4.963	8.729	-0.120	-0.065	-0.045	-0.030	0.043	0.077	0.123	0.174	0.133	
0.901	1.964	0.02	4.958	8.735	-0.145	-0.077	-0.049	-0.030	0.045	0.082	0.127	0.176	0.125	
0.902	2.959	0.04	4.967	8.723	-0.167	-0.090	-0.061	-0.041	0.038	0.076	0.122	0.171	0.121	
0.901	3.403	0.00	4.963	8.727	-0.165	-0.088	-0.057	-0.040	0.041	0.077	0.123	0.171	0.124	
0.900	3.910	0.00	4.956	8.739	-0.170	-0.090	-0.058	-0.040	0.040	0.077	0.122	0.172	0.126	
0.901	5.937	0.01	4.960	8.732	-0.136	-0.076	-0.050	-0.035	0.042	0.077	0.123	0.177	0.140	
0.900	7.806	0.03	4.957	8.737	-0.099	-0.058	-0.041	-0.030	0.043	0.077	0.123	0.181	0.149	
0.898	3.430	8.03	4.945	8.759	-0.131	-0.119	-0.112	-0.098	0.007	0.057	0.117	0.172	0.115	
0.898	3.418	3.99	4.946	8.758	-0.077	-0.068	-0.070	-0.076	-0.027	0.009	0.062	0.129	0.099	
0.801	1.489	0.02	4.351	9.689	-0.156	-0.074	-0.037	-0.035	0.080	0.104	0.142	0.185	0.151	
0.801	1.965	0.04	4.354	9.684	-0.164	-0.083	-0.047	-0.049	0.075	0.101	0.141	0.183	0.137	
0.803	2.944	0.01	4.362	9.673	-0.169	-0.088	-0.053	-0.057	0.074	0.101	0.141	0.179	0.134	
0.802	3.402	0.02	4.361	9.676	-0.167	-0.087	-0.053	-0.057	0.072	0.101	0.141	0.180	0.136	
0.802	3.914	0.03	4.358	9.679	-0.166	-0.086	-0.051	-0.055	0.072	0.100	0.140	0.181	0.140	
0.802	5.862	0.03	4.358	9.679	-0.157	-0.077	-0.039	-0.038	0.079	0.104	0.141	0.187	0.156	
0.803	7.807	0.00	4.368	9.665	-0.142	-0.058	-0.018	-0.012	0.084	0.106	0.141	0.192	0.165	
0.799	3.389	8.04	4.342	9.706	-0.136	-0.075	-0.048	-0.047	0.030	0.069	0.121	0.181	0.124	
0.800	3.383	4.00	4.345	9.698	-0.144	-0.068	-0.032	-0.028	0.042	0.068	0.109	0.165	0.121	
0.599	1.500	0.00	2.918	11.631	-0.136	-0.080	-0.061	-0.080	0.095	0.143	0.175	0.207	0.160	
0.599	1.975	0.00	2.917	11.632	-0.140	-0.085	-0.068	-0.090	0.086	0.139	0.170	0.196	0.135	
0.598	2.926	0.00	2.915	11.636	-0.143	-0.089	-0.071	-0.095	0.085	0.139	0.169	0.194	0.134	
0.599	3.372	0.01	2.919	11.629	-0.141	-0.086	-0.069	-0.092	0.087	0.141	0.171	0.197	0.139	
0.598	3.922	0.01	2.916	11.634	-0.141	-0.088	-0.070	-0.093	0.089	0.143	0.171	0.200	0.146	
0.599	5.826	0.03	2.921	11.626	-0.136	-0.082	-0.061	-0.080	0.099	0.146	0.177	0.213	0.168	
0.599	7.770	-0.01	2.919	11.629	-0.128	-0.071	-0.048	-0.061	0.112	0.148	0.182	0.224	0.185	
0.599	3.411	8.01	2.918	11.633	-0.176	-0.109	-0.084	-0.110	0.082	0.140	0.173	0.199	0.131	
0.600	3.469	3.99	2.929	11.617	-0.154	-0.095	-0.070	-0.084	0.077	0.129	0.159	0.186	0.131	

Table 3. Continued

(I) $\theta = 315^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.202	1.501	0.05	6.148	6.079	-0.183	-0.195	-0.243	-0.227	-0.299	-0.316	-0.109	-0.123	-0.107	-0.100
1.199	1.971	0.01	6.142	6.099	-0.184	-0.193	-0.243	-0.226	-0.300	-0.372	-0.129	-0.146	-0.130	-0.120
1.200	2.948	0.02	6.142	6.098	-0.185	-0.193	-0.244	-0.228	-0.301	-0.402	-0.147	-0.164	-0.151	-0.139
1.200	3.392	0.03	6.143	6.094	-0.184	-0.191	-0.243	-0.227	-0.300	-0.408	-0.149	-0.166	-0.153	-0.143
1.198	3.919	0.01	6.140	6.109	-0.186	-0.192	-0.245	-0.228	-0.302	-0.414	-0.151	-0.168	-0.154	-0.146
1.200	5.871	0.02	6.143	6.092	-0.184	-0.190	-0.243	-0.227	-0.300	-0.424	-0.164	-0.177	-0.148	-0.132
1.200	7.788	0.04	6.143	6.093	-0.185	-0.189	-0.243	-0.227	-0.300	-0.427	-0.225	-0.161	-0.086	-0.031
1.199	3.394	8.02	6.141	6.100	-0.312	-0.316	-0.343	-0.309	-0.354	-0.481	-0.416	-0.257	-0.190	-0.119
1.199	3.449	8.03	6.141	6.101	-0.312	-0.316	-0.344	-0.309	-0.354	-0.482	-0.416	-0.256	-0.189	-0.118
1.198	3.395	4.02	6.138	6.109	-0.270	-0.279	-0.316	-0.288	-0.355	-0.474	-0.284	-0.208	-0.118	-0.069
0.901	1.476	0.02	4.963	8.729	-0.392	-0.201	-0.083	-0.052	-0.034	-0.005	0.054	0.084	0.128	0.171
0.901	1.964	0.02	4.958	8.735	-0.423	-0.234	-0.094	-0.055	-0.037	-0.012	0.061	0.091	0.130	0.172
0.902	2.959	0.04	4.967	8.723	-0.441	-0.262	-0.110	-0.064	-0.047	-0.020	0.056	0.087	0.127	0.167
0.901	3.403	0.00	4.963	8.727	-0.441	-0.260	-0.107	-0.062	-0.044	-0.021	0.059	0.090	0.130	0.169
0.900	3.910	0.00	4.956	8.739	-0.445	-0.266	-0.109	-0.063	-0.046	-0.020	0.059	0.088	0.131	0.170
0.901	5.937	0.01	4.960	8.732	-0.417	-0.229	-0.093	-0.058	-0.040	-0.010	0.057	0.087	0.135	0.175
0.900	7.806	0.03	4.957	8.737	-0.365	-0.177	-0.076	-0.052	-0.035	0.001	0.053	0.083	0.136	0.179
0.898	3.430	8.03	4.945	8.759	-0.367	-0.190	-0.128	-0.110	-0.088	-0.043	0.020	0.067	0.124	0.167
0.898	3.418	3.99	4.946	8.758	-0.258	-0.129	-0.088	-0.081	-0.076	-0.047	-0.018	0.017	0.086	0.138
0.801	1.489	0.02	4.351	9.689	-0.241	-0.160	-0.085	-0.053	-0.063	-0.082	0.082	0.122	0.152	0.184
0.801	1.965	0.04	4.354	9.684	-0.248	-0.167	-0.092	-0.061	-0.074	-0.098	0.072	0.119	0.147	0.179
0.803	2.944	0.01	4.362	9.673	-0.252	-0.170	-0.097	-0.066	-0.080	-0.106	0.069	0.117	0.146	0.175
0.802	3.402	0.02	4.361	9.676	-0.250	-0.170	-0.095	-0.065	-0.079	-0.107	0.068	0.117	0.146	0.174
0.802	3.914	0.03	4.358	9.679	-0.250	-0.170	-0.095	-0.064	-0.078	-0.106	0.068	0.118	0.148	0.176
0.802	5.862	0.03	4.358	9.679	-0.244	-0.162	-0.087	-0.055	-0.066	-0.085	0.080	0.122	0.154	0.187
0.803	7.807	0.00	4.368	9.665	-0.232	-0.151	-0.073	-0.039	-0.045	-0.056	0.097	0.126	0.159	0.197
0.799	3.389	8.04	4.342	9.706	-0.263	-0.161	-0.091	-0.067	-0.080	-0.097	0.059	0.094	0.138	0.178
0.800	3.383	4.00	4.345	9.698	-0.252	-0.163	-0.086	-0.056	-0.060	-0.062	0.059	0.083	0.123	0.160
0.599	1.500	0.00	2.918	11.631	-0.169	-0.133	-0.084	-0.073	-0.097	-0.124	0.055	0.131	0.168	0.191
0.599	1.975	0.00	2.917	11.632	-0.172	-0.137	-0.089	-0.079	-0.106	-0.138	0.044	0.122	0.156	0.178
0.598	2.926	0.00	2.915	11.636	-0.176	-0.140	-0.092	-0.083	-0.110	-0.143	0.039	0.117	0.153	0.172
0.599	3.372	0.01	2.919	11.629	-0.172	-0.138	-0.090	-0.081	-0.108	-0.142	0.041	0.119	0.155	0.174
0.598	3.922	0.01	2.916	11.634	-0.173	-0.139	-0.091	-0.082	-0.109	-0.141	0.041	0.120	0.157	0.176
0.599	5.826	0.03	2.921	11.626	-0.171	-0.135	-0.086	-0.075	-0.098	-0.125	0.054	0.130	0.169	0.193
0.599	7.770	-0.01	2.919	11.629	-0.164	-0.128	-0.077	-0.063	-0.083	-0.104	0.070	0.141	0.179	0.210
0.599	3.411	8.01	2.918	11.633	-0.224	-0.175	-0.114	-0.099	-0.128	-0.168	0.039	0.125	0.161	0.181
0.600	3.469	3.99	2.929	11.617	-0.195	-0.152	-0.099	-0.085	-0.103	-0.121	0.046	0.116	0.148	0.170

Table 3. Continued

(m) Top interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—			
					0.709	0.723	0.898	0.941
1.202	1.501	0.05	6.148	6.079	-0.040	-0.038	-0.224	-0.262
1.199	1.971	0.01	6.142	6.099	-0.041	-0.037	-0.226	-0.267
1.200	2.948	0.02	6.142	6.098	-0.043	-0.039	-0.228	-0.271
1.200	3.392	0.03	6.143	6.094	-0.044	-0.038	-0.228	-0.270
1.198	3.919	0.01	6.140	6.109	-0.045	-0.038	-0.231	-0.272
1.200	5.871	0.02	6.143	6.092	-0.045	-0.037	-0.229	-0.272
1.200	7.788	0.04	6.143	6.093	-0.046	-0.038	-0.231	-0.271
1.199	3.394	8.02	6.141	6.100	-0.178	-0.167	-0.346	-0.295
1.199	3.449	8.03	6.141	6.100	-0.179	-0.166	-0.347	-0.295
1.198	3.395	4.02	6.138	6.109	-0.128	-0.125	-0.311	-0.298
0.901	1.476	0.02	4.963	8.729	-0.040	-0.016	-0.122	-0.024
0.901	1.964	0.02	4.958	8.735	-0.044	-0.019	-0.148	-0.027
0.902	2.959	0.04	4.967	8.723	-0.043	-0.018	-0.168	-0.038
0.901	3.403	0.00	4.963	8.727	-0.042	-0.017	-0.167	-0.035
0.900	3.910	0.00	4.956	8.739	-0.042	-0.018	-0.172	-0.036
0.901	5.937	0.01	4.960	8.732	-0.042	-0.018	-0.139	-0.032
0.900	7.806	0.03	4.957	8.737	-0.041	-0.017	-0.102	-0.023
0.898	3.430	8.03	4.945	8.759	-0.096	-0.072	-0.139	-0.086
0.898	3.418	3.99	4.946	8.758	-0.044	-0.029	-0.086	-0.063
0.801	1.489	0.02	4.351	9.689	-0.042	-0.030	-0.154	0.012
0.801	1.965	0.04	4.354	9.684	-0.042	-0.031	-0.161	-0.002
0.803	2.944	0.01	4.362	9.673	-0.042	-0.031	-0.166	-0.011
0.802	3.402	0.02	4.361	9.676	-0.042	-0.030	-0.164	-0.010
0.802	3.914	0.03	4.358	9.679	-0.041	-0.031	-0.164	-0.009
0.802	5.862	0.03	4.358	9.679	-0.043	-0.032	-0.155	0.008
0.803	7.807	0.00	4.368	9.665	-0.040	-0.030	-0.140	0.032
0.799	3.389	8.04	4.342	9.706	-0.092	-0.073	-0.138	-0.025
0.800	3.383	4.00	4.345	9.698	-0.061	-0.049	-0.143	0.000
0.599	1.500	0.00	2.918	11.631	-0.035	-0.035	-0.133	-0.050
0.599	1.975	0.00	2.917	11.632	-0.033	-0.035	-0.136	-0.063
0.598	2.926	0.00	2.915	11.636	-0.035	-0.036	-0.139	-0.069
0.599	3.372	0.01	2.919	11.629	-0.031	-0.034	-0.137	-0.067
0.598	3.922	0.01	2.916	11.634	-0.035	-0.036	-0.138	-0.067
0.599	5.826	0.03	2.921	11.626	-0.034	-0.036	-0.133	-0.051
0.599	7.770	-0.01	2.919	11.629	-0.031	-0.034	-0.125	-0.027
0.599	3.411	8.01	2.918	11.633	-0.075	-0.074	-0.171	-0.073
0.600	3.469	3.99	2.929	11.617	-0.051	-0.052	-0.151	-0.052

Table 3. Concluded

(n) Bottom interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.709	0.723	0.738	0.752	0.767	0.781	0.869	0.883	0.941	
1.202	1.501	0.05	6.148	6.079	-0.022	-0.047	-0.071	-0.085	-0.096	-0.098	-0.183	-0.200	-0.167	
1.199	1.971	0.01	6.142	6.099	-0.024	-0.048	-0.072	-0.087	-0.098	-0.099	-0.185	-0.202	-0.167	
1.200	2.948	0.02	6.142	6.098	-0.024	-0.047	-0.072	-0.086	-0.098	-0.101	-0.185	-0.202	-0.170	
1.200	3.392	0.03	6.143	6.094	-0.022	-0.046	-0.072	-0.085	-0.099	-0.101	-0.184	-0.202	-0.171	
1.198	3.919	0.01	6.140	6.109	-0.023	-0.047	-0.072	-0.085	-0.099	-0.102	-0.186	-0.204	-0.170	
1.200	5.871	0.02	6.143	6.092	-0.020	-0.044	-0.070	-0.084	-0.098	-0.102	-0.183	-0.202	-0.174	
1.200	7.788	0.04	6.143	6.093	-0.018	-0.043	-0.069	-0.083	-0.099	-0.103	-0.183	-0.202	-0.173	
1.199	3.394	8.02	6.141	6.100	0.179	0.147	0.113	0.089	0.058	0.020	-0.076	-0.075	-0.129	
1.199	3.449	8.03	6.141	6.100	0.179	0.147	0.113	0.089	0.057	0.020	-0.075	-0.075	-0.129	
1.198	3.395	4.02	6.138	6.109	0.065	0.050	0.034	0.028	0.007	-0.024	-0.163	-0.170	-0.146	
0.901	1.476	0.02	4.963	8.729	-0.053	-0.053	-0.049	-0.046	-0.053	-0.077	-0.113	-0.115	-0.062	
0.901	1.964	0.02	4.958	8.735	-0.055	-0.055	-0.051	-0.046	-0.054	-0.079	-0.119	-0.116	-0.072	
0.902	2.959	0.04	4.967	8.723	-0.056	-0.056	-0.052	-0.048	-0.056	-0.079	-0.117	-0.115	-0.069	
0.901	3.403	0.00	4.963	8.727	-0.054	-0.055	-0.051	-0.047	-0.054	-0.078	-0.117	-0.115	-0.070	
0.900	3.910	0.00	4.956	8.739	-0.054	-0.055	-0.051	-0.047	-0.054	-0.078	-0.118	-0.115	-0.065	
0.901	5.937	0.01	4.960	8.732	-0.056	-0.056	-0.051	-0.047	-0.053	-0.077	-0.116	-0.110	-0.061	
0.900	7.806	0.03	4.957	8.737	-0.056	-0.055	-0.051	-0.047	-0.053	-0.077	-0.114	-0.110	-0.061	
0.898	3.430	8.03	4.945	8.759	-0.032	-0.061	-0.073	-0.077	-0.084	-0.109	-0.095	-0.092	-0.125	
0.898	3.418	3.99	4.946	8.758	-0.039	-0.054	-0.056	-0.056	-0.061	-0.085	-0.109	-0.110	-0.127	
0.801	1.489	0.02	4.351	9.689	-0.040	-0.052	-0.054	-0.055	-0.064	-0.087	-0.093	-0.087	-0.029	
0.801	1.965	0.04	4.354	9.684	-0.041	-0.053	-0.054	-0.056	-0.064	-0.088	-0.095	-0.090	-0.041	
0.803	2.944	0.01	4.362	9.673	-0.041	-0.053	-0.055	-0.056	-0.065	-0.089	-0.097	-0.091	-0.047	
0.802	3.402	0.02	4.361	9.676	-0.042	-0.053	-0.055	-0.056	-0.065	-0.088	-0.096	-0.091	-0.046	
0.802	3.914	0.03	4.358	9.679	-0.042	-0.053	-0.055	-0.057	-0.065	-0.089	-0.096	-0.091	-0.046	
0.802	5.862	0.03	4.358	9.679	-0.042	-0.053	-0.055	-0.056	-0.065	-0.088	-0.096	-0.091	-0.046	
0.803	7.807	0.00	4.368	9.665	-0.043	-0.055	-0.056	-0.057	-0.064	-0.087	-0.094	-0.088	-0.031	
0.799	3.389	8.04	4.342	9.706	0.003	-0.020	-0.028	-0.033	-0.042	-0.068	-0.070	-0.068	-0.088	
0.800	3.383	4.00	4.345	9.698	-0.015	-0.031	-0.036	-0.039	-0.046	-0.071	-0.084	-0.081	-0.063	
0.599	1.500	0.00	2.918	11.631	-0.034	-0.045	-0.048	-0.048	-0.057	-0.075	-0.075	-0.069	-0.056	
0.599	1.975	0.00	2.917	11.632	-0.034	-0.046	-0.049	-0.049	-0.057	-0.075	-0.076	-0.072	-0.069	
0.598	2.926	0.00	2.915	11.636	-0.034	-0.045	-0.049	-0.049	-0.057	-0.075	-0.076	-0.073	-0.074	
0.599	3.372	0.01	2.919	11.629	-0.032	-0.044	-0.048	-0.048	-0.055	-0.073	-0.075	-0.071	-0.071	
0.598	3.922	0.01	2.916	11.634	-0.033	-0.045	-0.048	-0.048	-0.056	-0.073	-0.075	-0.071	-0.071	
0.599	5.826	0.03	2.921	11.626	-0.033	-0.045	-0.048	-0.048	-0.056	-0.073	-0.074	-0.069	-0.057	
0.599	7.770	-0.01	2.919	11.629	-0.032	-0.044	-0.048	-0.048	-0.055	-0.073	-0.071	-0.066	-0.036	
0.599	3.411	8.01	2.918	11.633	-0.032	-0.044	-0.048	-0.048	-0.055	-0.073	-0.071	-0.066	-0.089	
0.600	3.469	3.99	2.929	11.617	0.003	-0.012	-0.020	-0.022	-0.030	-0.050	-0.059	-0.058	-0.073	

Table 4. Static Pressure Coefficients of Afterbody Surface With Horizontal Tail in Mid Position and Vertical Tail in Aft Position

(a) $\theta = 0^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
1.199	1.483	-0.02	6.151	6.115	-0.058	-0.079	-0.084	-0.078	-0.050	-0.068	-0.095	-0.097	-0.106	-0.062	-0.063
1.199	1.983	0.00	6.151	6.117	-0.060	-0.080	-0.085	-0.079	-0.051	-0.069	-0.096	-0.098	-0.106	-0.063	-0.066
1.200	2.938	0.00	6.154	6.107	-0.060	-0.079	-0.086	-0.081	-0.052	-0.069	-0.096	-0.098	-0.105	-0.063	-0.065
1.198	3.413	0.00	6.150	6.118	-0.061	-0.079	-0.086	-0.081	-0.053	-0.070	-0.097	-0.099	-0.107	-0.064	-0.067
1.199	3.907	0.01	6.152	6.113	-0.062	-0.080	-0.086	-0.082	-0.053	-0.070	-0.096	-0.099	-0.106	-0.064	-0.067
1.199	5.894	0.03	6.151	6.115	-0.062	-0.079	-0.086	-0.083	-0.055	-0.071	-0.097	-0.100	-0.107	-0.064	-0.068
1.198	7.810	0.01	6.150	6.118	-0.063	-0.080	-0.087	-0.084	-0.055	-0.071	-0.097	-0.100	-0.108	-0.065	-0.069
1.200	3.430	8.00	6.154	6.109	-0.201	-0.220	-0.213	-0.193	-0.152	-0.145	-0.123	-0.100	-0.087	-0.082	-0.133
1.202	3.410	4.00	6.158	6.093	-0.135	-0.154	-0.150	-0.138	-0.108	-0.118	-0.130	-0.123	-0.129	-0.113	-0.145
0.899	1.480	0.02	4.958	8.769	-0.041	-0.029	-0.022	-0.015	0.006	-0.007	-0.005	0.019	0.028	-0.004	-0.102
0.901	1.989	-0.01	4.971	8.747	-0.044	-0.033	-0.024	-0.016	0.005	-0.008	-0.005	0.020	0.029	-0.002	-0.099
0.903	2.939	0.00	4.980	8.732	-0.042	-0.031	-0.022	-0.014	0.006	-0.007	-0.004	0.021	0.030	-0.001	-0.098
0.899	3.377	0.00	4.960	8.767	-0.041	-0.032	-0.023	-0.016	0.005	-0.009	-0.006	0.019	0.027	-0.004	-0.101
0.900	3.932	0.00	4.968	8.754	-0.040	-0.031	-0.023	-0.014	0.007	-0.006	-0.003	0.021	0.030	-0.002	-0.100
0.899	5.811	0.01	4.960	8.768	-0.041	-0.032	-0.023	-0.016	0.006	-0.008	-0.005	0.019	0.028	-0.003	-0.101
0.899	7.794	0.02	4.957	8.772	-0.042	-0.032	-0.023	-0.016	0.005	-0.008	-0.005	0.020	0.028	-0.004	-0.102
0.901	3.433	8.03	4.970	8.752	-0.121	-0.126	-0.127	-0.123	-0.098	-0.112	-0.101	-0.065	-0.043	-0.065	-0.160
0.901	3.406	4.01	4.969	8.753	-0.054	-0.054	-0.050	-0.044	-0.025	-0.037	-0.033	-0.007	0.004	-0.029	-0.130
0.800	1.483	-0.01	4.354	9.720	-0.036	-0.037	-0.035	-0.032	-0.014	-0.030	-0.033	-0.014	-0.018	-0.064	-0.166
0.800	1.972	0.00	4.357	9.716	-0.036	-0.038	-0.035	-0.032	-0.013	-0.030	-0.033	-0.014	-0.017	-0.063	-0.166
0.800	2.946	0.01	4.355	9.718	-0.037	-0.039	-0.035	-0.033	-0.014	-0.031	-0.034	-0.014	-0.018	-0.065	-0.168
0.801	3.397	0.01	4.361	9.710	-0.038	-0.039	-0.036	-0.033	-0.013	-0.031	-0.034	-0.015	-0.018	-0.065	-0.168
0.801	3.914	0.01	4.360	9.711	-0.037	-0.039	-0.036	-0.033	-0.013	-0.031	-0.032	-0.014	-0.018	-0.064	-0.166
0.802	5.871	0.03	4.368	9.700	-0.036	-0.038	-0.035	-0.032	-0.012	-0.030	-0.032	-0.014	-0.018	-0.064	-0.166
0.802	7.776	0.01	4.365	9.704	-0.035	-0.038	-0.034	-0.031	-0.011	-0.030	-0.032	-0.014	-0.017	-0.061	-0.161
0.800	3.450	8.03	4.358	9.717	-0.104	-0.107	-0.102	-0.096	-0.072	-0.094	-0.097	-0.077	-0.082	-0.136	-0.246
0.800	3.405	4.00	4.354	9.723	-0.062	-0.063	-0.058	-0.051	-0.034	-0.051	-0.055	-0.039	-0.048	-0.102	-0.209
0.601	1.486	0.01	2.940	11.613	-0.031	-0.036	-0.039	-0.037	-0.020	-0.038	-0.045	-0.032	-0.043	-0.088	-0.168
0.602	1.961	0.01	2.945	11.607	-0.031	-0.037	-0.040	-0.038	-0.021	-0.039	-0.046	-0.034	-0.045	-0.092	-0.174
0.602	2.922	0.02	2.943	11.610	-0.031	-0.038	-0.041	-0.039	-0.023	-0.040	-0.048	-0.036	-0.048	-0.095	-0.176
0.602	3.384	0.02	2.947	11.605	-0.031	-0.038	-0.040	-0.038	-0.021	-0.040	-0.047	-0.035	-0.047	-0.093	-0.174
0.602	3.957	0.02	2.947	11.605	-0.031	-0.039	-0.041	-0.038	-0.021	-0.041	-0.049	-0.035	-0.048	-0.095	-0.176
0.603	5.882	0.02	2.949	11.602	-0.032	-0.039	-0.041	-0.039	-0.022	-0.040	-0.048	-0.035	-0.047	-0.093	-0.172
0.603	7.782	0.02	2.950	11.601	-0.030	-0.039	-0.041	-0.038	-0.021	-0.040	-0.046	-0.033	-0.043	-0.088	-0.165
0.601	3.390	8.04	2.935	11.623	-0.086	-0.096	-0.090	-0.085	-0.064	-0.089	-0.100	-0.088	-0.109	-0.167	-0.252
0.601	3.427	4.00	2.940	11.615	-0.054	-0.064	-0.061	-0.053	-0.037	-0.058	-0.068	-0.058	-0.077	-0.130	-0.211

Table 4. Continued

(a) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.869	0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	
1.199	1.483	-0.02	6.151	6.115	-0.124	-0.143	-0.160	-0.207	-0.268	-0.312	-0.356	-0.500	-0.183	
1.199	1.983	0.00	6.151	6.117	-0.127	-0.145	-0.162	-0.208	-0.269	-0.314	-0.358	-0.512	-0.203	
1.200	2.938	0.00	6.154	6.107	-0.127	-0.145	-0.163	-0.208	-0.269	-0.315	-0.360	-0.513	-0.234	
1.198	3.413	0.00	6.150	6.118	-0.129	-0.146	-0.164	-0.208	-0.270	-0.316	-0.362	-0.515	-0.243	
1.199	3.907	0.01	6.152	6.113	-0.129	-0.147	-0.165	-0.208	-0.270	-0.316	-0.362	-0.515	-0.256	
1.199	5.894	0.03	6.151	6.115	-0.131	-0.148	-0.167	-0.209	-0.271	-0.318	-0.363	-0.516	-0.219	
1.198	7.810	0.01	6.150	6.118	-0.132	-0.150	-0.169	-0.209	-0.273	-0.320	-0.365	-0.507	-0.159	
1.200	3.430	8.00	6.154	6.109	-0.163	-0.194	-0.221	-0.267	-0.335	-0.394	-0.433	-0.507	-0.280	
1.202	3.410	4.00	6.158	6.093	-0.183	-0.213	-0.230	-0.265	-0.331	-0.381	-0.404	-0.535	-0.222	
0.899	1.480	0.02	4.958	8.769	-0.183	-0.245	-0.322	-0.416	-0.340	-0.100	-0.045	0.015	0.046	
0.901	1.989	-0.01	4.971	8.747	-0.180	-0.242	-0.319	-0.416	-0.382	-0.115	-0.053	0.005	0.038	
0.903	2.939	0.00	4.980	8.732	-0.178	-0.240	-0.317	-0.415	-0.401	-0.126	-0.056	-0.001	0.031	
0.899	3.377	0.00	4.960	8.767	-0.182	-0.244	-0.321	-0.419	-0.402	-0.126	-0.060	-0.002	0.032	
0.900	3.932	0.00	4.968	8.754	-0.181	-0.243	-0.320	-0.418	-0.399	-0.126	-0.059	0.000	0.032	
0.899	5.811	0.01	4.960	8.768	-0.181	-0.244	-0.321	-0.418	-0.371	-0.111	-0.051	0.009	0.042	
0.899	7.794	0.02	4.957	8.772	-0.182	-0.245	-0.320	-0.413	-0.309	-0.087	-0.033	0.026	0.056	
0.901	3.433	8.03	4.970	8.752	-0.242	-0.312	-0.381	-0.454	-0.261	-0.078	-0.027	-0.004	0.012	
0.901	3.406	4.01	4.969	8.753	-0.213	-0.280	-0.347	-0.449	-0.341	-0.104	-0.047	-0.009	0.016	
0.800	1.483	-0.01	4.354	9.720	-0.237	-0.287	-0.320	-0.316	-0.250	-0.156	-0.134	0.023	0.079	
0.800	1.972	0.00	4.357	9.716	-0.239	-0.291	-0.328	-0.325	-0.260	-0.165	-0.144	0.016	0.072	
0.800	2.946	0.01	4.355	9.718	-0.241	-0.294	-0.333	-0.333	-0.267	-0.171	-0.152	0.010	0.068	
0.801	3.397	0.01	4.361	9.710	-0.242	-0.296	-0.336	-0.334	-0.269	-0.172	-0.153	0.009	0.067	
0.801	3.914	0.01	4.360	9.711	-0.241	-0.295	-0.334	-0.334	-0.268	-0.171	-0.151	0.011	0.068	
0.802	5.871	0.03	4.368	9.700	-0.237	-0.289	-0.324	-0.320	-0.251	-0.155	-0.133	0.022	0.076	
0.802	7.776	0.01	4.365	9.704	-0.229	-0.278	-0.306	-0.298	-0.229	-0.133	-0.109	0.039	0.088	
0.800	3.450	8.03	4.358	9.717	-0.322	-0.372	-0.381	-0.336	-0.238	-0.138	-0.091	-0.005	0.024	
0.800	3.405	4.00	4.354	9.723	-0.281	-0.331	-0.351	-0.329	-0.246	-0.149	-0.121	0.005	0.045	
0.601	1.486	0.01	2.940	11.613	-0.208	-0.231	-0.243	-0.246	-0.223	-0.170	-0.170	-0.006	0.087	
0.602	1.961	0.01	2.945	11.607	-0.213	-0.238	-0.251	-0.256	-0.234	-0.183	-0.185	-0.020	0.075	
0.602	2.922	0.02	2.943	11.610	-0.215	-0.241	-0.254	-0.261	-0.240	-0.189	-0.191	-0.026	0.069	
0.602	3.384	0.02	2.947	11.605	-0.214	-0.240	-0.255	-0.262	-0.240	-0.189	-0.192	-0.026	0.069	
0.602	3.957	0.02	2.947	11.605	-0.216	-0.242	-0.256	-0.264	-0.241	-0.189	-0.190	-0.025	0.069	
0.603	5.882	0.02	2.949	11.602	-0.211	-0.236	-0.248	-0.255	-0.231	-0.178	-0.175	-0.012	0.081	
0.603	7.782	0.02	2.950	11.601	-0.203	-0.227	-0.237	-0.242	-0.215	-0.159	-0.153	0.006	0.096	
0.601	3.390	8.04	2.935	11.623	-0.286	-0.306	-0.304	-0.302	-0.256	-0.187	-0.165	-0.023	0.052	
0.601	3.427	4.00	2.940	11.615	-0.243	-0.267	-0.271	-0.278	-0.240	-0.180	-0.167	-0.020	0.063	

Table 4. Continued

(b) $\theta = 45^\circ$

M	NPR	α, deg	q _∞ , psi	p _∞ , psi	C _p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.941	0.951	0.963	0.974	0.997	
1.199	1.483	-0.02	6.151	6.115	-0.056	-0.004	0.008	-0.032	-0.059	-0.091	-0.431	-0.371	-0.281	-0.194	-0.040	
1.199	1.983	0.00	6.151	6.117	-0.057	-0.004	0.008	-0.032	-0.059	-0.091	-0.433	-0.422	-0.354	-0.221	-0.063	
1.200	2.938	0.00	6.154	6.107	-0.057	-0.005	0.008	-0.031	-0.057	-0.090	-0.434	-0.445	-0.421	-0.236	-0.078	
1.198	3.413	0.00	6.150	6.118	-0.059	-0.005	0.008	-0.031	-0.058	-0.091	-0.434	-0.446	-0.427	-0.237	-0.074	
1.199	3.907	0.01	6.152	6.113	-0.060	-0.006	0.007	-0.031	-0.058	-0.091	-0.435	-0.446	-0.424	-0.237	-0.063	
1.199	5.894	0.03	6.151	6.115	-0.061	-0.008	0.007	-0.032	-0.058	-0.091	-0.436	-0.426	-0.364	-0.221	-0.019	
1.198	7.810	0.01	6.150	6.118	-0.062	-0.008	0.007	-0.031	-0.057	-0.092	-0.435	-0.376	-0.286	-0.192	0.010	
1.200	3.430	8.00	6.154	6.109	-0.177	-0.085	-0.087	-0.103	-0.114	-0.108	-0.496	-0.446	-0.310	-0.288	-0.080	
1.202	3.410	4.00	6.158	6.093	-0.116	-0.042	-0.047	-0.077	-0.111	-0.136	-0.482	-0.393	-0.325	-0.242	-0.079	
0.899	1.480	0.02	4.958	8.769	0.004	0.023	0.009	0.006	0.016	0.032	-0.062	-0.061	-0.096	-0.047	0.124	
0.901	1.989	-0.01	4.971	8.747	0.004	0.024	0.010	0.007	0.018	0.033	-0.070	-0.069	-0.103	-0.058	0.112	
0.903	2.939	0.00	4.980	8.732	0.001	0.022	0.009	0.006	0.018	0.033	-0.078	-0.076	-0.111	-0.063	0.112	
0.899	3.377	0.00	4.960	8.767	0.002	0.021	0.007	0.004	0.015	0.030	-0.077	-0.077	-0.112	-0.061	0.113	
0.900	3.932	0.00	4.968	8.754	0.003	0.023	0.009	0.006	0.018	0.033	-0.077	-0.076	-0.111	-0.061	0.117	
0.899	5.811	0.01	4.960	8.768	0.003	0.023	0.009	0.005	0.017	0.032	-0.068	-0.067	-0.102	-0.053	0.130	
0.899	7.794	0.02	4.957	8.772	0.007	0.026	0.009	0.006	0.017	0.033	-0.054	-0.051	-0.084	-0.035	0.141	
0.901	3.433	8.03	4.970	8.752	-0.078	-0.069	-0.105	-0.119	-0.101	-0.078	-0.083	-0.083	-0.108	-0.062	0.100	
0.901	3.406	4.01	4.969	8.753	-0.008	0.002	-0.025	-0.034	-0.021	-0.003	-0.080	-0.083	-0.116	-0.070	0.105	
0.800	1.483	-0.01	4.354	9.720	-0.002	0.008	-0.009	-0.014	-0.004	0.007	-0.063	-0.037	-0.063	0.006	0.137	
0.800	1.972	0.00	4.357	9.716	-0.001	0.010	-0.007	-0.012	-0.003	0.008	-0.068	-0.045	-0.069	0.001	0.125	
0.800	2.946	0.01	4.355	9.718	-0.003	0.010	-0.008	-0.012	-0.004	0.007	-0.078	-0.050	-0.074	-0.002	0.123	
0.801	3.397	0.01	4.361	9.710	-0.001	0.010	-0.008	-0.013	-0.004	0.007	-0.077	-0.050	-0.074	-0.003	0.123	
0.801	3.914	0.01	4.360	9.711	0.000	0.010	-0.008	-0.013	-0.003	0.008	-0.079	-0.051	-0.075	-0.003	0.127	
0.802	5.871	0.03	4.368	9.700	-0.002	0.009	-0.008	-0.014	-0.004	0.007	-0.066	-0.039	-0.065	0.005	0.141	
0.802	7.776	0.01	4.365	9.704	0.000	0.011	-0.007	-0.012	-0.002	0.009	-0.047	-0.023	-0.051	0.017	0.152	
0.800	3.450	8.03	4.358	9.717	-0.057	-0.046	-0.069	-0.076	-0.067	-0.060	-0.077	-0.069	-0.098	-0.035	0.112	
0.800	3.405	4.00	4.354	9.723	-0.009	-0.001	-0.024	-0.031	-0.024	-0.016	-0.072	-0.060	-0.097	-0.037	0.116	
0.601	1.486	0.01	2.940	11.613	0.001	0.007	-0.011	-0.015	-0.010	-0.001	-0.061	-0.041	-0.034	0.043	0.133	
0.602	1.961	0.01	2.945	11.607	0.000	0.006	-0.011	-0.016	-0.010	-0.002	-0.072	-0.051	-0.044	0.033	0.107	
0.602	2.922	0.02	2.943	11.610	-0.001	0.006	-0.014	-0.017	-0.012	-0.004	-0.078	-0.058	-0.049	0.030	0.102	
0.602	3.384	0.02	2.947	11.605	0.001	0.005	-0.013	-0.018	-0.012	-0.003	-0.078	-0.058	-0.051	0.027	0.104	
0.602	3.957	0.02	2.947	11.605	0.000	0.006	-0.013	-0.018	-0.013	-0.003	-0.079	-0.058	-0.051	0.028	0.109	
0.603	5.882	0.02	2.949	11.602	-0.001	0.007	-0.014	-0.018	-0.012	-0.003	-0.065	-0.048	-0.042	0.033	0.132	
0.603	7.782	0.02	2.950	11.601	0.002	0.007	-0.012	-0.019	-0.012	-0.001	-0.051	-0.035	-0.030	0.044	0.148	
0.601	3.390	8.04	2.935	11.623	-0.036	-0.029	-0.051	-0.058	-0.054	-0.053	-0.085	-0.068	-0.077	0.000	0.100	
0.601	3.427	4.00	2.940	11.615	-0.003	0.001	-0.024	-0.031	-0.027	-0.023	-0.080	-0.059	-0.071	-0.005	0.102	

Table 4. Continued

(c) $\theta = 90^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.951	0.963	0.974	0.986	0.997
1.199	1.483	-0.02	6.151	6.115	-0.061	-0.012	0.030	-0.065	-0.100	0.149	-0.184	-0.281	-0.187	-0.100	-0.034
1.199	1.983	0.00	6.151	6.117	-0.060	-0.012	0.030	-0.067	-0.101	0.149	-0.189	-0.312	-0.194	-0.114	-0.060
1.200	2.938	0.00	6.154	6.107	-0.060	-0.012	0.029	-0.066	-0.100	0.149	-0.190	-0.327	-0.201	-0.121	-0.070
1.198	3.413	0.00	6.150	6.118	-0.062	-0.011	0.028	-0.068	-0.101	0.148	-0.190	-0.329	-0.202	-0.121	-0.066
1.199	3.907	0.01	6.152	6.113	-0.061	-0.011	0.028	-0.068	-0.101	0.147	-0.190	-0.329	-0.203	-0.120	-0.059
1.199	5.894	0.03	6.151	6.115	-0.064	-0.014	0.026	-0.069	-0.102	0.145	-0.189	-0.315	-0.196	-0.103	-0.014
1.198	7.810	0.01	6.150	6.118	-0.064	-0.014	0.025	-0.070	-0.101	0.144	-0.183	-0.284	-0.189	-0.083	-0.027
1.200	3.430	8.00	6.154	6.109	0.097	0.060	0.003	-0.075	-0.099	0.272	-0.239	-0.326	-0.201	-0.120	-0.077
1.202	3.410	4.00	6.158	6.093	0.042	0.015	0.016	-0.076	-0.132	0.180	-0.166	-0.295	-0.214	-0.124	-0.072
0.899	1.480	0.02	4.958	8.769	0.008	0.025	0.014	-0.021	0.046	0.205	-0.007	-0.032	0.024	0.125	0.184
0.901	1.989	-0.01	4.971	8.747	0.006	0.024	0.014	-0.020	0.046	0.205	-0.012	-0.039	0.011	0.114	0.165
0.903	2.939	0.00	4.980	8.732	0.005	0.024	0.015	-0.020	0.046	0.204	-0.012	-0.039	0.013	0.114	0.165
0.899	3.377	0.00	4.960	8.767	0.006	0.023	0.016	-0.019	0.047	0.206	-0.011	-0.038	0.013	0.116	0.168
0.900	3.932	0.00	4.968	8.754	0.007	0.025	0.015	-0.020	0.046	0.207	-0.006	-0.030	0.020	0.125	0.188
0.899	5.811	0.01	4.960	8.768	0.007	0.024	0.015	-0.019	0.047	0.207	-0.005	-0.012	0.036	0.141	0.210
0.899	7.794	0.02	4.957	8.772	0.009	0.025	0.016	-0.019	-0.062	0.254	-0.011	-0.035	0.010	0.102	0.152
0.901	3.433	8.03	4.970	8.752	-0.051	-0.057	-0.088	-0.184	0.011	0.241	-0.012	-0.041	0.007	0.109	0.156
0.901	3.406	4.01	4.969	8.753	-0.010	-0.004	-0.019	-0.070	0.027	0.197	0.007	-0.009	0.049	0.145	0.187
0.800	1.483	-0.01	4.354	9.720	0.006	0.011	0.000	-0.038	0.028	0.199	0.003	-0.015	0.044	0.140	0.179
0.800	1.972	0.00	4.357	9.716	0.007	0.012	0.000	-0.036	0.027	0.198	0.000	-0.020	0.042	0.136	0.170
0.800	2.946	0.01	4.355	9.718	0.006	0.011	-0.001	-0.037	0.027	0.198	-0.001	-0.020	0.041	0.136	0.169
0.801	3.397	0.01	4.361	9.710	0.006	0.011	0.000	-0.036	0.027	0.199	-0.002	-0.021	0.042	0.137	0.172
0.801	3.914	0.01	4.360	9.711	0.008	0.012	0.000	-0.037	0.027	0.198	0.004	-0.011	0.048	0.146	0.190
0.802	5.871	0.03	4.368	9.700	0.006	0.010	-0.001	-0.035	0.029	0.200	0.013	0.004	0.060	0.157	0.209
0.802	7.776	0.01	4.365	9.704	0.007	0.012	0.000	-0.035	-0.013	0.278	-0.004	-0.027	0.026	0.117	0.157
0.800	3.450	8.03	4.358	9.717	-0.008	-0.020	-0.039	-0.101	0.022	0.228	-0.003	-0.027	0.022	0.122	0.158
0.800	3.405	4.00	4.354	9.723	0.009	0.005	-0.007	-0.047	0.022	0.190	0.015	-0.001	0.070	0.152	0.181
0.601	1.486	0.01	2.940	11.613	0.008	0.007	-0.004	-0.034	0.025	0.189	0.009	-0.010	0.063	0.144	0.167
0.602	1.961	0.01	2.945	11.607	0.009	0.007	-0.005	-0.035	0.023	0.190	0.005	-0.015	0.060	0.139	0.156
0.602	2.922	0.02	2.943	11.610	0.007	0.005	-0.005	-0.035	0.023	0.190	0.005	-0.017	0.058	0.138	0.156
0.602	3.384	0.02	2.947	11.605	0.007	0.005	-0.005	-0.036	0.022	0.190	0.003	-0.015	0.059	0.139	0.159
0.602	3.957	0.02	2.947	11.605	0.006	0.004	-0.005	-0.034	0.024	0.191	0.007	-0.007	0.064	0.148	0.178
0.603	5.882	0.02	2.949	11.602	0.007	0.005	-0.005	-0.032	0.025	0.192	0.014	0.005	0.074	0.158	0.196
0.603	7.782	0.02	2.950	11.601	0.007	0.005	-0.003	-0.032	0.025	0.192	-0.003	-0.029	0.044	0.129	0.148
0.601	3.390	8.04	2.935	11.623	0.020	0.002	-0.009	-0.055	0.013	0.276	0.000	-0.025	0.041	0.130	0.149
0.601	3.427	4.00	2.940	11.615	0.022	0.012	0.005	-0.026	0.031	0.210	0.000	-0.025	0.041	0.130	0.149

Table 4. Continued

(d) $\theta = 135^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.869
1.199	1.483	-0.02	6.151	6.115	-0.071	-0.084	-0.084	-0.072	-0.080	-0.108	-0.081	-0.086	-0.097	-0.123	-0.161
1.199	1.983	0.00	6.151	6.117	-0.071	-0.083	-0.084	-0.071	-0.082	-0.110	-0.081	-0.086	-0.097	-0.122	-0.161
1.200	2.938	0.00	6.154	6.107	-0.070	-0.083	-0.084	-0.070	-0.080	-0.110	-0.081	-0.086	-0.097	-0.122	-0.162
1.198	3.413	0.00	6.150	6.118	-0.072	-0.085	-0.085	-0.071	-0.082	-0.112	-0.082	-0.087	-0.097	-0.123	-0.163
1.199	3.907	0.01	6.152	6.113	-0.071	-0.085	-0.085	-0.071	-0.082	-0.111	-0.082	-0.086	-0.096	-0.122	-0.164
1.199	5.894	0.03	6.151	6.115	-0.073	-0.086	-0.086	-0.071	-0.083	-0.112	-0.082	-0.088	-0.098	-0.122	-0.165
1.198	7.810	0.01	6.150	6.118	-0.074	-0.085	-0.085	-0.070	-0.083	-0.113	-0.082	-0.088	-0.098	-0.123	-0.166
1.200	3.430	8.00	6.154	6.109	0.144	0.100	0.054	0.000	-0.037	-0.066	0.002	0.028	0.037	0.013	-0.048
1.202	3.410	4.00	6.158	6.093	0.057	0.037	0.000	-0.033	-0.050	-0.075	-0.049	-0.041	-0.037	-0.060	-0.111
0.899	1.480	0.02	4.958	8.769	-0.040	-0.034	-0.043	-0.032	-0.013	0.002	-0.004	-0.030	-0.042	-0.067	-0.112
0.901	1.989	-0.01	4.971	8.747	-0.043	-0.035	-0.044	-0.033	-0.013	0.002	-0.004	-0.030	-0.043	-0.068	-0.113
0.903	2.939	0.00	4.980	8.732	-0.041	-0.034	-0.041	-0.032	-0.012	0.004	-0.003	-0.029	-0.042	-0.067	-0.114
0.899	3.377	0.00	4.960	8.767	-0.039	-0.034	-0.043	-0.034	-0.013	0.002	-0.005	-0.031	-0.044	-0.069	-0.114
0.900	3.932	0.00	4.968	8.754	-0.041	-0.034	-0.043	-0.033	-0.013	0.004	-0.003	-0.029	-0.042	-0.067	-0.113
0.899	5.811	0.01	4.960	8.768	-0.040	-0.034	-0.043	-0.034	-0.013	0.003	-0.005	-0.031	-0.043	-0.068	-0.112
0.899	7.794	0.02	4.957	8.772	-0.040	-0.034	-0.043	-0.034	-0.013	0.004	-0.003	-0.029	-0.041	-0.066	-0.108
0.901	3.433	8.03	4.970	8.752	-0.071	-0.094	-0.125	-0.135	-0.103	-0.063	-0.042	-0.040	-0.030	-0.040	-0.067
0.901	3.406	4.01	4.969	8.753	-0.051	-0.056	-0.070	-0.065	-0.038	-0.012	-0.012	-0.028	-0.034	-0.055	-0.094
0.800	1.483	-0.01	4.354	9.720	-0.032	-0.039	-0.053	-0.044	-0.028	-0.015	-0.019	-0.040	-0.046	-0.065	-0.093
0.800	1.972	0.00	4.357	9.716	-0.031	-0.039	-0.053	-0.044	-0.027	-0.014	-0.019	-0.038	-0.045	-0.064	-0.092
0.800	2.946	0.01	4.355	9.718	-0.032	-0.039	-0.053	-0.045	-0.027	-0.014	-0.019	-0.039	-0.046	-0.065	-0.093
0.801	3.397	0.01	4.361	9.710	-0.033	-0.040	-0.054	-0.046	-0.028	-0.015	-0.020	-0.040	-0.047	-0.066	-0.093
0.801	3.914	0.01	4.360	9.711	-0.032	-0.039	-0.053	-0.046	-0.027	-0.014	-0.019	-0.039	-0.046	-0.066	-0.093
0.802	5.871	0.03	4.368	9.700	-0.032	-0.039	-0.053	-0.046	-0.027	-0.014	-0.019	-0.039	-0.046	-0.065	-0.092
0.802	7.776	0.01	4.365	9.704	-0.031	-0.038	-0.052	-0.045	-0.026	-0.012	-0.018	-0.038	-0.045	-0.064	-0.089
0.800	3.450	8.03	4.358	9.717	-0.022	-0.043	-0.067	-0.070	-0.046	-0.019	-0.005	-0.006	0.000	-0.015	-0.040
0.800	3.405	4.00	4.354	9.723	-0.018	-0.030	-0.048	-0.045	-0.022	-0.001	-0.001	-0.014	-0.019	-0.039	-0.068
0.601	1.486	0.01	2.940	11.613	-0.025	-0.035	-0.049	-0.041	-0.026	-0.016	-0.019	-0.034	-0.035	-0.049	-0.065
0.602	1.961	0.01	2.945	11.607	-0.024	-0.036	-0.050	-0.044	-0.028	-0.017	-0.019	-0.036	-0.035	-0.050	-0.066
0.602	2.922	0.02	2.943	11.610	-0.024	-0.036	-0.050	-0.045	-0.029	-0.017	-0.020	-0.036	-0.037	-0.051	-0.068
0.602	3.384	0.02	2.947	11.605	-0.025	-0.037	-0.051	-0.046	-0.029	-0.018	-0.022	-0.037	-0.038	-0.052	-0.068
0.602	3.957	0.02	2.947	11.605	-0.026	-0.037	-0.051	-0.046	-0.029	-0.017	-0.021	-0.037	-0.039	-0.053	-0.069
0.603	5.882	0.02	2.949	11.602	-0.026	-0.037	-0.051	-0.046	-0.029	-0.017	-0.021	-0.037	-0.039	-0.053	-0.067
0.603	7.782	0.02	2.950	11.601	-0.026	-0.035	-0.049	-0.043	-0.027	-0.015	-0.019	-0.035	-0.036	-0.051	-0.065
0.601	3.390	8.04	2.935	11.623	0.003	-0.018	-0.038	-0.043	-0.018	0.006	0.015	0.013	0.021	0.007	-0.011
0.601	3.427	4.00	2.940	11.615	0.000	-0.016	-0.032	-0.035	-0.012	0.007	0.007	-0.005	-0.006	-0.023	-0.039

Table 4. Continued

(d) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.199	1.483	-0.02	6.151	6.115	-0.154	-0.134	-0.167	-0.202	-0.211	-0.268	-0.335	-0.243	-0.143	-0.056
1.199	1.983	0.00	6.151	6.117	-0.155	-0.133	-0.167	-0.202	-0.212	-0.268	-0.340	-0.253	-0.167	-0.082
1.200	2.938	0.00	6.154	6.107	-0.157	-0.134	-0.167	-0.202	-0.212	-0.269	-0.345	-0.260	-0.181	-0.100
1.198	3.413	0.00	6.150	6.118	-0.156	-0.134	-0.169	-0.203	-0.214	-0.271	-0.346	-0.260	-0.180	-0.100
1.199	3.907	0.01	6.152	6.113	-0.157	-0.134	-0.169	-0.203	-0.213	-0.271	-0.347	-0.262	-0.180	-0.095
1.199	5.894	0.03	6.151	6.115	-0.158	-0.135	-0.170	-0.204	-0.215	-0.272	-0.347	-0.259	-0.153	-0.037
1.198	7.810	0.01	6.150	6.118	-0.157	-0.135	-0.170	-0.205	-0.216	-0.272	-0.343	-0.240	-0.101	0.009
1.200	3.430	8.00	6.154	6.109	-0.070	-0.105	-0.153	-0.173	-0.165	-0.225	-0.333	-0.256	-0.196	-0.117
1.202	3.410	4.00	6.158	6.093	-0.118	-0.110	-0.132	-0.161	-0.163	-0.223	-0.293	-0.229	-0.188	-0.111
0.899	1.480	0.02	4.958	8.769	-0.133	-0.153	-0.130	-0.096	-0.041	-0.075	0.013	0.083	0.136	0.159
0.901	1.989	-0.01	4.971	8.747	-0.136	-0.158	-0.135	-0.101	-0.046	-0.082	0.007	0.076	0.126	0.146
0.903	2.939	0.00	4.980	8.732	-0.137	-0.161	-0.138	-0.104	-0.048	-0.085	0.003	0.072	0.123	0.140
0.899	3.377	0.00	4.960	8.767	-0.136	-0.159	-0.136	-0.103	-0.048	-0.086	0.004	0.073	0.123	0.141
0.900	3.932	0.00	4.968	8.754	-0.136	-0.158	-0.135	-0.101	-0.047	-0.084	0.006	0.075	0.126	0.147
0.899	5.811	0.01	4.960	8.768	-0.134	-0.154	-0.131	-0.097	-0.041	-0.076	0.015	0.087	0.143	0.173
0.899	7.794	0.02	4.957	8.772	-0.129	-0.147	-0.122	-0.088	-0.032	-0.062	0.030	0.104	0.163	0.198
0.901	3.433	8.03	4.970	8.752	-0.084	-0.103	-0.096	-0.082	-0.041	-0.081	0.007	0.074	0.119	0.132
0.901	3.406	4.01	4.969	8.753	-0.115	-0.137	-0.122	-0.098	-0.050	-0.074	0.007	0.071	0.114	0.135
0.800	1.483	-0.01	4.354	9.720	-0.099	-0.107	-0.093	-0.075	-0.032	-0.069	0.014	0.082	0.135	0.158
0.800	1.972	0.00	4.357	9.716	-0.098	-0.107	-0.094	-0.077	-0.034	-0.073	0.008	0.076	0.127	0.148
0.800	2.946	0.01	4.355	9.718	-0.100	-0.109	-0.095	-0.079	-0.036	-0.076	0.006	0.073	0.123	0.140
0.801	3.397	0.01	4.361	9.710	-0.101	-0.109	-0.096	-0.079	-0.036	-0.076	0.006	0.073	0.123	0.140
0.801	3.914	0.01	4.360	9.711	-0.100	-0.109	-0.095	-0.079	-0.035	-0.076	0.007	0.075	0.125	0.144
0.802	5.871	0.03	4.368	9.700	-0.099	-0.108	-0.093	-0.075	-0.031	-0.069	0.016	0.086	0.139	0.168
0.802	7.776	0.01	4.365	9.704	-0.095	-0.103	-0.087	-0.069	-0.023	-0.057	0.029	0.102	0.158	0.193
0.800	3.450	8.03	4.358	9.717	-0.053	-0.069	-0.064	-0.059	-0.026	-0.069	0.011	0.075	0.118	0.132
0.800	3.405	4.00	4.354	9.723	-0.078	-0.093	-0.082	-0.072	-0.035	-0.068	0.009	0.071	0.113	0.133
0.601	1.486	0.01	2.940	11.613	-0.067	-0.071	-0.062	-0.052	-0.018	-0.053	0.021	0.083	0.131	0.150
0.602	1.961	0.01	2.945	11.607	-0.068	-0.073	-0.064	-0.055	-0.022	-0.059	0.014	0.074	0.118	0.129
0.602	2.922	0.02	2.943	11.610	-0.070	-0.076	-0.067	-0.058	-0.025	-0.062	0.010	0.070	0.113	0.120
0.602	3.384	0.02	2.947	11.605	-0.069	-0.075	-0.067	-0.058	-0.025	-0.063	0.011	0.071	0.114	0.122
0.602	3.957	0.02	2.947	11.605	-0.071	-0.077	-0.067	-0.059	-0.024	-0.063	0.011	0.071	0.115	0.126
0.603	5.882	0.02	2.949	11.602	-0.070	-0.075	-0.066	-0.056	-0.021	-0.058	0.019	0.082	0.129	0.153
0.603	7.782	0.02	2.950	11.601	-0.066	-0.073	-0.061	-0.051	-0.015	-0.048	0.033	0.097	0.147	0.179
0.601	3.390	8.04	2.935	11.623	-0.019	-0.035	-0.031	-0.034	-0.008	-0.050	0.025	0.078	0.111	0.118
0.601	3.427	4.00	2.940	11.615	-0.045	-0.059	-0.051	-0.048	-0.019	-0.053	0.020	0.073	0.104	0.118

Table 4. Continued

(e) $\theta = 180^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
1.199	1.483	-0.02	6.151	6.115	-0.030	-0.061	-0.088	-0.092	-0.089	-0.090	-0.111	-0.131	-0.142	-0.149	-0.165
1.199	1.983	0.00	6.151	6.117	-0.031	-0.060	-0.088	-0.093	-0.089	-0.090	-0.111	-0.131	-0.142	-0.149	-0.166
1.200	2.938	0.00	6.154	6.107	-0.030	-0.059	-0.088	-0.092	-0.088	-0.089	-0.112	-0.131	-0.141	-0.149	-0.166
1.198	3.413	0.00	6.150	6.118	-0.032	-0.060	-0.089	-0.094	-0.089	-0.090	-0.113	-0.131	-0.141	-0.150	-0.168
1.199	3.907	0.01	6.152	6.113	-0.031	-0.058	-0.088	-0.093	-0.089	-0.090	-0.113	-0.130	-0.140	-0.149	-0.168
1.199	5.894	0.03	6.151	6.115	-0.031	-0.057	-0.088	-0.093	-0.089	-0.091	-0.115	-0.131	-0.141	-0.150	-0.169
1.198	7.810	0.01	6.150	6.118	-0.031	-0.056	-0.087	-0.093	-0.088	-0.090	-0.114	-0.131	-0.140	-0.151	-0.170
1.200	3.430	8.00	6.154	6.109	0.183	0.137	0.102	0.079	0.052	0.020	-0.030	-0.063	-0.069	-0.065	-0.065
1.202	3.410	4.00	6.158	6.093	0.074	0.050	0.030	0.021	0.009	-0.013	-0.054	-0.086	-0.108	-0.120	-0.135
0.899	1.480	0.02	4.958	8.769	-0.050	-0.054	-0.050	-0.040	-0.034	-0.042	-0.055	-0.061	-0.070	-0.084	-0.096
0.901	1.989	-0.01	4.971	8.747	-0.053	-0.056	-0.052	-0.041	-0.034	-0.043	-0.055	-0.061	-0.071	-0.084	-0.097
0.903	2.939	0.00	4.980	8.732	-0.052	-0.055	-0.051	-0.040	-0.033	-0.042	-0.054	-0.061	-0.070	-0.085	-0.098
0.899	3.377	0.00	4.960	8.767	-0.050	-0.055	-0.052	-0.041	-0.035	-0.043	-0.055	-0.063	-0.072	-0.086	-0.098
0.900	3.932	0.00	4.968	8.754	-0.051	-0.055	-0.051	-0.040	-0.033	-0.042	-0.053	-0.060	-0.070	-0.084	-0.096
0.899	5.811	0.01	4.960	8.768	-0.050	-0.055	-0.052	-0.041	-0.035	-0.043	-0.054	-0.062	-0.071	-0.084	-0.096
0.899	7.794	0.02	4.957	8.772	-0.051	-0.056	-0.051	-0.040	-0.034	-0.042	-0.053	-0.060	-0.069	-0.081	-0.092
0.901	3.433	8.03	4.970	8.752	-0.039	-0.076	-0.089	-0.086	-0.082	-0.087	-0.087	-0.079	-0.072	-0.072	-0.084
0.901	3.406	4.01	4.969	8.753	-0.045	-0.065	-0.065	-0.055	-0.050	-0.056	-0.062	-0.065	-0.069	-0.077	-0.084
0.800	1.483	-0.01	4.354	9.720	-0.039	-0.055	-0.056	-0.049	-0.044	-0.053	-0.064	-0.067	-0.072	-0.078	-0.083
0.800	1.972	0.00	4.357	9.716	-0.039	-0.055	-0.056	-0.048	-0.044	-0.052	-0.063	-0.066	-0.071	-0.077	-0.083
0.800	2.946	0.01	4.355	9.718	-0.039	-0.055	-0.056	-0.048	-0.044	-0.053	-0.063	-0.067	-0.072	-0.078	-0.084
0.801	3.397	0.01	4.361	9.710	-0.039	-0.055	-0.056	-0.049	-0.044	-0.053	-0.063	-0.067	-0.072	-0.079	-0.084
0.801	3.914	0.01	4.360	9.711	-0.040	-0.056	-0.057	-0.049	-0.045	-0.054	-0.064	-0.068	-0.073	-0.079	-0.084
0.802	5.871	0.03	4.368	9.700	-0.039	-0.055	-0.056	-0.048	-0.044	-0.053	-0.063	-0.067	-0.072	-0.078	-0.083
0.802	7.776	0.01	4.365	9.704	-0.039	-0.056	-0.057	-0.049	-0.044	-0.053	-0.062	-0.067	-0.072	-0.078	-0.083
0.800	3.450	8.03	4.358	9.717	-0.001	-0.030	-0.036	-0.032	-0.032	-0.039	-0.042	-0.042	-0.043	-0.043	-0.045
0.800	3.405	4.00	4.354	9.723	-0.013	-0.038	-0.040	-0.033	-0.032	-0.039	-0.045	-0.050	-0.055	-0.058	-0.061
0.601	1.486	0.01	2.940	11.613	-0.033	-0.047	-0.048	-0.042	-0.036	-0.046	-0.056	-0.056	-0.060	-0.061	-0.062
0.602	1.961	0.01	2.945	11.607	-0.033	-0.048	-0.049	-0.042	-0.037	-0.047	-0.056	-0.058	-0.062	-0.062	-0.064
0.602	2.922	0.02	2.943	11.610	-0.034	-0.050	-0.050	-0.044	-0.039	-0.048	-0.056	-0.059	-0.063	-0.064	-0.065
0.602	3.384	0.02	2.947	11.605	-0.034	-0.050	-0.050	-0.044	-0.039	-0.048	-0.057	-0.059	-0.063	-0.065	-0.065
0.602	3.957	0.02	2.947	11.605	-0.035	-0.051	-0.051	-0.045	-0.040	-0.049	-0.057	-0.059	-0.063	-0.065	-0.065
0.603	5.882	0.02	2.949	11.602	-0.034	-0.052	-0.051	-0.044	-0.040	-0.049	-0.057	-0.060	-0.064	-0.065	-0.064
0.603	7.782	0.02	2.950	11.601	-0.032	-0.050	-0.049	-0.043	-0.039	-0.048	-0.055	-0.058	-0.062	-0.062	-0.062
0.601	3.390	8.04	2.935	11.623	0.023	-0.005	-0.007	-0.002	-0.003	-0.009	-0.014	-0.018	-0.021	-0.019	-0.017
0.601	3.427	4.00	2.940	11.615	0.001	-0.025	-0.024	-0.017	-0.018	-0.025	-0.030	-0.037	-0.042	-0.039	-0.038

Table 4. Continued

(e) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—								
					0.883	0.898	0.912	0.927	0.941	0.951	0.974	0.986	
1.199	1.483	-0.02	6.151	6.115	-0.183	-0.184	-0.194	-0.206	-0.231	-0.248	-0.286	-0.143	
1.199	1.983	0.00	6.151	6.117	-0.181	-0.183	-0.194	-0.207	-0.230	-0.248	-0.302	-0.188	
1.200	2.938	0.00	6.154	6.107	-0.180	-0.183	-0.195	-0.208	-0.232	-0.250	-0.307	-0.221	
1.198	3.413	0.00	6.150	6.118	-0.181	-0.183	-0.195	-0.209	-0.231	-0.249	-0.307	-0.223	
1.199	3.907	0.01	6.152	6.113	-0.180	-0.183	-0.195	-0.209	-0.231	-0.250	-0.308	-0.219	
1.199	5.894	0.03	6.151	6.115	-0.180	-0.184	-0.196	-0.211	-0.233	-0.252	-0.306	-0.178	
1.198	7.810	0.01	6.150	6.118	-0.180	-0.185	-0.197	-0.212	-0.232	-0.251	-0.285	-0.104	
1.200	3.430	8.00	6.154	6.109	-0.058	-0.067	-0.105	-0.148	-0.205	-0.213	-0.333	-0.264	
1.202	3.410	4.00	6.158	6.093	-0.144	-0.141	-0.153	-0.159	-0.187	-0.202	-0.280	-0.249	
0.899	1.480	0.02	4.958	8.769	-0.117	-0.121	-0.125	-0.113	-0.126	-0.151	0.095	0.144	
0.901	1.989	-0.01	4.971	8.747	-0.121	-0.126	-0.131	-0.119	-0.134	-0.162	0.085	0.134	
0.903	2.939	0.00	4.980	8.732	-0.123	-0.128	-0.133	-0.121	-0.137	-0.166	0.082	0.130	
0.899	3.377	0.00	4.960	8.767	-0.122	-0.127	-0.132	-0.121	-0.138	-0.167	0.081	0.130	
0.900	3.932	0.00	4.968	8.754	-0.122	-0.126	-0.131	-0.119	-0.135	-0.165	0.084	0.133	
0.899	5.811	0.01	4.960	8.768	-0.120	-0.123	-0.127	-0.114	-0.128	-0.153	0.096	0.150	
0.899	7.794	0.02	4.957	8.772	-0.115	-0.116	-0.118	-0.104	-0.115	-0.133	0.115	0.170	
0.901	3.433	8.03	4.970	8.752	-0.087	-0.090	-0.105	-0.106	-0.145	-0.211	0.086	0.132	
0.901	3.406	4.01	4.969	8.753	-0.109	-0.113	-0.123	-0.117	-0.144	-0.188	0.080	0.124	
0.800	1.483	-0.01	4.354	9.720	-0.092	-0.091	-0.097	-0.094	-0.115	-0.139	0.087	0.138	
0.800	1.972	0.00	4.357	9.716	-0.093	-0.092	-0.098	-0.097	-0.120	-0.146	0.080	0.129	
0.800	2.946	0.01	4.355	9.718	-0.094	-0.093	-0.101	-0.099	-0.123	-0.150	0.076	0.125	
0.801	3.397	0.01	4.361	9.710	-0.095	-0.094	-0.101	-0.099	-0.123	-0.149	0.078	0.127	
0.801	3.914	0.01	4.360	9.711	-0.095	-0.093	-0.100	-0.098	-0.123	-0.149	0.078	0.127	
0.802	5.871	0.03	4.368	9.700	-0.093	-0.092	-0.098	-0.094	-0.117	-0.139	0.089	0.143	
0.802	7.776	0.01	4.365	9.704	-0.090	-0.087	-0.091	-0.086	-0.105	-0.123	0.106	0.163	
0.800	3.450	8.03	4.358	9.717	-0.063	-0.064	-0.078	-0.085	-0.128	-0.173	0.078	0.127	
0.800	3.405	4.00	4.354	9.723	-0.081	-0.080	-0.091	-0.093	-0.128	-0.163	0.074	0.120	
0.601	1.486	0.01	2.940	11.613	-0.068	-0.065	-0.073	-0.073	-0.097	-0.116	0.080	0.130	
0.602	1.961	0.01	2.945	11.607	-0.071	-0.068	-0.078	-0.078	-0.104	-0.125	0.069	0.116	
0.602	2.922	0.02	2.943	11.610	-0.073	-0.071	-0.079	-0.081	-0.107	-0.129	0.064	0.110	
0.602	3.384	0.02	2.947	11.605	-0.073	-0.071	-0.079	-0.081	-0.107	-0.129	0.065	0.111	
0.602	3.957	0.02	2.947	11.605	-0.075	-0.072	-0.080	-0.080	-0.108	-0.129	0.065	0.112	
0.603	5.882	0.02	2.949	11.602	-0.074	-0.070	-0.077	-0.076	-0.103	-0.122	0.077	0.129	
0.603	7.782	0.02	2.950	11.601	-0.073	-0.067	-0.073	-0.071	-0.095	-0.109	0.093	0.150	
0.601	3.390	8.04	2.935	11.623	-0.042	-0.039	-0.053	-0.059	-0.105	-0.139	0.069	0.115	
0.601	3.427	4.00	2.940	11.615	-0.063	-0.057	-0.067	-0.069	-0.109	-0.136	0.064	0.109	

Table 4. Continued

(f) $\theta = 225^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.199	1.483	-0.02	6.151	6.115	-0.209	-0.213	-0.189	-0.145	-0.211	-0.322	-0.443	-0.214	-0.097	-0.041	
1.199	1.983	0.00	6.151	6.117	-0.207	-0.212	-0.189	-0.145	-0.211	-0.323	-0.443	-0.286	-0.138	-0.076	
1.200	2.938	0.00	6.154	6.107	-0.206	-0.212	-0.189	-0.147	-0.214	-0.325	-0.443	-0.352	-0.155	-0.093	
1.198	3.413	0.00	6.150	6.118	-0.206	-0.213	-0.190	-0.144	-0.212	-0.325	-0.444	-0.355	-0.153	-0.092	
1.199	3.907	0.01	6.152	6.113	-0.205	-0.213	-0.189	-0.145	-0.213	-0.326	-0.444	-0.337	-0.147	-0.088	
1.199	5.894	0.03	6.151	6.115	-0.205	-0.214	-0.191	-0.146	-0.215	-0.328	-0.445	-0.251	-0.115	-0.057	
1.198	7.810	0.01	6.150	6.118	-0.206	-0.215	-0.191	-0.145	-0.215	-0.329	-0.445	-0.166	-0.074	-0.020	
1.200	3.430	8.00	6.154	6.109	-0.074	-0.067	-0.054	-0.054	-0.178	-0.300	-0.446	-0.433	-0.233	-0.133	
1.202	3.410	4.00	6.158	6.093	-0.163	-0.163	-0.139	-0.105	-0.184	-0.297	-0.424	-0.233	-0.185	-0.149	
0.899	1.480	0.02	4.958	8.769	-0.125	-0.107	-0.086	-0.090	-0.169	-0.203	0.042	0.109	0.150	0.177	
0.901	1.989	-0.01	4.971	8.747	-0.131	-0.112	-0.092	-0.097	-0.179	-0.216	0.033	0.099	0.142	0.167	
0.903	2.939	0.00	4.980	8.732	-0.133	-0.114	-0.094	-0.100	-0.185	-0.226	0.026	0.094	0.139	0.162	
0.899	3.377	0.00	4.960	8.767	-0.133	-0.114	-0.094	-0.100	-0.186	-0.228	0.025	0.094	0.140	0.162	
0.900	3.932	0.00	4.968	8.754	-0.131	-0.113	-0.093	-0.099	-0.183	-0.226	0.027	0.097	0.142	0.166	
0.899	5.811	0.01	4.960	8.768	-0.130	-0.110	-0.089	-0.094	-0.173	-0.208	0.037	0.106	0.150	0.181	
0.899	7.794	0.02	4.957	8.772	-0.124	-0.102	-0.079	-0.081	-0.152	-0.177	0.056	0.120	0.162	0.199	
0.901	3.433	8.03	4.970	8.752	-0.108	-0.088	-0.074	-0.094	-0.214	-0.337	0.015	0.103	0.149	0.166	
0.901	3.406	4.01	4.969	8.753	-0.125	-0.105	-0.088	-0.101	-0.203	-0.277	0.017	0.092	0.134	0.154	
0.800	1.483	-0.01	4.354	9.720	-0.103	-0.084	-0.068	-0.079	-0.163	-0.205	0.028	0.110	0.154	0.180	
0.800	1.972	0.00	4.357	9.716	-0.105	-0.086	-0.071	-0.083	-0.171	-0.218	0.019	0.102	0.147	0.170	
0.800	2.946	0.01	4.355	9.718	-0.106	-0.088	-0.073	-0.087	-0.176	-0.225	0.013	0.098	0.144	0.164	
0.801	3.397	0.01	4.361	9.710	-0.107	-0.089	-0.074	-0.087	-0.176	-0.224	0.013	0.099	0.143	0.166	
0.801	3.914	0.01	4.360	9.711	-0.106	-0.088	-0.072	-0.086	-0.174	-0.223	0.014	0.099	0.145	0.168	
0.802	5.871	0.03	4.368	9.700	-0.105	-0.086	-0.069	-0.080	-0.163	-0.204	0.026	0.109	0.154	0.184	
0.802	7.776	0.01	4.365	9.704	-0.101	-0.079	-0.061	-0.069	-0.144	-0.177	0.045	0.122	0.166	0.204	
0.800	3.450	8.03	4.358	9.717	-0.084	-0.066	-0.056	-0.083	-0.200	-0.288	-0.002	0.100	0.150	0.169	
0.800	3.405	4.00	4.354	9.723	-0.099	-0.079	-0.066	-0.087	-0.189	-0.257	0.002	0.091	0.136	0.159	
0.601	1.486	0.01	2.940	11.613	-0.082	-0.066	-0.054	-0.067	-0.149	-0.189	0.014	0.104	0.152	0.174	
0.602	1.961	0.01	2.945	11.607	-0.086	-0.069	-0.059	-0.073	-0.159	-0.203	0.003	0.092	0.139	0.155	
0.602	2.922	0.02	2.943	11.610	-0.089	-0.072	-0.061	-0.077	-0.164	-0.207	-0.002	0.088	0.134	0.148	
0.602	3.384	0.02	2.947	11.605	-0.089	-0.073	-0.062	-0.078	-0.163	-0.207	-0.003	0.087	0.135	0.149	
0.602	3.957	0.02	2.947	11.605	-0.091	-0.074	-0.063	-0.078	-0.163	-0.206	-0.001	0.088	0.136	0.154	
0.603	5.882	0.02	2.949	11.602	-0.090	-0.071	-0.059	-0.072	-0.153	-0.193	0.011	0.099	0.148	0.177	
0.603	7.782	0.02	2.950	11.601	-0.087	-0.067	-0.053	-0.065	-0.140	-0.173	0.026	0.114	0.165	0.201	
0.601	3.390	8.04	2.935	11.623	-0.068	-0.048	-0.043	-0.071	-0.174	-0.240	-0.017	0.084	0.136	0.154	
0.601	3.427	4.00	2.940	11.615	-0.083	-0.062	-0.054	-0.077	-0.169	-0.225	-0.013	0.079	0.127	0.148	

Table 4. Continued

(g) $\theta = 240^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986
1.199	1.483	-0.02	6.151	6.115	-0.221	-0.164	-0.128	-0.166	-0.424	-0.428	-0.222	-0.133
1.199	1.983	0.00	6.151	6.117	-0.222	-0.164	-0.127	-0.166	-0.424	-0.525	-0.247	-0.170
1.200	2.938	0.00	6.154	6.107	-0.223	-0.164	-0.129	-0.168	-0.426	-0.547	-0.266	-0.172
1.198	3.413	0.00	6.150	6.118	-0.226	-0.164	-0.127	-0.166	-0.425	-0.548	-0.267	-0.169
1.199	3.907	0.01	6.152	6.113	-0.226	-0.164	-0.128	-0.168	-0.426	-0.547	-0.262	-0.163
1.199	5.894	0.03	6.151	6.115	-0.227	-0.165	-0.129	-0.169	-0.426	-0.543	-0.224	-0.123
1.198	7.810	0.01	6.150	6.118	-0.229	-0.165	-0.129	-0.168	-0.427	-0.461	-0.177	-0.081
1.200	3.430	8.00	6.154	6.109	-0.076	-0.037	-0.034	-0.115	-0.405	-0.559	-0.305	-0.280
1.202	3.410	4.00	6.158	6.093	-0.172	-0.114	-0.085	-0.136	-0.405	-0.332	-0.247	-0.226
0.899	1.480	0.02	4.958	8.769	-0.110	-0.075	-0.071	-0.114	-0.215	0.055	0.094	0.142
0.901	1.989	-0.01	4.971	8.747	-0.116	-0.082	-0.079	-0.126	-0.229	0.044	0.083	0.136
0.903	2.939	0.00	4.980	8.732	-0.118	-0.085	-0.082	-0.132	-0.242	0.037	0.078	0.133
0.899	3.377	0.00	4.960	8.767	-0.117	-0.085	-0.083	-0.134	-0.247	0.038	0.080	0.133
0.900	3.932	0.00	4.968	8.754	-0.115	-0.083	-0.081	-0.130	-0.243	0.040	0.082	0.136
0.899	5.811	0.01	4.960	8.768	-0.112	-0.079	-0.076	-0.120	-0.228	0.051	0.093	0.142
0.899	7.794	0.02	4.957	8.772	-0.102	-0.069	-0.061	-0.097	-0.190	0.069	0.104	0.150
0.901	3.433	8.03	4.970	8.752	-0.092	-0.069	-0.081	-0.163	-0.368	0.040	0.100	0.148
0.901	3.406	4.01	4.969	8.753	-0.107	-0.080	-0.084	-0.148	-0.304	0.037	0.081	0.128
0.800	1.483	-0.01	4.354	9.720	-0.088	-0.059	-0.062	-0.115	-0.248	0.053	0.107	0.151
0.800	1.972	0.00	4.357	9.716	-0.090	-0.062	-0.067	-0.124	-0.266	0.044	0.101	0.147
0.800	2.946	0.01	4.355	9.718	-0.092	-0.065	-0.071	-0.130	-0.278	0.038	0.099	0.145
0.801	3.397	0.01	4.361	9.710	-0.092	-0.065	-0.071	-0.130	-0.277	0.039	0.100	0.145
0.801	3.914	0.01	4.360	9.711	-0.090	-0.064	-0.070	-0.128	-0.275	0.040	0.099	0.145
0.802	5.871	0.03	4.368	9.700	-0.088	-0.061	-0.063	-0.115	-0.251	0.052	0.107	0.151
0.802	7.776	0.01	4.365	9.704	-0.080	-0.052	-0.051	-0.094	-0.214	0.069	0.116	0.158
0.800	3.450	8.03	4.358	9.717	-0.070	-0.052	-0.069	-0.156	-0.370	0.033	0.110	0.157
0.800	3.405	4.00	4.354	9.723	-0.081	-0.060	-0.071	-0.141	-0.324	0.034	0.094	0.139
0.601	1.486	0.01	2.940	11.613	-0.068	-0.045	-0.053	-0.109	-0.257	0.042	0.118	0.162
0.602	1.961	0.01	2.945	11.607	-0.071	-0.050	-0.060	-0.120	-0.276	0.029	0.108	0.151
0.602	2.922	0.02	2.943	11.610	-0.073	-0.053	-0.063	-0.125	-0.284	0.024	0.104	0.147
0.602	3.384	0.02	2.947	11.605	-0.074	-0.054	-0.064	-0.126	-0.283	0.025	0.104	0.146
0.602	3.957	0.02	2.947	11.605	-0.073	-0.054	-0.065	-0.125	-0.283	0.026	0.106	0.148
0.603	5.882	0.02	2.949	11.602	-0.069	-0.050	-0.058	-0.115	-0.265	0.039	0.115	0.158
0.603	7.782	0.02	2.950	11.601	-0.063	-0.044	-0.049	-0.099	-0.239	0.056	0.128	0.170
0.601	3.390	8.04	2.935	11.623	-0.046	-0.039	-0.058	-0.139	-0.342	0.018	0.109	0.154
0.601	3.427	4.00	2.940	11.615	-0.057	-0.048	-0.063	-0.132	-0.321	0.021	0.102	0.142

Table 4. Continued

(h) $\theta = 255^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.941	0.963	0.974	0.986	0.997
1.199	1.483	-0.02	6.151	6.115	-0.173	-0.253	-0.230	-0.188	-0.115
1.199	1.983	0.00	6.151	6.117	-0.173	-0.281	-0.262	-0.209	-0.139
1.200	2.938	0.00	6.154	6.107	-0.176	-0.302	-0.285	-0.223	-0.155
1.198	3.413	0.00	6.150	6.118	-0.174	-0.304	-0.287	-0.224	-0.155
1.199	3.907	0.01	6.152	6.113	-0.175	-0.301	-0.287	-0.227	-0.158
1.199	5.894	0.03	6.151	6.115	-0.177	-0.295	-0.275	-0.209	-0.134
1.198	7.810	0.01	6.150	6.118	-0.177	-0.273	-0.242	-0.171	-0.089
1.200	3.430	8.00	6.154	6.109	-0.118	-0.289	-0.292	-0.267	-0.218
1.202	3.410	4.00	6.158	6.093	-0.144	-0.352	-0.230	-0.171	-0.122
0.899	1.480	0.02	4.958	8.769	-0.087	0.054	0.093	0.136	0.176
0.901	1.989	-0.01	4.971	8.747	-0.100	0.044	0.091	0.132	0.169
0.903	2.939	0.00	4.980	8.732	-0.108	0.039	0.089	0.130	0.164
0.899	3.377	0.00	4.960	8.767	-0.109	0.039	0.088	0.130	0.165
0.900	3.932	0.00	4.968	8.754	-0.106	0.040	0.089	0.131	0.166
0.899	5.811	0.01	4.960	8.768	-0.094	0.049	0.092	0.136	0.176
0.899	7.794	0.02	4.957	8.772	-0.069	0.063	0.100	0.142	0.186
0.901	3.433	8.03	4.970	8.752	-0.147	0.034	0.091	0.140	0.172
0.901	3.406	4.01	4.969	8.753	-0.123	0.033	0.077	0.118	0.152
0.800	1.483	-0.01	4.354	9.720	-0.094	0.051	0.095	0.145	0.182
0.800	1.972	0.00	4.357	9.716	-0.105	0.043	0.092	0.141	0.176
0.800	2.946	0.01	4.355	9.718	-0.113	0.040	0.090	0.140	0.173
0.801	3.397	0.01	4.361	9.710	-0.111	0.038	0.089	0.139	0.172
0.801	3.914	0.01	4.360	9.711	-0.109	0.040	0.090	0.140	0.173
0.802	5.871	0.03	4.368	9.700	-0.093	0.046	0.091	0.142	0.182
0.802	7.776	0.01	4.365	9.704	-0.069	0.062	0.101	0.149	0.192
0.800	3.450	8.03	4.358	9.717	-0.146	0.040	0.101	0.150	0.179
0.800	3.405	4.00	4.354	9.723	-0.124	0.038	0.090	0.132	0.162
0.601	1.486	0.01	2.940	11.613	-0.098	0.058	0.114	0.159	0.183
0.602	1.961	0.01	2.945	11.607	-0.111	0.047	0.105	0.149	0.165
0.602	2.922	0.02	2.943	11.610	-0.116	0.043	0.103	0.144	0.161
0.602	3.384	0.02	2.947	11.605	-0.116	0.043	0.103	0.145	0.161
0.602	3.957	0.02	2.947	11.605	-0.115	0.044	0.103	0.144	0.163
0.603	5.882	0.02	2.949	11.602	-0.104	0.055	0.112	0.154	0.180
0.603	7.782	0.02	2.950	11.601	-0.086	0.070	0.122	0.165	0.196
0.601	3.390	8.04	2.935	11.623	-0.135	0.044	0.115	0.152	0.164
0.601	3.427	4.00	2.940	11.615	-0.123	0.042	0.107	0.140	0.154

Table 4. Continued

(i) $\theta = 270^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.951	0.963	0.974	0.986	
1.199	1.483	-0.02	6.151	6.115	-0.345	-0.185	-0.218	-0.205	
1.199	1.983	0.00	6.151	6.117	-0.372	-0.220	-0.254	-0.230	
1.200	2.938	0.00	6.154	6.107	-0.379	-0.248	-0.280	-0.246	
1.198	3.413	0.00	6.150	6.118	-0.380	-0.249	-0.282	-0.247	
1.199	3.907	0.01	6.152	6.113	-0.380	-0.245	-0.280	-0.248	
1.199	5.894	0.03	6.151	6.115	-0.379	-0.233	-0.267	-0.236	
1.198	7.810	0.01	6.150	6.118	-0.358	-0.200	-0.235	-0.203	
1.200	3.430	8.00	6.154	6.109	-0.373	-0.227	-0.273	-0.276	
1.202	3.410	4.00	6.158	6.093	-0.386	-0.230	-0.197	-0.180	
0.899	1.480	0.02	4.958	8.769	-0.004	0.052	0.093	0.134	
0.901	1.989	-0.01	4.971	8.747	-0.020	0.042	0.089	0.129	
0.903	2.939	0.00	4.980	8.732	-0.031	0.037	0.088	0.127	
0.899	3.377	0.00	4.960	8.767	-0.031	0.036	0.087	0.127	
0.900	3.932	0.00	4.968	8.754	-0.030	0.038	0.088	0.128	
0.899	5.811	0.01	4.960	8.768	-0.012	0.046	0.091	0.133	
0.899	7.794	0.02	4.957	8.772	0.014	0.060	0.098	0.140	
0.901	3.433	8.03	4.970	8.752	-0.062	0.029	0.092	0.136	
0.901	3.406	4.01	4.969	8.753	-0.037	0.027	0.075	0.115	
0.800	1.483	-0.01	4.354	9.720	-0.023	0.046	0.095	0.143	
0.800	1.972	0.00	4.357	9.716	-0.037	0.038	0.091	0.141	
0.800	2.946	0.01	4.355	9.718	-0.046	0.034	0.089	0.139	
0.801	3.397	0.01	4.361	9.710	-0.045	0.032	0.087	0.138	
0.801	3.914	0.01	4.360	9.711	-0.046	0.035	0.090	0.139	
0.802	5.871	0.03	4.368	9.700	-0.019	0.039	0.089	0.141	
0.802	7.776	0.01	4.365	9.704	0.007	0.055	0.100	0.147	
0.800	3.450	8.03	4.358	9.717	-0.085	0.033	0.101	0.151	
0.800	3.405	4.00	4.354	9.723	-0.060	0.030	0.088	0.131	
0.601	1.486	0.01	2.940	11.613	-0.060	0.055	0.116	0.159	
0.602	1.961	0.01	2.945	11.607	-0.075	0.043	0.108	0.150	
0.602	2.922	0.02	2.943	11.610	-0.082	0.039	0.105	0.147	
0.602	3.384	0.02	2.947	11.605	-0.083	0.039	0.105	0.147	
0.602	3.957	0.02	2.947	11.605	-0.082	0.039	0.104	0.148	
0.603	5.882	0.02	2.949	11.602	-0.068	0.049	0.114	0.156	
0.603	7.782	0.02	2.950	11.601	-0.043	0.063	0.123	0.166	
0.601	3.390	8.04	2.935	11.623	-0.108	0.035	0.116	0.155	
0.601	3.427	4.00	2.940	11.615	-0.091	0.033	0.106	0.143	

Table 4. Continued

(i) $\theta = 285^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—						
					0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.199	1.483	-0.02	6.151	6.115	-0.152	-0.229	-0.507	-0.260	-0.181	-0.143	-0.065
1.199	1.983	0.00	6.151	6.117	-0.153	-0.231	-0.510	-0.327	-0.224	-0.169	-0.088
1.200	2.938	0.00	6.154	6.107	-0.154	-0.231	-0.510	-0.379	-0.251	-0.185	-0.100
1.198	3.413	0.00	6.150	6.118	-0.155	-0.232	-0.511	-0.378	-0.254	-0.187	-0.098
1.199	3.907	0.01	6.152	6.113	-0.155	-0.232	-0.511	-0.372	-0.251	-0.187	-0.093
1.199	5.894	0.03	6.151	6.115	-0.156	-0.234	-0.511	-0.347	-0.228	-0.163	-0.054
1.198	7.810	0.01	6.150	6.118	-0.157	-0.235	-0.511	-0.278	-0.184	-0.129	-0.020
1.200	3.430	8.00	6.154	6.109	-0.202	-0.263	-0.502	-0.360	-0.242	-0.213	-0.145
1.202	3.410	4.00	6.158	6.093	-0.218	-0.284	-0.434	-0.219	-0.246	-0.253	-0.211
0.899	1.480	0.02	4.958	8.769	-0.160	-0.094	-0.013	0.040	0.077	0.136	0.186
0.901	1.989	-0.01	4.971	8.747	-0.183	-0.110	-0.017	0.027	0.065	0.131	0.181
0.903	2.939	0.00	4.980	8.732	-0.199	-0.124	-0.025	0.017	0.059	0.130	0.179
0.899	3.377	0.00	4.960	8.767	-0.197	-0.123	-0.027	0.018	0.060	0.129	0.179
0.900	3.932	0.00	4.968	8.754	-0.196	-0.122	-0.027	0.022	0.064	0.131	0.179
0.899	5.811	0.01	4.960	8.768	-0.175	-0.105	-0.022	0.034	0.076	0.135	0.181
0.899	7.794	0.02	4.957	8.772	-0.134	-0.069	-0.001	0.053	0.090	0.144	0.189
0.901	3.433	8.03	4.970	8.752	-0.187	-0.147	-0.117	0.030	0.089	0.140	0.164
0.901	3.406	4.01	4.969	8.753	-0.185	-0.114	-0.040	0.016	0.063	0.123	0.163
0.800	1.483	-0.01	4.354	9.720	-0.137	-0.120	-0.064	0.044	0.091	0.143	0.191
0.800	1.972	0.00	4.357	9.716	-0.149	-0.134	-0.078	0.033	0.086	0.141	0.185
0.800	2.946	0.01	4.355	9.718	-0.157	-0.146	-0.089	0.029	0.083	0.140	0.184
0.801	3.397	0.01	4.361	9.710	-0.157	-0.144	-0.089	0.028	0.084	0.139	0.183
0.801	3.914	0.01	4.360	9.711	-0.155	-0.142	-0.085	0.029	0.084	0.140	0.183
0.802	5.871	0.03	4.368	9.700	-0.137	-0.119	-0.068	0.042	0.092	0.143	0.186
0.802	7.776	0.01	4.365	9.704	-0.112	-0.086	-0.034	0.055	0.100	0.150	0.194
0.800	3.450	8.03	4.358	9.717	-0.166	-0.173	-0.152	0.033	0.102	0.153	0.180
0.800	3.405	4.00	4.354	9.723	-0.151	-0.134	-0.078	0.018	0.076	0.136	0.174
0.601	1.486	0.01	2.940	11.613	-0.130	-0.140	-0.124	0.050	0.108	0.157	0.190
0.602	1.961	0.01	2.945	11.607	-0.141	-0.154	-0.143	0.041	0.100	0.147	0.176
0.602	2.922	0.02	2.943	11.610	-0.146	-0.161	-0.149	0.037	0.097	0.144	0.172
0.602	3.384	0.02	2.947	11.605	-0.147	-0.161	-0.149	0.035	0.097	0.144	0.171
0.602	3.957	0.02	2.947	11.605	-0.148	-0.161	-0.149	0.037	0.099	0.146	0.172
0.603	5.882	0.02	2.949	11.602	-0.135	-0.144	-0.130	0.044	0.105	0.155	0.184
0.603	7.782	0.02	2.950	11.601	-0.118	-0.122	-0.102	0.059	0.116	0.166	0.196
0.601	3.390	8.04	2.935	11.623	-0.168	-0.184	-0.185	0.031	0.109	0.157	0.173
0.601	3.427	4.00	2.940	11.615	-0.149	-0.154	-0.139	0.028	0.095	0.145	0.165

Table 4. Continued

(k) $\theta = 300^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.199	1.483	-0.02	6.151	6.115	-0.163	-0.168	-0.159	-0.207	-0.206	-0.210	-0.181	-0.131	-0.034	
1.199	1.983	0.00	6.151	6.117	-0.164	-0.169	-0.161	-0.210	-0.249	-0.245	-0.203	-0.151	-0.055	
1.200	2.938	0.00	6.154	6.107	-0.165	-0.169	-0.162	-0.210	-0.280	-0.270	-0.217	-0.163	-0.068	
1.198	3.413	0.00	6.150	6.118	-0.166	-0.170	-0.164	-0.211	-0.280	-0.270	-0.219	-0.163	-0.064	
1.199	3.907	0.01	6.152	6.113	-0.166	-0.170	-0.164	-0.211	-0.273	-0.271	-0.223	-0.163	-0.053	
1.199	5.894	0.03	6.151	6.115	-0.167	-0.170	-0.165	-0.213	-0.259	-0.258	-0.207	-0.146	-0.007	
1.198	7.810	0.01	6.150	6.118	-0.169	-0.171	-0.167	-0.214	-0.226	-0.221	-0.168	-0.120	0.023	
1.200	3.430	8.00	6.154	6.109	-0.233	-0.252	-0.213	-0.245	-0.250	-0.272	-0.262	-0.213	-0.075	
1.202	3.410	4.00	6.158	6.093	-0.230	-0.236	-0.224	-0.269	-0.360	-0.218	-0.173	-0.125	-0.067	
0.899	1.480	0.02	4.958	8.769	-0.332	-0.281	-0.204	-0.132	0.068	0.101	0.135	0.171	0.124	
0.901	1.989	-0.01	4.971	8.747	-0.334	-0.294	-0.228	-0.151	0.058	0.098	0.131	0.166	0.111	
0.903	2.939	0.00	4.980	8.732	-0.334	-0.301	-0.246	-0.167	0.053	0.095	0.127	0.161	0.111	
0.899	3.377	0.00	4.960	8.767	-0.338	-0.303	-0.243	-0.165	0.053	0.093	0.126	0.161	0.112	
0.900	3.932	0.00	4.968	8.754	-0.337	-0.301	-0.242	-0.164	0.054	0.094	0.127	0.163	0.115	
0.899	5.811	0.01	4.960	8.768	-0.335	-0.290	-0.219	-0.145	0.061	0.096	0.132	0.170	0.128	
0.899	7.794	0.02	4.957	8.772	-0.325	-0.264	-0.177	-0.109	0.074	0.103	0.139	0.180	0.139	
0.901	3.433	8.03	4.970	8.752	-0.400	-0.339	-0.221	-0.159	0.054	0.081	0.104	0.134	0.096	
0.901	3.406	4.01	4.969	8.753	-0.357	-0.308	-0.227	-0.149	0.044	0.079	0.109	0.148	0.101	
0.800	1.483	-0.01	4.354	9.720	-0.279	-0.203	-0.160	-0.153	0.057	0.100	0.143	0.182	0.133	
0.800	1.972	0.00	4.357	9.716	-0.288	-0.212	-0.170	-0.166	0.050	0.096	0.138	0.178	0.121	
0.800	2.946	0.01	4.355	9.718	-0.294	-0.220	-0.179	-0.176	0.046	0.093	0.136	0.175	0.119	
0.801	3.397	0.01	4.361	9.710	-0.296	-0.221	-0.178	-0.174	0.044	0.092	0.136	0.174	0.119	
0.801	3.914	0.01	4.360	9.711	-0.295	-0.220	-0.176	-0.172	0.045	0.092	0.135	0.175	0.122	
0.802	5.871	0.03	4.368	9.700	-0.282	-0.206	-0.160	-0.152	0.053	0.094	0.138	0.181	0.135	
0.802	7.776	0.01	4.365	9.704	-0.261	-0.184	-0.136	-0.123	0.069	0.103	0.144	0.190	0.147	
0.800	3.450	8.03	4.358	9.717	-0.339	-0.236	-0.182	-0.194	0.045	0.101	0.135	0.163	0.105	
0.800	3.405	4.00	4.354	9.723	-0.302	-0.222	-0.169	-0.161	0.040	0.088	0.123	0.165	0.108	
0.601	1.486	0.01	2.940	11.613	-0.211	-0.160	-0.143	-0.157	0.056	0.115	0.153	0.184	0.130	
0.602	1.961	0.01	2.945	11.607	-0.220	-0.171	-0.155	-0.172	0.043	0.105	0.142	0.169	0.102	
0.602	2.922	0.02	2.943	11.610	-0.223	-0.175	-0.159	-0.178	0.038	0.101	0.138	0.163	0.098	
0.602	3.384	0.02	2.947	11.605	-0.224	-0.175	-0.160	-0.178	0.036	0.101	0.137	0.164	0.099	
0.602	3.957	0.02	2.947	11.605	-0.225	-0.177	-0.160	-0.178	0.038	0.101	0.137	0.165	0.104	
0.603	5.882	0.02	2.949	11.602	-0.216	-0.168	-0.148	-0.163	0.051	0.110	0.145	0.182	0.125	
0.603	7.782	0.02	2.950	11.601	-0.204	-0.154	-0.132	-0.142	0.069	0.120	0.156	0.198	0.141	
0.601	3.390	8.04	2.935	11.623	-0.269	-0.207	-0.178	-0.203	0.033	0.102	0.133	0.163	0.088	
0.601	3.427	4.00	2.940	11.615	-0.238	-0.187	-0.159	-0.172	0.030	0.092	0.123	0.161	0.089	

Table 4. Continued

(1) $\theta = 315^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.199	1.483	-0.02	6.151	6.115	-0.099	-0.125	-0.176	-0.168	-0.246	-0.381	-0.526	-0.203	-0.141	-0.113	
1.199	1.983	0.00	6.151	6.117	-0.100	-0.125	-0.177	-0.169	-0.247	-0.382	-0.532	-0.267	-0.166	-0.135	
1.200	2.938	0.00	6.154	6.107	-0.101	-0.124	-0.177	-0.169	-0.247	-0.382	-0.532	-0.310	-0.185	-0.148	
1.198	3.413	0.00	6.150	6.118	-0.102	-0.124	-0.177	-0.169	-0.247	-0.383	-0.533	-0.315	-0.188	-0.150	
1.199	3.907	0.01	6.152	6.113	-0.101	-0.123	-0.177	-0.168	-0.247	-0.383	-0.534	-0.315	-0.188	-0.151	
1.199	5.894	0.03	6.151	6.115	-0.102	-0.122	-0.177	-0.169	-0.248	-0.384	-0.534	-0.256	-0.158	-0.129	
1.198	7.810	0.01	6.150	6.118	-0.104	-0.123	-0.178	-0.170	-0.251	-0.384	-0.527	-0.200	-0.131	-0.107	
1.200	3.430	8.00	6.154	6.109	-0.173	-0.191	-0.254	-0.241	-0.299	-0.432	-0.568	-0.262	-0.204	-0.185	
1.202	3.410	4.00	6.158	6.093	-0.168	-0.195	-0.242	-0.236	-0.304	-0.432	-0.568	-0.253	-0.216	-0.192	
0.899	1.480	0.02	4.958	8.769	-0.249	-0.302	-0.313	-0.250	-0.161	-0.108	0.067	0.102	0.136	0.172	
0.901	1.989	-0.01	4.971	8.747	-0.248	-0.303	-0.322	-0.278	-0.180	-0.115	0.058	0.093	0.129	0.166	
0.903	2.939	0.00	4.980	8.732	-0.245	-0.302	-0.327	-0.297	-0.196	-0.125	0.051	0.087	0.126	0.163	
0.899	3.377	0.00	4.960	8.767	-0.250	-0.308	-0.330	-0.295	-0.195	-0.127	0.051	0.088	0.126	0.162	
0.900	3.932	0.00	4.968	8.754	-0.249	-0.306	-0.329	-0.294	-0.194	-0.125	0.052	0.089	0.127	0.163	
0.899	5.811	0.01	4.960	8.768	-0.250	-0.306	-0.321	-0.269	-0.175	-0.115	0.062	0.098	0.134	0.168	
0.899	7.794	0.02	4.957	8.772	-0.249	-0.301	-0.299	-0.223	-0.142	-0.091	0.078	0.110	0.145	0.178	
0.901	3.433	8.03	4.970	8.752	-0.315	-0.367	-0.386	-0.263	-0.174	-0.163	0.041	0.080	0.107	0.137	
0.901	3.406	4.01	4.969	8.753	-0.280	-0.334	-0.341	-0.277	-0.176	-0.111	0.044	0.081	0.117	0.150	
0.800	1.483	-0.01	4.354	9.720	-0.266	-0.267	-0.222	-0.186	-0.181	-0.166	0.048	0.109	0.142	0.180	
0.800	1.972	0.00	4.357	9.716	-0.272	-0.275	-0.231	-0.196	-0.192	-0.181	0.039	0.103	0.137	0.174	
0.800	2.946	0.01	4.355	9.718	-0.278	-0.282	-0.238	-0.204	-0.201	-0.190	0.033	0.100	0.134	0.170	
0.801	3.397	0.01	4.361	9.710	-0.279	-0.285	-0.240	-0.203	-0.200	-0.190	0.033	0.099	0.134	0.169	
0.801	3.914	0.01	4.360	9.711	-0.279	-0.283	-0.239	-0.203	-0.199	-0.189	0.034	0.100	0.135	0.170	
0.802	5.871	0.03	4.368	9.700	-0.271	-0.273	-0.226	-0.188	-0.182	-0.169	0.046	0.107	0.142	0.177	
0.802	7.776	0.01	4.365	9.704	-0.255	-0.254	-0.205	-0.166	-0.158	-0.139	0.064	0.117	0.151	0.188	
0.800	3.450	8.03	4.358	9.717	-0.354	-0.336	-0.256	-0.208	-0.216	-0.229	0.023	0.101	0.134	0.161	
0.800	3.405	4.00	4.354	9.723	-0.306	-0.300	-0.241	-0.199	-0.189	-0.166	0.029	0.089	0.127	0.162	
0.601	1.486	0.01	2.940	11.613	-0.203	-0.198	-0.170	-0.164	-0.177	-0.188	0.023	0.110	0.145	0.177	
0.602	1.961	0.01	2.945	11.607	-0.211	-0.206	-0.180	-0.174	-0.191	-0.204	0.010	0.100	0.133	0.162	
0.602	2.922	0.02	2.943	11.610	-0.214	-0.211	-0.185	-0.179	-0.196	-0.211	0.004	0.095	0.130	0.157	
0.602	3.384	0.02	2.947	11.605	-0.215	-0.213	-0.185	-0.180	-0.198	-0.212	0.004	0.094	0.129	0.156	
0.602	3.957	0.02	2.947	11.605	-0.216	-0.215	-0.187	-0.181	-0.198	-0.211	0.003	0.095	0.131	0.159	
0.603	5.882	0.02	2.949	11.602	-0.209	-0.208	-0.178	-0.170	-0.185	-0.195	0.018	0.106	0.144	0.175	
0.603	7.782	0.02	2.950	11.601	-0.201	-0.197	-0.166	-0.155	-0.167	-0.171	0.035	0.119	0.157	0.191	
0.601	3.390	8.04	2.935	11.623	-0.276	-0.269	-0.223	-0.208	-0.228	-0.249	0.007	0.093	0.138	0.163	
0.601	3.427	4.00	2.940	11.615	-0.239	-0.238	-0.200	-0.188	-0.198	-0.195	0.005	0.088	0.128	0.157	

Table 4. Continued

(m) Top interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—			
					0.709	0.723	0.898	0.941
1.199	1.483	-0.02	6.151	6.115	-0.046	-0.043	-0.153	-0.238
1.199	1.983	0.00	6.151	6.117	-0.048	-0.045	-0.156	-0.239
1.200	2.938	0.00	6.154	6.107	-0.049	-0.044	-0.156	-0.239
1.198	3.413	0.00	6.150	6.118	-0.050	-0.045	-0.158	-0.239
1.199	3.907	0.01	6.152	6.113	-0.051	-0.045	-0.157	-0.239
1.199	5.894	0.03	6.151	6.115	-0.052	-0.044	-0.159	-0.240
1.198	7.810	0.01	6.150	6.118	-0.053	-0.043	-0.160	-0.240
1.200	3.430	8.00	6.154	6.109	-0.179	-0.168	-0.222	-0.262
1.202	3.410	4.00	6.158	6.093	-0.125	-0.124	-0.222	-0.285
0.899	1.480	0.02	4.958	8.769	-0.053	-0.033	-0.317	-0.055
0.901	1.989	-0.01	4.971	8.747	-0.058	-0.036	-0.320	-0.070
0.903	2.939	0.00	4.980	8.732	-0.054	-0.032	-0.320	-0.082
0.899	3.377	0.00	4.960	8.767	-0.053	-0.033	-0.324	-0.082
0.900	3.932	0.00	4.968	8.754	-0.053	-0.033	-0.322	-0.082
0.899	5.811	0.01	4.960	8.768	-0.053	-0.034	-0.320	-0.065
0.899	7.794	0.02	4.957	8.772	-0.053	-0.034	-0.311	-0.032
0.901	3.433	8.03	4.970	8.752	-0.112	-0.093	-0.384	-0.119
0.901	3.406	4.01	4.969	8.753	-0.059	-0.049	-0.344	-0.081
0.800	1.483	-0.01	4.354	9.720	-0.045	-0.038	-0.267	-0.087
0.800	1.972	0.00	4.357	9.716	-0.047	-0.038	-0.276	-0.102
0.800	2.946	0.01	4.355	9.718	-0.048	-0.039	-0.283	-0.113
0.801	3.397	0.01	4.361	9.710	-0.049	-0.041	-0.284	-0.111
0.801	3.914	0.01	4.360	9.711	-0.048	-0.039	-0.283	-0.109
0.802	5.871	0.03	4.368	9.700	-0.045	-0.038	-0.270	-0.085
0.802	7.776	0.01	4.365	9.704	-0.044	-0.038	-0.250	-0.054
0.800	3.450	8.03	4.358	9.717	-0.099	-0.086	-0.324	-0.143
0.800	3.405	4.00	4.354	9.723	-0.066	-0.060	-0.291	-0.107
0.601	1.486	0.01	2.940	11.613	-0.043	-0.038	-0.204	-0.114
0.602	1.961	0.01	2.945	11.607	-0.042	-0.038	-0.212	-0.130
0.602	2.922	0.02	2.943	11.610	-0.043	-0.038	-0.216	-0.137
0.602	3.384	0.02	2.947	11.605	-0.043	-0.039	-0.216	-0.138
0.602	3.957	0.02	2.947	11.605	-0.042	-0.041	-0.218	-0.137
0.603	5.882	0.02	2.949	11.602	-0.042	-0.040	-0.208	-0.121
0.603	7.782	0.02	2.950	11.601	-0.042	-0.039	-0.196	-0.098
0.601	3.390	8.04	2.935	11.623	-0.082	-0.080	-0.260	-0.157
0.601	3.427	4.00	2.940	11.615	-0.056	-0.056	-0.229	-0.134

Table 4. Concluded

(n) Bottom interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.709	0.723	0.738	0.752	0.767	0.781	0.869	0.883	0.941	
1.199	1.483	-0.02	6.151	6.115	-0.031	-0.055	-0.079	-0.093	-0.104	-0.105	-0.196	-0.215	-0.181	
1.199	1.983	0.00	6.151	6.117	-0.030	-0.054	-0.078	-0.092	-0.104	-0.105	-0.195	-0.214	-0.182	
1.200	2.938	0.00	6.154	6.107	-0.030	-0.053	-0.077	-0.091	-0.103	-0.106	-0.194	-0.214	-0.183	
1.198	3.413	0.00	6.150	6.118	-0.031	-0.054	-0.078	-0.092	-0.106	-0.107	-0.195	-0.215	-0.182	
1.199	3.907	0.01	6.152	6.113	-0.028	-0.052	-0.077	-0.091	-0.105	-0.107	-0.195	-0.214	-0.182	
1.199	5.894	0.03	6.151	6.115	-0.027	-0.051	-0.077	-0.090	-0.105	-0.109	-0.195	-0.215	-0.184	
1.198	7.810	0.01	6.150	6.118	-0.025	-0.049	-0.076	-0.089	-0.105	-0.108	-0.195	-0.216	-0.183	
1.200	3.430	8.00	6.154	6.109	0.179	0.146	0.113	0.089	0.058	0.021	-0.079	-0.079	-0.128	
1.202	3.410	4.00	6.158	6.093	0.065	0.050	0.035	0.031	0.012	-0.017	-0.160	-0.169	-0.155	
0.899	1.480	0.02	4.958	8.769	-0.054	-0.056	-0.052	-0.049	-0.055	-0.079	-0.120	-0.121	-0.074	
0.901	1.989	-0.01	4.971	8.747	-0.058	-0.058	-0.053	-0.049	-0.056	-0.079	-0.124	-0.126	-0.088	
0.903	2.939	0.00	4.980	8.732	-0.058	-0.058	-0.053	-0.049	-0.055	-0.078	-0.125	-0.128	-0.096	
0.899	3.377	0.00	4.960	8.767	-0.056	-0.056	-0.054	-0.051	-0.057	-0.080	-0.126	-0.128	-0.098	
0.900	3.932	0.00	4.968	8.754	-0.057	-0.057	-0.053	-0.048	-0.054	-0.077	-0.124	-0.126	-0.095	
0.899	5.811	0.01	4.960	8.768	-0.057	-0.059	-0.055	-0.050	-0.056	-0.078	-0.123	-0.125	-0.084	
0.899	7.794	0.02	4.957	8.772	-0.057	-0.057	-0.053	-0.049	-0.054	-0.076	-0.118	-0.118	-0.057	
0.901	3.433	8.03	4.970	8.752	-0.038	-0.069	-0.083	-0.087	-0.094	-0.119	-0.101	-0.101	-0.139	
0.901	3.406	4.01	4.969	8.753	-0.050	-0.064	-0.066	-0.063	-0.068	-0.091	-0.115	-0.118	-0.112	
0.800	1.483	-0.01	4.354	9.720	-0.043	-0.055	-0.057	-0.057	-0.066	-0.089	-0.101	-0.100	-0.086	
0.800	1.972	0.00	4.357	9.716	-0.043	-0.054	-0.057	-0.057	-0.065	-0.089	-0.102	-0.101	-0.098	
0.800	2.946	0.01	4.355	9.718	-0.043	-0.055	-0.057	-0.057	-0.066	-0.089	-0.103	-0.103	-0.107	
0.801	3.397	0.01	4.361	9.710	-0.044	-0.056	-0.058	-0.058	-0.067	-0.090	-0.104	-0.103	-0.105	
0.801	3.914	0.01	4.360	9.711	-0.043	-0.056	-0.057	-0.058	-0.066	-0.088	-0.103	-0.102	-0.103	
0.802	5.871	0.03	4.368	9.700	-0.044	-0.056	-0.057	-0.058	-0.066	-0.089	-0.103	-0.101	-0.087	
0.802	7.776	0.01	4.365	9.704	-0.044	-0.056	-0.058	-0.059	-0.066	-0.088	-0.099	-0.096	-0.061	
0.800	3.450	8.03	4.358	9.717	-0.044	-0.056	-0.058	-0.058	-0.066	-0.089	-0.076	-0.078	-0.143	
0.800	3.405	4.00	4.354	9.723	-0.001	-0.025	-0.033	-0.038	-0.046	-0.071	-0.091	-0.092	-0.117	
0.601	1.486	0.01	2.940	11.613	-0.019	-0.037	-0.041	-0.044	-0.051	-0.074	-0.091	-0.092	-0.117	
0.602	1.961	0.01	2.945	11.607	-0.035	-0.048	-0.049	-0.053	-0.061	-0.081	-0.080	-0.080	-0.094	
0.602	2.922	0.02	2.943	11.610	-0.036	-0.049	-0.051	-0.054	-0.062	-0.082	-0.084	-0.083	-0.108	
0.602	3.384	0.02	2.947	11.605	-0.037	-0.050	-0.052	-0.055	-0.063	-0.083	-0.086	-0.085	-0.113	
0.602	3.957	0.02	2.947	11.605	-0.038	-0.051	-0.053	-0.055	-0.064	-0.083	-0.086	-0.085	-0.114	
0.603	5.882	0.02	2.947	11.605	-0.039	-0.052	-0.053	-0.056	-0.064	-0.083	-0.087	-0.086	-0.114	
0.603	7.782	0.02	2.949	11.602	-0.039	-0.052	-0.054	-0.056	-0.063	-0.083	-0.086	-0.084	-0.101	
0.603	3.390	8.04	2.950	11.601	-0.038	-0.051	-0.053	-0.055	-0.062	-0.081	-0.083	-0.080	-0.082	
0.601	3.427	4.00	2.940	11.615	-0.009	-0.025	-0.028	-0.033	-0.036	-0.055	-0.055	-0.058	-0.137	
0.601													-0.122	

Table 5. Static Pressure Coefficients of Afterbody Surface With Horizontal Tail in Aft Position and Vertical Tail in Forward Position

(a) $\theta = 0^\circ$

M	NPR	α , deg	q_∞ , psi	P_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
1.202	1.477	0.03	6.124	6.056	-0.049	-0.052	0.047	0.059	0.068	0.040	0.003	-0.029	-0.075	-0.124	-0.179
1.199	1.939	0.01	6.119	6.077	-0.050	-0.052	0.047	0.058	0.068	0.039	0.002	-0.030	-0.076	-0.127	-0.182
1.200	2.923	0.02	6.120	6.072	-0.051	-0.053	0.046	0.058	0.066	0.038	0.002	-0.030	-0.076	-0.126	-0.182
1.201	3.410	0.03	6.122	6.062	-0.051	-0.052	0.046	0.056	0.065	0.038	0.002	-0.030	-0.076	-0.125	-0.182
1.202	3.914	0.02	6.123	6.057	-0.051	-0.052	0.045	0.053	0.063	0.036	0.001	-0.031	-0.076	-0.125	-0.182
1.201	5.875	0.03	6.122	6.061	-0.052	-0.050	0.046	0.055	0.063	0.037	0.002	-0.030	-0.076	-0.125	-0.183
1.201	7.816	0.02	6.123	6.062	-0.053	-0.051	0.046	0.054	0.063	0.036	0.001	-0.031	-0.076	-0.125	-0.183
1.200	3.407	8.04	6.118	6.074	-0.195	-0.202	-0.127	-0.132	-0.154	-0.176	-0.188	-0.212	-0.245	-0.275	-0.330
1.202	3.378	4.03	6.124	6.051	-0.131	-0.136	-0.055	-0.060	-0.062	-0.089	-0.118	-0.141	-0.172	-0.216	-0.273
0.902	1.513	0.00	4.946	8.689	0.023	0.058	0.042	-0.019	0.068	-0.164	-0.252	-0.312	-0.388	-0.463	-0.449
0.902	1.951	0.01	4.949	8.684	0.020	0.057	0.042	-0.018	0.067	-0.164	-0.251	-0.312	-0.387	-0.462	-0.454
0.900	2.937	0.03	4.934	8.711	0.023	0.058	0.041	-0.021	0.069	-0.167	-0.255	-0.317	-0.392	-0.468	-0.450
0.898	3.375	0.02	4.925	8.724	0.019	0.053	0.037	-0.025	0.074	-0.171	-0.259	-0.320	-0.396	-0.471	-0.446
0.900	3.913	0.01	4.935	8.707	0.021	0.057	0.041	-0.020	0.069	-0.167	-0.255	-0.315	-0.391	-0.467	-0.448
0.900	5.836	0.03	4.934	8.708	0.020	0.056	0.040	-0.021	0.070	-0.168	-0.256	-0.317	-0.392	-0.467	-0.444
0.900	7.817	0.04	4.938	8.704	0.022	0.057	0.041	-0.020	0.068	-0.166	-0.254	-0.315	-0.391	-0.466	-0.424
0.900	3.431	8.08	4.935	8.708	-0.043	-0.020	0.045	-0.115	-0.165	-0.268	-0.365	-0.430	-0.500	-0.575	-0.622
0.902	3.384	3.98	4.945	8.690	0.009	0.032	0.005	-0.060	-0.110	-0.210	-0.300	-0.361	-0.433	-0.509	-0.505
0.802	1.470	-0.01	4.340	9.642	-0.001	0.007	-0.030	-0.103	-0.153	-0.249	-0.311	-0.307	-0.270	-0.203	-0.134
0.801	1.953	0.01	4.334	9.650	-0.002	0.008	-0.029	-0.103	-0.151	-0.247	-0.309	-0.306	-0.270	-0.205	-0.137
0.801	2.935	0.03	4.336	9.648	-0.001	0.008	-0.029	-0.103	-0.152	-0.249	-0.311	-0.308	-0.272	-0.206	-0.137
0.801	3.395	0.03	4.332	9.653	-0.001	0.008	-0.028	-0.101	-0.151	-0.246	-0.307	-0.306	-0.272	-0.205	-0.136
0.800	3.902	0.03	4.330	9.657	-0.002	0.006	-0.031	-0.104	-0.154	-0.249	-0.310	-0.304	-0.268	-0.203	-0.135
0.801	5.864	0.04	4.331	9.655	-0.003	0.005	-0.032	-0.104	-0.154	-0.250	-0.311	-0.305	-0.268	-0.203	-0.135
0.802	7.714	-0.03	4.338	9.645	0.001	0.008	-0.027	-0.099	-0.148	-0.245	-0.306	-0.302	-0.265	-0.199	-0.131
0.800	3.412	8.00	4.329	9.662	-0.068	-0.068	-0.117	-0.206	-0.266	-0.391	-0.496	-0.554	-0.464	-0.280	-0.185
0.800	3.401	3.99	4.328	9.661	-0.032	-0.031	-0.077	-0.158	-0.215	-0.324	-0.397	-0.389	-0.327	-0.240	-0.160
0.603	1.498	0.01	2.933	11.527	-0.014	-0.020	-0.060	-0.111	-0.134	-0.185	-0.209	-0.189	-0.168	-0.139	-0.109
0.603	1.945	0.02	2.933	11.528	-0.016	-0.023	-0.063	-0.114	-0.138	-0.188	-0.211	-0.192	-0.171	-0.142	-0.110
0.601	2.930	-0.02	2.919	11.545	-0.014	-0.023	-0.062	-0.114	-0.136	-0.187	-0.211	-0.192	-0.170	-0.142	-0.112
0.601	3.389	-0.02	2.921	11.543	-0.016	-0.024	-0.062	-0.114	-0.136	-0.188	-0.211	-0.191	-0.171	-0.141	-0.111
0.601	3.896	-0.01	2.920	11.543	-0.015	-0.024	-0.063	-0.114	-0.136	-0.188	-0.211	-0.189	-0.169	-0.140	-0.109
0.601	5.869	0.00	2.920	11.542	-0.013	-0.022	-0.061	-0.110	-0.133	-0.185	-0.208	-0.187	-0.167	-0.137	-0.107
0.601	7.809	0.01	2.921	11.540	-0.012	-0.022	-0.060	-0.109	-0.132	-0.185	-0.207	-0.187	-0.167	-0.137	-0.107
0.600	3.408	8.00	2.911	11.554	-0.081	-0.104	-0.157	-0.221	-0.242	-0.299	-0.317	-0.283	-0.252	-0.209	-0.164
0.601	3.406	4.00	2.918	11.543	-0.048	-0.068	-0.114	-0.168	-0.191	-0.244	-0.263	-0.237	-0.211	-0.174	-0.136

Table 5. Continued

(a) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.869	0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	
1.202	1.477	0.03	6.124	6.056	-0.242	-0.243	-0.169	-0.180	-0.230	-0.209	-0.214	-0.290	-0.273	
1.199	1.939	0.01	6.119	6.077	-0.245	-0.244	-0.170	-0.180	-0.232	-0.202	-0.211	-0.288	-0.289	
1.200	2.923	0.02	6.120	6.072	-0.246	-0.245	-0.170	-0.179	-0.231	-0.205	-0.214	-0.290	-0.292	
1.201	3.410	0.03	6.122	6.062	-0.246	-0.246	-0.172	-0.178	-0.231	-0.213	-0.218	-0.291	-0.291	
1.202	3.914	0.02	6.123	6.057	-0.248	-0.247	-0.175	-0.179	-0.232	-0.218	-0.221	-0.292	-0.290	
1.201	5.875	0.03	6.122	6.061	-0.250	-0.247	-0.174	-0.177	-0.231	-0.215	-0.220	-0.292	-0.281	
1.201	7.816	0.02	6.123	6.062	-0.249	-0.248	-0.175	-0.178	-0.232	-0.215	-0.220	-0.291	-0.177	
1.200	3.407	8.04	6.118	6.074	-0.381	-0.352	-0.274	-0.304	-0.363	-0.381	-0.335	-0.366	-0.277	
1.202	3.378	4.03	6.124	6.051	-0.322	-0.320	-0.229	-0.222	-0.270	-0.285	-0.276	-0.318	-0.283	
0.902	1.513	0.00	4.946	6.689	-0.079	-0.020	-0.025	-0.052	-0.080	-0.091	-0.144	0.016	0.108	
0.902	1.951	0.01	4.949	8.684	-0.081	-0.021	-0.027	-0.054	-0.085	-0.097	-0.151	0.010	0.103	
0.900	2.937	0.03	4.934	8.711	-0.082	-0.024	-0.031	-0.059	-0.090	-0.103	-0.160	0.002	0.097	
0.898	3.375	0.02	4.925	8.724	-0.082	-0.026	-0.033	-0.061	-0.092	-0.106	-0.162	0.001	0.096	
0.900	3.913	0.01	4.935	8.707	-0.081	-0.024	-0.030	-0.058	-0.089	-0.103	-0.160	0.003	0.099	
0.900	5.836	0.03	4.934	8.708	-0.080	-0.023	-0.028	-0.055	-0.084	-0.094	-0.145	0.016	0.110	
0.900	7.817	0.04	4.938	8.704	-0.073	-0.018	-0.022	-0.047	-0.073	-0.079	-0.123	0.034	0.126	
0.900	3.431	8.08	4.935	8.708	-0.200	-0.087	-0.062	-0.074	-0.089	-0.091	-0.105	0.014	0.094	
0.902	3.384	3.98	4.945	8.690	-0.113	-0.039	-0.034	-0.057	-0.081	-0.088	-0.125	0.013	0.094	
0.802	1.470	-0.01	4.340	9.642	-0.068	-0.045	-0.052	-0.074	-0.095	-0.099	-0.157	-0.003	0.094	
0.801	1.953	0.01	4.334	9.650	-0.069	-0.047	-0.054	-0.077	-0.098	-0.104	-0.163	-0.010	0.087	
0.801	2.935	0.03	4.336	9.648	-0.070	-0.048	-0.056	-0.079	-0.101	-0.108	-0.168	-0.015	0.081	
0.801	3.395	0.03	4.332	9.653	-0.069	-0.047	-0.055	-0.079	-0.100	-0.107	-0.168	-0.014	0.082	
0.800	3.902	0.03	4.330	9.657	-0.068	-0.047	-0.054	-0.078	-0.099	-0.106	-0.165	-0.012	0.085	
0.801	5.864	0.04	4.331	9.655	-0.067	-0.045	-0.052	-0.075	-0.094	-0.098	-0.152	0.000	0.097	
0.802	7.714	-0.03	4.338	9.645	-0.063	-0.041	-0.046	-0.068	-0.085	-0.085	-0.134	0.018	0.115	
0.800	3.412	8.00	4.329	9.662	-0.109	-0.083	-0.084	-0.105	-0.118	-0.116	-0.152	-0.011	0.078	
0.800	3.401	3.99	4.328	9.661	-0.087	-0.064	-0.066	-0.089	-0.105	-0.106	-0.151	-0.008	0.082	
0.603	1.498	0.01	2.933	11.527	-0.066	-0.050	-0.051	-0.065	-0.079	-0.083	-0.128	-0.004	0.084	
0.603	1.945	0.02	2.933	11.528	-0.069	-0.053	-0.055	-0.071	-0.085	-0.089	-0.137	-0.012	0.073	
0.601	2.930	-0.02	2.919	11.545	-0.068	-0.053	-0.054	-0.074	-0.086	-0.090	-0.140	-0.016	0.068	
0.601	3.389	-0.02	2.921	11.543	-0.068	-0.053	-0.056	-0.074	-0.088	-0.092	-0.142	-0.017	0.066	
0.601	3.896	-0.01	2.920	11.543	-0.067	-0.053	-0.054	-0.073	-0.087	-0.091	-0.139	-0.015	0.069	
0.601	5.869	0.00	2.920	11.542	-0.064	-0.050	-0.051	-0.070	-0.083	-0.084	-0.130	-0.005	0.082	
0.601	7.809	0.01	2.921	11.540	-0.061	-0.047	-0.047	-0.066	-0.077	-0.075	-0.117	0.008	0.098	
0.600	3.408	8.00	2.911	11.554	-0.109	-0.091	-0.087	-0.109	-0.113	-0.106	-0.142	-0.021	0.060	
0.601	3.406	4.00	2.918	11.543	-0.084	-0.070	-0.066	-0.090	-0.096	-0.092	-0.131	-0.015	0.066	

Table 5. Continued

(b) $\theta = 45^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.941	0.951	0.963	0.974	0.997	
1.202	1.477	0.03	6.124	6.056	-0.054	0.117	0.087	-0.001	-0.037	-0.100	-0.176	-0.249	-0.379	-0.306	0.033	
1.199	1.939	0.01	6.119	6.077	-0.057	0.117	0.087	-0.003	-0.040	-0.103	-0.179	-0.250	-0.377	-0.320	0.021	
1.200	2.923	0.02	6.120	6.072	-0.057	0.116	0.088	-0.001	-0.039	-0.102	-0.179	-0.250	-0.377	-0.324	0.006	
1.201	3.410	0.03	6.122	6.062	-0.056	0.116	0.088	0.000	-0.036	-0.101	-0.177	-0.250	-0.378	-0.327	0.002	
1.202	3.914	0.02	6.123	6.057	-0.058	0.114	0.087	0.000	-0.037	-0.101	-0.178	-0.250	-0.378	-0.328	0.003	
1.201	5.875	0.03	6.122	6.061	-0.057	0.115	0.087	-0.001	-0.037	-0.101	-0.178	-0.250	-0.377	-0.320	0.026	
1.201	7.816	0.02	6.123	6.062	-0.059	0.114	0.086	-0.001	-0.038	-0.101	-0.178	-0.250	-0.376	-0.320	0.050	
1.200	3.407	8.04	6.118	6.074	-0.175	0.003	0.026	0.007	0.004	0.003	-0.334	-0.370	-0.487	-0.241	-0.040	
1.202	3.378	4.03	6.124	6.051	-0.115	0.056	0.062	0.004	-0.040	-0.083	-0.225	-0.289	-0.424	-0.342	0.000	
0.902	1.513	0.00	4.946	8.689	0.049	0.121	-0.024	-0.060	-0.068	-0.068	-0.088	-0.124	-0.008	0.096	0.140	
0.900	2.937	0.03	4.934	8.711	0.046	0.119	-0.026	-0.062	-0.070	-0.071	-0.096	-0.135	-0.018	0.087	0.134	
0.898	3.375	0.02	4.925	8.724	0.045	0.117	-0.029	-0.064	-0.073	-0.072	-0.096	-0.136	-0.019	0.086	0.136	
0.900	5.836	0.03	4.934	8.708	0.048	0.121	-0.026	-0.062	-0.070	-0.069	-0.095	-0.135	-0.017	0.088	0.139	
0.900	7.817	0.04	4.938	8.704	0.046	0.119	-0.028	-0.063	-0.070	-0.069	-0.087	-0.124	-0.006	0.099	0.148	
0.900	3.431	8.08	4.935	8.708	-0.021	0.050	-0.089	-0.147	-0.069	-0.067	-0.075	-0.107	0.009	0.114	0.156	
0.902	3.384	3.98	4.945	8.690	0.042	0.112	-0.030	-0.077	-0.091	-0.090	-0.094	-0.111	-0.004	0.090	0.128	
0.802	1.470	-0.01	4.340	9.642	0.033	0.096	-0.036	-0.064	-0.069	-0.063	-0.077	-0.113	-0.012	0.081	0.118	
0.801	1.953	0.01	4.334	9.650	0.034	0.096	-0.035	-0.063	-0.068	-0.063	-0.080	-0.117	-0.017	0.080	0.147	
0.801	2.935	0.03	4.336	9.648	0.032	0.096	-0.035	-0.064	-0.069	-0.064	-0.084	-0.121	-0.021	0.075	0.144	
0.801	3.395	0.03	4.332	9.653	0.033	0.097	-0.033	-0.064	-0.069	-0.063	-0.083	-0.121	-0.021	0.075	0.145	
0.800	3.902	0.03	4.330	9.657	0.036	0.098	-0.032	-0.062	-0.068	-0.062	-0.081	-0.119	-0.020	0.077	0.149	
0.801	5.864	0.04	4.331	9.655	0.033	0.096	-0.035	-0.065	-0.070	-0.064	-0.076	-0.111	-0.011	0.087	0.162	
0.802	7.714	-0.03	4.338	9.645	0.035	0.098	-0.034	-0.063	-0.068	-0.062	-0.066	-0.098	0.003	0.102	0.173	
0.800	3.412	8.00	4.329	9.662	-0.014	0.055	-0.049	-0.083	-0.091	-0.085	-0.101	-0.118	-0.019	0.074	0.138	
0.800	3.401	3.99	4.328	9.661	0.028	0.096	-0.017	-0.049	-0.057	-0.053	-0.088	-0.123	-0.023	0.073	0.137	
0.603	1.498	0.01	2.933	11.527	0.030	0.081	-0.027	-0.048	-0.054	-0.045	-0.056	-0.086	-0.004	0.079	0.149	
0.603	1.945	0.02	2.933	11.528	0.030	0.079	-0.031	-0.051	-0.056	-0.046	-0.062	-0.093	-0.012	0.069	0.121	
0.601	2.930	-0.02	2.919	11.545	0.031	0.080	-0.030	-0.051	-0.056	-0.046	-0.062	-0.095	-0.016	0.064	0.118	
0.601	3.389	-0.02	2.921	11.543	0.028	0.077	-0.033	-0.054	-0.058	-0.049	-0.065	-0.098	-0.018	0.062	0.118	
0.601	3.896	-0.01	2.920	11.543	0.029	0.079	-0.031	-0.052	-0.056	-0.048	-0.063	-0.097	-0.017	0.062	0.124	
0.601	5.869	0.00	2.920	11.542	0.029	0.079	-0.032	-0.053	-0.056	-0.048	-0.063	-0.089	-0.009	0.073	0.147	
0.601	7.809	0.01	2.921	11.540	0.031	0.080	-0.032	-0.053	-0.056	-0.046	-0.063	-0.089	-0.001	0.086	0.164	
0.600	3.408	8.00	2.911	11.554	-0.004	0.059	-0.017	-0.043	-0.049	-0.043	-0.089	-0.109	-0.029	0.054	0.118	
0.601	3.406	4.00	2.918	11.543	0.026	0.085	-0.005	-0.031	-0.038	-0.033	-0.070	-0.106	-0.026	0.057	0.120	

Table 5. Continued

(c) $\theta = 90^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—													
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.825	0.951	0.963	0.974	0.986	0.997	
1.202	1.477	0.03	6.124	6.056	-0.058	0.001	0.055	-0.035	-0.069	-0.078	-0.131	-0.148	-0.436	-0.379	-0.227	-0.055	0.006	
1.199	1.939	0.01	6.119	6.077	-0.059	0.001	0.054	-0.039	-0.072	-0.080	-0.134	-0.151	-0.438	-0.381	-0.272	-0.079	-0.003	
1.200	2.923	0.02	6.120	6.072	-0.057	0.003	0.055	-0.038	-0.072	-0.079	-0.133	-0.152	-0.436	-0.381	-0.295	-0.134	-0.018	
1.201	3.410	0.03	6.122	6.062	-0.058	0.002	0.054	-0.038	-0.071	-0.078	-0.131	-0.156	-0.433	-0.379	-0.300	-0.141	-0.021	
1.202	3.914	0.02	6.123	6.057	-0.057	0.002	0.053	-0.039	-0.071	-0.079	-0.131	-0.156	-0.433	-0.380	-0.301	-0.138	-0.019	
1.201	5.875	0.03	6.122	6.061	-0.058	0.002	0.052	-0.039	-0.072	-0.080	-0.132	-0.156	-0.432	-0.379	-0.267	-0.069	0.002	
1.201	7.816	0.02	6.123	6.062	-0.059	0.001	0.051	-0.040	-0.072	-0.079	-0.131	-0.156	-0.432	-0.376	-0.144	-0.048	0.021	
1.200	3.407	8.04	6.118	6.074	0.102	0.063	0.012	-0.053	-0.062	0.003	-0.031	-0.082	-0.556	-0.474	-0.211	-0.180	-0.123	
1.202	3.378	4.03	6.124	6.051	0.041	0.020	0.027	-0.052	-0.085	-0.066	-0.102	-0.127	-0.481	-0.449	-0.282	-0.116	-0.063	
0.902	1.513	0.00	4.946	8.689	0.014	0.027	0.003	-0.061	-0.045	-0.045	-0.053	0.059	-0.202	-0.059	0.102	0.171	0.216	
0.902	1.951	0.01	4.949	8.684	0.012	0.025	0.001	-0.062	-0.046	-0.047	-0.054	0.058	-0.207	-0.066	0.097	0.166	0.209	
0.900	2.937	0.03	4.934	8.711	0.013	0.025	0.000	-0.064	-0.048	-0.048	-0.055	0.057	-0.212	-0.073	0.092	0.161	0.201	
0.898	3.375	0.02	4.925	8.724	0.011	0.023	-0.001	-0.065	-0.050	-0.050	-0.057	0.056	-0.213	-0.074	0.091	0.161	0.201	
0.900	3.913	0.01	4.935	8.707	0.013	0.025	0.001	-0.063	-0.048	-0.047	-0.055	0.058	-0.212	-0.072	0.093	0.162	0.204	
0.900	5.836	0.03	4.934	8.708	0.011	0.023	0.000	-0.064	-0.048	-0.048	-0.055	0.058	-0.204	-0.055	0.104	0.176	0.221	
0.900	7.817	0.04	4.938	8.704	0.012	0.025	0.001	-0.062	-0.046	-0.046	-0.053	0.060	-0.196	-0.035	0.119	0.191	0.239	
0.900	3.431	8.08	4.935	8.708	-0.036	-0.041	-0.078	-0.202	-0.146	-0.134	-0.144	0.030	-0.196	-0.003	0.116	0.175	0.211	
0.902	3.384	3.98	4.945	8.690	0.006	0.009	-0.018	-0.099	-0.070	-0.066	-0.076	0.055	-0.196	-0.074	0.092	0.164	0.200	
0.802	1.470	-0.01	4.340	9.642	0.010	0.010	-0.015	-0.075	-0.051	-0.046	-0.047	0.061	-0.185	-0.053	0.100	0.164	0.208	
0.801	1.953	0.01	4.334	9.650	0.010	0.011	-0.015	-0.074	-0.051	-0.047	-0.047	0.060	-0.189	-0.057	0.095	0.158	0.200	
0.801	2.935	0.03	4.336	9.648	0.009	0.008	-0.016	-0.075	-0.052	-0.048	-0.048	0.059	-0.194	-0.060	0.089	0.152	0.191	
0.801	3.395	0.03	4.332	9.653	0.009	0.010	-0.014	-0.073	-0.051	-0.046	-0.047	0.060	-0.194	-0.060	0.090	0.153	0.191	
0.800	3.902	0.03	4.330	9.657	0.010	0.010	-0.014	-0.073	-0.050	-0.046	-0.047	0.061	-0.192	-0.058	0.092	0.154	0.194	
0.801	5.864	0.04	4.331	9.655	0.009	0.008	-0.015	-0.073	-0.051	-0.046	-0.048	0.059	-0.185	-0.051	0.101	0.167	0.211	
0.802	7.714	-0.03	4.338	9.645	0.011	0.010	-0.013	-0.072	-0.049	-0.044	-0.046	0.062	-0.171	-0.039	0.115	0.183	0.228	
0.800	3.412	8.00	4.329	9.662	0.002	-0.012	-0.038	-0.121	-0.078	-0.067	-0.075	0.059	-0.218	-0.007	0.109	0.158	0.191	
0.800	3.401	3.99	4.328	9.661	0.018	0.012	-0.009	-0.073	-0.044	-0.037	-0.043	0.066	-0.191	-0.037	0.094	0.156	0.192	
0.603	1.498	0.01	2.933	11.527	0.009	0.005	-0.015	-0.063	-0.040	-0.035	-0.031	0.064	-0.157	-0.018	0.102	0.154	0.192	
0.603	1.945	0.02	2.933	11.528	0.007	0.003	-0.016	-0.065	-0.042	-0.036	-0.033	0.062	-0.167	-0.026	0.093	0.144	0.175	
0.601	2.930	-0.02	2.919	11.545	0.007	0.002	-0.017	-0.065	-0.042	-0.036	-0.032	0.062	-0.171	-0.027	0.091	0.138	0.165	
0.601	3.389	-0.02	2.921	11.543	0.006	0.000	-0.017	-0.065	-0.044	-0.037	-0.034	0.060	-0.175	-0.031	0.088	0.136	0.163	
0.601	3.896	-0.01	2.920	11.543	0.007	0.001	-0.017	-0.065	-0.042	-0.036	-0.033	0.060	-0.174	-0.030	0.090	0.138	0.167	
0.601	5.869	0.00	2.920	11.542	0.006	0.001	-0.016	-0.063	-0.042	-0.034	-0.033	0.061	-0.164	-0.023	0.099	0.153	0.189	
0.601	7.809	0.01	2.921	11.540	0.007	0.001	-0.014	-0.062	-0.040	-0.033	-0.032	0.062	-0.154	-0.013	0.110	0.168	0.209	
0.600	3.408	8.00	2.911	11.554	0.026	0.008	-0.009	-0.072	-0.040	-0.029	-0.036	0.070	-0.220	-0.008	0.098	0.139	0.160	
0.601	3.406	4.00	2.918	11.543	0.024	0.012	-0.001	-0.050	-0.028	-0.019	-0.023	0.068	-0.199	-0.029	0.091	0.139	0.160	

Table 5. Continued

(d) $\theta = 135^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.869	
1.202	1.477	0.03	6.124	6.056	-0.065	-0.074	-0.081	-0.058	-0.064	-0.086	-0.095	-0.130	-0.145	-0.138	-0.108	
1.199	1.939	0.01	6.119	6.077	-0.067	-0.076	-0.084	-0.060	-0.066	-0.089	-0.097	-0.132	-0.147	-0.141	-0.111	
1.200	2.923	0.02	6.120	6.072	-0.065	-0.074	-0.083	-0.058	-0.065	-0.089	-0.096	-0.131	-0.146	-0.141	-0.110	
1.201	3.410	0.03	6.122	6.062	-0.066	-0.076	-0.085	-0.056	-0.065	-0.089	-0.096	-0.132	-0.146	-0.141	-0.112	
1.202	3.914	0.02	6.123	6.057	-0.066	-0.076	-0.084	-0.055	-0.065	-0.090	-0.096	-0.132	-0.146	-0.141	-0.112	
1.201	5.875	0.03	6.122	6.061	-0.068	-0.077	-0.085	-0.055	-0.066	-0.090	-0.096	-0.132	-0.145	-0.141	-0.113	
1.201	7.816	0.02	6.123	6.062	-0.069	-0.078	-0.085	-0.055	-0.066	-0.091	-0.097	-0.133	-0.146	-0.142	-0.114	
1.200	3.407	8.04	6.118	6.074	0.141	0.100	0.046	0.001	-0.034	-0.051	-0.040	-0.087	-0.081	-0.070	-0.027	
1.202	3.378	4.03	6.124	6.051	0.051	0.034	-0.009	-0.029	-0.043	-0.064	-0.075	-0.130	-0.141	-0.140	-0.079	
0.902	1.513	0.00	4.946	8.689	-0.045	-0.039	-0.062	-0.053	-0.051	-0.047	-0.041	-0.032	0.001	0.010	-0.032	
0.902	1.951	0.01	4.949	8.684	-0.047	-0.040	-0.063	-0.053	-0.051	-0.047	-0.041	-0.034	0.000	0.008	-0.034	
0.900	2.937	0.03	4.934	8.711	-0.044	-0.039	-0.064	-0.055	-0.053	-0.050	-0.046	-0.036	-0.004	0.007	-0.035	
0.898	3.375	0.02	4.925	8.724	-0.046	-0.041	-0.066	-0.057	-0.055	-0.051	-0.046	-0.035	-0.002	0.008	-0.035	
0.900	3.913	0.01	4.935	8.707	-0.045	-0.040	-0.064	-0.055	-0.053	-0.049	-0.044	-0.035	-0.003	0.007	-0.035	
0.900	5.836	0.03	4.934	8.708	-0.047	-0.041	-0.065	-0.056	-0.054	-0.050	-0.045	-0.035	-0.003	0.007	-0.035	
0.900	7.817	0.04	4.938	8.704	-0.044	-0.040	-0.063	-0.054	-0.052	-0.047	-0.042	-0.033	0.000	0.009	-0.032	
0.900	3.431	8.08	4.935	8.708	-0.067	-0.089	-0.135	-0.145	-0.147	-0.142	-0.128	-0.103	-0.042	-0.006	-0.002	
0.902	3.384	3.98	4.945	8.690	-0.048	-0.054	-0.086	-0.081	-0.078	-0.071	-0.063	-0.048	-0.007	0.009	-0.021	
0.802	1.470	-0.01	4.340	9.642	-0.036	-0.045	-0.072	-0.062	-0.055	-0.048	-0.042	-0.034	-0.007	-0.003	-0.037	
0.801	1.953	0.01	4.334	9.650	-0.037	-0.045	-0.072	-0.062	-0.056	-0.048	-0.042	-0.033	-0.007	-0.002	-0.037	
0.801	2.935	0.03	4.336	9.648	-0.036	-0.045	-0.073	-0.063	-0.056	-0.048	-0.042	-0.034	-0.007	-0.003	-0.038	
0.801	3.395	0.03	4.332	9.653	-0.036	-0.044	-0.072	-0.062	-0.056	-0.048	-0.042	-0.033	-0.007	-0.002	-0.037	
0.800	3.902	0.03	4.330	9.657	-0.036	-0.044	-0.071	-0.062	-0.055	-0.046	-0.040	-0.032	-0.006	-0.002	-0.037	
0.801	5.864	0.04	4.331	9.655	-0.037	-0.045	-0.072	-0.063	-0.055	-0.047	-0.042	-0.034	-0.007	-0.003	-0.037	
0.802	7.714	-0.03	4.338	9.645	-0.036	-0.044	-0.071	-0.063	-0.054	-0.045	-0.041	-0.032	-0.006	-0.001	-0.034	
0.800	3.412	8.00	4.329	9.662	-0.024	-0.045	-0.083	-0.085	-0.079	-0.070	-0.063	-0.051	-0.010	0.012	-0.015	
0.800	3.401	3.99	4.328	9.661	-0.018	-0.043	-0.065	-0.061	-0.052	-0.042	-0.038	-0.029	0.002	0.013	-0.009	
0.603	1.498	0.01	2.933	11.527	-0.028	-0.041	-0.067	-0.054	-0.044	-0.037	-0.029	-0.023	-0.003	-0.001	-0.028	
0.603	1.945	0.02	2.933	11.528	-0.030	-0.042	-0.069	-0.056	-0.047	-0.039	-0.032	-0.026	-0.007	-0.005	-0.031	
0.601	2.930	-0.02	2.919	11.545	-0.029	-0.042	-0.067	-0.055	-0.046	-0.037	-0.030	-0.025	-0.005	-0.004	-0.031	
0.601	3.389	-0.02	2.921	11.543	-0.030	-0.044	-0.069	-0.058	-0.047	-0.039	-0.033	-0.027	-0.008	-0.007	-0.032	
0.601	3.896	-0.01	2.920	11.543	-0.030	-0.043	-0.069	-0.058	-0.047	-0.038	-0.032	-0.026	-0.007	-0.006	-0.031	
0.601	5.869	0.00	2.920	11.542	-0.028	-0.042	-0.068	-0.057	-0.046	-0.036	-0.030	-0.024	-0.005	-0.004	-0.029	
0.601	7.809	0.01	2.921	11.540	-0.028	-0.041	-0.066	-0.058	-0.045	-0.034	-0.030	-0.025	-0.006	-0.004	-0.028	
0.600	3.408	8.00	2.911	11.554	0.000	-0.022	-0.056	-0.056	-0.043	-0.034	-0.031	-0.024	0.004	0.020	0.030	
0.601	3.406	4.00	2.918	11.543	-0.004	-0.021	-0.051	-0.048	-0.034	-0.023	-0.020	-0.015	0.008	0.014	0.001	

Table 5. Continued

(d) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.202	1.477	0.03	6.124	6.056	-0.110	-0.088	-0.108	-0.148	-0.168	-0.241	-0.375	-0.316	-0.057	0.016
1.199	1.939	0.01	6.119	6.077	-0.108	-0.089	-0.111	-0.150	-0.170	-0.242	-0.376	-0.328	-0.092	0.005
1.200	2.923	0.02	6.120	6.072	-0.108	-0.088	-0.110	-0.149	-0.170	-0.242	-0.379	-0.335	-0.156	-0.014
1.201	3.410	0.03	6.122	6.062	-0.110	-0.088	-0.111	-0.149	-0.171	-0.243	-0.381	-0.339	-0.170	-0.017
1.202	3.914	0.02	6.123	6.057	-0.111	-0.089	-0.111	-0.149	-0.171	-0.243	-0.382	-0.340	-0.170	-0.015
1.201	5.875	0.03	6.122	6.061	-0.111	-0.089	-0.112	-0.149	-0.172	-0.243	-0.384	-0.336	-0.082	0.013
1.201	7.816	0.02	6.123	6.062	-0.112	-0.090	-0.113	-0.150	-0.173	-0.243	-0.385	-0.236	-0.018	0.034
1.200	3.407	8.04	6.118	6.074	-0.052	-0.082	-0.109	-0.125	-0.129	-0.204	-0.316	-0.322	-0.255	-0.101
1.202	3.378	4.03	6.124	6.051	-0.079	-0.068	-0.084	-0.118	-0.138	-0.212	-0.340	-0.320	-0.252	-0.039
0.902	1.513	0.00	4.946	8.689	-0.049	-0.069	-0.091	-0.122	-0.107	-0.141	0.000	0.100	0.163	0.184
0.902	1.951	0.01	4.949	8.684	-0.050	-0.071	-0.094	-0.126	-0.113	-0.149	-0.007	0.093	0.155	0.173
0.900	2.937	0.03	4.934	8.711	-0.052	-0.073	-0.096	-0.129	-0.116	-0.153	-0.011	0.090	0.152	0.170
0.898	3.375	0.02	4.925	8.724	-0.054	-0.075	-0.098	-0.131	-0.117	-0.155	-0.012	0.089	0.152	0.171
0.900	3.913	0.01	4.935	8.707	-0.052	-0.072	-0.095	-0.128	-0.114	-0.152	-0.009	0.092	0.154	0.174
0.900	5.836	0.03	4.934	8.708	-0.051	-0.071	-0.093	-0.122	-0.106	-0.139	0.003	0.105	0.168	0.191
0.900	7.817	0.04	4.938	8.704	-0.047	-0.066	-0.086	-0.113	-0.093	-0.122	0.020	0.121	0.184	0.206
0.900	3.431	8.08	4.935	8.708	-0.008	-0.024	-0.044	-0.075	-0.075	-0.126	0.003	0.097	0.155	0.169
0.902	3.384	3.98	4.945	8.690	-0.035	-0.056	-0.079	-0.110	-0.105	-0.137	-0.006	0.088	0.141	0.157
0.802	1.470	-0.01	4.340	9.642	-0.049	-0.066	-0.082	-0.106	-0.088	-0.128	-0.007	0.088	0.152	0.180
0.801	1.953	0.01	4.334	9.650	-0.050	-0.067	-0.085	-0.109	-0.091	-0.133	-0.013	0.080	0.143	0.168
0.801	2.935	0.03	4.336	9.648	-0.050	-0.069	-0.086	-0.111	-0.094	-0.137	-0.017	0.076	0.138	0.161
0.801	3.395	0.03	4.332	9.653	-0.050	-0.068	-0.085	-0.110	-0.092	-0.136	-0.015	0.079	0.141	0.166
0.800	3.902	0.03	4.330	9.657	-0.049	-0.067	-0.084	-0.109	-0.093	-0.137	-0.004	0.091	0.155	0.185
0.801	5.864	0.04	4.331	9.655	-0.049	-0.066	-0.083	-0.105	-0.086	-0.126	-0.013	0.109	0.173	0.204
0.802	7.714	-0.03	4.338	9.645	-0.046	-0.062	-0.077	-0.098	-0.076	-0.112	-0.013	0.084	0.139	0.159
0.800	3.412	8.00	4.329	9.662	0.006	-0.016	-0.034	-0.063	-0.060	-0.113	-0.003	0.077	0.131	0.155
0.800	3.401	3.99	4.328	9.661	-0.023	-0.046	-0.063	-0.091	-0.082	-0.121	-0.003	0.077	0.136	0.164
0.603	1.498	0.01	2.933	11.527	-0.035	-0.049	-0.061	-0.078	-0.060	-0.101	-0.011	0.067	0.123	0.142
0.603	1.945	0.02	2.933	11.528	-0.038	-0.053	-0.065	-0.083	-0.067	-0.109	-0.011	0.067	0.123	0.142
0.601	2.930	-0.02	2.919	11.545	-0.038	-0.054	-0.065	-0.085	-0.070	-0.114	-0.015	0.062	0.116	0.133
0.601	3.389	-0.02	2.921	11.543	-0.038	-0.054	-0.065	-0.084	-0.068	-0.112	-0.012	0.065	0.119	0.139
0.601	3.896	-0.01	2.920	11.543	-0.038	-0.054	-0.065	-0.085	-0.061	-0.104	-0.003	0.078	0.134	0.165
0.601	5.869	0.00	2.920	11.542	-0.035	-0.052	-0.061	-0.080	-0.061	-0.096	0.011	0.092	0.152	0.188
0.601	7.809	0.01	2.921	11.540	-0.033	-0.050	-0.059	-0.075	-0.035	-0.087	0.006	0.075	0.118	0.134
0.600	3.408	8.00	2.911	11.554	0.024	0.002	-0.013	-0.040	-0.035	-0.097	-0.003	0.068	0.110	0.132
0.601	3.406	4.00	2.918	11.543	-0.007	-0.030	-0.042	-0.067	-0.056					

Table 5. Continued

(e) $\theta = 180^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
1.202	1.477	0.03	6.124	6.056	-0.021	-0.055	-0.080	-0.083	-0.079	-0.078	-0.099	-0.122	-0.144	-0.159	-0.178
1.199	1.939	0.01	6.119	6.077	-0.024	-0.057	-0.082	-0.085	-0.080	-0.080	-0.100	-0.123	-0.145	-0.161	-0.181
1.200	2.923	0.02	6.120	6.072	-0.023	-0.055	-0.081	-0.084	-0.079	-0.079	-0.100	-0.122	-0.144	-0.161	-0.182
1.201	3.410	0.03	6.122	6.062	-0.024	-0.054	-0.081	-0.086	-0.079	-0.078	-0.101	-0.123	-0.144	-0.160	-0.181
1.202	3.914	0.02	6.123	6.057	-0.024	-0.052	-0.081	-0.085	-0.078	-0.078	-0.102	-0.123	-0.143	-0.159	-0.181
1.201	5.875	0.03	6.122	6.061	-0.025	-0.052	-0.081	-0.086	-0.079	-0.079	-0.103	-0.123	-0.143	-0.159	-0.182
1.201	7.816	0.02	6.123	6.062	-0.027	-0.052	-0.081	-0.086	-0.078	-0.079	-0.104	-0.123	-0.143	-0.160	-0.183
1.200	3.407	8.04	6.118	6.074	0.181	0.135	0.100	0.077	0.052	0.021	-0.029	-0.062	-0.089	-0.112	-0.122
1.202	3.378	4.03	6.124	6.051	0.073	0.048	0.027	0.018	0.007	-0.013	-0.053	-0.087	-0.120	-0.151	-0.174
0.902	1.513	0.00	4.946	8.689	-0.057	-0.061	-0.060	-0.053	-0.049	-0.058	-0.065	-0.056	-0.047	-0.045	-0.047
0.902	1.951	0.01	4.949	8.684	-0.059	-0.062	-0.060	-0.053	-0.049	-0.058	-0.066	-0.056	-0.048	-0.045	-0.047
0.900	2.937	0.03	4.934	8.711	-0.056	-0.061	-0.062	-0.055	-0.051	-0.060	-0.068	-0.058	-0.049	-0.048	-0.050
0.898	3.375	0.02	4.925	8.724	-0.058	-0.063	-0.064	-0.057	-0.054	-0.062	-0.069	-0.060	-0.051	-0.049	-0.051
0.900	3.913	0.01	4.935	8.707	-0.058	-0.062	-0.062	-0.056	-0.051	-0.060	-0.068	-0.058	-0.049	-0.047	-0.049
0.900	5.836	0.03	4.934	8.708	-0.059	-0.063	-0.063	-0.056	-0.052	-0.061	-0.068	-0.058	-0.049	-0.048	-0.049
0.900	7.817	0.04	4.938	8.704	-0.057	-0.061	-0.060	-0.054	-0.050	-0.058	-0.065	-0.056	-0.046	-0.045	-0.045
0.900	3.431	8.08	4.935	8.708	-0.041	-0.077	-0.095	-0.102	-0.107	-0.121	-0.125	-0.103	-0.077	-0.061	-0.047
0.902	3.384	3.98	4.945	8.690	-0.045	-0.066	-0.071	-0.069	-0.067	-0.077	-0.081	-0.068	-0.055	-0.048	-0.044
0.802	1.470	-0.01	4.340	9.642	-0.045	-0.060	-0.063	-0.057	-0.052	-0.060	-0.065	-0.059	-0.053	-0.051	-0.051
0.801	1.953	0.01	4.334	9.650	-0.044	-0.060	-0.063	-0.057	-0.052	-0.060	-0.064	-0.059	-0.053	-0.052	-0.052
0.801	2.935	0.03	4.336	9.648	-0.044	-0.060	-0.063	-0.057	-0.052	-0.060	-0.065	-0.060	-0.054	-0.052	-0.053
0.801	3.395	0.03	4.332	9.653	-0.044	-0.060	-0.063	-0.057	-0.052	-0.060	-0.065	-0.059	-0.054	-0.052	-0.052
0.800	3.902	0.03	4.330	9.657	-0.043	-0.059	-0.062	-0.055	-0.051	-0.059	-0.063	-0.059	-0.053	-0.051	-0.052
0.801	5.864	0.04	4.331	9.655	-0.044	-0.061	-0.063	-0.057	-0.053	-0.060	-0.065	-0.060	-0.055	-0.052	-0.051
0.802	7.714	-0.03	4.338	9.645	-0.043	-0.060	-0.062	-0.055	-0.052	-0.059	-0.062	-0.058	-0.052	-0.049	-0.048
0.800	3.412	8.00	4.329	9.662	-0.003	-0.035	-0.044	-0.043	-0.045	-0.055	-0.059	-0.053	-0.043	-0.034	-0.026
0.800	3.401	3.99	4.328	9.661	-0.016	-0.041	-0.046	-0.042	-0.041	-0.049	-0.052	-0.049	-0.044	-0.038	-0.035
0.603	1.498	0.01	2.933	11.527	-0.034	-0.051	-0.051	-0.045	-0.038	-0.046	-0.054	-0.049	-0.044	-0.043	-0.044
0.603	1.945	0.02	2.933	11.528	-0.037	-0.054	-0.054	-0.048	-0.041	-0.049	-0.057	-0.052	-0.050	-0.047	-0.046
0.601	2.930	-0.02	2.919	11.545	-0.035	-0.052	-0.052	-0.046	-0.039	-0.047	-0.054	-0.050	-0.050	-0.046	-0.046
0.601	3.389	-0.02	2.921	11.543	-0.037	-0.055	-0.054	-0.048	-0.041	-0.049	-0.056	-0.051	-0.052	-0.047	-0.047
0.601	3.896	-0.01	2.920	11.543	-0.035	-0.054	-0.053	-0.047	-0.041	-0.049	-0.054	-0.051	-0.051	-0.046	-0.045
0.601	5.869	0.00	2.920	11.542	-0.035	-0.054	-0.053	-0.046	-0.041	-0.048	-0.054	-0.050	-0.051	-0.045	-0.044
0.601	7.809	0.01	2.921	11.540	-0.034	-0.055	-0.051	-0.046	-0.040	-0.048	-0.053	-0.050	-0.050	-0.045	-0.042
0.600	3.408	8.00	2.911	11.554	0.019	-0.012	-0.015	-0.010	-0.012	-0.019	-0.025	-0.025	-0.024	-0.014	-0.008
0.601	3.406	4.00	2.918	11.543	-0.003	-0.030	-0.030	-0.024	-0.024	-0.030	-0.035	-0.036	-0.036	-0.028	-0.024

Table 5. Continued

(e) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—								
					0.883	0.898	0.912	0.927	0.941	0.951	0.974	0.986	
1.202	1.477	0.03	6.124	6.056	-0.172	-0.165	-0.169	-0.163	-0.164	-0.212	-0.234	-0.037	
1.199	1.939	0.01	6.119	6.077	-0.174	-0.167	-0.170	-0.163	-0.160	-0.209	-0.274	-0.061	
1.200	2.923	0.02	6.120	6.072	-0.172	-0.165	-0.170	-0.165	-0.161	-0.210	-0.290	-0.113	
1.201	3.410	0.03	6.122	6.062	-0.170	-0.164	-0.170	-0.167	-0.164	-0.213	-0.291	-0.125	
1.202	3.914	0.02	6.123	6.057	-0.169	-0.163	-0.170	-0.167	-0.164	-0.213	-0.291	-0.124	
1.201	5.875	0.03	6.122	6.061	-0.168	-0.163	-0.170	-0.168	-0.164	-0.212	-0.269	-0.056	
1.201	7.816	0.02	6.123	6.062	-0.169	-0.164	-0.170	-0.168	-0.164	-0.213	-0.143	-0.021	
1.200	3.407	8.04	6.118	6.074	-0.095	-0.094	-0.122	-0.132	-0.155	-0.171	-0.294	-0.314	
1.202	3.378	4.03	6.124	6.051	-0.168	-0.153	-0.157	-0.143	-0.135	-0.178	-0.269	-0.199	
0.902	1.513	0.00	4.946	8.689	-0.060	-0.072	-0.095	-0.108	-0.139	-0.155	0.110	0.162	
0.902	1.951	0.01	4.949	8.684	-0.061	-0.074	-0.097	-0.112	-0.144	-0.163	0.104	0.155	
0.900	2.937	0.03	4.934	8.711	-0.063	-0.076	-0.100	-0.116	-0.150	-0.170	0.099	0.151	
0.898	3.375	0.02	4.925	8.724	-0.065	-0.078	-0.102	-0.118	-0.152	-0.171	0.099	0.150	
0.900	3.913	0.01	4.935	8.707	-0.062	-0.076	-0.100	-0.116	-0.150	-0.170	0.100	0.153	
0.900	5.836	0.03	4.934	8.708	-0.062	-0.074	-0.097	-0.109	-0.140	-0.155	0.112	0.164	
0.900	7.817	0.04	4.938	8.704	-0.057	-0.069	-0.089	-0.099	-0.125	-0.136	0.126	0.179	
0.900	3.431	8.08	4.935	8.708	-0.039	-0.045	-0.067	-0.085	-0.136	-0.193	0.108	0.157	
0.902	3.384	3.98	4.945	8.690	-0.053	-0.064	-0.089	-0.105	-0.149	-0.185	0.097	0.143	
0.802	1.470	-0.01	4.340	9.642	-0.062	-0.069	-0.086	-0.096	-0.128	-0.146	0.096	0.154	
0.801	1.953	0.01	4.334	9.650	-0.064	-0.071	-0.089	-0.100	-0.134	-0.153	0.089	0.145	
0.801	2.935	0.03	4.336	9.648	-0.065	-0.073	-0.091	-0.103	-0.138	-0.159	0.085	0.141	
0.801	3.395	0.03	4.332	9.653	-0.065	-0.073	-0.091	-0.102	-0.137	-0.158	0.085	0.141	
0.800	3.902	0.03	4.330	9.657	-0.065	-0.072	-0.090	-0.101	-0.136	-0.157	0.086	0.143	
0.801	5.864	0.04	4.331	9.655	-0.065	-0.070	-0.087	-0.096	-0.129	-0.145	0.098	0.156	
0.802	7.714	-0.03	4.338	9.645	-0.061	-0.065	-0.079	-0.086	-0.116	-0.128	0.115	0.174	
0.800	3.412	8.00	4.329	9.662	-0.034	-0.038	-0.058	-0.074	-0.127	-0.170	0.091	0.145	
0.800	3.401	3.99	4.328	9.661	-0.051	-0.056	-0.076	-0.088	-0.135	-0.166	0.084	0.137	
0.603	1.498	0.01	2.933	11.527	-0.052	-0.054	-0.068	-0.075	-0.106	-0.121	0.085	0.139	
0.603	1.945	0.02	2.933	11.528	-0.057	-0.059	-0.073	-0.081	-0.113	-0.130	0.074	0.124	
0.601	2.930	-0.02	2.919	11.545	-0.057	-0.060	-0.074	-0.082	-0.115	-0.133	0.069	0.118	
0.601	3.389	-0.02	2.921	11.543	-0.058	-0.061	-0.075	-0.082	-0.116	-0.136	0.068	0.118	
0.601	3.896	-0.01	2.920	11.543	-0.058	-0.061	-0.074	-0.081	-0.115	-0.134	0.070	0.120	
0.601	5.869	0.00	2.920	11.542	-0.057	-0.057	-0.070	-0.076	-0.108	-0.125	0.082	0.137	
0.601	7.809	0.01	2.921	11.540	-0.056	-0.055	-0.066	-0.070	-0.101	-0.113	0.096	0.155	
0.600	3.408	8.00	2.911	11.554	-0.025	-0.025	-0.040	-0.052	-0.105	-0.139	0.077	0.126	
0.601	3.406	4.00	2.918	11.543	-0.045	-0.044	-0.058	-0.066	-0.113	-0.139	0.070	0.118	

Table 5. Continued

(f) $\theta = 225^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.477	0.03	6.124	6.056	-0.223	-0.212	-0.156	-0.103	-0.162	-0.283	-0.381	-0.100	-0.036	0.009	
1.199	1.939	0.01	6.119	6.077	-0.226	-0.214	-0.156	-0.100	-0.160	-0.284	-0.399	-0.116	-0.051	-0.005	
1.200	2.923	0.02	6.120	6.072	-0.224	-0.213	-0.156	-0.100	-0.160	-0.284	-0.402	-0.144	-0.070	-0.022	
1.201	3.410	0.03	6.122	6.062	-0.221	-0.213	-0.158	-0.104	-0.164	-0.287	-0.402	-0.144	-0.072	-0.026	
1.202	3.914	0.02	6.123	6.057	-0.219	-0.212	-0.158	-0.104	-0.166	-0.287	-0.402	-0.143	-0.071	-0.026	
1.201	5.875	0.03	6.122	6.061	-0.218	-0.213	-0.158	-0.102	-0.164	-0.288	-0.393	-0.109	-0.044	-0.004	
1.201	7.816	0.02	6.123	6.062	-0.218	-0.213	-0.158	-0.104	-0.166	-0.290	-0.259	-0.065	-0.010	0.021	
1.200	3.407	8.04	6.118	6.074	-0.148	-0.111	-0.060	-0.048	-0.160	-0.280	-0.423	-0.391	-0.160	-0.095	
1.202	3.378	4.03	6.124	6.051	-0.218	-0.213	-0.152	-0.089	-0.146	-0.266	-0.393	-0.171	-0.074	-0.021	
0.902	1.513	0.00	4.946	8.689	-0.081	-0.069	-0.057	-0.070	-0.150	-0.158	0.071	0.120	0.151	0.176	
0.902	1.951	0.01	4.949	8.684	-0.083	-0.072	-0.061	-0.076	-0.159	-0.170	0.064	0.120	0.152	0.175	
0.900	2.937	0.03	4.934	8.711	-0.085	-0.075	-0.065	-0.082	-0.171	-0.189	0.054	0.116	0.151	0.172	
0.898	3.375	0.02	4.925	8.724	-0.087	-0.077	-0.066	-0.083	-0.173	-0.190	0.053	0.117	0.152	0.172	
0.900	3.913	0.01	4.935	8.707	-0.085	-0.075	-0.065	-0.082	-0.170	-0.187	0.055	0.117	0.153	0.174	
0.900	5.836	0.03	4.934	8.708	-0.083	-0.072	-0.060	-0.075	-0.156	-0.168	0.065	0.122	0.157	0.183	
0.900	7.817	0.04	4.938	8.704	-0.079	-0.065	-0.050	-0.061	-0.135	-0.138	0.080	0.128	0.160	0.193	
0.900	3.431	8.08	4.935	8.708	-0.069	-0.050	-0.037	-0.063	-0.180	-0.250	0.044	0.117	0.156	0.176	
0.902	3.384	3.98	4.945	8.690	-0.082	-0.067	-0.055	-0.076	-0.177	-0.216	0.046	0.104	0.133	0.151	
0.802	1.470	-0.01	4.340	9.642	-0.083	-0.067	-0.056	-0.072	-0.158	-0.184	0.048	0.128	0.169	0.194	
0.801	1.953	0.01	4.334	9.650	-0.085	-0.070	-0.059	-0.077	-0.167	-0.199	0.038	0.122	0.166	0.187	
0.801	2.935	0.03	4.336	9.648	-0.087	-0.073	-0.062	-0.082	-0.174	-0.208	0.033	0.119	0.162	0.182	
0.801	3.395	0.03	4.332	9.653	-0.087	-0.072	-0.062	-0.081	-0.173	-0.206	0.033	0.119	0.162	0.182	
0.800	3.902	0.03	4.330	9.657	-0.086	-0.071	-0.060	-0.079	-0.171	-0.205	0.034	0.120	0.165	0.185	
0.801	5.864	0.04	4.331	9.655	-0.085	-0.069	-0.056	-0.072	-0.156	-0.182	0.046	0.126	0.169	0.195	
0.802	7.714	-0.03	4.338	9.645	-0.080	-0.062	-0.047	-0.059	-0.135	-0.152	0.066	0.137	0.175	0.208	
0.800	3.412	8.00	4.329	9.662	-0.066	-0.046	-0.038	-0.067	-0.184	-0.251	0.022	0.116	0.164	0.185	
0.800	3.401	3.99	4.328	9.661	-0.079	-0.060	-0.049	-0.074	-0.175	-0.223	0.027	0.109	0.151	0.173	
0.603	1.498	0.01	2.933	11.527	-0.072	-0.056	-0.044	-0.059	-0.142	-0.168	0.035	0.123	0.169	0.187	
0.603	1.945	0.02	2.933	11.528	-0.077	-0.060	-0.050	-0.066	-0.153	-0.182	0.022	0.111	0.155	0.168	
0.601	2.930	-0.02	2.919	11.545	-0.079	-0.062	-0.052	-0.068	-0.156	-0.185	0.019	0.107	0.152	0.163	
0.601	3.389	-0.02	2.921	11.543	-0.081	-0.064	-0.053	-0.071	-0.158	-0.188	0.017	0.105	0.150	0.162	
0.601	3.896	-0.01	2.920	11.543	-0.080	-0.063	-0.053	-0.069	-0.155	-0.185	0.019	0.107	0.153	0.167	
0.601	5.869	0.00	2.920	11.542	-0.078	-0.059	-0.048	-0.064	-0.145	-0.170	0.030	0.118	0.165	0.188	
0.601	7.809	0.01	2.921	11.540	-0.076	-0.056	-0.043	-0.056	-0.131	-0.152	0.045	0.131	0.178	0.207	
0.600	3.408	8.00	2.911	11.554	-0.058	-0.037	-0.029	-0.058	-0.164	-0.216	0.006	0.104	0.156	0.171	
0.601	3.406	4.00	2.918	11.543	-0.073	-0.051	-0.042	-0.066	-0.161	-0.202	0.008	0.098	0.145	0.163	

Table 5. Continued

(g) $\theta = 240^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986
1.202	1.477	0.03	6.124	6.056	-0.212	-0.130	-0.089	-0.124	-0.393	-0.218	-0.114	-0.039
1.199	1.939	0.01	6.119	6.077	-0.215	-0.129	-0.087	-0.122	-0.393	-0.277	-0.127	-0.053
1.200	2.923	0.02	6.120	6.072	-0.215	-0.129	-0.086	-0.122	-0.393	-0.394	-0.145	-0.073
1.201	3.410	0.03	6.122	6.062	-0.217	-0.131	-0.089	-0.125	-0.392	-0.396	-0.148	-0.075
1.202	3.914	0.02	6.123	6.057	-0.218	-0.131	-0.090	-0.126	-0.392	-0.385	-0.149	-0.075
1.201	5.875	0.03	6.122	6.061	-0.219	-0.130	-0.089	-0.126	-0.393	-0.246	-0.128	-0.048
1.201	7.816	0.02	6.123	6.062	-0.220	-0.131	-0.090	-0.126	-0.392	-0.165	-0.083	-0.012
1.200	3.407	8.04	6.118	6.074	-0.113	-0.045	-0.029	-0.102	-0.392	-0.538	-0.259	-0.168
1.202	3.378	4.03	6.124	6.051	-0.219	-0.121	-0.071	-0.108	-0.382	-0.387	-0.164	-0.086
0.902	1.513	0.00	4.946	8.689	-0.077	-0.047	-0.046	-0.081	-0.154	0.081	0.105	0.139
0.902	1.951	0.01	4.949	8.684	-0.080	-0.052	-0.052	-0.090	-0.170	0.077	0.106	0.142
0.900	2.937	0.03	4.934	8.711	-0.084	-0.057	-0.059	-0.103	-0.196	0.070	0.104	0.141
0.898	3.375	0.02	4.925	8.724	-0.085	-0.058	-0.061	-0.105	-0.199	0.070	0.104	0.141
0.900	3.913	0.01	4.935	8.707	-0.083	-0.057	-0.059	-0.104	-0.196	0.071	0.104	0.141
0.900	5.836	0.03	4.934	8.708	-0.078	-0.051	-0.050	-0.088	-0.171	0.079	0.109	0.144
0.900	7.817	0.04	4.938	8.704	-0.070	-0.041	-0.036	-0.066	-0.135	0.090	0.111	0.143
0.900	3.431	8.08	4.935	8.708	-0.058	-0.035	-0.043	-0.109	-0.265	0.066	0.103	0.140
0.902	3.384	3.98	4.945	8.690	-0.073	-0.048	-0.052	-0.102	-0.223	0.063	0.083	0.112
0.802	1.470	-0.01	4.340	9.642	-0.075	-0.049	-0.052	-0.101	-0.219	0.074	0.126	0.165
0.801	1.953	0.01	4.334	9.650	-0.077	-0.052	-0.057	-0.112	-0.240	0.067	0.123	0.164
0.801	2.935	0.03	4.336	9.648	-0.080	-0.056	-0.063	-0.120	-0.253	0.062	0.121	0.162
0.801	3.395	0.03	4.332	9.653	-0.079	-0.056	-0.063	-0.119	-0.251	0.063	0.121	0.161
0.800	3.902	0.03	4.330	9.657	-0.077	-0.054	-0.060	-0.116	-0.251	0.064	0.125	0.164
0.801	5.864	0.04	4.331	9.655	-0.074	-0.050	-0.052	-0.100	-0.220	0.074	0.127	0.165
0.802	7.714	-0.03	4.338	9.645	-0.065	-0.039	-0.037	-0.075	-0.178	0.092	0.133	0.167
0.800	3.412	8.00	4.329	9.662	-0.053	-0.036	-0.050	-0.128	-0.317	0.057	0.122	0.165
0.800	3.401	3.99	4.328	9.661	-0.064	-0.046	-0.055	-0.116	-0.273	0.060	0.107	0.146
0.603	1.498	0.01	2.933	11.527	-0.060	-0.035	-0.042	-0.093	-0.224	0.065	0.136	0.172
0.603	1.945	0.02	2.933	11.528	-0.064	-0.041	-0.049	-0.104	-0.246	0.053	0.126	0.162
0.601	2.930	-0.02	2.919	11.545	-0.065	-0.044	-0.052	-0.107	-0.249	0.051	0.124	0.160
0.601	3.389	-0.02	2.921	11.543	-0.066	-0.045	-0.053	-0.108	-0.252	0.050	0.122	0.157
0.601	3.896	-0.01	2.920	11.543	-0.064	-0.045	-0.052	-0.106	-0.251	0.051	0.124	0.159
0.601	5.869	0.00	2.920	11.542	-0.059	-0.039	-0.045	-0.096	-0.233	0.064	0.134	0.169
0.601	7.809	0.01	2.921	11.540	-0.054	-0.034	-0.037	-0.081	-0.208	0.079	0.145	0.178
0.600	3.408	8.00	2.911	11.554	-0.035	-0.028	-0.042	-0.117	-0.309	0.045	0.130	0.169
0.601	3.406	4.00	2.918	11.543	-0.048	-0.038	-0.049	-0.111	-0.286	0.048	0.121	0.154

Table 5. Continued

(h) $\theta = 255^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.941	0.963	0.974	0.986	0.997
1.202	1.477	0.03	6.124	6.056	-0.134	-0.253	-0.097	-0.003	0.040
1.199	1.939	0.01	6.119	6.077	-0.133	-0.283	-0.123	-0.018	0.028
1.200	2.923	0.02	6.120	6.072	-0.133	-0.334	-0.157	-0.037	0.013
1.201	3.410	0.03	6.122	6.062	-0.136	-0.340	-0.161	-0.039	0.012
1.202	3.914	0.02	6.123	6.057	-0.137	-0.334	-0.159	-0.037	0.015
1.201	5.875	0.03	6.122	6.061	-0.136	-0.270	-0.114	-0.007	0.047
1.201	7.816	0.02	6.123	6.062	-0.137	-0.201	-0.062	0.030	0.084
1.200	3.407	8.04	6.118	6.074	-0.108	-0.528	-0.284	-0.182	-0.088
1.202	3.378	4.03	6.124	6.051	-0.120	-0.313	-0.167	-0.053	0.000
0.902	1.513	0.00	4.946	8.689	-0.044	0.075	0.102	0.137	0.174
0.902	1.951	0.01	4.949	8.684	-0.054	0.070	0.100	0.137	0.174
0.900	2.937	0.03	4.934	8.711	-0.069	0.063	0.096	0.137	0.174
0.898	3.375	0.02	4.925	8.724	-0.072	0.063	0.097	0.138	0.175
0.900	3.913	0.01	4.935	8.707	-0.071	0.065	0.099	0.139	0.177
0.900	5.836	0.03	4.934	8.708	-0.053	0.072	0.102	0.139	0.179
0.900	7.817	0.04	4.938	8.704	-0.028	0.082	0.108	0.141	0.180
0.900	3.431	8.08	4.935	8.708	-0.074	0.051	0.088	0.131	0.171
0.902	3.384	3.98	4.945	8.690	-0.062	0.048	0.073	0.103	0.137
0.802	1.470	-0.01	4.340	9.642	-0.076	0.076	0.116	0.160	0.198
0.801	1.953	0.01	4.334	9.650	-0.089	0.072	0.115	0.160	0.194
0.801	2.935	0.03	4.336	9.648	-0.098	0.068	0.116	0.160	0.191
0.801	3.395	0.03	4.332	9.653	-0.097	0.069	0.117	0.160	0.192
0.800	3.902	0.03	4.330	9.657	-0.094	0.070	0.114	0.160	0.193
0.801	5.864	0.04	4.331	9.655	-0.073	0.074	0.112	0.159	0.197
0.802	7.714	-0.03	4.338	9.645	-0.045	0.085	0.118	0.163	0.202
0.800	3.412	8.00	4.329	9.662	-0.108	0.065	0.115	0.158	0.192
0.800	3.401	3.99	4.328	9.661	-0.089	0.062	0.105	0.141	0.173
0.603	1.498	0.01	2.933	11.527	-0.074	0.079	0.131	0.172	0.195
0.603	1.945	0.02	2.933	11.528	-0.086	0.068	0.123	0.161	0.177
0.601	2.930	-0.02	2.919	11.545	-0.090	0.065	0.121	0.160	0.173
0.601	3.389	-0.02	2.921	11.543	-0.091	0.064	0.119	0.157	0.172
0.601	3.896	-0.01	2.920	11.543	-0.089	0.066	0.122	0.159	0.175
0.601	5.869	0.00	2.920	11.542	-0.076	0.077	0.130	0.168	0.192
0.601	7.809	0.01	2.921	11.540	-0.060	0.091	0.140	0.177	0.206
0.600	3.408	8.00	2.911	11.554	-0.104	0.066	0.133	0.168	0.182
0.601	3.406	4.00	2.918	11.543	-0.092	0.063	0.124	0.155	0.168

Table 5. Continued

(i) $\theta = 270^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.951	0.963	0.974	0.986	
1.202	1.477	0.03	6.124	6.056	-0.331	-0.174	-0.080	0.001	
1.199	1.939	0.01	6.119	6.077	-0.333	-0.202	-0.095	-0.009	
1.200	2.923	0.02	6.120	6.072	-0.332	-0.238	-0.114	-0.021	
1.201	3.410	0.03	6.122	6.062	-0.332	-0.241	-0.117	-0.022	
1.202	3.914	0.02	6.123	6.057	-0.331	-0.236	-0.117	-0.022	
1.201	5.875	0.03	6.122	6.061	-0.330	-0.185	-0.092	-0.002	
1.201	7.816	0.02	6.123	6.062	-0.310	-0.128	-0.054	0.029	
1.200	3.407	8.04	6.118	6.074	-0.358	-0.390	-0.227	-0.148	
1.202	3.378	4.03	6.124	6.051	-0.349	-0.214	-0.113	-0.038	
0.902	1.513	0.00	4.946	8.689	0.044	0.075	0.102	0.132	
0.902	1.951	0.01	4.949	8.684	0.037	0.069	0.099	0.133	
0.900	2.937	0.03	4.934	8.711	0.022	0.062	0.096	0.133	
0.898	3.375	0.02	4.925	8.724	0.019	0.061	0.096	0.134	
0.900	3.913	0.01	4.935	8.707	0.020	0.063	0.098	0.136	
0.900	5.836	0.03	4.934	8.708	0.038	0.068	0.101	0.135	
0.900	7.817	0.04	4.938	8.704	0.058	0.080	0.107	0.136	
0.900	3.431	8.08	4.935	8.708	0.019	0.050	0.085	0.126	
0.902	3.384	3.98	4.945	8.690	0.032	0.051	0.071	0.096	
0.802	1.470	-0.01	4.340	9.642	-0.005	0.073	0.117	0.160	
0.801	1.953	0.01	4.334	9.650	-0.022	0.068	0.117	0.160	
0.801	2.935	0.03	4.336	9.648	-0.034	0.065	0.117	0.159	
0.801	3.395	0.03	4.332	9.653	-0.035	0.066	0.119	0.161	
0.800	3.902	0.03	4.330	9.657	-0.029	0.066	0.116	0.160	
0.801	5.864	0.04	4.331	9.655	0.000	0.068	0.112	0.157	
0.802	7.714	-0.03	4.338	9.645	0.033	0.078	0.118	0.160	
0.800	3.412	8.00	4.329	9.662	-0.038	0.056	0.114	0.158	
0.800	3.401	3.99	4.328	9.661	-0.018	0.057	0.101	0.137	
0.603	1.498	0.01	2.933	11.527	-0.029	0.076	0.133	0.172	
0.603	1.945	0.02	2.933	11.528	-0.043	0.068	0.126	0.165	
0.601	2.930	-0.02	2.919	11.545	-0.048	0.065	0.124	0.162	
0.601	3.389	-0.02	2.921	11.543	-0.050	0.062	0.123	0.160	
0.601	3.896	-0.01	2.920	11.543	-0.048	0.064	0.125	0.162	
0.601	5.869	0.00	2.920	11.542	-0.034	0.073	0.133	0.170	
0.601	7.809	0.01	2.921	11.540	-0.015	0.085	0.143	0.177	
0.600	3.408	8.00	2.911	11.554	-0.068	0.062	0.135	0.171	
0.601	3.406	4.00	2.918	11.543	-0.052	0.058	0.125	0.156	

Table 5. Continued

(j) $\theta = 285^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.477	0.03	6.124	6.056	-0.095	-0.129	-0.407	-0.160	-0.144	-0.073	0.019	
1.199	1.939	0.01	6.119	6.077	-0.092	-0.129	-0.410	-0.180	-0.161	-0.086	0.008	
1.200	2.923	0.02	6.120	6.072	-0.094	-0.130	-0.409	-0.202	-0.181	-0.103	-0.006	
1.201	3.410	0.03	6.122	6.062	-0.098	-0.132	-0.410	-0.204	-0.185	-0.107	-0.007	
1.202	3.914	0.02	6.123	6.057	-0.101	-0.134	-0.411	-0.202	-0.185	-0.109	-0.006	
1.201	5.875	0.03	6.122	6.061	-0.099	-0.134	-0.410	-0.169	-0.158	-0.088	0.014	
1.201	7.816	0.02	6.123	6.062	-0.100	-0.135	-0.401	-0.120	-0.113	-0.054	0.044	
1.200	3.407	8.04	6.118	6.074	-0.214	-0.212	-0.448	-0.302	-0.252	-0.190	-0.099	
1.202	3.378	4.03	6.124	6.051	-0.176	-0.191	-0.426	-0.198	-0.176	-0.102	-0.018	
0.902	1.513	0.00	4.946	8.689	-0.004	-0.008	0.016	0.072	0.098	0.136	0.178	
0.902	1.951	0.01	4.949	8.684	-0.012	-0.019	0.001	0.068	0.098	0.137	0.181	
0.900	2.937	0.03	4.934	8.711	-0.024	-0.039	-0.021	0.062	0.095	0.137	0.179	
0.898	3.375	0.02	4.925	8.724	-0.028	-0.043	-0.023	0.062	0.095	0.137	0.180	
0.900	3.913	0.01	4.935	8.707	-0.025	-0.042	-0.021	0.063	0.095	0.138	0.181	
0.900	5.836	0.03	4.934	8.708	-0.013	-0.020	-0.001	0.071	0.100	0.140	0.182	
0.900	7.817	0.04	4.938	8.704	0.009	0.012	0.032	0.080	0.105	0.141	0.182	
0.900	3.431	8.08	4.935	8.708	-0.016	-0.007	0.005	0.050	0.083	0.132	0.182	
0.902	3.384	3.98	4.945	8.690	0.000	0.011	0.028	0.049	0.069	0.101	0.147	
0.802	1.470	-0.01	4.340	9.642	-0.043	-0.073	-0.063	0.073	0.115	0.161	0.201	
0.801	1.953	0.01	4.334	9.650	-0.053	-0.088	-0.083	0.066	0.113	0.160	0.199	
0.801	2.935	0.03	4.336	9.648	-0.058	-0.098	-0.095	0.063	0.112	0.159	0.197	
0.801	3.395	0.03	4.332	9.653	-0.057	-0.097	-0.095	0.063	0.112	0.159	0.197	
0.800	3.902	0.03	4.330	9.657	-0.055	-0.093	-0.095	0.065	0.116	0.161	0.197	
0.801	5.864	0.04	4.331	9.655	-0.042	-0.070	-0.065	0.072	0.118	0.161	0.199	
0.802	7.714	-0.03	4.338	9.645	-0.023	-0.038	-0.024	0.087	0.124	0.163	0.204	
0.800	3.412	8.00	4.329	9.662	-0.065	-0.096	-0.092	0.051	0.111	0.164	0.201	
0.800	3.401	3.99	4.328	9.661	-0.048	-0.062	-0.037	0.045	0.094	0.144	0.184	
0.603	1.498	0.01	2.933	11.527	-0.046	-0.080	-0.096	0.077	0.127	0.172	0.200	
0.603	1.945	0.02	2.933	11.528	-0.053	-0.092	-0.114	0.064	0.119	0.162	0.184	
0.601	2.930	-0.02	2.919	11.545	-0.056	-0.097	-0.119	0.062	0.118	0.160	0.180	
0.601	3.389	-0.02	2.921	11.543	-0.057	-0.097	-0.121	0.060	0.116	0.159	0.178	
0.601	3.896	-0.01	2.920	11.543	-0.055	-0.095	-0.118	0.061	0.117	0.161	0.180	
0.601	5.869	0.00	2.920	11.542	-0.047	-0.083	-0.101	0.072	0.127	0.172	0.195	
0.601	7.809	0.01	2.921	11.540	-0.036	-0.065	-0.076	0.084	0.136	0.182	0.209	
0.600	3.408	8.00	2.911	11.554	-0.073	-0.115	-0.145	0.058	0.128	0.174	0.185	
0.601	3.406	4.00	2.918	11.543	-0.059	-0.088	-0.103	0.056	0.114	0.159	0.173	

Table 5. Continued

(k) $\theta = 300^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.477	0.03	6.124	6.056	-0.363	-0.246	-0.095	-0.099	-0.384	-0.100	-0.018	0.025	0.038	
1.199	1.939	0.01	6.119	6.077	-0.366	-0.233	-0.092	-0.098	-0.461	-0.124	-0.038	0.008	0.027	
1.200	2.923	0.02	6.120	6.072	-0.366	-0.238	-0.094	-0.100	-0.487	-0.152	-0.060	-0.011	0.014	
1.201	3.410	0.03	6.122	6.062	-0.366	-0.247	-0.099	-0.102	-0.487	-0.154	-0.061	-0.013	0.013	
1.202	3.914	0.02	6.123	6.057	-0.366	-0.253	-0.102	-0.104	-0.485	-0.150	-0.057	-0.011	0.016	
1.201	5.875	0.03	6.122	6.061	-0.367	-0.247	-0.101	-0.104	-0.421	-0.104	-0.017	0.022	0.040	
1.201	7.816	0.02	6.123	6.062	-0.368	-0.248	-0.103	-0.105	-0.230	-0.057	-0.023	0.060	0.065	
1.200	3.407	8.04	6.118	6.074	-0.447	-0.408	-0.221	-0.187	-0.502	-0.227	-0.171	-0.092	-0.033	
1.202	3.378	4.03	6.124	6.051	-0.410	-0.326	-0.180	-0.160	-0.327	-0.117	-0.052	-0.017	0.009	
0.902	1.513	0.00	4.946	8.689	-0.039	-0.016	-0.017	-0.039	0.084	0.107	0.135	0.174	0.139	
0.902	1.951	0.01	4.949	8.684	-0.043	-0.020	-0.024	-0.050	0.080	0.106	0.137	0.174	0.132	
0.900	2.937	0.03	4.934	8.711	-0.049	-0.028	-0.035	-0.066	0.073	0.104	0.137	0.174	0.133	
0.898	3.375	0.02	4.925	8.724	-0.052	-0.030	-0.038	-0.070	0.072	0.103	0.137	0.175	0.135	
0.900	3.913	0.01	4.935	8.707	-0.049	-0.028	-0.036	-0.069	0.075	0.105	0.139	0.177	0.137	
0.900	5.836	0.03	4.934	8.708	-0.044	-0.021	-0.024	-0.050	0.081	0.106	0.138	0.180	0.146	
0.900	7.817	0.04	4.938	8.704	-0.032	-0.006	-0.004	-0.020	0.090	0.110	0.138	0.182	0.153	
0.900	3.431	8.08	4.935	8.708	-0.081	-0.042	-0.024	-0.037	0.057	0.090	0.126	0.175	0.125	
0.902	3.384	3.98	4.945	8.690	-0.044	-0.018	-0.010	-0.018	0.054	0.074	0.099	0.144	0.114	
0.802	1.470	-0.01	4.340	9.642	-0.072	-0.045	-0.052	-0.092	0.081	0.122	0.159	0.197	0.157	
0.801	1.953	0.01	4.334	9.650	-0.077	-0.051	-0.059	-0.104	0.074	0.121	0.158	0.194	0.145	
0.801	2.935	0.03	4.336	9.648	-0.079	-0.054	-0.064	-0.111	0.069	0.119	0.157	0.190	0.141	
0.801	3.395	0.03	4.332	9.653	-0.078	-0.053	-0.062	-0.110	0.068	0.121	0.158	0.190	0.142	
0.800	3.902	0.03	4.330	9.657	-0.077	-0.052	-0.061	-0.108	0.071	0.118	0.156	0.191	0.146	
0.801	5.864	0.04	4.331	9.655	-0.072	-0.045	-0.050	-0.089	0.079	0.117	0.156	0.198	0.158	
0.802	7.714	-0.03	4.338	9.645	-0.063	-0.033	-0.032	-0.063	0.093	0.123	0.160	0.205	0.169	
0.800	3.412	8.00	4.329	9.662	-0.099	-0.071	-0.071	-0.119	0.065	0.116	0.152	0.193	0.131	
0.800	3.401	3.99	4.328	9.661	-0.082	-0.057	-0.054	-0.086	0.060	0.105	0.135	0.178	0.131	
0.603	1.498	0.01	2.933	11.527	-0.073	-0.043	-0.050	-0.087	0.073	0.135	0.169	0.192	0.146	
0.603	1.945	0.02	2.933	11.528	-0.077	-0.048	-0.057	-0.099	0.060	0.125	0.158	0.177	0.118	
0.601	2.930	-0.02	2.919	11.545	-0.078	-0.051	-0.059	-0.102	0.056	0.122	0.155	0.171	0.113	
0.601	3.389	-0.02	2.921	11.543	-0.078	-0.051	-0.059	-0.103	0.055	0.121	0.152	0.170	0.114	
0.601	3.896	-0.01	2.920	11.543	-0.078	-0.051	-0.058	-0.101	0.057	0.122	0.154	0.175	0.119	
0.601	5.869	0.00	2.920	11.542	-0.073	-0.045	-0.050	-0.090	0.069	0.131	0.164	0.194	0.143	
0.601	7.809	0.01	2.921	11.540	-0.067	-0.039	-0.040	-0.074	0.086	0.141	0.173	0.210	0.157	
0.600	3.408	8.00	2.911	11.554	-0.109	-0.076	-0.075	-0.125	0.053	0.124	0.153	0.182	0.108	
0.601	3.406	4.00	2.918	11.543	-0.088	-0.061	-0.059	-0.096	0.051	0.114	0.141	0.172	0.109	

Table 5. Continued

(1) $\theta = 315^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.202	1.477	0.03	6.124	6.056	-0.298	-0.320	-0.311	-0.092	-0.097	-0.250	-0.290	-0.110	-0.045	0.015
1.199	1.939	0.01	6.119	6.077	-0.300	-0.320	-0.305	-0.086	-0.096	-0.251	-0.355	-0.126	-0.059	0.002
1.200	2.923	0.02	6.120	6.072	-0.300	-0.319	-0.307	-0.088	-0.096	-0.251	-0.391	-0.151	-0.076	-0.015
1.201	3.410	0.03	6.122	6.062	-0.299	-0.316	-0.311	-0.092	-0.097	-0.251	-0.391	-0.154	-0.080	-0.020
1.202	3.914	0.02	6.123	6.057	-0.299	-0.316	-0.314	-0.095	-0.099	-0.252	-0.389	-0.153	-0.081	-0.021
1.201	5.875	0.03	6.122	6.061	-0.299	-0.315	-0.311	-0.092	-0.097	-0.252	-0.322	-0.118	-0.059	-0.001
1.201	7.816	0.02	6.123	6.062	-0.300	-0.314	-0.311	-0.093	-0.098	-0.252	-0.180	-0.077	-0.024	0.023
1.200	3.407	8.04	6.118	6.074	-0.389	-0.403	-0.412	-0.263	-0.203	-0.327	-0.461	-0.239	-0.167	-0.096
1.202	3.378	4.03	6.124	6.051	-0.346	-0.367	-0.374	-0.179	-0.154	-0.289	-0.331	-0.148	-0.078	-0.029
0.902	1.513	0.00	4.946	8.689	-0.054	-0.031	-0.020	-0.034	-0.070	-0.102	0.081	0.117	0.137	0.172
0.902	1.951	0.01	4.949	8.684	-0.056	-0.033	-0.024	-0.039	-0.079	-0.115	0.073	0.117	0.137	0.171
0.900	2.937	0.03	4.934	8.711	-0.061	-0.039	-0.031	-0.048	-0.092	-0.135	0.063	0.113	0.137	0.171
0.898	3.375	0.02	4.925	8.724	-0.062	-0.042	-0.034	-0.051	-0.096	-0.138	0.061	0.113	0.137	0.173
0.900	3.913	0.01	4.935	8.707	-0.060	-0.039	-0.031	-0.049	-0.094	-0.136	0.064	0.114	0.139	0.173
0.900	5.836	0.03	4.934	8.708	-0.058	-0.035	-0.025	-0.040	-0.079	-0.114	0.074	0.119	0.143	0.181
0.900	7.817	0.04	4.938	8.704	-0.049	-0.026	-0.012	-0.022	-0.053	-0.079	0.092	0.123	0.146	0.188
0.900	3.431	8.08	4.935	8.708	-0.129	-0.075	-0.043	-0.042	-0.075	-0.118	0.067	0.106	0.134	0.171
0.902	3.384	3.98	4.945	8.690	-0.074	-0.041	-0.023	-0.030	-0.053	-0.064	0.063	0.081	0.104	0.139
0.802	1.470	-0.01	4.340	9.642	-0.075	-0.061	-0.048	-0.063	-0.106	-0.156	0.051	0.126	0.156	0.193
0.801	1.953	0.01	4.334	9.650	-0.077	-0.064	-0.053	-0.069	-0.116	-0.170	0.042	0.121	0.152	0.189
0.801	2.935	0.03	4.336	9.648	-0.079	-0.067	-0.055	-0.073	-0.121	-0.179	0.037	0.117	0.149	0.182
0.801	3.395	0.03	4.332	9.653	-0.079	-0.067	-0.055	-0.072	-0.121	-0.179	0.037	0.117	0.150	0.183
0.800	3.902	0.03	4.330	9.657	-0.077	-0.066	-0.054	-0.071	-0.119	-0.177	0.037	0.119	0.153	0.185
0.801	5.864	0.04	4.331	9.655	-0.076	-0.063	-0.048	-0.062	-0.105	-0.155	0.050	0.126	0.158	0.196
0.802	7.714	-0.03	4.338	9.645	-0.069	-0.055	-0.038	-0.048	-0.085	-0.123	0.072	0.136	0.165	0.207
0.800	3.412	8.00	4.329	9.662	-0.109	-0.093	-0.074	-0.086	-0.139	-0.205	0.037	0.117	0.154	0.187
0.800	3.401	3.99	4.328	9.661	-0.092	-0.079	-0.060	-0.072	-0.107	-0.140	0.044	0.105	0.137	0.173
0.603	1.498	0.01	2.933	11.527	-0.074	-0.060	-0.044	-0.060	-0.097	-0.143	0.039	0.121	0.155	0.188
0.603	1.945	0.02	2.933	11.528	-0.078	-0.065	-0.049	-0.066	-0.105	-0.157	0.026	0.109	0.144	0.172
0.601	2.930	-0.02	2.919	11.545	-0.078	-0.067	-0.051	-0.069	-0.109	-0.162	0.023	0.105	0.141	0.165
0.601	3.389	-0.02	2.921	11.543	-0.079	-0.069	-0.052	-0.070	-0.110	-0.164	0.020	0.104	0.139	0.165
0.601	3.896	-0.01	2.920	11.543	-0.079	-0.070	-0.052	-0.069	-0.110	-0.162	0.023	0.105	0.143	0.169
0.601	5.869	0.00	2.920	11.542	-0.075	-0.066	-0.048	-0.063	-0.101	-0.148	0.034	0.117	0.158	0.191
0.601	7.809	0.01	2.921	11.540	-0.072	-0.063	-0.042	-0.055	-0.089	-0.130	0.050	0.131	0.172	0.211
0.600	3.408	8.00	2.911	11.554	-0.119	-0.107	-0.080	-0.093	-0.141	-0.201	0.013	0.106	0.150	0.175
0.601	3.406	4.00	2.918	11.543	-0.095	-0.088	-0.065	-0.077	-0.113	-0.150	0.024	0.102	0.143	0.167

Table 5. Continued

(m) Top interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—				
					0.709	0.723	0.898	0.941	
1.202	1.477	0.03	6.124	6.056	-0.042	-0.036	-0.353	-0.145	
1.199	1.939	0.01	6.119	6.077	-0.044	-0.037	-0.354	-0.144	
1.200	2.923	0.02	6.120	6.072	-0.045	-0.038	-0.355	-0.145	
1.201	3.410	0.03	6.122	6.062	-0.044	-0.037	-0.355	-0.147	
1.202	3.914	0.02	6.123	6.057	-0.045	-0.036	-0.356	-0.148	
1.201	5.875	0.03	6.122	6.061	-0.046	-0.036	-0.357	-0.147	
1.201	7.816	0.02	6.123	6.062	-0.047	-0.036	-0.357	-0.147	
1.200	3.407	8.04	6.118	6.074	-0.180	-0.170	-0.436	-0.217	
1.202	3.378	4.03	6.124	6.051	-0.127	-0.123	-0.400	-0.199	
0.902	1.513	0.00	4.946	8.689	0.001	0.024	-0.040	0.011	
0.902	1.951	0.01	4.949	8.684	-0.002	0.022	-0.043	0.000	
0.900	2.937	0.03	4.934	8.711	0.000	0.022	-0.050	-0.018	
0.898	3.375	0.02	4.925	8.724	-0.004	0.020	-0.052	-0.022	
0.900	3.913	0.01	4.935	8.707	-0.002	0.021	-0.049	-0.022	
0.900	5.836	0.03	4.934	8.708	-0.003	0.020	-0.045	-0.001	
0.900	7.817	0.04	4.938	8.704	-0.001	0.022	-0.034	0.030	
0.900	3.431	8.08	4.935	8.708	-0.042	-0.014	-0.088	0.001	
0.902	3.384	3.98	4.945	8.690	-0.005	0.008	-0.048	0.020	
0.802	1.470	-0.01	4.340	9.642	-0.022	-0.021	-0.070	-0.054	
0.801	1.953	0.01	4.334	9.650	-0.024	-0.022	-0.074	-0.069	
0.801	2.935	0.03	4.336	9.648	-0.023	-0.021	-0.077	-0.080	
0.801	3.395	0.03	4.332	9.653	-0.024	-0.022	-0.075	-0.079	
0.800	3.902	0.03	4.330	9.657	-0.024	-0.023	-0.074	-0.076	
0.801	5.864	0.04	4.331	9.655	-0.027	-0.024	-0.070	-0.051	
0.802	7.714	-0.03	4.338	9.645	-0.021	-0.022	-0.061	-0.019	
0.800	3.412	8.00	4.329	9.662	-0.073	-0.070	-0.097	-0.076	
0.800	3.401	3.99	4.328	9.661	-0.049	-0.053	-0.081	-0.047	
0.603	1.498	0.01	2.933	11.527	-0.039	-0.044	-0.070	-0.064	
0.603	1.945	0.02	2.933	11.528	-0.040	-0.045	-0.074	-0.077	
0.601	2.930	-0.02	2.919	11.545	-0.037	-0.044	-0.075	-0.081	
0.601	3.389	-0.02	2.921	11.543	-0.038	-0.045	-0.076	-0.082	
0.601	3.896	-0.01	2.920	11.543	-0.038	-0.045	-0.076	-0.082	
0.601	5.869	0.00	2.920	11.542	-0.037	-0.046	-0.071	-0.068	
0.601	7.809	0.01	2.921	11.540	-0.034	-0.044	-0.064	-0.050	
0.600	3.408	8.00	2.911	11.554	-0.091	-0.108	-0.105	-0.098	
0.601	3.406	4.00	2.918	11.543	-0.063	-0.078	-0.084	-0.076	

Table 5. Concluded

(n) Bottom interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.709	0.723	0.738	0.752	0.767	0.781	0.869	0.883	0.941	
1.202	1.477	0.03	6.124	6.056	-0.021	-0.047	-0.072	-0.086	-0.095	-0.097	-0.215	-0.231	-0.144	
1.199	1.939	0.01	6.119	6.077	-0.023	-0.048	-0.073	-0.087	-0.096	-0.098	-0.218	-0.235	-0.143	
1.200	2.923	0.02	6.120	6.072	-0.022	-0.047	-0.072	-0.085	-0.096	-0.097	-0.217	-0.233	-0.143	
1.201	3.410	0.03	6.122	6.062	-0.021	-0.046	-0.072	-0.085	-0.098	-0.100	-0.214	-0.232	-0.145	
1.202	3.914	0.02	6.123	6.057	-0.019	-0.044	-0.070	-0.083	-0.096	-0.100	-0.213	-0.230	-0.145	
1.201	5.875	0.03	6.122	6.061	-0.019	-0.044	-0.070	-0.083	-0.097	-0.101	-0.213	-0.231	-0.146	
1.201	7.816	0.02	6.123	6.062	-0.019	-0.042	-0.069	-0.082	-0.096	-0.101	-0.213	-0.231	-0.145	
1.200	3.407	8.04	6.118	6.074	0.181	0.146	0.111	0.087	0.057	0.022	-0.157	-0.154	-0.120	
1.202	3.378	4.03	6.124	6.051	0.069	0.050	0.033	0.027	0.010	-0.018	-0.209	-0.230	-0.134	
0.902	1.513	0.00	4.946	8.689	-0.061	-0.062	-0.060	-0.059	-0.067	-0.090	-0.074	-0.078	-0.027	
0.902	1.951	0.01	4.949	8.684	-0.061	-0.062	-0.061	-0.058	-0.066	-0.090	-0.075	-0.080	-0.038	
0.900	2.937	0.03	4.934	8.711	-0.060	-0.063	-0.062	-0.061	-0.069	-0.092	-0.077	-0.083	-0.054	
0.898	3.375	0.02	4.925	8.724	-0.062	-0.065	-0.064	-0.062	-0.071	-0.093	-0.079	-0.084	-0.057	
0.900	3.913	0.01	4.935	8.707	-0.062	-0.064	-0.063	-0.061	-0.069	-0.092	-0.077	-0.082	-0.056	
0.900	5.836	0.03	4.934	8.708	-0.063	-0.065	-0.064	-0.061	-0.070	-0.092	-0.075	-0.079	-0.038	
0.900	7.817	0.04	4.938	8.704	-0.063	-0.063	-0.061	-0.060	-0.067	-0.089	-0.071	-0.074	-0.011	
0.900	3.431	8.08	4.935	8.708	-0.038	-0.072	-0.089	-0.100	-0.115	-0.148	-0.067	-0.063	-0.058	
0.902	3.384	3.98	4.945	8.690	-0.048	-0.065	-0.071	-0.073	-0.081	-0.107	-0.073	-0.076	-0.043	
0.802	1.470	-0.01	4.340	9.642	-0.047	-0.059	-0.062	-0.062	-0.070	-0.091	-0.078	-0.079	-0.066	
0.801	1.953	0.01	4.334	9.650	-0.046	-0.058	-0.061	-0.063	-0.070	-0.092	-0.078	-0.080	-0.080	
0.801	2.935	0.03	4.336	9.648	-0.047	-0.060	-0.063	-0.063	-0.071	-0.092	-0.080	-0.083	-0.090	
0.801	3.395	0.03	4.332	9.653	-0.047	-0.060	-0.063	-0.063	-0.071	-0.092	-0.080	-0.083	-0.090	
0.800	3.902	0.03	4.330	9.657	-0.046	-0.059	-0.062	-0.063	-0.070	-0.091	-0.079	-0.081	-0.086	
0.801	5.864	0.04	4.331	9.655	-0.048	-0.061	-0.064	-0.064	-0.071	-0.092	-0.079	-0.080	-0.064	
0.802	7.714	-0.03	4.338	9.645	-0.047	-0.060	-0.062	-0.063	-0.069	-0.089	-0.075	-0.074	-0.034	
0.800	3.412	8.00	4.329	9.662	-0.004	-0.029	-0.039	-0.046	-0.056	-0.083	-0.058	-0.059	-0.102	
0.800	3.401	3.99	4.328	9.661	-0.021	-0.039	-0.045	-0.049	-0.056	-0.080	-0.068	-0.071	-0.079	
0.603	1.498	0.01	2.933	11.527	-0.035	-0.049	-0.052	-0.054	-0.060	-0.080	-0.067	-0.068	-0.068	
0.603	1.945	0.02	2.933	11.528	-0.038	-0.051	-0.054	-0.055	-0.062	-0.082	-0.071	-0.072	-0.081	
0.601	2.930	-0.02	2.919	11.545	-0.038	-0.052	-0.055	-0.056	-0.062	-0.082	-0.072	-0.073	-0.085	
0.601	3.389	-0.02	2.921	11.543	-0.039	-0.053	-0.056	-0.057	-0.062	-0.082	-0.073	-0.075	-0.086	
0.601	3.896	-0.01	2.920	11.543	-0.039	-0.052	-0.055	-0.057	-0.062	-0.081	-0.072	-0.074	-0.084	
0.601	5.869	0.00	2.920	11.542	-0.038	-0.052	-0.055	-0.057	-0.061	-0.080	-0.070	-0.071	-0.071	
0.601	7.809	0.01	2.921	11.540	-0.040	-0.053	-0.055	-0.057	-0.061	-0.079	-0.068	-0.069	-0.056	
0.600	3.408	8.00	2.911	11.554	0.014	-0.005	-0.012	-0.018	-0.022	-0.046	-0.045	-0.047	-0.103	
0.601	3.406	4.00	2.918	11.543	-0.010	-0.027	-0.031	-0.035	-0.038	-0.058	-0.060	-0.062	-0.088	

Table 6. Static Pressure Coefficients of Afterbody Surface With Horizontal Tail in Aft Position and Vertical Tail in Mid Position

(a) $\theta = 0^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854	
1.202	1.485	0.01	6.141	6.068	-0.047	-0.072	-0.077	-0.067	-0.038	-0.022	0.029	0.019	-0.006	-0.033	-0.081	
1.203	1.986	0.02	6.142	6.063	-0.046	-0.070	-0.077	-0.067	-0.039	-0.023	0.029	0.019	-0.006	-0.033	-0.082	
1.203	2.955	0.03	6.143	6.062	-0.046	-0.070	-0.077	-0.068	-0.039	-0.022	0.029	0.019	-0.006	-0.033	-0.083	
1.201	3.398	0.04	6.138	6.082	-0.049	-0.071	-0.078	-0.070	-0.041	-0.024	0.029	0.018	-0.008	-0.035	-0.086	
1.200	3.912	0.04	6.137	6.084	-0.049	-0.071	-0.079	-0.071	-0.042	-0.025	0.028	0.017	-0.009	-0.036	-0.087	
1.200	5.854	0.06	6.136	6.087	-0.051	-0.072	-0.080	-0.073	-0.043	-0.026	0.026	0.015	-0.010	-0.038	-0.089	
1.201	7.886	0.07	6.138	6.079	-0.051	-0.071	-0.080	-0.075	-0.045	-0.026	0.026	0.016	-0.009	-0.037	-0.088	
1.200	7.867	0.02	6.137	6.086	-0.052	-0.071	-0.080	-0.075	-0.045	-0.026	0.027	0.016	-0.009	-0.037	-0.089	
1.200	3.417	8.05	6.135	6.089	-0.192	-0.211	-0.210	-0.191	-0.146	-0.113	-0.111	-0.135	-0.168	-0.170	-0.207	
1.198	3.387	4.04	6.133	6.101	-0.131	-0.149	-0.149	-0.137	-0.105	-0.084	-0.070	-0.084	-0.117	-0.136	-0.177	
0.900	1.481	0.02	4.949	8.734	-0.034	-0.018	-0.004	0.013	0.049	0.045	-0.004	-0.067	-0.145	-0.223	-0.319	
0.901	1.971	0.03	4.955	8.724	-0.036	-0.020	-0.006	0.012	0.047	0.045	-0.003	-0.066	-0.143	-0.221	-0.318	
0.900	2.943	0.01	4.951	8.732	-0.035	-0.018	-0.005	0.012	0.048	0.045	-0.004	-0.067	-0.144	-0.223	-0.319	
0.901	3.369	0.02	4.954	8.727	-0.035	-0.018	-0.005	0.012	0.048	0.045	-0.003	-0.067	-0.144	-0.222	-0.319	
0.899	3.919	0.03	4.943	8.746	-0.035	-0.018	-0.005	0.012	0.048	0.044	-0.005	-0.069	-0.146	-0.225	-0.322	
0.900	5.836	0.02	4.949	8.735	-0.036	-0.020	-0.007	0.011	0.047	0.043	-0.005	-0.069	-0.146	-0.224	-0.320	
0.899	7.828	0.04	4.947	8.739	-0.034	-0.019	-0.005	0.012	0.048	0.044	-0.004	-0.068	-0.145	-0.223	-0.320	
0.900	3.439	8.03	4.949	8.736	-0.108	-0.105	-0.097	-0.077	-0.034	-0.034	-0.084	-0.153	-0.233	-0.311	-0.406	
0.900	3.411	4.02	4.948	8.736	-0.046	-0.038	-0.029	-0.013	0.019	0.012	-0.041	-0.108	-0.186	-0.265	-0.359	
0.800	1.481	0.01	4.339	9.694	-0.033	-0.031	-0.027	-0.020	0.005	-0.021	-0.090	-0.165	-0.250	-0.331	-0.404	
0.801	1.971	0.02	4.345	9.685	-0.033	-0.032	-0.028	-0.021	0.004	-0.022	-0.091	-0.166	-0.252	-0.333	-0.408	
0.801	2.944	0.03	4.348	9.680	-0.036	-0.033	-0.028	-0.021	0.004	-0.022	-0.092	-0.166	-0.253	-0.337	-0.415	
0.800	3.400	0.02	4.340	9.691	-0.036	-0.033	-0.028	-0.021	0.003	-0.023	-0.092	-0.167	-0.253	-0.335	-0.412	
0.800	3.936	0.02	4.339	9.692	-0.037	-0.034	-0.029	-0.023	0.002	-0.024	-0.094	-0.170	-0.256	-0.339	-0.416	
0.801	5.970	0.01	4.345	9.684	-0.035	-0.034	-0.028	-0.020	0.006	-0.021	-0.091	-0.165	-0.251	-0.333	-0.407	
0.801	7.851	0.02	4.350	9.677	-0.034	-0.032	-0.027	-0.018	0.007	-0.020	-0.089	-0.162	-0.248	-0.328	-0.398	
0.799	3.419	8.01	4.334	9.704	-0.097	-0.094	-0.086	-0.076	-0.050	-0.089	-0.175	-0.262	-0.359	-0.454	-0.564	
0.799	3.383	4.03	4.338	9.696	-0.056	-0.053	-0.046	-0.037	-0.018	-0.054	-0.137	-0.219	-0.311	-0.402	-0.499	
0.601	1.488	0.00	2.928	11.578	-0.029	-0.032	-0.035	-0.032	-0.017	-0.051	-0.115	-0.155	-0.198	-0.225	-0.249	
0.602	1.957	0.01	2.939	11.567	-0.029	-0.033	-0.035	-0.032	-0.019	-0.054	-0.119	-0.160	-0.203	-0.229	-0.253	
0.601	2.949	0.02	2.925	11.582	-0.030	-0.034	-0.037	-0.033	-0.020	-0.054	-0.118	-0.160	-0.203	-0.230	-0.254	
0.602	3.368	0.02	2.936	11.568	-0.030	-0.035	-0.037	-0.034	-0.020	-0.055	-0.120	-0.161	-0.204	-0.231	-0.255	
0.601	3.922	0.02	2.929	11.577	-0.029	-0.035	-0.037	-0.032	-0.019	-0.056	-0.121	-0.161	-0.205	-0.232	-0.255	
0.601	5.881	0.04	2.929	11.576	-0.029	-0.035	-0.036	-0.030	-0.017	-0.054	-0.118	-0.158	-0.200	-0.227	-0.249	
0.601	7.860	0.02	2.931	11.575	-0.027	-0.036	-0.037	-0.030	-0.016	-0.054	-0.118	-0.157	-0.200	-0.225	-0.245	
0.599	3.402	8.01	2.913	11.598	-0.081	-0.090	-0.085	-0.079	-0.070	-0.130	-0.219	-0.269	-0.313	-0.335	-0.343	
0.601	3.397	4.00	2.929	11.577	-0.050	-0.059	-0.058	-0.049	-0.042	-0.092	-0.170	-0.213	-0.254	-0.278	-0.292	

Table 6. Continued

(a) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.869	0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	
1.202	1.485	0.01	6.141	6.068	-0.127	-0.170	-0.215	-0.273	-0.340	-0.285	-0.238	-0.337	-0.209	
1.203	1.986	0.02	6.142	6.063	-0.128	-0.170	-0.215	-0.272	-0.340	-0.287	-0.239	-0.337	-0.271	
1.203	2.955	0.03	6.143	6.062	-0.130	-0.172	-0.217	-0.273	-0.341	-0.291	-0.241	-0.338	-0.303	
1.201	3.398	0.04	6.138	6.082	-0.134	-0.174	-0.220	-0.275	-0.344	-0.294	-0.245	-0.343	-0.307	
1.200	3.912	0.04	6.137	6.084	-0.135	-0.175	-0.221	-0.274	-0.344	-0.295	-0.246	-0.344	-0.304	
1.200	5.854	0.06	6.136	6.087	-0.137	-0.177	-0.224	-0.276	-0.346	-0.297	-0.249	-0.346	-0.245	
1.201	7.886	0.07	6.138	6.079	-0.138	-0.177	-0.224	-0.275	-0.345	-0.300	-0.250	-0.344	-0.145	
1.200	7.867	0.02	6.137	6.086	-0.139	-0.177	-0.225	-0.275	-0.346	-0.300	-0.250	-0.346	-0.146	
1.200	3.417	8.05	6.135	6.089	-0.254	-0.280	-0.321	-0.375	-0.444	-0.477	-0.359	-0.428	-0.280	
1.198	3.387	4.04	6.133	6.101	-0.226	-0.254	-0.293	-0.345	-0.415	-0.343	-0.303	-0.381	-0.238	
0.900	1.481	0.02	4.949	8.734	-0.404	-0.475	-0.545	-0.264	-0.083	-0.053	-0.082	0.038	0.104	
0.901	1.971	0.03	4.955	8.724	-0.403	-0.473	-0.547	-0.294	-0.086	-0.057	-0.087	0.033	0.101	
0.900	2.943	0.01	4.951	8.732	-0.405	-0.475	-0.553	-0.324	-0.091	-0.061	-0.095	0.028	0.098	
0.901	3.369	0.02	4.954	8.727	-0.404	-0.475	-0.553	-0.344	-0.093	-0.062	-0.096	0.027	0.098	
0.899	3.919	0.03	4.943	8.746	-0.408	-0.478	-0.556	-0.318	-0.091	-0.062	-0.096	0.028	0.099	
0.900	5.836	0.02	4.949	8.735	-0.406	-0.477	-0.551	-0.286	-0.086	-0.055	-0.085	0.037	0.105	
0.899	7.828	0.04	4.947	8.739	-0.405	-0.475	-0.538	-0.223	-0.076	-0.047	-0.070	0.048	0.113	
0.900	3.439	8.03	4.949	8.736	-0.487	-0.554	-0.576	-0.250	-0.119	-0.074	-0.062	0.026	0.073	
0.900	3.411	4.02	4.948	8.736	-0.444	-0.507	-0.527	-0.214	-0.097	-0.071	-0.087	0.009	0.060	
0.800	1.481	0.01	4.339	9.694	-0.382	-0.308	-0.225	-0.152	-0.102	-0.087	-0.140	0.006	0.095	
0.801	1.971	0.02	4.345	9.685	-0.390	-0.313	-0.229	-0.156	-0.107	-0.092	-0.148	-0.002	0.087	
0.801	2.944	0.03	4.348	9.680	-0.397	-0.318	-0.233	-0.159	-0.111	-0.097	-0.154	-0.007	0.081	
0.800	3.400	0.02	4.340	9.691	-0.394	-0.317	-0.233	-0.159	-0.111	-0.097	-0.152	-0.007	0.082	
0.800	3.936	0.02	4.339	9.692	-0.398	-0.318	-0.233	-0.160	-0.111	-0.097	-0.152	-0.006	0.082	
0.801	5.970	0.01	4.345	9.684	-0.385	-0.310	-0.226	-0.154	-0.103	-0.087	-0.138	0.006	0.096	
0.801	7.851	0.02	4.350	9.677	-0.375	-0.301	-0.217	-0.145	-0.094	-0.074	-0.120	0.024	0.112	
0.799	3.419	8.01	4.334	9.704	-0.633	-0.410	-0.253	-0.176	-0.123	-0.098	-0.126	0.003	0.081	
0.799	3.383	4.03	4.338	9.696	-0.468	-0.335	-0.236	-0.164	-0.111	-0.091	-0.129	0.001	0.081	
0.601	1.488	0.00	2.928	11.578	-0.233	-0.206	-0.169	-0.134	-0.102	-0.087	-0.132	-0.003	0.083	
0.602	1.957	0.01	2.939	11.567	-0.237	-0.210	-0.173	-0.139	-0.108	-0.094	-0.141	-0.011	0.072	
0.601	2.949	0.02	2.925	11.582	-0.238	-0.211	-0.175	-0.142	-0.111	-0.097	-0.145	-0.016	0.067	
0.602	3.368	0.02	2.936	11.568	-0.237	-0.212	-0.174	-0.142	-0.111	-0.096	-0.143	-0.015	0.068	
0.601	3.922	0.02	2.929	11.577	-0.236	-0.212	-0.173	-0.143	-0.111	-0.095	-0.141	-0.015	0.069	
0.601	5.881	0.04	2.929	11.576	-0.230	-0.206	-0.167	-0.139	-0.104	-0.088	-0.131	-0.004	0.082	
0.601	7.860	0.02	2.931	11.575	-0.225	-0.202	-0.161	-0.134	-0.099	-0.079	-0.117	0.008	0.097	
0.599	3.402	8.01	2.913	11.598	-0.308	-0.275	-0.220	-0.185	-0.139	-0.111	-0.141	-0.020	0.062	
0.601	3.397	4.00	2.929	11.577	-0.263	-0.237	-0.189	-0.163	-0.121	-0.097	-0.130	-0.015	0.067	

Table 6. Continued

(b) $\theta = 45^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.912	0.941	0.951	0.963	0.974	0.997
1.202	1.485	0.01	6.141	6.068	-0.049	0.009	0.017	-0.023	-0.057	1.147	-0.365	-0.121	-0.191	-0.317	-0.262	0.011
1.203	1.986	0.02	6.142	6.063	-0.049	0.009	0.017	-0.022	-0.055	1.154	-0.365	-0.122	-0.191	-0.316	-0.285	-0.012
1.203	2.955	0.03	6.143	6.062	-0.050	0.008	0.017	-0.021	-0.054	1.158	-0.366	-0.123	-0.191	-0.316	-0.296	-0.028
1.201	3.398	0.04	6.138	6.082	-0.051	0.007	0.017	-0.023	-0.057	1.159	-0.369	-0.125	-0.194	-0.318	-0.300	-0.030
1.200	3.912	0.04	6.137	6.084	-0.053	0.006	0.016	-0.023	-0.057	1.162	-0.369	-0.127	-0.194	-0.318	-0.299	-0.024
1.200	5.854	0.06	6.136	6.087	-0.056	0.004	0.015	-0.024	-0.057	1.163	-0.369	-0.128	-0.195	-0.318	-0.291	0.009
1.201	7.886	0.07	6.138	6.079	-0.056	0.002	0.015	-0.023	-0.057	1.164	-0.368	-0.129	-0.196	-0.318	-0.240	0.035
1.200	7.867	0.02	6.137	6.086	-0.056	0.002	0.014	-0.024	-0.057	1.163	-0.367	-0.129	-0.197	-0.318	-0.240	0.035
1.200	3.417	8.05	6.135	6.089	-0.175	-0.086	-0.087	-0.107	-0.119	0.981	-0.488	-0.299	-0.339	-0.434	-0.268	-0.059
1.198	3.387	4.04	6.133	6.101	-0.117	-0.045	-0.050	-0.080	-0.110	0.973	-0.442	-0.187	-0.245	-0.362	-0.297	-0.045
0.900	1.481	0.02	4.949	8.734	0.007	0.030	0.020	0.024	0.062	0.509	-0.276	-0.011	-0.044	0.019	0.097	0.129
0.901	1.971	0.03	4.955	8.724	0.005	0.030	0.020	0.025	0.063	0.501	-0.282	-0.014	-0.048	0.015	0.093	0.120
0.900	2.943	0.01	4.951	8.732	0.006	0.028	0.018	0.023	0.060	0.489	-0.290	-0.017	-0.052	0.011	0.090	0.121
0.901	3.369	0.02	4.954	8.727	0.004	0.028	0.017	0.022	0.061	0.483	-0.290	-0.017	-0.052	0.011	0.089	0.122
0.899	3.919	0.03	4.943	8.746	0.008	0.030	0.019	0.023	0.062	0.476	-0.289	-0.016	-0.051	0.012	0.090	0.126
0.900	5.836	0.02	4.949	8.735	0.004	0.028	0.018	0.023	0.062	0.471	-0.279	-0.011	-0.043	0.020	0.099	0.133
0.899	7.828	0.04	4.947	8.739	0.005	0.029	0.017	0.022	0.061	0.464	-0.267	-0.004	-0.032	0.031	0.110	0.139
0.900	3.439	8.03	4.949	8.736	-0.077	-0.062	-0.087	-0.086	-0.036	0.471	-0.308	-0.037	-0.032	0.019	0.077	0.100
0.900	3.411	4.02	4.948	8.736	0.000	0.014	-0.008	-0.007	0.034	0.470	-0.272	-0.025	-0.053	0.003	0.072	0.095
0.800	1.481	0.01	4.339	9.694	0.000	0.012	-0.003	-0.001	0.033	0.242	-0.129	-0.042	-0.075	0.000	0.085	0.150
0.801	1.971	0.02	4.345	9.685	0.000	0.013	-0.002	-0.001	0.032	0.236	-0.132	-0.046	-0.081	-0.006	0.078	0.138
0.801	2.944	0.03	4.348	9.680	-0.002	0.011	-0.004	-0.002	0.031	0.229	-0.135	-0.049	-0.085	-0.011	0.073	0.136
0.800	3.400	0.02	4.340	9.691	-0.001	0.012	-0.003	-0.001	0.032	0.222	-0.135	-0.049	-0.085	-0.011	0.074	0.137
0.800	3.936	0.02	4.339	9.692	-0.001	0.011	-0.005	-0.004	0.029	0.216	-0.136	-0.050	-0.086	-0.011	0.073	0.139
0.801	5.970	0.01	4.345	9.684	0.001	0.013	-0.003	-0.002	0.032	0.211	-0.130	-0.041	-0.075	0.001	0.086	0.153
0.801	7.851	0.02	4.350	9.677	0.002	0.014	-0.003	-0.002	0.033	0.205	-0.123	-0.032	-0.063	0.014	0.101	0.163
0.799	3.419	8.01	4.334	9.704	-0.049	-0.036	-0.056	-0.055	-0.017	0.211	-0.178	-0.060	-0.068	-0.003	0.070	0.121
0.799	3.383	4.03	4.338	9.696	-0.007	0.005	-0.016	-0.016	0.021	0.212	-0.149	-0.051	-0.081	-0.011	0.069	0.124
0.601	1.488	0.00	2.928	11.578	0.002	0.010	-0.006	-0.004	0.024	-0.245	-0.117	-0.033	-0.065	0.002	0.080	0.149
0.602	1.957	0.01	2.939	11.567	0.001	0.010	-0.005	-0.004	0.023	-0.253	-0.122	-0.037	-0.071	-0.005	0.071	0.125
0.601	2.949	0.02	2.925	11.582	0.001	0.009	-0.008	-0.007	0.021	-0.282	-0.125	-0.040	-0.076	-0.010	0.064	0.118
0.602	3.368	0.02	2.936	11.568	0.001	0.009	-0.008	-0.007	0.021	-0.282	-0.125	-0.040	-0.076	-0.010	0.063	0.121
0.601	3.922	0.02	2.929	11.577	0.004	0.011	-0.006	-0.005	0.024	-0.291	-0.124	-0.038	-0.076	-0.010	0.064	0.128
0.601	5.881	0.04	2.929	11.576	0.006	0.012	-0.006	-0.005	0.024	-0.304	-0.121	-0.033	-0.068	-0.003	0.074	0.152
0.601	7.860	0.02	2.931	11.575	0.006	0.013	-0.006	-0.006	0.024	-0.311	-0.116	-0.026	-0.060	0.005	0.085	0.166
0.599	3.402	8.01	2.913	11.598	-0.028	-0.018	-0.037	-0.038	-0.007	-0.305	-0.160	-0.065	-0.083	-0.021	0.054	0.122
0.601	3.397	4.00	2.929	11.577	0.003	0.008	-0.016	-0.016	0.015	-0.298	-0.139	-0.046	-0.080	-0.019	0.057	0.124

Table 6. Continued

(c) $\theta = 90^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—												
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.825	0.951	0.963	0.974	0.986	0.997
1.202	1.485	0.01	6.141	6.068	-0.057	-0.003	0.039	-0.058	-0.099	-0.098	-0.120	-0.075	-0.412	-0.337	-0.214	-0.092	-0.023
1.203	1.986	0.02	6.142	6.063	-0.057	-0.003	0.039	-0.059	-0.099	-0.098	-0.120	-0.075	-0.410	-0.336	-0.230	-0.134	-0.045
1.203	2.955	0.03	6.143	6.062	-0.057	-0.004	0.039	-0.059	-0.099	-0.098	-0.121	-0.077	-0.409	-0.337	-0.235	-0.156	-0.055
1.201	3.398	0.04	6.138	6.082	-0.059	-0.004	0.036	-0.062	-0.101	-0.100	-0.122	-0.082	-0.412	-0.339	-0.238	-0.155	-0.054
1.200	3.912	0.04	6.137	6.084	-0.059	-0.004	0.035	-0.063	-0.102	-0.100	-0.122	-0.083	-0.411	-0.339	-0.238	-0.142	-0.050
1.200	5.854	0.06	6.136	6.087	-0.061	-0.006	0.034	-0.065	-0.103	-0.101	-0.123	-0.086	-0.410	-0.341	-0.220	-0.078	-0.029
1.201	7.886	0.07	6.138	6.079	-0.061	-0.006	0.033	-0.065	-0.103	-0.101	-0.123	-0.086	-0.409	-0.337	-0.160	-0.052	0.004
1.200	7.867	0.02	6.137	6.086	-0.064	-0.008	0.031	-0.066	-0.103	-0.102	-0.123	-0.089	-0.409	-0.338	-0.161	-0.053	0.005
1.200	3.417	8.05	6.135	6.089	0.098	0.064	0.004	-0.078	-0.113	-0.064	-0.078	-0.031	-0.468	-0.381	-0.218	-0.142	-0.093
1.198	3.387	4.04	6.133	6.101	0.035	0.011	0.013	-0.078	-0.125	-0.103	-0.128	-0.068	-0.432	-0.351	-0.246	-0.156	-0.070
0.900	1.481	0.02	4.949	8.734	0.006	0.025	0.012	-0.034	-0.001	0.008	-0.001	0.086	-0.202	-0.056	0.092	0.159	0.199
0.901	1.971	0.03	4.955	8.724	0.006	0.024	0.012	-0.034	-0.001	0.008	-0.002	0.086	-0.208	-0.060	0.088	0.154	0.192
0.900	2.943	0.01	4.951	8.732	0.007	0.024	0.011	-0.034	-0.001	0.007	-0.003	0.084	-0.206	-0.062	0.087	0.152	0.187
0.901	3.369	0.02	4.954	8.727	0.006	0.023	0.011	-0.035	-0.002	0.007	-0.004	0.084	-0.245	-0.048	0.087	0.152	0.188
0.899	3.919	0.03	4.943	8.746	0.007	0.024	0.011	-0.035	-0.002	0.007	-0.003	0.085	-0.243	-0.047	0.088	0.153	0.190
0.900	5.836	0.02	4.949	8.735	0.004	0.022	0.011	-0.035	-0.002	0.006	-0.003	0.085	-0.232	-0.038	0.098	0.164	0.205
0.899	7.828	0.04	4.947	8.739	0.006	0.023	0.011	-0.034	-0.001	0.008	-0.002	0.086	-0.213	-0.023	0.108	0.175	0.219
0.900	3.439	8.03	4.949	8.736	-0.052	-0.056	-0.086	-0.187	-0.103	-0.074	-0.080	0.064	-0.223	-0.027	0.093	0.144	0.175
0.900	3.411	4.02	4.948	8.736	-0.005	0.003	-0.015	-0.076	-0.027	-0.010	-0.019	0.087	-0.224	-0.066	0.084	0.148	0.178
0.800	1.481	0.01	4.339	9.694	0.004	0.010	-0.008	-0.055	-0.021	-0.010	-0.013	0.078	-0.191	-0.045	0.100	0.162	0.204
0.801	1.971	0.02	4.345	9.685	0.004	0.009	-0.009	-0.056	-0.022	-0.011	-0.014	0.076	-0.197	-0.051	0.094	0.155	0.195
0.801	2.944	0.03	4.348	9.680	0.002	0.008	-0.009	-0.057	-0.022	-0.012	-0.015	0.076	-0.202	-0.055	0.089	0.150	0.187
0.800	3.400	0.02	4.340	9.691	0.002	0.008	-0.009	-0.056	-0.023	-0.012	-0.016	0.076	-0.204	-0.055	0.090	0.150	0.188
0.800	3.936	0.02	4.339	9.692	0.002	0.007	-0.010	-0.057	-0.023	-0.012	-0.016	0.075	-0.202	-0.056	0.090	0.151	0.189
0.801	5.970	0.01	4.345	9.684	0.003	0.007	-0.008	-0.055	-0.021	-0.011	-0.014	0.077	-0.193	-0.044	0.101	0.165	0.207
0.801	7.851	0.02	4.350	9.677	0.004	0.008	-0.006	-0.054	-0.020	-0.009	-0.013	0.077	-0.182	-0.030	0.114	0.179	0.224
0.799	3.419	8.01	4.334	9.704	-0.008	-0.019	-0.042	-0.114	-0.058	-0.038	-0.041	0.078	-0.143	-0.001	0.101	0.148	0.181
0.799	3.383	4.03	4.338	9.696	0.010	0.007	-0.009	-0.061	-0.021	-0.007	-0.010	0.084	-0.159	-0.042	0.092	0.150	0.184
0.601	1.488	0.00	2.928	11.578	0.006	0.006	-0.009	-0.048	-0.019	-0.012	-0.006	0.078	-0.142	-0.015	0.107	0.157	0.194
0.602	1.957	0.01	2.939	11.567	0.007	0.004	-0.010	-0.050	-0.020	-0.013	-0.007	0.076	-0.149	-0.020	0.100	0.147	0.179
0.601	2.949	0.02	2.925	11.582	0.006	0.004	-0.010	-0.050	-0.022	-0.013	-0.009	0.075	-0.153	-0.022	0.095	0.140	0.169
0.602	3.368	0.02	2.936	11.568	0.005	0.003	-0.010	-0.050	-0.021	-0.012	-0.009	0.074	-0.155	-0.023	0.096	0.141	0.169
0.601	3.922	0.02	2.929	11.577	0.006	0.003	-0.009	-0.048	-0.020	-0.010	-0.008	0.075	-0.155	-0.024	0.096	0.143	0.172
0.601	5.881	0.04	2.929	11.576	0.007	0.003	-0.006	-0.045	-0.018	-0.008	-0.007	0.076	-0.149	-0.018	0.105	0.156	0.191
0.601	7.860	0.02	2.931	11.575	0.007	0.004	-0.005	-0.044	-0.017	-0.007	-0.006	0.075	-0.139	-0.010	0.116	0.170	0.208
0.599	3.402	8.01	2.913	11.598	0.025	0.006	-0.007	-0.061	-0.024	-0.007	-0.011	0.086	-0.158	0.002	0.101	0.141	0.165
0.601	3.397	4.00	2.929	11.577	0.024	0.012	0.004	-0.036	-0.009	0.004	0.003	0.084	-0.149	-0.020	0.098	0.144	0.168

Table 6. Continued

(d) $\theta = 135^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.869	
1.202	1.485	0.01	6.141	6.068	-0.059	-0.067	-0.079	-0.060	-0.070	-0.099	-0.108	-0.142	-0.142	-0.111	-0.094	
1.203	1.986	0.02	6.142	6.063	-0.059	-0.069	-0.082	-0.061	-0.070	-0.100	-0.107	-0.141	-0.141	-0.111	-0.095	
1.203	2.955	0.03	6.143	6.062	-0.060	-0.070	-0.083	-0.061	-0.070	-0.101	-0.109	-0.142	-0.141	-0.112	-0.097	
1.201	3.398	0.04	6.138	6.082	-0.061	-0.071	-0.083	-0.060	-0.073	-0.103	-0.110	-0.144	-0.144	-0.115	-0.098	
1.200	3.912	0.04	6.137	6.084	-0.062	-0.072	-0.084	-0.060	-0.073	-0.105	-0.111	-0.145	-0.144	-0.115	-0.100	
1.200	5.854	0.06	6.136	6.087	-0.064	-0.073	-0.085	-0.060	-0.074	-0.106	-0.112	-0.145	-0.144	-0.116	-0.100	
1.201	7.886	0.07	6.138	6.079	-0.065	-0.074	-0.085	-0.060	-0.073	-0.106	-0.112	-0.144	-0.144	-0.116	-0.102	
1.200	7.867	0.02	6.137	6.086	-0.067	-0.075	-0.087	-0.060	-0.075	-0.107	-0.112	-0.146	-0.145	-0.118	-0.104	
1.200	3.417	8.05	6.135	6.089	0.144	0.102	0.045	0.002	-0.038	-0.073	-0.073	-0.126	-0.119	-0.059	-0.017	
1.198	3.387	4.04	6.133	6.101	0.057	0.036	-0.011	-0.031	-0.050	-0.079	-0.097	-0.157	-0.165	-0.136	-0.070	
0.900	1.481	0.02	4.949	8.734	-0.039	-0.033	-0.055	-0.038	-0.030	-0.023	-0.016	-0.011	0.012	0.008	-0.049	
0.901	1.971	0.03	4.955	8.724	-0.040	-0.034	-0.056	-0.039	-0.030	-0.023	-0.017	-0.011	0.012	0.008	-0.049	
0.900	2.943	0.01	4.951	8.732	-0.039	-0.034	-0.056	-0.038	-0.031	-0.024	-0.018	-0.013	0.010	0.007	-0.052	
0.901	3.369	0.02	4.954	8.727	-0.040	-0.035	-0.056	-0.039	-0.032	-0.025	-0.018	-0.013	0.011	0.008	-0.050	
0.899	3.919	0.03	4.943	8.746	-0.039	-0.034	-0.056	-0.039	-0.031	-0.024	-0.018	-0.013	0.010	0.008	-0.052	
0.900	5.836	0.02	4.949	8.735	-0.041	-0.036	-0.056	-0.040	-0.032	-0.025	-0.019	-0.014	0.010	0.008	-0.050	
0.899	7.828	0.04	4.947	8.739	-0.038	-0.033	-0.054	-0.038	-0.030	-0.023	-0.017	-0.012	0.012	0.010	-0.049	
0.900	3.439	8.03	4.949	8.736	-0.071	-0.093	-0.138	-0.139	-0.129	-0.114	-0.096	-0.073	-0.021	0.003	-0.015	
0.900	3.411	4.02	4.948	8.736	-0.047	-0.052	-0.082	-0.068	-0.057	-0.044	-0.034	-0.022	0.010	0.013	-0.035	
0.800	1.481	0.01	4.339	9.694	-0.032	-0.039	-0.067	-0.050	-0.041	-0.032	-0.025	-0.019	0.003	0.001	-0.042	
0.801	1.971	0.02	4.345	9.685	-0.033	-0.041	-0.068	-0.052	-0.043	-0.034	-0.026	-0.020	0.003	0.000	-0.042	
0.801	2.944	0.03	4.348	9.680	-0.032	-0.040	-0.068	-0.052	-0.043	-0.034	-0.026	-0.020	0.001	-0.002	-0.044	
0.800	3.400	0.02	4.340	9.691	-0.033	-0.041	-0.068	-0.053	-0.043	-0.034	-0.027	-0.021	0.001	-0.002	-0.044	
0.800	3.936	0.02	4.339	9.692	-0.034	-0.041	-0.069	-0.053	-0.044	-0.035	-0.028	-0.022	0.000	-0.003	-0.046	
0.801	5.970	0.01	4.345	9.684	-0.032	-0.040	-0.067	-0.052	-0.042	-0.032	-0.026	-0.020	0.002	-0.001	-0.043	
0.801	7.851	0.02	4.350	9.677	-0.032	-0.039	-0.067	-0.052	-0.041	-0.030	-0.025	-0.019	0.003	0.001	-0.040	
0.799	3.419	8.01	4.334	9.704	-0.025	-0.046	-0.085	-0.081	-0.071	-0.059	-0.049	-0.038	0.001	0.017	0.010	
0.799	3.383	4.03	4.338	9.696	-0.016	-0.029	-0.062	-0.052	-0.040	-0.027	-0.022	-0.013	0.014	0.017	-0.015	
0.601	1.488	0.00	2.928	11.578	-0.023	-0.035	-0.061	-0.045	-0.034	-0.027	-0.017	-0.015	0.004	0.002	-0.028	
0.602	1.957	0.01	2.939	11.567	-0.024	-0.035	-0.062	-0.047	-0.036	-0.028	-0.018	-0.017	0.003	0.000	-0.030	
0.601	2.949	0.02	2.925	11.582	-0.025	-0.037	-0.063	-0.048	-0.036	-0.029	-0.019	-0.018	0.002	-0.002	-0.032	
0.602	3.368	0.02	2.936	11.568	-0.024	-0.036	-0.062	-0.048	-0.036	-0.027	-0.019	-0.017	0.002	-0.002	-0.031	
0.601	3.922	0.02	2.929	11.577	-0.024	-0.035	-0.061	-0.049	-0.036	-0.026	-0.018	-0.016	0.002	-0.002	-0.031	
0.601	5.881	0.04	2.929	11.576	-0.023	-0.034	-0.061	-0.051	-0.035	-0.024	-0.017	-0.015	0.003	-0.001	-0.027	
0.601	7.860	0.02	2.931	11.575	-0.023	-0.034	-0.061	-0.051	-0.034	-0.022	-0.017	-0.015	0.003	-0.001	-0.025	
0.599	3.402	8.01	2.913	11.598	0.004	-0.016	-0.052	-0.052	-0.035	-0.022	-0.018	-0.013	0.016	0.027	0.034	
0.601	3.397	4.00	2.929	11.577	0.002	-0.014	-0.046	-0.042	-0.023	-0.009	-0.007	-0.003	0.018	0.020	0.006	

Table 6. Continued

(d) Concluded

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.485	0.01	6.141	6.068	-0.093	-0.084	-0.105	-0.143	-0.156	-0.226	-0.356	-0.290	-0.114	-0.001	
1.203	1.986	0.02	6.142	6.063	-0.096	-0.083	-0.106	-0.143	-0.157	-0.226	-0.358	-0.295	-0.175	-0.033	
1.203	2.955	0.03	6.143	6.062	-0.098	-0.085	-0.107	-0.144	-0.158	-0.228	-0.361	-0.299	-0.206	-0.057	
1.201	3.398	0.04	6.138	6.082	-0.096	-0.085	-0.109	-0.146	-0.161	-0.229	-0.365	-0.301	-0.210	-0.059	
1.200	3.912	0.04	6.137	6.084	-0.097	-0.086	-0.111	-0.147	-0.162	-0.230	-0.366	-0.302	-0.202	-0.053	
1.200	5.854	0.06	6.136	6.087	-0.096	-0.086	-0.112	-0.148	-0.163	-0.231	-0.368	-0.299	-0.132	-0.007	
1.201	7.886	0.07	6.138	6.079	-0.100	-0.087	-0.112	-0.148	-0.164	-0.232	-0.370	-0.273	-0.051	0.032	
1.200	7.867	0.02	6.137	6.086	-0.100	-0.087	-0.113	-0.149	-0.166	-0.233	-0.372	-0.276	-0.052	0.032	
1.200	3.417	8.05	6.135	6.089	-0.046	-0.078	-0.107	-0.129	-0.130	-0.203	-0.322	-0.319	-0.241	-0.124	
1.198	3.387	4.04	6.133	6.101	-0.066	-0.061	-0.082	-0.118	-0.133	-0.206	-0.345	-0.306	-0.239	-0.100	
0.900	1.481	0.02	4.949	8.734	-0.065	-0.084	-0.103	-0.131	-0.110	-0.141	0.002	0.100	0.157	0.173	
0.901	1.971	0.03	4.955	8.724	-0.066	-0.085	-0.105	-0.134	-0.116	-0.148	-0.005	0.093	0.149	0.161	
0.900	2.943	0.01	4.951	8.732	-0.067	-0.086	-0.106	-0.135	-0.117	-0.150	-0.007	0.091	0.147	0.160	
0.901	3.369	0.02	4.954	8.727	-0.069	-0.087	-0.108	-0.137	-0.118	-0.150	-0.007	0.091	0.147	0.160	
0.899	3.919	0.03	4.943	8.746	-0.066	-0.085	-0.105	-0.133	-0.115	-0.148	-0.006	0.093	0.149	0.163	
0.900	5.836	0.02	4.949	8.735	-0.067	-0.085	-0.104	-0.131	-0.110	-0.139	0.005	0.104	0.160	0.176	
0.899	7.828	0.04	4.947	8.739	-0.064	-0.081	-0.100	-0.125	-0.101	-0.124	0.019	0.118	0.172	0.190	
0.900	3.439	8.03	4.949	8.736	-0.023	-0.040	-0.060	-0.092	-0.090	-0.146	-0.014	0.078	0.133	0.133	
0.900	3.411	4.02	4.948	8.736	-0.051	-0.072	-0.094	-0.125	-0.118	-0.153	-0.020	0.074	0.127	0.126	
0.800	1.481	0.01	4.339	9.694	-0.052	-0.067	-0.082	-0.101	-0.081	-0.115	0.003	0.094	0.153	0.175	
0.801	1.971	0.02	4.345	9.685	-0.054	-0.069	-0.084	-0.105	-0.086	-0.122	-0.004	0.086	0.143	0.163	
0.801	2.944	0.03	4.348	9.680	-0.056	-0.071	-0.086	-0.108	-0.089	-0.126	-0.008	0.083	0.140	0.157	
0.800	3.400	0.02	4.340	9.691	-0.055	-0.071	-0.085	-0.107	-0.088	-0.125	-0.007	0.084	0.140	0.158	
0.800	3.936	0.02	4.339	9.692	-0.056	-0.072	-0.087	-0.108	-0.088	-0.126	-0.007	0.084	0.141	0.160	
0.801	5.970	0.01	4.345	9.684	-0.053	-0.068	-0.082	-0.102	-0.080	-0.114	0.005	0.097	0.155	0.178	
0.801	7.851	0.02	4.350	9.677	-0.050	-0.065	-0.077	-0.095	-0.071	-0.101	0.021	0.114	0.172	0.196	
0.799	3.419	8.01	4.334	9.704	0.001	-0.019	-0.036	-0.064	-0.059	-0.109	-0.001	0.080	0.132	0.144	
0.799	3.383	4.03	4.338	9.696	-0.028	-0.048	-0.065	-0.090	-0.079	-0.113	-0.005	0.078	0.128	0.143	
0.601	1.488	0.00	2.928	11.578	-0.034	-0.047	-0.057	-0.074	-0.054	-0.089	-0.007	0.086	0.143	0.166	
0.602	1.957	0.01	2.939	11.567	-0.037	-0.050	-0.061	-0.078	-0.059	-0.096	-0.001	0.076	0.130	0.146	
0.601	2.949	0.02	2.925	11.582	-0.038	-0.052	-0.062	-0.081	-0.063	-0.101	-0.004	0.072	0.124	0.137	
0.602	3.368	0.02	2.936	11.568	-0.038	-0.052	-0.062	-0.080	-0.062	-0.101	-0.003	0.073	0.124	0.139	
0.601	3.922	0.02	2.929	11.577	-0.037	-0.052	-0.061	-0.079	-0.061	-0.100	0.000	0.075	0.125	0.144	
0.601	5.881	0.04	2.929	11.576	-0.034	-0.049	-0.057	-0.075	-0.055	-0.091	0.011	0.088	0.139	0.168	
0.601	7.860	0.02	2.931	11.575	-0.032	-0.047	-0.054	-0.071	-0.048	-0.082	0.023	0.102	0.154	0.187	
0.599	3.402	8.01	2.913	11.598	0.028	0.006	-0.007	-0.035	-0.026	-0.077	0.018	0.082	0.120	0.136	
0.601	3.397	4.00	2.929	11.577	-0.004	-0.026	-0.036	-0.061	-0.048	-0.085	0.012	0.077	0.115	0.135	

Table 6. Continued

(e) $\theta = 180^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
1.202	1.485	0.01	6.141	6.068	-0.018	-0.052	-0.073	-0.076	-0.074	-0.076	-0.099	-0.126	-0.153	-0.167	-0.175
1.203	1.986	0.02	6.142	6.063	-0.018	-0.052	-0.075	-0.079	-0.076	-0.077	-0.099	-0.125	-0.152	-0.166	-0.176
1.203	2.955	0.03	6.143	6.062	-0.019	-0.052	-0.076	-0.080	-0.077	-0.078	-0.101	-0.125	-0.151	-0.166	-0.178
1.201	3.398	0.04	6.138	6.082	-0.021	-0.052	-0.076	-0.081	-0.077	-0.079	-0.103	-0.127	-0.153	-0.168	-0.181
1.200	3.912	0.04	6.137	6.084	-0.022	-0.052	-0.077	-0.082	-0.078	-0.080	-0.105	-0.128	-0.153	-0.169	-0.182
1.200	5.854	0.06	6.136	6.087	-0.022	-0.052	-0.077	-0.082	-0.076	-0.080	-0.105	-0.128	-0.153	-0.169	-0.183
1.201	7.886	0.07	6.138	6.079	-0.023	-0.050	-0.077	-0.082	-0.076	-0.080	-0.106	-0.127	-0.152	-0.168	-0.184
1.200	7.867	0.02	6.137	6.086	-0.025	-0.051	-0.078	-0.084	-0.078	-0.081	-0.107	-0.129	-0.153	-0.170	-0.185
1.200	3.417	8.05	6.135	6.089	0.184	0.138	0.103	0.080	0.055	0.023	-0.030	-0.068	-0.100	-0.127	-0.140
1.198	3.387	4.04	6.133	6.101	0.077	0.052	0.030	0.021	0.009	-0.015	-0.058	-0.093	-0.128	-0.164	-0.190
0.900	1.481	0.02	4.949	8.734	-0.050	-0.054	-0.052	-0.043	-0.036	-0.043	-0.050	-0.044	-0.039	-0.043	-0.050
0.901	1.971	0.03	4.955	8.724	-0.052	-0.055	-0.052	-0.043	-0.037	-0.044	-0.051	-0.045	-0.039	-0.043	-0.050
0.900	2.943	0.01	4.951	8.732	-0.050	-0.054	-0.052	-0.044	-0.037	-0.044	-0.052	-0.045	-0.040	-0.044	-0.051
0.901	3.369	0.02	4.954	8.727	-0.052	-0.055	-0.053	-0.044	-0.038	-0.045	-0.052	-0.046	-0.041	-0.044	-0.052
0.899	3.919	0.03	4.943	8.746	-0.050	-0.054	-0.052	-0.044	-0.037	-0.044	-0.052	-0.045	-0.040	-0.044	-0.051
0.900	5.836	0.02	4.949	8.735	-0.054	-0.056	-0.055	-0.045	-0.039	-0.046	-0.053	-0.046	-0.041	-0.044	-0.051
0.899	7.828	0.04	4.947	8.739	-0.050	-0.053	-0.052	-0.043	-0.037	-0.044	-0.051	-0.044	-0.040	-0.042	-0.049
0.900	3.439	8.03	4.949	8.736	-0.039	-0.075	-0.092	-0.094	-0.095	-0.104	-0.107	-0.088	-0.067	-0.056	-0.048
0.900	3.411	4.02	4.948	8.736	-0.042	-0.060	-0.065	-0.058	-0.053	-0.060	-0.064	-0.054	-0.045	-0.043	-0.046
0.800	1.481	0.01	4.339	9.694	-0.038	-0.053	-0.056	-0.048	-0.044	-0.050	-0.056	-0.050	-0.047	-0.047	-0.049
0.801	1.971	0.02	4.345	9.685	-0.040	-0.054	-0.056	-0.047	-0.044	-0.050	-0.057	-0.051	-0.047	-0.048	-0.051
0.801	2.944	0.03	4.348	9.680	-0.040	-0.055	-0.057	-0.050	-0.046	-0.052	-0.059	-0.053	-0.049	-0.049	-0.052
0.800	3.400	0.02	4.340	9.691	-0.041	-0.055	-0.057	-0.050	-0.045	-0.052	-0.058	-0.052	-0.049	-0.049	-0.052
0.800	3.936	0.02	4.339	9.692	-0.040	-0.056	-0.058	-0.051	-0.046	-0.052	-0.059	-0.053	-0.050	-0.050	-0.052
0.801	5.970	0.01	4.345	9.684	-0.039	-0.055	-0.057	-0.049	-0.045	-0.051	-0.056	-0.052	-0.048	-0.047	-0.050
0.801	7.851	0.02	4.350	9.677	-0.039	-0.055	-0.056	-0.049	-0.045	-0.051	-0.055	-0.050	-0.047	-0.046	-0.047
0.799	3.419	8.01	4.334	9.704	-0.001	-0.032	-0.041	-0.039	-0.041	-0.050	-0.054	-0.047	-0.039	-0.031	-0.025
0.799	3.383	4.03	4.338	9.696	-0.012	-0.036	-0.041	-0.034	-0.034	-0.041	-0.046	-0.042	-0.038	-0.034	-0.034
0.601	1.488	0.00	2.928	11.578	-0.030	-0.045	-0.046	-0.038	-0.032	-0.039	-0.047	-0.043	-0.042	-0.039	-0.040
0.602	1.957	0.01	2.939	11.567	-0.032	-0.046	-0.047	-0.039	-0.033	-0.041	-0.049	-0.044	-0.043	-0.041	-0.042
0.601	2.949	0.02	2.925	11.582	-0.033	-0.048	-0.048	-0.040	-0.034	-0.041	-0.049	-0.045	-0.045	-0.042	-0.043
0.602	3.368	0.02	2.936	11.568	-0.032	-0.047	-0.047	-0.039	-0.034	-0.041	-0.048	-0.045	-0.045	-0.042	-0.042
0.601	3.922	0.02	2.929	11.577	-0.031	-0.048	-0.047	-0.039	-0.034	-0.042	-0.048	-0.045	-0.045	-0.041	-0.041
0.601	5.881	0.04	2.929	11.576	-0.030	-0.048	-0.047	-0.038	-0.034	-0.041	-0.045	-0.044	-0.044	-0.040	-0.038
0.601	7.860	0.02	2.931	11.575	-0.028	-0.049	-0.047	-0.038	-0.034	-0.040	-0.044	-0.044	-0.045	-0.038	-0.036
0.599	3.402	8.01	2.913	11.598	0.023	-0.008	-0.010	-0.003	-0.006	-0.013	-0.017	-0.019	-0.018	-0.008	-0.001
0.601	3.397	4.00	2.929	11.577	0.004	-0.025	-0.023	-0.015	-0.017	-0.022	-0.025	-0.028	-0.029	-0.021	-0.016

Table 6. Continued

(e) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—							
					0.883	0.898	0.912	0.927	0.941	0.951	0.974	0.986
1.202	1.485	0.01	6.141	6.068	-0.166	-0.157	-0.156	-0.142	-0.139	-0.193	-0.268	-0.088
1.203	1.986	0.02	6.142	6.063	-0.164	-0.156	-0.157	-0.144	-0.142	-0.194	-0.276	-0.169
1.203	2.955	0.03	6.143	6.062	-0.163	-0.156	-0.158	-0.146	-0.145	-0.197	-0.278	-0.217
1.201	3.398	0.04	6.138	6.082	-0.164	-0.156	-0.158	-0.147	-0.143	-0.197	-0.280	-0.224
1.200	3.912	0.04	6.137	6.084	-0.164	-0.156	-0.159	-0.149	-0.144	-0.197	-0.281	-0.213
1.200	5.854	0.06	6.136	6.087	-0.163	-0.156	-0.159	-0.149	-0.143	-0.197	-0.277	-0.126
1.201	7.886	0.07	6.138	6.079	-0.162	-0.156	-0.160	-0.152	-0.146	-0.199	-0.233	-0.034
1.200	7.867	0.02	6.137	6.086	-0.163	-0.157	-0.160	-0.153	-0.147	-0.200	-0.235	-0.035
1.200	3.417	8.05	6.135	6.089	-0.096	-0.097	-0.121	-0.123	-0.142	-0.168	-0.290	-0.306
1.198	3.387	4.04	6.133	6.101	-0.167	-0.147	-0.144	-0.125	-0.117	-0.174	-0.268	-0.267
0.900	1.481	0.02	4.949	8.734	-0.070	-0.083	-0.106	-0.121	-0.156	-0.178	0.094	0.147
0.901	1.971	0.03	4.955	8.724	-0.070	-0.085	-0.108	-0.126	-0.161	-0.185	0.088	0.138
0.900	2.943	0.01	4.951	8.732	-0.071	-0.086	-0.110	-0.127	-0.162	-0.187	0.086	0.136
0.901	3.369	0.02	4.954	8.727	-0.073	-0.087	-0.111	-0.127	-0.163	-0.187	0.086	0.136
0.899	3.919	0.03	4.943	8.746	-0.071	-0.085	-0.108	-0.125	-0.160	-0.185	0.087	0.138
0.900	5.836	0.02	4.949	8.735	-0.071	-0.084	-0.107	-0.122	-0.155	-0.176	0.096	0.148
0.899	7.828	0.04	4.947	8.739	-0.068	-0.081	-0.103	-0.116	-0.146	-0.163	0.108	0.161
0.900	3.439	8.03	4.949	8.736	-0.049	-0.057	-0.082	-0.104	-0.165	-0.247	0.075	0.124
0.900	3.411	4.02	4.948	8.736	-0.063	-0.077	-0.103	-0.125	-0.178	-0.229	0.070	0.117
0.800	1.481	0.01	4.339	9.694	-0.062	-0.068	-0.085	-0.094	-0.124	-0.141	0.094	0.148
0.801	1.971	0.02	4.345	9.685	-0.064	-0.071	-0.089	-0.099	-0.130	-0.149	0.086	0.138
0.801	2.944	0.03	4.348	9.680	-0.066	-0.073	-0.091	-0.101	-0.134	-0.154	0.083	0.134
0.800	3.400	0.02	4.340	9.691	-0.066	-0.072	-0.090	-0.101	-0.133	-0.152	0.083	0.135
0.800	3.936	0.02	4.339	9.692	-0.067	-0.073	-0.091	-0.101	-0.134	-0.153	0.083	0.135
0.801	5.970	0.01	4.345	9.684	-0.064	-0.070	-0.086	-0.094	-0.125	-0.141	0.095	0.149
0.801	7.851	0.02	4.350	9.677	-0.061	-0.065	-0.080	-0.086	-0.115	-0.127	0.110	0.166
0.799	3.419	8.01	4.334	9.704	-0.036	-0.039	-0.060	-0.077	-0.132	-0.178	0.078	0.128
0.799	3.383	4.03	4.338	9.696	-0.051	-0.056	-0.077	-0.090	-0.136	-0.167	0.077	0.125
0.601	1.488	0.00	2.928	11.578	-0.048	-0.050	-0.066	-0.071	-0.102	-0.116	0.086	0.138
0.602	1.957	0.01	2.939	11.567	-0.051	-0.054	-0.070	-0.076	-0.108	-0.125	0.076	0.124
0.601	2.949	0.02	2.925	11.582	-0.054	-0.056	-0.072	-0.079	-0.111	-0.129	0.072	0.119
0.602	3.368	0.02	2.936	11.568	-0.055	-0.056	-0.071	-0.077	-0.112	-0.129	0.071	0.120
0.601	3.922	0.02	2.929	11.577	-0.056	-0.056	-0.071	-0.077	-0.111	-0.128	0.073	0.121
0.601	5.881	0.04	2.929	11.576	-0.055	-0.054	-0.067	-0.070	-0.105	-0.119	0.085	0.137
0.601	7.860	0.02	2.931	11.575	-0.054	-0.051	-0.064	-0.064	-0.097	-0.108	0.097	0.154
0.599	3.402	8.01	2.913	11.598	-0.024	-0.021	-0.037	-0.046	-0.103	-0.134	0.077	0.125
0.601	3.397	4.00	2.929	11.577	-0.043	-0.039	-0.055	-0.060	-0.110	-0.133	0.073	0.119

Table 6. Continued

(f) $\theta = 225^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.833	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.485	0.01	6.141	6.068	-0.220	-0.192	-0.128	-0.077	-0.139	-0.271	-0.279	-0.126	-0.100	-0.052	
1.203	1.986	0.02	6.142	6.063	-0.219	-0.191	-0.129	-0.079	-0.142	-0.273	-0.340	-0.146	-0.119	-0.072	
1.203	2.955	0.03	6.143	6.062	-0.218	-0.192	-0.131	-0.082	-0.145	-0.275	-0.374	-0.168	-0.133	-0.084	
1.201	3.398	0.04	6.138	6.082	-0.219	-0.194	-0.130	-0.079	-0.143	-0.276	-0.379	-0.171	-0.136	-0.087	
1.200	3.912	0.04	6.137	6.084	-0.218	-0.193	-0.130	-0.080	-0.145	-0.278	-0.378	-0.169	-0.135	-0.087	
1.200	5.854	0.06	6.136	6.087	-0.218	-0.194	-0.129	-0.078	-0.144	-0.279	-0.368	-0.156	-0.112	-0.057	
1.201	7.886	0.07	6.138	6.079	-0.217	-0.194	-0.132	-0.081	-0.147	-0.281	-0.295	-0.123	-0.074	-0.022	
1.200	7.867	0.02	6.137	6.086	-0.217	-0.196	-0.132	-0.082	-0.148	-0.282	-0.294	-0.122	-0.073	-0.022	
1.200	3.417	8.05	6.135	6.089	-0.166	-0.110	-0.052	-0.040	-0.150	-0.274	-0.416	-0.247	-0.206	-0.165	
1.198	3.387	4.04	6.133	6.101	-0.231	-0.212	-0.129	-0.065	-0.123	-0.258	-0.385	-0.187	-0.147	-0.102	
0.900	1.481	0.02	4.949	8.734	-0.090	-0.082	-0.075	-0.095	-0.196	-0.237	0.025	0.101	0.144	0.169	
0.901	1.971	0.03	4.955	8.724	-0.092	-0.084	-0.077	-0.098	-0.201	-0.243	0.022	0.097	0.139	0.161	
0.900	2.943	0.01	4.951	8.732	-0.093	-0.085	-0.078	-0.099	-0.202	-0.246	0.019	0.095	0.138	0.158	
0.901	3.369	0.02	4.954	8.727	-0.094	-0.086	-0.078	-0.100	-0.202	-0.245	0.021	0.096	0.139	0.159	
0.899	3.919	0.03	4.943	8.746	-0.092	-0.084	-0.077	-0.098	-0.200	-0.244	0.021	0.097	0.140	0.162	
0.900	5.836	0.02	4.949	8.735	-0.092	-0.083	-0.075	-0.096	-0.195	-0.236	0.024	0.100	0.144	0.172	
0.899	7.828	0.04	4.947	8.739	-0.089	-0.080	-0.070	-0.090	-0.185	-0.222	0.031	0.107	0.152	0.184	
0.900	3.439	8.03	4.949	8.736	-0.080	-0.066	-0.058	-0.093	-0.238	-0.434	-0.022	0.070	0.121	0.139	
0.900	3.411	4.02	4.948	8.736	-0.091	-0.081	-0.075	-0.108	-0.243	-0.361	-0.041	0.069	0.113	0.126	
0.800	1.481	0.01	4.339	9.694	-0.080	-0.065	-0.053	-0.068	-0.152	-0.185	0.041	0.114	0.154	0.182	
0.801	1.971	0.02	4.345	9.685	-0.083	-0.069	-0.058	-0.074	-0.162	-0.198	0.032	0.109	0.150	0.174	
0.801	2.944	0.03	4.348	9.680	-0.084	-0.071	-0.060	-0.077	-0.167	-0.206	0.026	0.104	0.146	0.168	
0.800	3.400	0.02	4.340	9.691	-0.085	-0.071	-0.061	-0.077	-0.166	-0.204	0.026	0.104	0.146	0.169	
0.800	3.936	0.02	4.339	9.692	-0.086	-0.072	-0.061	-0.078	-0.167	-0.205	0.025	0.104	0.147	0.170	
0.801	5.970	0.01	4.345	9.684	-0.082	-0.067	-0.054	-0.069	-0.153	-0.185	0.038	0.112	0.154	0.186	
0.801	7.851	0.02	4.350	9.677	-0.080	-0.062	-0.048	-0.060	-0.137	-0.163	0.052	0.121	0.162	0.200	
0.799	3.419	8.01	4.334	9.704	-0.065	-0.048	-0.040	-0.071	-0.194	-0.279	-0.009	0.081	0.133	0.159	
0.799	3.383	4.03	4.338	9.696	-0.077	-0.060	-0.051	-0.076	-0.181	-0.242	0.007	0.086	0.129	0.154	
0.601	1.488	0.00	2.928	11.578	-0.068	-0.053	-0.042	-0.056	-0.139	-0.171	0.030	0.116	0.162	0.182	
0.602	1.957	0.01	2.939	11.567	-0.072	-0.057	-0.048	-0.064	-0.149	-0.186	0.019	0.105	0.152	0.165	
0.601	2.949	0.02	2.925	11.582	-0.074	-0.059	-0.050	-0.067	-0.155	-0.191	0.013	0.100	0.147	0.158	
0.602	3.368	0.02	2.936	11.568	-0.076	-0.060	-0.051	-0.068	-0.154	-0.190	0.013	0.100	0.147	0.160	
0.601	3.922	0.02	2.929	11.577	-0.076	-0.060	-0.051	-0.068	-0.152	-0.187	0.015	0.101	0.149	0.165	
0.601	5.881	0.04	2.929	11.576	-0.075	-0.057	-0.046	-0.062	-0.140	-0.171	0.026	0.110	0.159	0.188	
0.601	7.860	0.02	2.931	11.575	-0.074	-0.054	-0.042	-0.055	-0.126	-0.152	0.040	0.121	0.170	0.206	
0.599	3.402	8.01	2.913	11.598	-0.056	-0.035	-0.028	-0.058	-0.161	-0.220	-0.001	0.093	0.147	0.167	
0.601	3.397	4.00	2.929	11.577	-0.070	-0.049	-0.040	-0.065	-0.156	-0.204	0.004	0.089	0.138	0.161	

Table 6. Continued

(g) $\theta = 240^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—								
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	
1.202	1.485	0.01	6.141	6.068	-0.187	-0.103	-0.066	-0.104	-0.384	-0.159	-0.151	-0.123	
1.203	1.986	0.02	6.142	6.063	-0.188	-0.104	-0.068	-0.108	-0.384	-0.190	-0.173	-0.148	
1.203	2.955	0.03	6.143	6.062	-0.190	-0.105	-0.070	-0.110	-0.385	-0.222	-0.189	-0.163	
1.201	3.398	0.04	6.138	6.082	-0.193	-0.104	-0.068	-0.109	-0.386	-0.229	-0.192	-0.165	
1.200	3.912	0.04	6.137	6.084	-0.193	-0.104	-0.069	-0.110	-0.387	-0.227	-0.192	-0.164	
1.200	5.854	0.06	6.136	6.087	-0.195	-0.103	-0.068	-0.110	-0.387	-0.215	-0.182	-0.142	
1.201	7.886	0.07	6.138	6.079	-0.197	-0.106	-0.071	-0.112	-0.387	-0.176	-0.157	-0.095	
1.200	7.867	0.02	6.137	6.086	-0.199	-0.107	-0.072	-0.113	-0.388	-0.176	-0.157	-0.094	
1.200	3.417	8.05	6.135	6.089	-0.111	-0.037	-0.023	-0.095	-0.387	-0.365	-0.248	-0.242	
1.198	3.387	4.04	6.133	6.101	-0.209	-0.098	-0.051	-0.089	-0.378	-0.251	-0.194	-0.177	
0.900	1.481	0.02	4.949	8.734	-0.091	-0.067	-0.077	-0.139	-0.266	0.039	0.083	0.127	
0.901	1.971	0.03	4.955	8.724	-0.093	-0.070	-0.081	-0.144	-0.271	0.036	0.080	0.123	
0.900	2.943	0.01	4.951	8.732	-0.094	-0.072	-0.082	-0.145	-0.276	0.035	0.079	0.123	
0.901	3.369	0.02	4.954	8.727	-0.095	-0.072	-0.082	-0.146	-0.273	0.037	0.081	0.125	
0.899	3.919	0.03	4.943	8.746	-0.092	-0.070	-0.081	-0.144	-0.273	0.037	0.082	0.126	
0.900	5.836	0.02	4.949	8.735	-0.091	-0.069	-0.078	-0.139	-0.266	0.040	0.083	0.128	
0.899	7.828	0.04	4.947	8.739	-0.087	-0.064	-0.072	-0.129	-0.255	0.045	0.088	0.132	
0.900	3.439	8.03	4.949	8.736	-0.075	-0.058	-0.082	-0.188	-0.475	-0.007	0.052	0.102	
0.900	3.411	4.02	4.948	8.736	-0.089	-0.073	-0.095	-0.188	-0.413	-0.005	0.048	0.086	
0.800	1.481	0.01	4.339	9.694	-0.073	-0.045	-0.048	-0.096	-0.215	0.064	0.104	0.146	
0.801	1.971	0.02	4.345	9.685	-0.077	-0.050	-0.055	-0.106	-0.232	0.056	0.101	0.143	
0.801	2.944	0.03	4.348	9.680	-0.079	-0.052	-0.058	-0.113	-0.244	0.051	0.098	0.142	
0.800	3.400	0.02	4.340	9.691	-0.078	-0.053	-0.058	-0.112	-0.243	0.052	0.099	0.142	
0.800	3.936	0.02	4.339	9.692	-0.078	-0.053	-0.059	-0.112	-0.246	0.051	0.099	0.141	
0.801	5.970	0.01	4.345	9.684	-0.072	-0.047	-0.050	-0.098	-0.220	0.062	0.104	0.146	
0.801	7.851	0.02	4.350	9.677	-0.066	-0.040	-0.039	-0.080	-0.190	0.075	0.109	0.150	
0.799	3.419	8.01	4.334	9.704	-0.055	-0.038	-0.056	-0.141	-0.353	0.020	0.080	0.132	
0.799	3.383	4.03	4.338	9.696	-0.066	-0.046	-0.058	-0.124	-0.299	0.036	0.080	0.121	
0.601	1.488	0.00	2.928	11.578	-0.058	-0.032	-0.038	-0.092	-0.229	0.059	0.126	0.166	
0.602	1.957	0.01	2.939	11.567	-0.062	-0.039	-0.046	-0.104	-0.250	0.048	0.120	0.159	
0.601	2.949	0.02	2.925	11.582	-0.063	-0.041	-0.050	-0.107	-0.258	0.043	0.116	0.155	
0.602	3.368	0.02	2.936	11.568	-0.062	-0.042	-0.051	-0.108	-0.259	0.044	0.116	0.154	
0.601	3.922	0.02	2.929	11.577	-0.060	-0.042	-0.050	-0.106	-0.256	0.046	0.118	0.156	
0.601	5.881	0.04	2.929	11.576	-0.055	-0.037	-0.043	-0.094	-0.236	0.059	0.125	0.163	
0.601	7.860	0.02	2.931	11.575	-0.049	-0.032	-0.034	-0.080	-0.210	0.074	0.134	0.171	
0.599	3.402	8.01	2.913	11.598	-0.030	-0.025	-0.041	-0.119	-0.318	0.037	0.120	0.161	
0.601	3.397	4.00	2.929	11.577	-0.042	-0.035	-0.046	-0.109	-0.290	0.042	0.111	0.148	

Table 6. Continued

(h) $\theta = 255^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—					
					0.941	0.963	0.974	0.986	0.997	
1.202	1.485	0.01	6.141	6.068	-0.119	-0.189	-0.109	-0.070	-0.013	
1.203	1.986	0.02	6.142	6.063	-0.122	-0.225	-0.134	-0.100	-0.042	
1.203	2.955	0.03	6.143	6.062	-0.125	-0.258	-0.156	-0.117	-0.060	
1.201	3.398	0.04	6.138	6.082	-0.124	-0.264	-0.160	-0.119	-0.062	
1.200	3.912	0.04	6.137	6.084	-0.126	-0.263	-0.158	-0.115	-0.057	
1.200	5.854	0.06	6.136	6.087	-0.126	-0.248	-0.143	-0.083	-0.014	
1.201	7.886	0.07	6.138	6.079	-0.126	-0.200	-0.109	-0.040	0.028	
1.200	7.867	0.02	6.137	6.086	-0.127	-0.199	-0.110	-0.040	0.028	
1.200	3.417	8.05	6.135	6.089	-0.102	-0.386	-0.264	-0.194	-0.132	
1.198	3.387	4.04	6.133	6.101	-0.107	-0.285	-0.179	-0.141	-0.088	
0.900	1.481	0.02	4.949	8.734	-0.116	0.015	0.055	0.108	0.154	
0.901	1.971	0.03	4.955	8.724	-0.121	0.014	0.054	0.107	0.150	
0.900	2.943	0.01	4.951	8.732	-0.122	0.014	0.055	0.109	0.151	
0.901	3.369	0.02	4.954	8.727	-0.123	0.014	0.057	0.109	0.151	
0.899	3.919	0.03	4.943	8.746	-0.121	0.016	0.058	0.112	0.154	
0.900	5.836	0.02	4.949	8.735	-0.116	0.016	0.058	0.110	0.156	
0.899	7.828	0.04	4.947	8.739	-0.106	0.019	0.059	0.112	0.160	
0.900	3.439	8.03	4.949	8.736	-0.183	-0.044	0.010	0.075	0.123	
0.900	3.411	4.02	4.948	8.736	-0.174	-0.043	0.005	0.055	0.095	
0.800	1.481	0.01	4.339	9.694	-0.069	0.054	0.090	0.138	0.180	
0.801	1.971	0.02	4.345	9.685	-0.080	0.048	0.086	0.136	0.176	
0.801	2.944	0.03	4.348	9.680	-0.087	0.044	0.084	0.134	0.173	
0.800	3.400	0.02	4.340	9.691	-0.087	0.046	0.084	0.134	0.172	
0.800	3.936	0.02	4.339	9.692	-0.087	0.044	0.084	0.134	0.172	
0.801	5.970	0.01	4.345	9.684	-0.070	0.052	0.089	0.137	0.178	
0.801	7.851	0.02	4.350	9.677	-0.050	0.064	0.096	0.141	0.185	
0.799	3.419	8.01	4.334	9.704	-0.124	0.008	0.055	0.115	0.157	
0.799	3.383	4.03	4.338	9.696	-0.098	0.023	0.058	0.105	0.144	
0.601	1.488	0.00	2.928	11.578	-0.075	0.070	0.119	0.164	0.192	
0.602	1.957	0.01	2.939	11.567	-0.090	0.062	0.115	0.156	0.178	
0.601	2.949	0.02	2.925	11.582	-0.094	0.058	0.112	0.153	0.173	
0.602	3.368	0.02	2.936	11.568	-0.094	0.058	0.112	0.153	0.172	
0.601	3.922	0.02	2.929	11.577	-0.092	0.061	0.115	0.154	0.174	
0.601	5.881	0.04	2.929	11.576	-0.077	0.071	0.122	0.161	0.188	
0.601	7.860	0.02	2.931	11.575	-0.060	0.082	0.126	0.164	0.197	
0.599	3.402	8.01	2.913	11.598	-0.109	0.057	0.120	0.156	0.174	
0.601	3.397	4.00	2.929	11.577	-0.094	0.056	0.114	0.144	0.163	

Table 6. Continued

(i) $\theta = 270^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.951	0.963	0.974	0.986	
1.202	1.485	0.01	6.141	6.068	-0.275	-0.123	-0.102	-0.082	
1.203	1.986	0.02	6.142	6.063	-0.311	-0.145	-0.125	-0.107	
1.203	2.955	0.03	6.143	6.062	-0.337	-0.167	-0.146	-0.122	
1.201	3.398	0.04	6.138	6.082	-0.340	-0.170	-0.150	-0.125	
1.200	3.912	0.04	6.137	6.084	-0.340	-0.169	-0.148	-0.123	
1.200	5.854	0.06	6.136	6.087	-0.328	-0.158	-0.136	-0.098	
1.201	7.886	0.07	6.138	6.079	-0.277	-0.129	-0.111	-0.060	
1.200	7.867	0.02	6.137	6.086	-0.277	-0.128	-0.110	-0.060	
1.200	3.417	8.05	6.135	6.089	-0.387	-0.251	-0.209	-0.196	
1.198	3.387	4.04	6.133	6.101	-0.330	-0.183	-0.164	-0.143	
0.900	1.481	0.02	4.949	8.734	-0.030	0.008	0.052	0.103	
0.901	1.971	0.03	4.955	8.724	-0.033	0.008	0.053	0.103	
0.900	2.943	0.01	4.951	8.732	-0.034	0.009	0.057	0.107	
0.901	3.369	0.02	4.954	8.727	-0.035	0.010	0.057	0.108	
0.899	3.919	0.03	4.943	8.746	-0.034	0.011	0.059	0.110	
0.900	5.836	0.02	4.949	8.735	-0.031	0.011	0.057	0.106	
0.899	7.828	0.04	4.947	8.739	-0.027	0.010	0.053	0.103	
0.900	3.439	8.03	4.949	8.736	-0.113	-0.067	-0.010	0.063	
0.900	3.411	4.02	4.948	8.736	-0.097	-0.072	-0.029	0.030	
0.800	1.481	0.01	4.339	9.694	0.015	0.048	0.089	0.134	
0.801	1.971	0.02	4.345	9.685	0.005	0.042	0.084	0.132	
0.801	2.944	0.03	4.348	9.680	-0.005	0.039	0.082	0.131	
0.800	3.400	0.02	4.340	9.691	-0.006	0.040	0.083	0.131	
0.800	3.936	0.02	4.339	9.692	-0.004	0.037	0.082	0.130	
0.801	5.970	0.01	4.345	9.684	0.013	0.046	0.088	0.133	
0.801	7.851	0.02	4.350	9.677	0.031	0.058	0.095	0.136	
0.799	3.419	8.01	4.334	9.704	-0.041	-0.007	0.053	0.113	
0.799	3.383	4.03	4.338	9.696	-0.013	0.013	0.056	0.102	
0.601	1.488	0.00	2.928	11.578	-0.025	0.067	0.120	0.161	
0.602	1.957	0.01	2.939	11.567	-0.041	0.060	0.116	0.156	
0.601	2.949	0.02	2.925	11.582	-0.048	0.056	0.114	0.154	
0.602	3.368	0.02	2.936	11.568	-0.049	0.055	0.115	0.153	
0.601	3.922	0.02	2.929	11.577	-0.048	0.057	0.117	0.154	
0.601	5.881	0.04	2.929	11.576	-0.031	0.064	0.124	0.159	
0.601	7.860	0.02	2.931	11.575	-0.008	0.072	0.126	0.162	
0.599	3.402	8.01	2.913	11.598	-0.070	0.046	0.121	0.156	
0.601	3.397	4.00	2.929	11.577	-0.051	0.046	0.112	0.143	

Table 6. Continued

(j) $\theta = 285^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.485	0.01	6.141	6.068	-0.194	-0.259	-0.171	-0.111	-0.136	-0.143	-0.100	
1.203	1.986	0.02	6.142	6.063	-0.194	-0.260	-0.211	-0.132	-0.161	-0.170	-0.119	
1.203	2.955	0.03	6.143	6.062	-0.195	-0.263	-0.258	-0.152	-0.186	-0.190	-0.126	
1.201	3.398	0.04	6.138	6.082	-0.199	-0.266	-0.265	-0.156	-0.190	-0.194	-0.128	
1.200	3.912	0.04	6.137	6.084	-0.199	-0.267	-0.266	-0.155	-0.189	-0.195	-0.129	
1.200	5.854	0.06	6.136	6.087	-0.202	-0.269	-0.238	-0.145	-0.178	-0.182	-0.115	
1.201	7.886	0.07	6.138	6.079	-0.201	-0.267	-0.179	-0.119	-0.153	-0.152	-0.082	
1.200	7.867	0.02	6.137	6.086	-0.202	-0.267	-0.178	-0.117	-0.152	-0.152	-0.082	
1.200	3.417	8.05	6.135	6.089	-0.261	-0.309	-0.466	-0.215	-0.249	-0.267	-0.225	
1.198	3.387	4.04	6.133	6.101	-0.263	-0.317	-0.235	-0.168	-0.204	-0.210	-0.151	
0.900	1.481	0.02	4.949	8.734	-0.084	-0.062	-0.028	0.013	0.053	0.108	0.167	
0.901	1.971	0.03	4.955	8.724	-0.088	-0.068	-0.030	0.011	0.053	0.107	0.166	
0.900	2.943	0.01	4.951	8.732	-0.094	-0.068	-0.032	0.012	0.056	0.110	0.166	
0.901	3.369	0.02	4.954	8.727	-0.097	-0.071	-0.029	0.015	0.058	0.111	0.166	
0.899	3.919	0.03	4.943	8.746	-0.094	-0.067	-0.030	0.014	0.059	0.114	0.169	
0.900	5.836	0.02	4.949	8.735	-0.089	-0.065	-0.027	0.015	0.056	0.111	0.167	
0.899	7.828	0.04	4.947	8.739	-0.077	-0.058	-0.024	0.014	0.053	0.109	0.167	
0.900	3.439	8.03	4.949	8.736	-0.150	-0.147	-0.113	-0.063	-0.008	0.070	0.142	
0.900	3.411	4.02	4.948	8.736	-0.108	-0.113	-0.100	-0.072	-0.032	0.037	0.105	
0.800	1.481	0.01	4.339	9.694	-0.050	-0.039	-0.011	0.052	0.087	0.134	0.186	
0.801	1.971	0.02	4.345	9.685	-0.059	-0.051	-0.025	0.045	0.083	0.132	0.184	
0.801	2.944	0.03	4.348	9.680	-0.067	-0.064	-0.036	0.041	0.080	0.131	0.182	
0.800	3.400	0.02	4.340	9.691	-0.069	-0.065	-0.038	0.041	0.081	0.130	0.181	
0.800	3.936	0.02	4.339	9.692	-0.068	-0.063	-0.039	0.040	0.081	0.132	0.181	
0.801	5.970	0.01	4.345	9.684	-0.051	-0.041	-0.016	0.050	0.087	0.135	0.185	
0.801	7.851	0.02	4.350	9.677	-0.029	-0.011	0.014	0.060	0.094	0.138	0.188	
0.799	3.419	8.01	4.334	9.704	-0.075	-0.055	-0.040	-0.004	0.050	0.118	0.180	
0.799	3.383	4.03	4.338	9.696	-0.044	-0.028	-0.014	0.014	0.053	0.105	0.161	
0.601	1.488	0.00	2.928	11.578	-0.067	-0.084	-0.076	0.065	0.112	0.161	0.198	
0.602	1.957	0.01	2.939	11.567	-0.078	-0.099	-0.100	0.057	0.108	0.155	0.186	
0.601	2.949	0.02	2.925	11.582	-0.083	-0.106	-0.108	0.054	0.105	0.152	0.181	
0.602	3.368	0.02	2.936	11.568	-0.082	-0.104	-0.108	0.053	0.106	0.153	0.180	
0.601	3.922	0.02	2.929	11.577	-0.081	-0.102	-0.104	0.054	0.108	0.155	0.181	
0.601	5.881	0.04	2.929	11.576	-0.070	-0.086	-0.080	0.062	0.114	0.163	0.192	
0.601	7.860	0.02	2.931	11.575	-0.055	-0.062	-0.051	0.072	0.121	0.171	0.201	
0.599	3.402	8.01	2.913	11.598	-0.100	-0.118	-0.125	0.044	0.114	0.167	0.186	
0.601	3.397	4.00	2.929	11.577	-0.081	-0.089	-0.078	0.037	0.099	0.153	0.173	

Table 6. Continued

(k) $\theta = 300^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.202	1.485	0.01	6.141	6.068	-0.229	-0.242	-0.203	-0.249	-0.280	-0.085	-0.058	-0.009	0.015	
1.203	1.986	0.02	6.142	6.063	-0.229	-0.241	-0.203	-0.249	-0.309	-0.108	-0.082	-0.036	-0.007	
1.203	2.955	0.03	6.143	6.062	-0.231	-0.243	-0.205	-0.250	-0.320	-0.133	-0.095	-0.053	-0.021	
1.201	3.398	0.04	6.138	6.082	-0.234	-0.246	-0.208	-0.254	-0.324	-0.136	-0.096	-0.056	-0.022	
1.200	3.912	0.04	6.137	6.084	-0.235	-0.246	-0.209	-0.254	-0.324	-0.133	-0.092	-0.053	-0.015	
1.200	5.854	0.06	6.136	6.087	-0.237	-0.248	-0.212	-0.257	-0.306	-0.117	-0.064	-0.016	0.020	
1.201	7.886	0.07	6.138	6.079	-0.237	-0.246	-0.212	-0.256	-0.267	-0.089	-0.030	0.023	0.047	
1.200	7.867	0.02	6.137	6.086	-0.238	-0.247	-0.213	-0.257	-0.268	-0.089	-0.030	0.022	0.048	
1.200	3.417	8.05	6.135	6.089	-0.363	-0.351	-0.278	-0.295	-0.406	-0.257	-0.173	-0.107	-0.053	
1.198	3.387	4.04	6.133	6.101	-0.318	-0.313	-0.273	-0.309	-0.309	-0.158	-0.121	-0.073	-0.038	
0.900	1.481	0.02	4.949	8.734	-0.230	-0.116	-0.083	-0.066	0.015	0.055	0.104	0.160	0.127	
0.901	1.971	0.03	4.955	8.724	-0.255	-0.125	-0.089	-0.070	0.014	0.055	0.104	0.158	0.118	
0.900	2.943	0.01	4.951	8.732	-0.302	-0.143	-0.098	-0.073	0.014	0.057	0.106	0.157	0.119	
0.901	3.369	0.02	4.954	8.727	-0.312	-0.148	-0.100	-0.075	0.015	0.058	0.105	0.156	0.119	
0.899	3.919	0.03	4.943	8.746	-0.307	-0.144	-0.097	-0.073	0.017	0.060	0.108	0.159	0.123	
0.900	5.836	0.02	4.949	8.735	-0.262	-0.128	-0.089	-0.069	0.015	0.056	0.105	0.161	0.129	
0.899	7.828	0.04	4.947	8.739	-0.196	-0.101	-0.075	-0.058	0.017	0.056	0.104	0.164	0.136	
0.900	3.439	8.03	4.949	8.736	-0.167	-0.142	-0.142	-0.139	-0.052	-0.002	0.069	0.143	0.096	
0.900	3.411	4.02	4.948	8.736	-0.137	-0.104	-0.102	-0.108	-0.058	-0.020	0.037	0.107	0.090	
0.800	1.481	0.01	4.339	9.694	-0.186	-0.102	-0.067	-0.070	0.064	0.093	0.135	0.179	0.148	
0.801	1.971	0.02	4.345	9.685	-0.191	-0.109	-0.075	-0.082	0.059	0.090	0.133	0.176	0.136	
0.801	2.944	0.03	4.348	9.680	-0.197	-0.115	-0.083	-0.092	0.055	0.088	0.131	0.172	0.134	
0.800	3.400	0.02	4.340	9.691	-0.197	-0.116	-0.084	-0.093	0.056	0.089	0.131	0.172	0.134	
0.800	3.936	0.02	4.339	9.692	-0.197	-0.116	-0.082	-0.091	0.054	0.087	0.130	0.173	0.136	
0.801	5.970	0.01	4.345	9.684	-0.186	-0.104	-0.067	-0.071	0.063	0.092	0.132	0.179	0.150	
0.801	7.851	0.02	4.350	9.677	-0.171	-0.087	-0.046	-0.044	0.071	0.097	0.134	0.186	0.159	
0.799	3.419	8.01	4.334	9.704	-0.172	-0.111	-0.083	-0.082	0.010	0.052	0.107	0.172	0.116	
0.799	3.383	4.03	4.338	9.696	-0.178	-0.098	-0.059	-0.059	0.027	0.056	0.098	0.155	0.117	
0.601	1.488	0.00	2.928	11.578	-0.152	-0.094	-0.079	-0.098	0.073	0.123	0.158	0.189	0.148	
0.602	1.957	0.01	2.939	11.567	-0.158	-0.102	-0.088	-0.112	0.063	0.117	0.151	0.177	0.123	
0.601	2.949	0.02	2.925	11.582	-0.161	-0.105	-0.092	-0.117	0.058	0.113	0.147	0.170	0.116	
0.602	3.368	0.02	2.936	11.568	-0.161	-0.105	-0.091	-0.117	0.057	0.112	0.145	0.171	0.117	
0.601	3.922	0.02	2.929	11.577	-0.159	-0.105	-0.089	-0.115	0.059	0.113	0.145	0.175	0.123	
0.601	5.881	0.04	2.929	11.576	-0.152	-0.098	-0.077	-0.100	0.072	0.119	0.152	0.191	0.144	
0.601	7.860	0.02	2.931	11.575	-0.143	-0.088	-0.064	-0.080	0.085	0.122	0.154	0.203	0.157	
0.599	3.402	8.01	2.913	11.598	-0.202	-0.137	-0.108	-0.137	0.054	0.109	0.139	0.181	0.107	
0.601	3.397	4.00	2.929	11.577	-0.172	-0.116	-0.088	-0.106	0.051	0.101	0.128	0.173	0.110	

Table 6. Continued

(1) $\theta = 315^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.202	1.485	0.01	6.141	6.068	-0.177	-0.191	-0.242	-0.226	-0.297	-0.350	-0.120	-0.140	-0.118	-0.084
1.203	1.986	0.02	6.142	6.063	-0.176	-0.190	-0.241	-0.225	-0.295	-0.402	-0.146	-0.171	-0.146	-0.103
1.203	2.955	0.03	6.143	6.062	-0.177	-0.190	-0.242	-0.227	-0.295	-0.418	-0.168	-0.195	-0.164	-0.115
1.201	3.398	0.04	6.138	6.082	-0.180	-0.191	-0.244	-0.229	-0.298	-0.422	-0.171	-0.197	-0.167	-0.120
1.200	3.912	0.04	6.137	6.084	-0.180	-0.190	-0.243	-0.228	-0.298	-0.423	-0.170	-0.195	-0.170	-0.122
1.200	5.854	0.06	6.136	6.087	-0.182	-0.191	-0.245	-0.230	-0.299	-0.418	-0.160	-0.182	-0.152	-0.103
1.201	7.886	0.07	6.138	6.079	-0.181	-0.188	-0.243	-0.229	-0.297	-0.352	-0.129	-0.143	-0.112	-0.067
1.200	7.867	0.02	6.137	6.086	-0.182	-0.188	-0.244	-0.229	-0.298	-0.353	-0.130	-0.143	-0.113	-0.067
1.200	3.417	8.05	6.135	6.089	-0.312	-0.320	-0.353	-0.324	-0.355	-0.479	-0.240	-0.259	-0.234	-0.210
1.198	3.387	4.04	6.133	6.101	-0.268	-0.278	-0.316	-0.289	-0.355	-0.386	-0.188	-0.218	-0.180	-0.142
0.900	1.481	0.02	4.949	8.734	-0.462	-0.329	-0.165	-0.092	-0.073	-0.043	0.028	0.061	0.102	0.154
0.901	1.971	0.03	4.955	8.724	-0.468	-0.349	-0.179	-0.096	-0.076	-0.051	0.029	0.061	0.102	0.152
0.900	2.943	0.01	4.951	8.732	-0.480	-0.386	-0.200	-0.100	-0.080	-0.059	0.034	0.065	0.105	0.152
0.901	3.369	0.02	4.954	8.727	-0.480	-0.393	-0.208	-0.103	-0.082	-0.058	0.036	0.067	0.106	0.153
0.899	3.919	0.03	4.943	8.746	-0.484	-0.391	-0.199	-0.099	-0.080	-0.060	0.036	0.065	0.107	0.154
0.900	5.836	0.02	4.949	8.735	-0.474	-0.357	-0.179	-0.095	-0.075	-0.049	0.033	0.065	0.108	0.157
0.899	7.828	0.04	4.947	8.739	-0.450	-0.298	-0.143	-0.085	-0.065	-0.033	0.028	0.060	0.108	0.162
0.900	3.439	8.03	4.949	8.736	-0.483	-0.261	-0.189	-0.149	-0.133	-0.094	-0.042	0.003	0.073	0.139
0.900	3.411	4.02	4.948	8.736	-0.396	-0.225	-0.145	-0.119	-0.108	-0.083	-0.050	-0.012	0.053	0.117
0.800	1.481	0.01	4.339	9.694	-0.267	-0.186	-0.113	-0.084	-0.096	-0.116	0.063	0.110	0.135	0.175
0.801	1.971	0.02	4.345	9.685	-0.273	-0.190	-0.118	-0.091	-0.105	-0.132	0.053	0.105	0.131	0.171
0.801	2.944	0.03	4.348	9.680	-0.278	-0.196	-0.123	-0.097	-0.113	-0.143	0.046	0.103	0.130	0.166
0.800	3.400	0.02	4.340	9.691	-0.278	-0.196	-0.123	-0.098	-0.114	-0.144	0.046	0.103	0.130	0.166
0.800	3.936	0.02	4.339	9.692	-0.280	-0.198	-0.125	-0.097	-0.114	-0.144	0.045	0.102	0.131	0.167
0.801	5.970	0.01	4.345	9.684	-0.269	-0.188	-0.114	-0.086	-0.097	-0.121	0.060	0.109	0.137	0.178
0.801	7.851	0.02	4.350	9.677	-0.258	-0.177	-0.101	-0.070	-0.075	-0.087	0.078	0.112	0.142	0.187
0.799	3.419	8.01	4.334	9.704	-0.324	-0.196	-0.125	-0.104	-0.114	-0.132	0.034	0.072	0.115	0.166
0.799	3.383	4.03	4.338	9.696	-0.287	-0.192	-0.114	-0.086	-0.090	-0.096	0.041	0.070	0.106	0.151
0.601	1.488	0.00	2.928	11.578	-0.181	-0.143	-0.099	-0.092	-0.110	-0.143	0.043	0.119	0.151	0.183
0.602	1.957	0.01	2.939	11.567	-0.186	-0.149	-0.106	-0.100	-0.122	-0.158	0.030	0.109	0.141	0.169
0.601	2.949	0.02	2.925	11.582	-0.189	-0.153	-0.110	-0.104	-0.128	-0.165	0.024	0.104	0.137	0.161
0.602	3.368	0.02	2.936	11.568	-0.189	-0.154	-0.110	-0.104	-0.127	-0.165	0.024	0.104	0.139	0.163
0.601	3.922	0.02	2.929	11.577	-0.189	-0.156	-0.110	-0.104	-0.126	-0.162	0.026	0.105	0.141	0.167
0.601	5.881	0.04	2.929	11.576	-0.184	-0.152	-0.105	-0.096	-0.115	-0.146	0.039	0.115	0.155	0.187
0.601	7.860	0.02	2.931	11.575	-0.180	-0.147	-0.096	-0.085	-0.101	-0.125	0.055	0.126	0.166	0.204
0.599	3.402	8.01	2.913	11.598	-0.250	-0.211	-0.146	-0.133	-0.158	-0.200	0.015	0.104	0.151	0.176
0.601	3.397	4.00	2.929	11.577	-0.213	-0.183	-0.127	-0.113	-0.127	-0.147	0.027	0.097	0.139	0.166

Table 6. Continued

(m) Top interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—			
					0.709	0.723	0.898	0.941
1.202	1.485	0.01	6.141	6.068	-0.039	-0.034	-0.218	-0.261
1.203	1.986	0.02	6.142	6.063	-0.039	-0.032	-0.218	-0.264
1.203	2.955	0.03	6.143	6.062	-0.040	-0.033	-0.220	-0.267
1.201	3.398	0.04	6.138	6.082	-0.042	-0.034	-0.223	-0.270
1.200	3.912	0.04	6.137	6.084	-0.043	-0.034	-0.224	-0.270
1.200	5.854	0.06	6.136	6.087	-0.045	-0.034	-0.227	-0.271
1.201	7.886	0.07	6.138	6.079	-0.046	-0.034	-0.226	-0.266
1.200	7.867	0.02	6.137	6.086	-0.046	-0.034	-0.227	-0.266
1.200	3.417	8.05	6.135	6.089	-0.178	-0.163	-0.351	-0.297
1.198	3.387	4.04	6.133	6.101	-0.128	-0.122	-0.309	-0.307
0.900	1.481	0.02	4.949	8.734	-0.044	-0.019	-0.229	-0.060
0.901	1.971	0.03	4.955	8.724	-0.046	-0.019	-0.252	-0.065
0.900	2.943	0.01	4.951	8.732	-0.045	-0.020	-0.295	-0.068
0.901	3.369	0.02	4.954	8.727	-0.045	-0.020	-0.304	-0.069
0.899	3.919	0.03	4.943	8.746	-0.046	-0.020	-0.300	-0.067
0.900	5.836	0.02	4.949	8.735	-0.046	-0.021	-0.257	-0.063
0.899	7.828	0.04	4.947	8.739	-0.044	-0.019	-0.195	-0.053
0.900	3.439	8.03	4.949	8.736	-0.096	-0.073	-0.170	-0.129
0.900	3.411	4.02	4.948	8.736	-0.049	-0.031	-0.141	-0.098
0.800	1.481	0.01	4.339	9.694	-0.041	-0.030	-0.182	-0.017
0.801	1.971	0.02	4.345	9.685	-0.042	-0.032	-0.188	-0.030
0.801	2.944	0.03	4.348	9.680	-0.044	-0.033	-0.193	-0.042
0.800	3.400	0.02	4.340	9.691	-0.044	-0.033	-0.194	-0.043
0.800	3.936	0.02	4.339	9.692	-0.045	-0.034	-0.194	-0.041
0.801	5.970	0.01	4.345	9.684	-0.044	-0.032	-0.183	-0.020
0.801	7.851	0.02	4.350	9.677	-0.041	-0.032	-0.168	0.007
0.799	3.419	8.01	4.334	9.704	-0.090	-0.072	-0.174	-0.056
0.799	3.383	4.03	4.338	9.696	-0.058	-0.049	-0.176	-0.022
0.601	1.488	0.00	2.928	11.578	-0.038	-0.033	-0.148	-0.063
0.602	1.957	0.01	2.939	11.567	-0.038	-0.033	-0.154	-0.080
0.601	2.949	0.02	2.925	11.582	-0.038	-0.036	-0.157	-0.086
0.602	3.368	0.02	2.936	11.568	-0.038	-0.035	-0.156	-0.086
0.601	3.922	0.02	2.929	11.577	-0.037	-0.036	-0.155	-0.084
0.601	5.881	0.04	2.929	11.576	-0.036	-0.035	-0.147	-0.067
0.601	7.860	0.02	2.931	11.575	-0.034	-0.035	-0.139	-0.044
0.599	3.402	8.01	2.913	11.598	-0.073	-0.073	-0.196	-0.099
0.601	3.397	4.00	2.929	11.577	-0.050	-0.052	-0.167	-0.073

Table 6. Concluded

(n) Bottom interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of									
					0.709	0.723	0.738	0.752	0.767	0.781	0.869	0.883	0.941	
1.202	1.485	0.01	6.141	6.068	-0.020	-0.044	-0.067	-0.080	-0.087	-0.090	-0.222	-0.226	-0.132	
1.203	1.986	0.02	6.142	6.063	-0.020	-0.044	-0.068	-0.082	-0.090	-0.091	-0.221	-0.225	-0.136	
1.203	2.955	0.03	6.143	6.062	-0.020	-0.045	-0.068	-0.082	-0.091	-0.093	-0.220	-0.226	-0.139	
1.201	3.398	0.04	6.138	6.082	-0.021	-0.045	-0.069	-0.082	-0.092	-0.094	-0.222	-0.228	-0.138	
1.200	3.912	0.04	6.137	6.084	-0.021	-0.045	-0.069	-0.082	-0.093	-0.096	-0.222	-0.228	-0.139	
1.200	5.854	0.06	6.136	6.087	-0.019	-0.043	-0.067	-0.080	-0.092	-0.096	-0.222	-0.228	-0.138	
1.201	7.886	0.07	6.138	6.079	-0.017	-0.041	-0.066	-0.079	-0.092	-0.097	-0.221	-0.228	-0.137	
1.200	7.867	0.02	6.137	6.086	-0.019	-0.042	-0.067	-0.080	-0.094	-0.099	-0.222	-0.229	-0.138	
1.200	3.417	8.05	6.135	6.089	0.183	0.148	0.114	0.089	0.060	0.022	-0.177	-0.171	-0.117	
1.198	3.387	4.04	6.133	6.101	0.070	0.053	0.038	0.030	0.010	-0.021	-0.224	-0.242	-0.125	
0.900	1.481	0.02	4.949	8.734	-0.054	-0.055	-0.053	-0.049	-0.056	-0.077	-0.080	-0.087	-0.106	
0.901	1.971	0.03	4.955	8.724	-0.055	-0.056	-0.054	-0.050	-0.056	-0.078	-0.081	-0.089	-0.109	
0.900	2.943	0.01	4.951	8.732	-0.054	-0.056	-0.054	-0.050	-0.057	-0.078	-0.082	-0.090	-0.111	
0.901	3.369	0.02	4.954	8.727	-0.056	-0.057	-0.054	-0.050	-0.057	-0.079	-0.084	-0.091	-0.112	
0.899	3.919	0.03	4.943	8.746	-0.054	-0.056	-0.053	-0.050	-0.057	-0.079	-0.081	-0.089	-0.110	
0.900	5.836	0.02	4.949	8.735	-0.059	-0.059	-0.056	-0.052	-0.058	-0.079	-0.081	-0.088	-0.106	
0.899	7.828	0.04	4.947	8.739	-0.054	-0.056	-0.053	-0.051	-0.057	-0.077	-0.079	-0.086	-0.096	
0.900	3.439	8.03	4.948	8.736	-0.036	-0.070	-0.086	-0.093	-0.105	-0.134	-0.074	-0.075	-0.179	
0.900	3.411	4.02	4.948	8.736	-0.043	-0.060	-0.063	-0.062	-0.069	-0.093	-0.079	-0.086	-0.167	
0.800	1.481	0.01	4.339	9.694	-0.041	-0.053	-0.056	-0.055	-0.064	-0.084	-0.074	-0.076	-0.057	
0.801	1.971	0.02	4.345	9.685	-0.041	-0.052	-0.056	-0.056	-0.065	-0.084	-0.077	-0.079	-0.069	
0.801	2.944	0.03	4.348	9.680	-0.042	-0.055	-0.058	-0.057	-0.066	-0.086	-0.079	-0.081	-0.077	
0.800	3.400	0.02	4.340	9.691	-0.042	-0.054	-0.058	-0.057	-0.066	-0.085	-0.078	-0.081	-0.077	
0.800	3.936	0.02	4.339	9.692	-0.043	-0.056	-0.059	-0.059	-0.067	-0.086	-0.080	-0.081	-0.077	
0.801	5.970	0.01	4.345	9.684	-0.043	-0.054	-0.058	-0.058	-0.066	-0.084	-0.077	-0.077	-0.059	
0.801	7.851	0.02	4.350	9.677	-0.043	-0.054	-0.058	-0.057	-0.064	-0.082	-0.074	-0.074	-0.038	
0.799	3.419	8.01	4.334	9.704	-0.003	-0.027	-0.038	-0.043	-0.053	-0.077	-0.057	-0.059	-0.117	
0.799	3.383	4.03	4.338	9.696	-0.016	-0.035	-0.040	-0.043	-0.052	-0.073	-0.067	-0.069	-0.087	
0.601	1.488	0.00	2.928	11.578	-0.031	-0.043	-0.048	-0.047	-0.056	-0.075	-0.064	-0.063	-0.069	
0.602	1.957	0.01	2.939	11.567	-0.031	-0.044	-0.049	-0.049	-0.057	-0.076	-0.068	-0.067	-0.084	
0.601	2.949	0.02	2.925	11.582	-0.033	-0.045	-0.050	-0.049	-0.058	-0.077	-0.069	-0.069	-0.089	
0.602	3.368	0.02	2.936	11.568	-0.032	-0.046	-0.050	-0.050	-0.057	-0.077	-0.070	-0.070	-0.090	
0.601	3.922	0.02	2.929	11.577	-0.033	-0.046	-0.051	-0.051	-0.057	-0.075	-0.070	-0.066	-0.073	
0.601	5.881	0.04	2.929	11.576	-0.033	-0.047	-0.051	-0.050	-0.055	-0.073	-0.067	-0.066	-0.088	
0.601	7.860	0.02	2.931	11.575	-0.034	-0.048	-0.052	-0.052	-0.055	-0.072	-0.067	-0.064	-0.054	
0.599	3.402	8.01	2.913	11.598	0.018	0.000	-0.008	-0.014	-0.017	-0.040	-0.042	-0.043	-0.108	
0.601	3.397	4.00	2.929	11.577	-0.005	-0.021	-0.026	-0.031	-0.032	-0.051	-0.057	-0.058	-0.090	

Table 7. Static Pressure Coefficients of Afterbody Surface With Horizontal Tail in Aft Position and Vertical Tail in Aft Position

(a) $\theta = 0^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854
1.201	1.459	0.01	6.135	6.078	-0.051	-0.074	-0.079	-0.075	-0.046	-0.065	-0.092	-0.102	-0.126	-0.086	-0.069
1.201	1.985	0.02	6.136	6.074	-0.051	-0.074	-0.079	-0.076	-0.047	-0.065	-0.092	-0.102	-0.126	-0.086	-0.068
1.201	2.960	0.00	6.136	6.076	-0.052	-0.073	-0.080	-0.077	-0.049	-0.066	-0.093	-0.103	-0.127	-0.086	-0.071
1.201	2.938	0.01	6.135	6.076	-0.053	-0.074	-0.081	-0.079	-0.050	-0.066	-0.093	-0.103	-0.127	-0.086	-0.071
1.201	3.442	0.01	6.136	6.076	-0.053	-0.073	-0.081	-0.079	-0.050	-0.067	-0.093	-0.103	-0.127	-0.087	-0.072
1.201	3.913	0.01	6.136	6.074	-0.055	-0.073	-0.081	-0.081	-0.052	-0.068	-0.094	-0.104	-0.128	-0.088	-0.074
1.202	5.876	0.03	6.138	6.071	-0.056	-0.073	-0.082	-0.083	-0.054	-0.069	-0.095	-0.105	-0.128	-0.088	-0.074
1.201	7.896	0.04	6.137	6.075	-0.058	-0.074	-0.083	-0.085	-0.056	-0.071	-0.096	-0.107	-0.129	-0.089	-0.076
1.200	3.427	8.05	6.132	6.084	-0.197	-0.215	-0.212	-0.193	-0.154	-0.151	-0.138	-0.121	-0.102	-0.094	-0.143
1.200	3.413	4.00	6.133	6.083	-0.133	-0.151	-0.150	-0.138	-0.108	-0.118	-0.134	-0.137	-0.147	-0.122	-0.149
0.900	1.480	0.00	4.947	8.726	-0.044	-0.033	-0.027	-0.021	-0.001	-0.012	-0.002	0.031	0.046	0.016	-0.082
0.899	1.971	0.02	4.943	8.732	-0.047	-0.036	-0.029	-0.023	-0.002	-0.013	-0.003	0.029	0.044	0.014	-0.085
0.900	2.955	0.04	4.950	8.721	-0.050	-0.039	-0.031	-0.024	-0.003	-0.014	-0.004	0.029	0.044	0.015	-0.084
0.901	3.389	-0.01	4.955	8.713	-0.046	-0.035	-0.027	-0.021	0.000	-0.012	-0.002	0.030	0.046	0.016	-0.081
0.902	3.929	0.00	4.957	8.709	-0.046	-0.036	-0.028	-0.021	0.000	-0.011	0.000	0.031	0.047	0.017	-0.081
0.901	5.880	0.00	4.952	8.715	-0.050	-0.038	-0.030	-0.023	-0.001	-0.012	-0.002	0.030	0.046	0.016	-0.082
0.901	7.835	0.02	4.952	8.717	-0.047	-0.037	-0.029	-0.022	-0.001	-0.012	-0.002	0.030	0.046	0.017	-0.081
0.900	3.431	8.03	4.950	8.723	-0.127	-0.129	-0.131	-0.126	-0.099	-0.103	-0.078	-0.033	-0.011	-0.042	-0.145
0.902	3.386	4.00	4.959	8.707	-0.054	-0.052	-0.050	-0.047	-0.027	-0.036	-0.024	0.010	0.028	-0.005	-0.109
0.800	1.474	0.01	4.336	9.687	-0.041	-0.043	-0.042	-0.039	-0.020	-0.035	-0.032	-0.009	-0.008	-0.056	-0.164
0.802	1.966	0.02	4.351	9.664	-0.041	-0.042	-0.040	-0.038	-0.019	-0.035	-0.032	-0.009	-0.008	-0.055	-0.165
0.802	2.940	0.03	4.350	9.665	-0.041	-0.042	-0.040	-0.038	-0.019	-0.035	-0.031	-0.009	-0.008	-0.057	-0.166
0.802	3.399	0.04	4.352	9.662	-0.042	-0.043	-0.041	-0.038	-0.018	-0.034	-0.031	-0.009	-0.008	-0.056	-0.166
0.800	3.904	0.02	4.340	9.681	-0.039	-0.041	-0.039	-0.037	-0.018	-0.034	-0.032	-0.009	-0.008	-0.057	-0.166
0.802	5.884	0.03	4.348	9.668	-0.043	-0.044	-0.041	-0.038	-0.018	-0.034	-0.032	-0.009	-0.008	-0.056	-0.164
0.801	7.775	0.01	4.346	9.671	-0.043	-0.044	-0.041	-0.037	-0.017	-0.034	-0.032	-0.008	-0.007	-0.054	-0.161
0.800	3.404	8.03	4.340	9.682	-0.104	-0.106	-0.100	-0.093	-0.068	-0.082	-0.076	-0.052	-0.059	-0.126	-0.250
0.800	3.400	4.02	4.335	9.687	-0.064	-0.064	-0.058	-0.052	-0.033	-0.047	-0.045	-0.025	-0.033	-0.094	-0.212
0.601	1.471	-0.01	2.923	11.573	-0.031	-0.043	-0.043	-0.037	-0.019	-0.040	-0.045	-0.028	-0.037	-0.087	-0.170
0.600	1.967	0.00	2.919	11.579	-0.031	-0.042	-0.042	-0.035	-0.018	-0.039	-0.044	-0.028	-0.038	-0.088	-0.172
0.601	2.956	0.01	2.929	11.566	-0.031	-0.043	-0.043	-0.036	-0.019	-0.040	-0.045	-0.030	-0.040	-0.090	-0.175
0.602	3.396	0.02	2.929	11.567	-0.034	-0.046	-0.045	-0.038	-0.020	-0.042	-0.047	-0.031	-0.041	-0.092	-0.176
0.601	3.957	0.02	2.928	11.567	-0.033	-0.045	-0.045	-0.038	-0.020	-0.041	-0.046	-0.030	-0.041	-0.091	-0.175
0.603	5.940	0.01	2.941	11.553	-0.032	-0.046	-0.044	-0.036	-0.019	-0.040	-0.045	-0.029	-0.039	-0.089	-0.172
0.602	7.831	0.02	2.932	11.564	-0.032	-0.048	-0.045	-0.037	-0.019	-0.040	-0.045	-0.029	-0.038	-0.085	-0.167
0.600	3.393	8.01	2.921	11.580	-0.083	-0.096	-0.087	-0.076	-0.054	-0.079	-0.087	-0.072	-0.096	-0.165	-0.258
0.602	3.403	4.02	2.934	11.561	-0.056	-0.069	-0.063	-0.051	-0.035	-0.057	-0.065	-0.051	-0.070	-0.133	-0.219

Table 7. Continued

(a) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.869	0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	
1.201	1.459	0.01	6.135	6.078	-0.133	-0.153	-0.163	-0.205	-0.267	-0.311	-0.355	-0.499	-0.293	
1.201	1.985	0.02	6.136	6.074	-0.133	-0.153	-0.163	-0.204	-0.267	-0.311	-0.355	-0.500	-0.435	
1.201	2.960	0.00	6.136	6.076	-0.137	-0.156	-0.166	-0.204	-0.268	-0.313	-0.357	-0.502	-0.471	
1.201	2.938	0.01	6.135	6.076	-0.137	-0.155	-0.166	-0.204	-0.268	-0.313	-0.358	-0.502	-0.471	
1.201	3.442	0.01	6.136	6.076	-0.138	-0.156	-0.168	-0.204	-0.269	-0.315	-0.359	-0.503	-0.474	
1.201	3.913	0.01	6.136	6.074	-0.141	-0.159	-0.171	-0.204	-0.270	-0.318	-0.361	-0.503	-0.474	
1.202	5.876	0.03	6.138	6.071	-0.142	-0.159	-0.172	-0.204	-0.270	-0.319	-0.362	-0.503	-0.349	
1.201	7.896	0.04	6.137	6.075	-0.144	-0.161	-0.175	-0.205	-0.272	-0.322	-0.365	-0.502	-0.204	
1.200	3.427	8.05	6.132	6.084	-0.170	-0.199	-0.218	-0.262	-0.331	-0.395	-0.430	-0.524	-0.284	
1.200	3.413	4.00	6.133	6.083	-0.192	-0.224	-0.237	-0.270	-0.335	-0.388	-0.410	-0.544	-0.312	
0.900	1.480	0.00	4.947	8.726	-0.167	-0.233	-0.311	-0.410	-0.464	-0.208	-0.118	-0.039	0.010	
0.899	1.971	0.02	4.943	8.732	-0.170	-0.235	-0.314	-0.413	-0.479	-0.231	-0.127	-0.047	-0.001	
0.900	2.955	0.04	4.950	8.721	-0.168	-0.233	-0.312	-0.412	-0.488	-0.257	-0.135	-0.055	-0.009	
0.901	3.389	-0.01	4.955	8.713	-0.165	-0.231	-0.310	-0.409	-0.488	-0.261	-0.137	-0.055	-0.009	
0.902	3.929	0.00	4.957	8.709	-0.165	-0.231	-0.309	-0.409	-0.488	-0.260	-0.137	-0.054	-0.007	
0.901	5.880	0.00	4.952	8.715	-0.166	-0.232	-0.310	-0.409	-0.482	-0.236	-0.127	-0.044	0.004	
0.901	7.835	0.02	4.952	8.717	-0.166	-0.232	-0.310	-0.409	-0.457	-0.194	-0.110	-0.027	0.023	
0.900	3.431	8.03	4.950	8.723	-0.234	-0.307	-0.378	-0.470	-0.391	-0.160	-0.095	-0.061	-0.033	
0.902	3.386	4.00	4.959	8.707	-0.198	-0.267	-0.336	-0.442	-0.483	-0.219	-0.124	-0.066	-0.022	
0.800	1.474	0.01	4.336	9.687	-0.245	-0.307	-0.353	-0.362	-0.295	-0.195	-0.180	-0.014	0.059	
0.802	1.966	0.02	4.351	9.664	-0.248	-0.312	-0.363	-0.377	-0.307	-0.206	-0.192	-0.023	0.051	
0.802	2.940	0.03	4.350	9.665	-0.250	-0.316	-0.369	-0.385	-0.313	-0.211	-0.199	-0.029	0.045	
0.802	3.399	0.04	4.352	9.662	-0.250	-0.316	-0.370	-0.387	-0.314	-0.212	-0.199	-0.029	0.045	
0.800	3.904	0.02	4.340	9.681	-0.249	-0.315	-0.367	-0.384	-0.315	-0.213	-0.199	-0.028	0.047	
0.802	5.884	0.03	4.348	9.668	-0.246	-0.308	-0.356	-0.367	-0.296	-0.196	-0.181	-0.016	0.056	
0.801	7.775	0.01	4.346	9.671	-0.239	-0.300	-0.340	-0.347	-0.277	-0.177	-0.160	0.001	0.070	
0.800	3.404	8.03	4.340	9.682	-0.339	-0.402	-0.426	-0.383	-0.278	-0.177	-0.137	-0.040	-0.004	
0.800	3.400	4.02	4.335	9.687	-0.296	-0.359	-0.397	-0.386	-0.297	-0.196	-0.174	-0.035	0.024	
0.601	1.471	-0.01	2.923	11.573	-0.209	-0.242	-0.249	-0.267	-0.243	-0.189	-0.190	-0.033	0.060	
0.600	1.967	0.00	2.919	11.579	-0.213	-0.246	-0.256	-0.275	-0.252	-0.200	-0.203	-0.045	0.049	
0.601	2.956	0.01	2.929	11.566	-0.215	-0.249	-0.259	-0.280	-0.258	-0.205	-0.209	-0.051	0.043	
0.602	3.396	0.02	2.929	11.567	-0.217	-0.251	-0.261	-0.283	-0.259	-0.207	-0.210	-0.051	0.042	
0.601	3.957	0.02	2.928	11.567	-0.215	-0.250	-0.260	-0.281	-0.258	-0.205	-0.207	-0.050	0.043	
0.603	5.940	0.01	2.941	11.553	-0.210	-0.244	-0.252	-0.273	-0.247	-0.192	-0.193	-0.037	0.056	
0.602	7.831	0.02	2.932	11.564	-0.203	-0.237	-0.242	-0.263	-0.235	-0.178	-0.177	-0.022	0.069	
0.600	3.393	8.01	2.921	11.580	-0.293	-0.321	-0.312	-0.321	-0.274	-0.203	-0.182	-0.051	0.025	
0.602	3.403	4.02	2.934	11.561	-0.253	-0.285	-0.284	-0.304	-0.267	-0.205	-0.196	-0.052	0.034	

Table 7. Continued

(b) $\theta = 45^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—													
					0.709	0.723	0.738	0.752	0.767	0.781	0.883	0.912	0.941	0.951	0.963	0.974	0.997	
1.201	1.459	0.01	6.135	6.078	-0.051	0.001	0.012	-0.030	-0.061	-0.096	1.000	1.000	-0.421	-0.351	-0.315	-0.278	-0.064	
1.201	1.985	0.02	6.136	6.074	-0.050	0.001	0.012	-0.030	-0.060	-0.096	1.000	1.000	-0.421	-0.378	-0.344	-0.304	-0.082	
1.201	2.960	0.00	6.136	6.076	-0.051	0.000	0.011	-0.030	-0.061	-0.096	1.000	1.000	-0.423	-0.405	-0.380	-0.321	-0.096	
1.201	2.938	0.01	6.135	6.076	-0.053	-0.001	0.011	-0.030	-0.061	-0.096	1.000	1.000	-0.423	-0.404	-0.379	-0.319	-0.096	
1.201	3.442	0.01	6.136	6.076	-0.052	0.000	0.012	-0.029	-0.060	-0.097	1.000	1.000	-0.424	-0.407	-0.382	-0.322	-0.096	
1.201	3.913	0.01	6.136	6.074	-0.053	-0.001	0.011	-0.029	-0.060	-0.097	1.000	1.000	-0.425	-0.406	-0.379	-0.320	-0.093	
1.202	5.876	0.03	6.138	6.071	-0.055	-0.003	0.010	-0.030	-0.061	-0.097	1.000	1.000	-0.424	-0.360	-0.321	-0.291	-0.062	
1.201	7.896	0.04	6.137	6.075	-0.056	-0.004	0.010	-0.030	-0.062	-0.098	1.000	1.000	-0.419	-0.304	-0.284	-0.257	-0.030	
1.200	3.427	8.05	6.132	6.084	-0.176	-0.085	-0.088	-0.108	-0.125	-0.128	1.000	1.000	-0.491	-0.445	-0.336	-0.298	-0.118	
1.200	3.413	4.00	6.133	6.083	-0.116	-0.042	-0.047	-0.078	-0.110	-0.138	1.000	1.000	-0.444	-0.336	-0.335	-0.332	-0.113	
0.900	1.480	0.00	4.947	8.726	-0.003	0.017	0.001	-0.008	-0.004	0.009	1.000	1.000	-0.093	-0.088	-0.129	-0.102	0.117	
0.899	1.971	0.02	4.943	8.732	-0.001	0.018	0.002	-0.008	-0.004	0.008	1.000	1.000	-0.100	-0.095	-0.137	-0.112	0.108	
0.900	2.955	0.04	4.950	8.721	-0.005	0.016	0.001	-0.009	-0.005	0.008	1.000	1.000	-0.112	-0.105	-0.148	-0.118	0.104	
0.901	3.389	-0.01	4.955	8.713	-0.004	0.018	0.002	-0.007	-0.003	0.009	1.000	1.000	-0.108	-0.103	-0.148	-0.120	0.104	
0.902	3.929	0.00	4.957	8.709	-0.002	0.018	0.002	-0.007	-0.003	0.010	1.000	1.000	-0.111	-0.104	-0.147	-0.119	0.105	
0.901	5.880	0.00	4.952	8.715	-0.003	0.019	0.002	-0.007	-0.003	0.009	1.000	1.000	-0.102	-0.096	-0.140	-0.111	0.117	
0.901	7.835	0.02	4.952	8.717	-0.004	0.016	0.001	-0.008	-0.004	0.009	1.000	1.000	-0.083	-0.080	-0.122	-0.096	0.128	
0.900	3.431	8.03	4.950	8.723	-0.086	-0.075	-0.111	-0.131	-0.117	-0.089	1.000	1.000	-0.100	-0.101	-0.140	-0.125	0.097	
0.902	3.386	4.00	4.959	8.707	-0.005	0.004	-0.024	-0.039	-0.034	-0.018	1.000	1.000	-0.116	-0.114	-0.161	-0.132	0.098	
0.800	1.474	0.01	4.336	9.687	-0.003	0.006	-0.016	-0.025	-0.021	-0.011	1.000	1.000	-0.087	-0.057	-0.104	-0.062	0.135	
0.802	1.966	0.02	4.351	9.664	-0.002	0.007	-0.016	-0.026	-0.021	-0.011	1.000	1.000	-0.094	-0.067	-0.113	-0.069	0.125	
0.802	2.940	0.03	4.350	9.665	-0.003	0.007	-0.016	-0.026	-0.021	-0.010	1.000	1.000	-0.102	-0.074	-0.122	-0.075	0.122	
0.802	3.399	0.04	4.352	9.662	-0.003	0.006	-0.016	-0.026	-0.021	-0.010	1.000	1.000	-0.102	-0.074	-0.121	-0.076	0.122	
0.800	3.904	0.02	4.340	9.681	-0.003	0.006	-0.017	-0.026	-0.021	-0.011	1.000	1.000	-0.103	-0.073	-0.121	-0.076	0.124	
0.802	5.884	0.03	4.348	9.668	-0.002	0.007	-0.017	-0.026	-0.021	-0.011	1.000	1.000	-0.091	-0.061	-0.110	-0.067	0.135	
0.801	7.775	0.01	4.346	9.671	-0.002	0.007	-0.016	-0.026	-0.021	-0.010	1.000	1.000	-0.074	-0.047	-0.096	-0.055	0.145	
0.800	3.404	8.03	4.340	9.682	-0.057	-0.046	-0.072	-0.082	-0.073	-0.057	1.000	1.000	-0.090	-0.084	-0.135	-0.107	0.112	
0.800	3.400	4.02	4.335	9.687	-0.011	-0.001	-0.029	-0.041	-0.034	-0.022	1.000	1.000	-0.089	-0.083	-0.139	-0.111	0.116	
0.601	1.471	-0.01	2.923	11.573	0.003	0.007	-0.019	-0.028	-0.022	-0.010	1.000	1.000	-0.074	-0.064	-0.089	-0.021	0.140	
0.600	1.967	0.00	2.919	11.579	0.004	0.008	-0.018	-0.027	-0.021	-0.010	1.000	1.000	-0.088	-0.075	-0.099	-0.028	0.117	
0.601	2.956	0.01	2.929	11.566	0.004	0.007	-0.019	-0.028	-0.022	-0.011	1.000	1.000	-0.093	-0.081	-0.104	-0.033	0.113	
0.602	3.396	0.02	2.929	11.567	0.003	0.006	-0.020	-0.029	-0.023	-0.012	1.000	1.000	-0.094	-0.081	-0.107	-0.034	0.113	
0.601	3.957	0.02	2.928	11.567	0.004	0.007	-0.019	-0.028	-0.022	-0.010	1.000	1.000	-0.092	-0.080	-0.107	-0.035	0.117	
0.603	5.940	0.01	2.941	11.553	0.004	0.007	-0.019	-0.029	-0.023	-0.011	1.000	1.000	-0.080	-0.069	-0.095	-0.026	0.136	
0.602	7.831	0.02	2.932	11.564	0.002	0.005	-0.022	-0.031	-0.023	-0.012	1.000	1.000	-0.066	-0.057	-0.087	-0.021	0.149	
0.600	3.393	8.01	2.921	11.580	-0.028	-0.021	-0.049	-0.058	-0.050	-0.039	1.000	1.000	-0.087	-0.072	-0.128	-0.083	0.107	
0.602	3.403	4.02	2.934	11.561	0.000	0.003	-0.028	-0.039	-0.032	-0.020	1.000	1.000	-0.086	-0.073	-0.123	-0.083	0.110	

Table 7. Continued

(c) $\theta = 90^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—												
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.825	0.951	0.963	0.974	0.986	0.997
1.201	1.459	0.01	6.135	6.078	-0.055	-0.006	0.033	-0.064	-0.104	-0.104	-0.147	-0.121	-0.272	-0.291	-0.304	-0.203	-0.105
1.201	1.985	0.02	6.136	6.074	-0.054	-0.006	0.033	-0.064	-0.104	-0.103	-0.146	-0.124	-0.305	-0.330	-0.326	-0.212	-0.115
1.201	2.960	0.00	6.136	6.076	-0.056	-0.007	0.031	-0.066	-0.105	-0.104	-0.146	-0.126	-0.338	-0.370	-0.347	-0.206	-0.112
1.201	2.938	0.01	6.135	6.076	-0.054	-0.006	0.031	-0.066	-0.106	-0.104	-0.146	-0.125	-0.337	-0.369	-0.346	-0.205	-0.112
1.201	3.442	0.01	6.136	6.076	-0.055	-0.006	0.031	-0.066	-0.106	-0.105	-0.147	-0.128	-0.339	-0.372	-0.349	-0.206	-0.110
1.201	3.913	0.01	6.136	6.074	-0.056	-0.006	0.029	-0.068	-0.107	-0.106	-0.147	-0.130	-0.335	-0.370	-0.349	-0.208	-0.108
1.202	5.876	0.03	6.138	6.071	-0.056	-0.006	0.028	-0.068	-0.106	-0.106	-0.146	-0.130	-0.277	-0.302	-0.318	-0.217	-0.108
1.201	7.896	0.04	6.137	6.075	-0.058	-0.007	0.026	-0.069	-0.107	-0.107	-0.147	-0.131	-0.220	-0.251	-0.281	-0.199	-0.095
1.200	3.427	8.05	6.132	6.084	0.095	0.059	0.002	-0.077	-0.112	-0.064	-0.086	-0.131	-0.377	-0.373	-0.321	-0.221	-0.141
1.200	3.413	4.00	6.133	6.083	0.039	0.011	0.012	-0.078	-0.128	-0.108	-0.138	-0.148	-0.301	-0.319	-0.332	-0.242	-0.156
1.200	1.480	0.00	4.947	8.726	0.004	0.019	0.004	-0.047	-0.016	-0.002	0.005	0.124	-0.095	-0.093	-0.081	0.037	0.137
0.900	1.971	0.02	4.943	8.732	0.003	0.018	0.004	-0.047	-0.016	-0.003	0.005	0.123	-0.102	-0.100	-0.090	0.029	0.131
0.899	2.955	0.04	4.950	8.721	0.002	0.017	0.002	-0.049	-0.017	-0.003	0.004	0.123	-0.109	-0.109	-0.096	0.024	0.125
0.901	3.389	-0.01	4.955	8.713	0.002	0.018	0.004	-0.046	-0.016	-0.002	0.005	0.124	-0.111	-0.110	-0.098	0.023	0.124
0.900	3.929	0.00	4.957	8.709	0.003	0.019	0.005	-0.045	-0.014	-0.001	0.006	0.124	-0.111	-0.110	-0.098	0.025	0.124
0.901	5.880	0.00	4.952	8.715	0.003	0.018	0.004	-0.046	-0.015	-0.002	0.005	0.124	-0.103	-0.102	-0.089	0.032	0.132
0.901	7.835	0.02	4.952	8.717	0.002	0.018	0.004	-0.046	-0.015	-0.002	0.005	0.124	-0.090	-0.087	-0.076	0.042	0.142
0.900	3.431	8.03	4.950	8.723	-0.063	-0.068	-0.102	-0.217	-0.134	-0.101	-0.089	0.094	-0.077	-0.092	-0.104	-0.018	0.095
0.902	3.386	4.00	4.959	8.707	-0.011	-0.004	-0.024	-0.090	-0.045	-0.025	-0.019	0.121	-0.096	-0.100	-0.100	0.006	0.112
0.800	1.474	0.01	4.336	9.687	0.000	0.003	-0.013	-0.061	-0.032	-0.018	-0.013	0.104	-0.085	-0.086	-0.048	0.080	0.159
0.802	1.966	0.02	4.351	9.664	0.000	0.002	-0.013	-0.061	-0.031	-0.018	-0.012	0.104	-0.093	-0.095	-0.055	0.075	0.155
0.802	2.940	0.03	4.350	9.665	-0.001	0.001	-0.013	-0.061	-0.032	-0.018	-0.013	0.104	-0.097	-0.102	-0.060	0.071	0.150
0.802	3.399	0.04	4.352	9.662	0.000	0.002	-0.012	-0.060	-0.031	-0.018	-0.014	0.104	-0.098	-0.102	-0.060	0.071	0.149
0.800	3.904	0.02	4.340	9.681	0.000	0.001	-0.013	-0.061	-0.032	-0.019	-0.014	0.103	-0.098	-0.103	-0.060	0.072	0.150
0.802	5.884	0.03	4.348	9.668	-0.001	0.001	-0.013	-0.061	-0.031	-0.018	-0.014	0.104	-0.089	-0.089	-0.051	0.077	0.157
0.801	7.775	0.01	4.346	9.671	-0.002	0.000	-0.013	-0.061	-0.031	-0.017	-0.013	0.104	-0.078	-0.076	-0.040	0.086	0.164
0.800	3.404	8.03	4.340	9.682	-0.014	-0.029	-0.050	-0.124	-0.074	-0.053	-0.053	0.097	-0.072	-0.094	-0.081	0.032	0.127
0.800	3.400	4.02	4.335	9.687	0.007	0.000	-0.014	-0.067	-0.033	-0.019	-0.017	0.104	-0.083	-0.095	-0.015	0.114	0.168
0.601	1.471	-0.01	2.923	11.573	0.003	-0.005	-0.009	-0.048	-0.025	-0.015	-0.010	0.094	-0.091	-0.107	-0.023	0.111	0.159
0.600	1.967	0.00	2.919	11.579	0.004	-0.004	-0.009	-0.047	-0.025	-0.013	-0.009	0.095	-0.098	-0.115	-0.027	0.105	0.152
0.601	2.956	0.01	2.929	11.566	0.004	-0.005	-0.009	-0.048	-0.025	-0.014	-0.010	0.094	-0.098	-0.116	-0.029	0.105	0.151
0.602	3.396	0.02	2.929	11.567	0.001	-0.007	-0.011	-0.049	-0.027	-0.015	-0.012	0.092	-0.098	-0.116	-0.020	0.106	0.152
0.601	3.957	0.02	2.928	11.567	0.003	-0.006	-0.009	-0.047	-0.025	-0.013	-0.010	0.093	-0.098	-0.103	-0.020	0.111	0.162
0.603	5.940	0.01	2.941	11.553	0.003	-0.006	-0.009	-0.047	-0.025	-0.014	-0.011	0.092	-0.083	-0.089	-0.014	0.116	0.168
0.602	7.831	0.02	2.932	11.564	0.001	-0.008	-0.010	-0.048	-0.026	-0.014	-0.012	0.091	-0.083	-0.089	-0.055	0.089	0.141
0.600	3.393	8.01	2.921	11.580	0.017	-0.006	-0.011	-0.066	-0.035	-0.020	-0.025	0.095	-0.083	-0.113	-0.055	0.089	0.141
0.602	3.403	4.02	2.934	11.561	0.016	-0.001	-0.001	-0.041	-0.020	-0.008	-0.009	0.094	-0.095	-0.118	-0.055	0.089	0.145

Table 7. Continued

(d) $\theta = 135^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.869	
1.201	1.459	0.01	6.135	6.078	-0.068	-0.080	-0.082	-0.070	-0.077	-0.108	-0.116	-0.158	-0.162	-0.144	-0.099	
1.201	1.985	0.02	6.136	6.074	-0.068	-0.081	-0.082	-0.069	-0.077	-0.108	-0.116	-0.158	-0.162	-0.146	-0.100	
1.201	2.960	0.00	6.136	6.076	-0.071	-0.083	-0.084	-0.069	-0.078	-0.110	-0.117	-0.160	-0.162	-0.146	-0.102	
1.201	2.938	0.01	6.135	6.076	-0.070	-0.083	-0.084	-0.068	-0.078	-0.110	-0.117	-0.159	-0.162	-0.147	-0.102	
1.201	3.442	0.01	6.136	6.076	-0.071	-0.084	-0.084	-0.068	-0.079	-0.111	-0.118	-0.160	-0.162	-0.147	-0.103	
1.201	3.913	0.01	6.136	6.074	-0.073	-0.085	-0.086	-0.067	-0.080	-0.113	-0.119	-0.162	-0.163	-0.147	-0.105	
1.202	5.876	0.03	6.138	6.071	-0.072	-0.085	-0.087	-0.066	-0.080	-0.114	-0.119	-0.162	-0.163	-0.148	-0.106	
1.201	7.896	0.04	6.137	6.075	-0.073	-0.085	-0.086	-0.065	-0.081	-0.115	-0.119	-0.162	-0.164	-0.148	-0.107	
1.200	3.427	8.05	6.132	6.084	0.143	0.098	0.053	-0.001	-0.039	-0.072	-0.072	-0.126	-0.133	-0.120	-0.025	
1.200	3.413	4.00	6.133	6.083	0.053	0.033	0.003	-0.034	-0.052	-0.079	-0.095	-0.157	-0.172	-0.177	-0.075	
0.900	1.480	0.00	4.947	8.726	-0.043	-0.038	-0.050	-0.045	-0.034	-0.024	-0.013	-0.001	0.026	0.022	-0.044	
0.899	1.971	0.02	4.943	8.732	-0.045	-0.040	-0.052	-0.046	-0.036	-0.026	-0.014	-0.002	0.025	0.021	-0.046	
0.900	2.955	0.04	4.950	8.721	-0.046	-0.041	-0.052	-0.047	-0.037	-0.027	-0.015	-0.003	0.024	0.020	-0.048	
0.901	3.389	-0.01	4.955	8.713	-0.045	-0.039	-0.051	-0.045	-0.035	-0.025	-0.013	-0.001	0.026	0.022	-0.046	
0.902	3.929	0.00	4.957	8.709	-0.043	-0.038	-0.049	-0.044	-0.034	-0.023	-0.012	0.000	0.027	0.023	-0.046	
0.901	5.880	0.00	4.952	8.715	-0.045	-0.039	-0.050	-0.045	-0.035	-0.025	-0.013	-0.001	0.026	0.022	-0.046	
0.901	7.835	0.02	4.952	8.717	-0.045	-0.039	-0.049	-0.044	-0.034	-0.024	-0.013	-0.001	0.026	0.023	-0.044	
0.900	3.431	8.03	4.950	8.723	-0.077	-0.101	-0.136	-0.154	-0.145	-0.128	-0.104	-0.072	-0.013	0.015	-0.006	
0.902	3.386	4.00	4.959	8.707	-0.053	-0.058	-0.077	-0.077	-0.064	-0.049	-0.034	-0.014	0.023	0.027	-0.029	
0.800	1.474	0.01	4.336	9.687	-0.032	-0.041	-0.057	-0.055	-0.044	-0.033	-0.026	-0.017	0.009	0.006	-0.044	
0.802	1.966	0.02	4.351	9.664	-0.031	-0.041	-0.057	-0.056	-0.044	-0.033	-0.025	-0.016	0.009	0.007	-0.044	
0.802	2.940	0.03	4.350	9.665	-0.031	-0.041	-0.057	-0.056	-0.044	-0.032	-0.025	-0.016	0.009	0.006	-0.044	
0.802	3.399	0.04	4.352	9.662	-0.032	-0.041	-0.058	-0.057	-0.045	-0.033	-0.025	-0.016	0.008	0.006	-0.045	
0.800	3.904	0.02	4.340	9.681	-0.032	-0.042	-0.058	-0.057	-0.045	-0.033	-0.026	-0.017	0.007	0.005	-0.045	
0.802	5.884	0.03	4.348	9.668	-0.032	-0.041	-0.058	-0.056	-0.044	-0.031	-0.025	-0.016	0.008	0.006	-0.044	
0.801	7.775	0.01	4.346	9.671	-0.033	-0.042	-0.058	-0.057	-0.044	-0.031	-0.025	-0.016	0.009	0.006	-0.043	
0.800	3.404	8.03	4.340	9.682	-0.026	-0.049	-0.077	-0.088	-0.078	-0.064	-0.054	-0.039	0.004	0.024	0.013	
0.800	3.400	4.02	4.335	9.687	-0.017	-0.033	-0.053	-0.058	-0.045	-0.031	-0.025	-0.014	0.018	0.024	-0.014	
0.601	1.471	-0.01	2.923	11.573	-0.021	-0.034	-0.047	-0.054	-0.035	-0.019	-0.018	-0.013	0.006	0.004	-0.027	
0.600	1.967	0.00	2.919	11.579	-0.020	-0.035	-0.048	-0.055	-0.035	-0.019	-0.019	-0.014	0.005	0.002	-0.028	
0.601	2.956	0.01	2.929	11.566	-0.020	-0.034	-0.049	-0.055	-0.036	-0.020	-0.019	-0.015	0.004	0.001	-0.028	
0.602	3.396	0.02	2.929	11.567	-0.021	-0.036	-0.050	-0.057	-0.038	-0.020	-0.021	-0.016	0.003	0.000	-0.030	
0.601	3.957	0.02	2.928	11.567	-0.021	-0.035	-0.050	-0.057	-0.037	-0.020	-0.020	-0.015	0.004	0.000	-0.029	
0.603	5.940	0.01	2.941	11.553	-0.020	-0.034	-0.048	-0.057	-0.036	-0.019	-0.019	-0.014	0.005	0.001	-0.028	
0.602	7.831	0.02	2.932	11.564	-0.021	-0.035	-0.049	-0.058	-0.037	-0.018	-0.020	-0.015	0.004	0.001	-0.027	
0.600	3.393	8.01	2.921	11.580	0.004	-0.020	-0.042	-0.063	-0.042	-0.023	-0.025	-0.018	0.012	0.028	0.034	
0.602	3.403	4.02	2.934	11.561	0.001	-0.018	-0.036	-0.053	-0.030	-0.010	-0.014	-0.007	0.017	0.021	0.004	

Table 7. Continued

(d) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997
1.201	1.459	0.01	6.135	6.078	-0.092	-0.084	-0.108	0.147	-0.163	-0.233	-0.358	-0.254	-0.175	-0.092
1.201	1.985	0.02	6.136	6.074	-0.094	-0.084	-0.108	0.147	-0.164	-0.234	-0.371	-0.289	-0.219	-0.121
1.201	2.960	0.00	6.136	6.076	-0.095	-0.085	-0.110	0.149	-0.165	-0.235	-0.377	-0.305	-0.236	-0.140
1.201	2.938	0.01	6.135	6.076	-0.095	-0.085	-0.110	0.148	-0.165	-0.234	-0.377	-0.305	-0.236	-0.140
1.201	3.442	0.01	6.136	6.076	-0.097	-0.085	-0.111	0.149	-0.166	-0.236	-0.380	-0.309	-0.238	-0.140
1.201	3.913	0.01	6.136	6.074	-0.098	-0.086	-0.112	0.150	-0.168	-0.236	-0.382	-0.309	-0.236	-0.137
1.202	5.876	0.03	6.138	6.071	-0.099	-0.087	-0.114	0.151	-0.169	-0.236	-0.377	-0.284	-0.198	-0.096
1.201	7.896	0.04	6.137	6.075	-0.099	-0.087	-0.114	0.150	-0.170	-0.237	-0.367	-0.246	-0.153	-0.064
1.200	3.427	8.05	6.132	6.084	-0.045	-0.074	-0.103	0.132	-0.138	-0.207	-0.332	-0.295	-0.258	-0.193
1.200	3.413	4.00	6.133	6.083	-0.068	-0.062	-0.081	0.120	-0.135	-0.204	-0.353	-0.282	-0.240	-0.175
0.900	1.480	0.00	4.947	8.726	-0.066	-0.091	-0.115	0.147	-0.131	-0.182	-0.061	0.022	0.087	0.131
0.899	1.971	0.02	4.943	8.732	-0.068	-0.093	-0.117	0.151	-0.137	-0.191	-0.070	0.014	0.077	0.119
0.900	2.955	0.04	4.950	8.721	-0.070	-0.095	-0.120	0.155	-0.144	-0.200	-0.077	0.009	0.074	0.113
0.901	3.389	-0.01	4.955	8.713	-0.069	-0.095	-0.120	0.155	-0.143	-0.201	-0.078	0.008	0.074	0.113
0.902	3.929	0.00	4.957	8.709	-0.069	-0.095	-0.120	0.155	-0.144	-0.201	-0.078	0.010	0.076	0.117
0.901	5.880	0.00	4.952	8.715	-0.068	-0.093	-0.117	0.150	-0.135	-0.188	-0.066	0.022	0.091	0.136
0.901	7.835	0.02	4.952	8.717	-0.066	-0.090	-0.112	0.142	-0.123	-0.169	-0.049	0.037	0.105	0.154
0.900	3.431	8.03	4.950	8.723	-0.020	-0.041	-0.064	0.101	-0.102	-0.176	-0.063	0.019	0.079	0.108
0.902	3.386	4.00	4.959	8.707	-0.050	-0.076	-0.100	0.135	-0.132	-0.184	-0.073	0.011	0.071	0.107
0.800	1.474	0.01	4.336	9.687	-0.058	-0.076	-0.092	0.115	-0.096	-0.143	-0.037	0.048	0.110	0.148
0.802	1.966	0.02	4.351	9.664	-0.058	-0.077	-0.094	0.118	-0.100	-0.152	-0.046	0.039	0.102	0.138
0.802	2.940	0.03	4.350	9.665	-0.059	-0.079	-0.095	0.120	-0.103	-0.156	-0.049	0.035	0.098	0.131
0.802	3.399	0.04	4.352	9.662	-0.059	-0.079	-0.095	0.120	-0.102	-0.156	-0.049	0.035	0.098	0.131
0.800	3.904	0.02	4.340	9.681	-0.059	-0.079	-0.096	0.120	-0.103	-0.154	-0.048	0.037	0.101	0.135
0.802	5.884	0.03	4.348	9.668	-0.058	-0.077	-0.092	0.115	-0.097	-0.145	-0.038	0.046	0.109	0.150
0.801	7.775	0.01	4.346	9.671	-0.056	-0.075	-0.089	0.110	-0.089	-0.133	-0.025	0.059	0.124	0.168
0.800	3.404	8.03	4.340	9.682	0.001	-0.024	-0.043	0.075	-0.072	-0.136	-0.040	0.035	0.092	0.122
0.800	3.400	4.02	4.335	9.687	-0.031	-0.055	-0.074	0.102	-0.094	-0.143	-0.048	0.029	0.086	0.122
0.601	1.471	-0.01	2.923	11.573	-0.034	-0.053	-0.061	0.081	-0.062	-0.110	-0.013	0.059	0.112	0.150
0.600	1.967	0.00	2.919	11.579	-0.035	-0.055	-0.064	0.084	-0.067	-0.116	-0.020	0.050	0.101	0.132
0.601	2.956	0.01	2.929	11.566	-0.037	-0.057	-0.065	0.086	-0.070	-0.121	-0.023	0.046	0.097	0.124
0.602	3.396	0.02	2.929	11.567	-0.037	-0.058	-0.067	0.088	-0.071	-0.122	-0.024	0.045	0.095	0.124
0.601	3.957	0.02	2.928	11.567	-0.037	-0.058	-0.066	0.087	-0.070	-0.121	-0.023	0.046	0.097	0.129
0.603	5.940	0.01	2.941	11.553	-0.036	-0.055	-0.063	0.083	-0.065	-0.114	-0.014	0.057	0.109	0.150
0.602	7.831	0.02	2.932	11.564	-0.034	-0.054	-0.061	0.080	-0.060	-0.106	-0.004	0.067	0.120	0.167
0.600	3.393	8.01	2.921	11.580	0.026	0.000	-0.013	0.044	-0.039	-0.101	-0.010	0.050	0.090	0.118
0.602	3.403	4.02	2.934	11.561	-0.007	-0.033	-0.043	0.070	-0.059	-0.109	-0.018	0.046	0.085	0.117

Table 7. Continued

(e) $\theta = 180^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.709	0.723	0.738	0.752	0.767	0.781	0.796	0.810	0.825	0.840	0.854	
1.201	1.459	0.01	6.135	6.078	-0.026	-0.057	-0.084	-0.089	-0.086	-0.087	-0.108	-0.132	-0.156	-0.175	-0.192	
1.201	1.985	0.02	6.136	6.074	-0.028	-0.057	-0.085	-0.090	-0.085	-0.086	-0.109	-0.133	-0.156	-0.175	-0.192	
1.201	2.960	0.00	6.136	6.076	-0.030	-0.058	-0.087	-0.091	-0.086	-0.088	-0.111	-0.134	-0.157	-0.176	-0.194	
1.201	2.938	0.01	6.135	6.076	-0.029	-0.057	-0.086	-0.091	-0.085	-0.088	-0.110	-0.133	-0.156	-0.175	-0.194	
1.201	3.442	0.01	6.136	6.076	-0.030	-0.056	-0.087	-0.092	-0.086	-0.088	-0.112	-0.134	-0.156	-0.175	-0.194	
1.201	3.913	0.01	6.136	6.074	-0.032	-0.055	-0.087	-0.094	-0.087	-0.089	-0.114	-0.135	-0.156	-0.176	-0.196	
1.202	5.876	0.03	6.138	6.071	-0.031	-0.053	-0.086	-0.094	-0.086	-0.089	-0.115	-0.135	-0.156	-0.176	-0.196	
1.201	7.896	0.04	6.137	6.075	-0.030	-0.052	-0.086	-0.093	-0.085	-0.089	-0.116	-0.135	-0.157	-0.176	-0.197	
1.200	3.427	8.05	6.132	6.084	0.181	0.136	0.102	0.078	0.052	0.020	-0.031	-0.069	-0.101	-0.128	-0.148	
1.200	3.413	4.00	6.133	6.083	0.070	0.047	0.027	0.017	0.007	-0.015	-0.058	-0.094	-0.129	-0.163	-0.192	
0.900	1.480	0.00	4.947	8.726	-0.052	-0.057	-0.055	-0.045	-0.038	-0.044	-0.047	-0.039	-0.033	-0.035	-0.044	
0.899	1.971	0.02	4.943	8.732	-0.054	-0.059	-0.056	-0.046	-0.039	-0.045	-0.049	-0.041	-0.035	-0.037	-0.045	
0.900	2.955	0.04	4.950	8.721	-0.057	-0.060	-0.058	-0.048	-0.040	-0.046	-0.050	-0.042	-0.035	-0.038	-0.047	
0.901	3.389	-0.01	4.955	8.713	-0.055	-0.059	-0.056	-0.046	-0.038	-0.044	-0.048	-0.039	-0.033	-0.037	-0.046	
0.902	3.929	0.00	4.957	8.709	-0.053	-0.057	-0.054	-0.045	-0.037	-0.043	-0.047	-0.039	-0.033	-0.037	-0.046	
0.901	5.880	0.00	4.952	8.715	-0.055	-0.058	-0.055	-0.046	-0.038	-0.043	-0.047	-0.038	-0.032	-0.036	-0.046	
0.901	7.835	0.02	4.952	8.717	-0.054	-0.058	-0.055	-0.045	-0.037	-0.043	-0.047	-0.038	-0.032	-0.035	-0.044	
0.900	3.431	8.03	4.950	8.723	-0.046	-0.084	-0.100	-0.103	-0.103	-0.112	-0.110	-0.087	-0.064	-0.052	-0.044	
0.902	3.386	4.00	4.959	8.707	-0.045	-0.065	-0.068	-0.062	-0.056	-0.062	-0.063	-0.050	-0.038	-0.037	-0.040	
0.800	1.474	0.01	4.336	9.687	-0.041	-0.056	-0.058	-0.051	-0.046	-0.052	-0.055	-0.049	-0.046	-0.047	-0.050	
0.802	1.966	0.02	4.351	9.664	-0.040	-0.056	-0.058	-0.050	-0.045	-0.051	-0.055	-0.049	-0.046	-0.048	-0.051	
0.802	2.940	0.03	4.350	9.665	-0.039	-0.055	-0.058	-0.051	-0.046	-0.052	-0.055	-0.050	-0.046	-0.048	-0.051	
0.802	3.399	0.04	4.352	9.662	-0.041	-0.056	-0.058	-0.051	-0.046	-0.052	-0.055	-0.050	-0.046	-0.048	-0.051	
0.800	3.904	0.02	4.340	9.681	-0.040	-0.056	-0.058	-0.051	-0.046	-0.052	-0.055	-0.050	-0.046	-0.049	-0.051	
0.802	5.884	0.03	4.348	9.668	-0.040	-0.057	-0.058	-0.051	-0.046	-0.052	-0.054	-0.050	-0.046	-0.048	-0.050	
0.801	7.775	0.01	4.346	9.671	-0.040	-0.058	-0.059	-0.051	-0.046	-0.052	-0.054	-0.049	-0.046	-0.047	-0.049	
0.800	3.404	8.03	4.340	9.682	-0.004	-0.035	-0.044	-0.043	-0.044	-0.052	-0.053	-0.047	-0.039	-0.032	-0.026	
0.800	3.400	4.02	4.335	9.687	-0.014	-0.039	-0.043	-0.038	-0.036	-0.043	-0.045	-0.042	-0.037	-0.036	-0.035	
0.601	1.471	-0.01	2.923	11.573	-0.027	-0.049	-0.048	-0.039	-0.035	-0.041	-0.042	-0.042	-0.042	-0.041	-0.038	
0.600	1.967	0.00	2.919	11.579	-0.027	-0.050	-0.048	-0.039	-0.035	-0.041	-0.042	-0.044	-0.043	-0.041	-0.039	
0.601	2.956	0.01	2.929	11.566	-0.026	-0.051	-0.048	-0.040	-0.036	-0.042	-0.043	-0.044	-0.044	-0.043	-0.040	
0.602	3.396	0.02	2.929	11.567	-0.027	-0.052	-0.050	-0.041	-0.037	-0.044	-0.044	-0.046	-0.045	-0.043	-0.041	
0.601	3.957	0.02	2.928	11.567	-0.026	-0.052	-0.049	-0.041	-0.037	-0.043	-0.042	-0.045	-0.046	-0.043	-0.041	
0.603	5.940	0.01	2.941	11.553	-0.026	-0.051	-0.049	-0.040	-0.037	-0.042	-0.042	-0.045	-0.045	-0.042	-0.040	
0.602	7.831	0.02	2.932	11.564	-0.028	-0.054	-0.051	-0.042	-0.039	-0.044	-0.042	-0.045	-0.045	-0.042	-0.039	
0.600	3.393	8.01	2.921	11.580	0.024	-0.012	-0.014	-0.008	-0.011	-0.018	-0.015	-0.021	-0.019	-0.012	-0.003	
0.602	3.403	4.02	2.934	11.561	0.003	-0.030	-0.029	-0.020	-0.022	-0.027	-0.025	-0.031	-0.031	-0.026	-0.019	

Table 7. Continued

(e) Concluded

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—								
					0.883	0.898	0.912	0.927	0.941	0.951	0.974	0.986	
1.201	1.459	0.01	6.135	6.078	-0.173	-0.159	-0.164	-0.156	-0.147	-0.202	-0.270	-0.248	
1.201	1.985	0.02	6.136	6.074	-0.173	-0.159	-0.165	-0.157	-0.149	-0.204	-0.281	-0.291	
1.201	2.960	0.00	6.136	6.076	-0.172	-0.159	-0.165	-0.159	-0.150	-0.205	-0.285	-0.300	
1.201	2.938	0.01	6.135	6.076	-0.172	-0.159	-0.165	-0.159	-0.149	-0.204	-0.285	-0.300	
1.201	3.442	0.01	6.136	6.076	-0.171	-0.159	-0.165	-0.160	-0.150	-0.205	-0.286	-0.302	
1.201	3.913	0.01	6.136	6.074	-0.170	-0.158	-0.165	-0.162	-0.150	-0.206	-0.286	-0.301	
1.202	5.876	0.03	6.138	6.071	-0.169	-0.158	-0.166	-0.164	-0.152	-0.208	-0.278	-0.273	
1.201	7.896	0.04	6.137	6.075	-0.168	-0.158	-0.166	-0.165	-0.152	-0.207	-0.266	-0.199	
1.200	3.427	8.05	6.132	6.084	-0.115	-0.106	-0.135	-0.142	-0.148	-0.170	-0.287	-0.302	
1.200	3.413	4.00	6.133	6.083	-0.185	-0.159	-0.158	-0.143	-0.126	-0.176	-0.261	-0.284	
0.900	1.480	0.00	4.947	8.726	-0.072	-0.089	-0.117	-0.139	-0.183	-0.225	0.056	0.116	
0.899	1.971	0.02	4.943	8.732	-0.073	-0.091	-0.119	-0.142	-0.190	-0.236	0.047	0.105	
0.900	2.955	0.04	4.950	8.721	-0.075	-0.094	-0.123	-0.147	-0.197	-0.246	0.041	0.099	
0.901	3.389	-0.01	4.955	8.713	-0.075	-0.093	-0.123	-0.148	-0.196	-0.246	0.041	0.100	
0.902	3.929	0.00	4.957	8.709	-0.075	-0.094	-0.123	-0.148	-0.196	-0.246	0.042	0.102	
0.901	5.880	0.00	4.952	8.715	-0.074	-0.092	-0.120	-0.142	-0.187	-0.230	0.056	0.118	
0.901	7.835	0.02	4.952	8.717	-0.071	-0.088	-0.114	-0.133	-0.174	-0.206	0.073	0.137	
0.900	3.431	8.03	4.950	8.723	-0.048	-0.059	-0.086	-0.112	-0.177	-0.277	0.051	0.106	
0.902	3.386	4.00	4.959	8.707	-0.063	-0.080	-0.109	-0.134	-0.193	-0.265	0.041	0.095	
0.800	1.474	0.01	4.336	9.687	-0.069	-0.079	-0.098	-0.110	-0.150	-0.176	0.064	0.122	
0.802	1.966	0.02	4.351	9.664	-0.070	-0.080	-0.100	-0.114	-0.155	-0.185	0.056	0.113	
0.802	2.940	0.03	4.350	9.665	-0.071	-0.081	-0.101	-0.116	-0.158	-0.191	0.051	0.108	
0.802	3.399	0.04	4.352	9.662	-0.072	-0.082	-0.102	-0.116	-0.158	-0.190	0.051	0.109	
0.800	3.904	0.02	4.340	9.681	-0.072	-0.082	-0.102	-0.115	-0.157	-0.189	0.052	0.110	
0.802	5.884	0.03	4.348	9.668	-0.071	-0.080	-0.098	-0.110	-0.150	-0.178	0.064	0.124	
0.801	7.775	0.01	4.346	9.671	-0.069	-0.077	-0.094	-0.103	-0.141	-0.163	0.078	0.142	
0.800	3.404	8.03	4.340	9.682	-0.040	-0.047	-0.069	-0.089	-0.150	-0.206	0.054	0.109	
0.800	3.400	4.02	4.335	9.687	-0.057	-0.066	-0.088	-0.105	-0.159	-0.201	0.048	0.103	
0.601	1.471	-0.01	2.923	11.573	-0.061	-0.062	-0.074	-0.079	-0.121	-0.140	0.063	0.117	
0.600	1.967	0.00	2.919	11.579	-0.062	-0.063	-0.076	-0.083	-0.127	-0.148	0.054	0.105	
0.601	2.956	0.01	2.929	11.566	-0.064	-0.066	-0.079	-0.085	-0.130	-0.152	0.049	0.099	
0.602	3.396	0.02	2.929	11.567	-0.065	-0.066	-0.079	-0.086	-0.131	-0.153	0.048	0.098	
0.601	3.957	0.02	2.928	11.567	-0.066	-0.067	-0.080	-0.085	-0.131	-0.153	0.048	0.100	
0.603	5.940	0.01	2.941	11.553	-0.065	-0.065	-0.076	-0.081	-0.125	-0.144	0.061	0.117	
0.602	7.831	0.02	2.932	11.564	-0.066	-0.066	-0.074	-0.077	-0.119	-0.136	0.073	0.133	
0.600	3.393	8.01	2.921	11.580	-0.033	-0.032	-0.047	-0.057	-0.121	-0.159	0.053	0.102	
0.602	3.403	4.02	2.934	11.561	-0.054	-0.051	-0.066	-0.072	-0.131	-0.160	0.047	0.096	

Table 7. Continued

(f) $\theta = 225^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—											
					0.833	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997		
1.201	1.459	0.01	6.135	6.078	-0.238	-0.214	-0.145	-0.085	-0.145	-0.279	-0.398	-0.284	-0.172	-0.101		
1.201	1.985	0.02	6.136	6.074	-0.236	-0.214	-0.147	-0.086	-0.147	-0.279	-0.398	-0.424	-0.211	-0.128		
1.201	2.960	0.00	6.136	6.076	-0.236	-0.215	-0.147	-0.087	-0.148	-0.281	-0.398	-0.427	-0.239	-0.154		
1.201	2.938	0.01	6.135	6.076	-0.235	-0.215	-0.148	-0.087	-0.149	-0.281	-0.398	-0.426	-0.238	-0.153		
1.201	3.442	0.01	6.136	6.076	-0.234	-0.215	-0.148	-0.086	-0.149	-0.283	-0.398	-0.426	-0.241	-0.154		
1.201	3.913	0.01	6.136	6.074	-0.232	-0.215	-0.149	-0.086	-0.150	-0.285	-0.398	-0.426	-0.234	-0.148		
1.202	5.876	0.03	6.138	6.071	-0.230	-0.215	-0.150	-0.088	-0.153	-0.286	-0.399	-0.401	-0.167	-0.101		
1.201	7.896	0.04	6.137	6.075	-0.229	-0.216	-0.150	-0.088	-0.153	-0.287	-0.398	-0.263	-0.119	-0.056		
1.200	3.427	8.05	6.132	6.084	-0.187	-0.152	-0.077	-0.051	-0.154	-0.276	-0.417	-0.455	-0.390	-0.184		
1.200	3.413	4.00	6.133	6.083	-0.242	-0.238	-0.164	-0.082	-0.134	-0.261	-0.389	-0.413	-0.287	-0.166		
0.900	1.480	0.00	4.947	8.726	-0.094	-0.089	-0.086	-0.111	-0.222	-0.277	0.004	0.076	0.127	0.159		
0.899	1.971	0.02	4.943	8.732	-0.096	-0.091	-0.089	-0.116	-0.233	-0.296	-0.007	0.069	0.120	0.148		
0.900	2.955	0.04	4.950	8.721	-0.099	-0.095	-0.093	-0.122	-0.242	-0.309	-0.014	0.060	0.112	0.141		
0.901	3.389	-0.01	4.955	8.713	-0.098	-0.094	-0.092	-0.121	-0.240	-0.308	-0.014	0.059	0.111	0.140		
0.902	3.929	0.00	4.957	8.709	-0.099	-0.095	-0.093	-0.121	-0.240	-0.307	-0.014	0.061	0.113	0.143		
0.901	5.880	0.00	4.952	8.715	-0.097	-0.091	-0.088	-0.115	-0.228	-0.286	-0.003	0.072	0.125	0.160		
0.901	7.835	0.02	4.952	8.717	-0.093	-0.087	-0.082	-0.105	-0.209	-0.256	0.015	0.093	0.144	0.182		
0.900	3.431	8.03	4.950	8.723	-0.080	-0.067	-0.062	-0.098	-0.247	-0.471	-0.017	0.081	0.132	0.150		
0.902	3.386	4.00	4.959	8.707	-0.092	-0.084	-0.081	-0.114	-0.250	-0.379	-0.020	0.060	0.110	0.135		
0.800	1.474	0.01	4.336	9.687	-0.091	-0.078	-0.069	-0.093	-0.191	-0.239	0.001	0.091	0.143	0.171		
0.802	1.966	0.02	4.351	9.664	-0.093	-0.081	-0.073	-0.098	-0.202	-0.255	-0.008	0.083	0.135	0.161		
0.802	2.940	0.03	4.350	9.665	-0.095	-0.082	-0.075	-0.101	-0.207	-0.263	-0.014	0.078	0.132	0.156		
0.802	3.399	0.04	4.352	9.662	-0.096	-0.083	-0.076	-0.102	-0.206	-0.262	-0.014	0.079	0.132	0.157		
0.800	3.904	0.02	4.340	9.681	-0.096	-0.082	-0.075	-0.101	-0.205	-0.260	-0.013	0.079	0.133	0.160		
0.802	5.884	0.03	4.348	9.668	-0.093	-0.079	-0.070	-0.094	-0.191	-0.240	0.000	0.090	0.143	0.176		
0.801	7.775	0.01	4.346	9.671	-0.091	-0.074	-0.064	-0.084	-0.174	-0.213	0.018	0.104	0.154	0.193		
0.800	3.404	8.03	4.340	9.682	-0.073	-0.056	-0.051	-0.088	-0.222	-0.322	-0.027	0.080	0.138	0.157		
0.800	3.400	4.02	4.335	9.687	-0.088	-0.072	-0.065	-0.098	-0.217	-0.293	-0.024	0.069	0.123	0.148		
0.601	1.471	-0.01	2.923	11.573	-0.087	-0.066	-0.056	-0.078	-0.160	-0.196	-0.001	0.087	0.143	0.175		
0.600	1.967	0.00	2.919	11.579	-0.088	-0.068	-0.059	-0.084	-0.168	-0.210	-0.014	0.075	0.131	0.157		
0.601	2.956	0.01	2.929	11.566	-0.090	-0.070	-0.062	-0.087	-0.173	-0.215	-0.018	0.071	0.126	0.150		
0.602	3.996	0.02	2.929	11.567	-0.092	-0.072	-0.064	-0.089	-0.176	-0.216	-0.019	0.070	0.125	0.150		
0.601	3.957	0.02	2.928	11.567	-0.091	-0.072	-0.063	-0.088	-0.173	-0.214	-0.018	0.071	0.127	0.155		
0.603	5.940	0.01	2.941	11.553	-0.090	-0.068	-0.059	-0.082	-0.163	-0.200	-0.006	0.082	0.141	0.179		
0.602	7.831	0.02	2.932	11.564	-0.091	-0.068	-0.056	-0.078	-0.153	-0.183	0.008	0.095	0.154	0.200		
0.600	3.393	8.01	2.921	11.580	-0.071	-0.047	-0.041	-0.077	-0.183	-0.248	-0.033	0.065	0.126	0.153		
0.602	3.403	4.02	2.934	11.561	-0.087	-0.063	-0.055	-0.088	-0.181	-0.236	-0.032	0.059	0.116	0.147		

Table 7. Continued

(g) $\theta = 240^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—								
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	
1.201	1.459	0.01	6.135	6.078	-0.211	-0.116	-0.075	-0.112	-0.391	-0.464	-0.236	-0.186	
1.201	1.985	0.02	6.136	6.074	-0.213	-0.118	-0.076	-0.114	-0.391	-0.503	-0.320	-0.200	
1.201	2.960	0.00	6.136	6.076	-0.215	-0.118	-0.077	-0.115	-0.391	-0.505	-0.372	-0.216	
1.201	2.938	0.01	6.135	6.076	-0.216	-0.118	-0.077	-0.115	-0.391	-0.505	-0.372	-0.215	
1.201	3.442	0.01	6.136	6.076	-0.217	-0.118	-0.077	-0.116	-0.391	-0.506	-0.377	-0.218	
1.201	3.913	0.01	6.136	6.074	-0.220	-0.118	-0.077	-0.116	-0.391	-0.508	-0.367	-0.215	
1.202	5.876	0.03	6.138	6.071	-0.221	-0.119	-0.077	-0.116	-0.392	-0.509	-0.266	-0.172	
1.201	7.896	0.04	6.137	6.075	-0.223	-0.119	-0.079	-0.118	-0.392	-0.487	-0.212	-0.131	
1.200	3.427	8.05	6.132	6.084	-0.149	-0.059	-0.035	-0.101	-0.389	-0.531	-0.555	-0.270	
1.200	3.413	4.00	6.133	6.083	-0.242	-0.127	-0.068	-0.101	-0.381	-0.500	-0.316	-0.247	
0.900	1.480	0.00	4.947	8.726	-0.099	-0.080	-0.096	-0.167	-0.299	0.016	0.060	0.121	
0.899	1.971	0.02	4.943	8.732	-0.101	-0.083	-0.101	-0.178	-0.320	0.007	0.054	0.115	
0.900	2.955	0.04	4.950	8.721	-0.105	-0.088	-0.106	-0.187	-0.332	-0.001	0.043	0.106	
0.901	3.389	-0.01	4.955	8.713	-0.104	-0.087	-0.106	-0.186	-0.330	-0.001	0.042	0.106	
0.902	3.929	0.00	4.957	8.709	-0.105	-0.088	-0.106	-0.186	-0.329	-0.001	0.043	0.108	
0.901	5.880	0.00	4.952	8.715	-0.101	-0.083	-0.099	-0.173	-0.310	0.010	0.057	0.119	
0.901	7.835	0.02	4.952	8.717	-0.095	-0.075	-0.088	-0.152	-0.283	0.031	0.084	0.135	
0.900	3.431	8.03	4.950	8.723	-0.077	-0.063	-0.088	-0.198	-0.498	0.009	0.082	0.134	
0.902	3.386	4.00	4.959	8.707	-0.092	-0.078	-0.100	-0.196	-0.402	-0.002	0.048	0.106	
0.800	1.474	0.01	4.336	9.687	-0.083	-0.065	-0.077	-0.142	-0.302	0.030	0.095	0.143	
0.802	1.966	0.02	4.351	9.664	-0.086	-0.070	-0.083	-0.154	-0.323	0.021	0.088	0.137	
0.802	2.940	0.03	4.350	9.665	-0.087	-0.072	-0.086	-0.159	-0.334	0.016	0.085	0.135	
0.802	3.399	0.04	4.352	9.662	-0.087	-0.072	-0.087	-0.159	-0.333	0.016	0.086	0.135	
0.800	3.904	0.02	4.340	9.681	-0.087	-0.072	-0.086	-0.158	-0.330	0.017	0.086	0.136	
0.802	5.884	0.03	4.348	9.668	-0.083	-0.066	-0.077	-0.142	-0.303	0.028	0.096	0.142	
0.801	7.775	0.01	4.346	9.671	-0.077	-0.059	-0.066	-0.122	-0.269	0.046	0.106	0.149	
0.800	3.404	8.03	4.340	9.682	-0.061	-0.053	-0.078	-0.177	-0.422	0.011	0.096	0.145	
0.800	3.400	4.02	4.335	9.687	-0.075	-0.065	-0.084	-0.169	-0.379	0.010	0.077	0.127	
0.601	1.471	-0.01	2.923	11.573	-0.059	-0.053	-0.063	-0.121	-0.286	0.032	0.109	0.151	
0.600	1.967	0.00	2.919	11.579	-0.062	-0.056	-0.068	-0.131	-0.305	0.021	0.099	0.141	
0.601	2.956	0.01	2.929	11.566	-0.063	-0.059	-0.072	-0.135	-0.313	0.017	0.095	0.138	
0.602	3.396	0.02	2.929	11.567	-0.065	-0.061	-0.073	-0.137	-0.313	0.016	0.095	0.135	
0.601	3.957	0.02	2.928	11.567	-0.064	-0.061	-0.072	-0.135	-0.312	0.018	0.096	0.137	
0.603	5.940	0.01	2.941	11.553	-0.059	-0.057	-0.066	-0.124	-0.293	0.031	0.107	0.149	
0.602	7.831	0.02	2.932	11.564	-0.056	-0.053	-0.060	-0.113	-0.270	0.046	0.117	0.159	
0.600	3.393	8.01	2.921	11.580	-0.037	-0.044	-0.064	-0.147	-0.372	0.009	0.098	0.143	
0.602	3.403	4.02	2.934	11.561	-0.050	-0.056	-0.072	-0.144	-0.354	0.010	0.089	0.129	

Table 7. Continued

(h) $\theta = 255^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—					
					0.941	0.963	0.974	0.986	0.997	
1.201	1.459	0.01	6.135	6.078	-0.128	-0.426	-0.242	-0.142	-0.074	
1.201	1.985	0.02	6.136	6.074	-0.129	-0.554	-0.302	-0.220	-0.128	
1.201	2.960	0.00	6.136	6.076	-0.130	-0.566	-0.318	-0.237	-0.150	
1.201	2.938	0.01	6.135	6.076	-0.130	-0.566	-0.318	-0.236	-0.149	
1.201	3.442	0.01	6.136	6.076	-0.131	-0.568	-0.321	-0.239	-0.150	
1.201	3.913	0.01	6.136	6.074	-0.132	-0.567	-0.319	-0.238	-0.144	
1.202	5.876	0.03	6.138	6.071	-0.134	-0.440	-0.285	-0.197	-0.102	
1.201	7.896	0.04	6.137	6.075	-0.133	-0.328	-0.241	-0.143	-0.056	
1.200	3.427	8.05	6.132	6.084	-0.109	-0.579	-0.353	-0.266	-0.187	
1.200	3.413	4.00	6.133	6.083	-0.119	-0.576	-0.316	-0.241	-0.159	
0.900	1.480	0.00	4.947	8.726	-0.146	0.015	0.077	0.125	0.163	
0.899	1.971	0.02	4.943	8.732	-0.159	0.004	0.069	0.119	0.155	
0.900	2.955	0.04	4.950	8.721	-0.169	-0.004	0.062	0.114	0.150	
0.901	3.389	-0.01	4.955	8.713	-0.168	-0.004	0.063	0.114	0.150	
0.902	3.929	0.00	4.957	8.709	-0.167	-0.003	0.064	0.115	0.152	
0.901	5.880	0.00	4.952	8.715	-0.153	0.011	0.075	0.123	0.164	
0.901	7.835	0.02	4.952	8.717	-0.129	0.022	0.070	0.129	0.177	
0.900	3.431	8.03	4.950	8.723	-0.197	0.003	0.073	0.128	0.162	
0.902	3.386	4.00	4.959	8.707	-0.181	-0.006	0.059	0.105	0.140	
0.800	1.474	0.01	4.336	9.687	-0.127	0.033	0.084	0.137	0.176	
0.802	1.966	0.02	4.351	9.664	-0.140	0.026	0.082	0.133	0.168	
0.802	2.940	0.03	4.350	9.665	-0.146	0.022	0.079	0.130	0.164	
0.802	3.399	0.04	4.352	9.662	-0.147	0.022	0.079	0.131	0.164	
0.800	3.904	0.02	4.340	9.681	-0.145	0.022	0.078	0.131	0.166	
0.802	5.884	0.03	4.348	9.668	-0.125	0.025	0.075	0.133	0.175	
0.801	7.775	0.01	4.346	9.671	-0.102	0.038	0.081	0.139	0.185	
0.800	3.404	8.03	4.340	9.682	-0.174	0.025	0.092	0.143	0.171	
0.800	3.400	4.02	4.335	9.687	-0.158	0.018	0.081	0.123	0.154	
0.601	1.471	-0.01	2.923	11.573	-0.111	0.050	0.113	0.148	0.172	
0.600	1.967	0.00	2.919	11.579	-0.123	0.040	0.105	0.139	0.156	
0.601	2.956	0.01	2.929	11.566	-0.128	0.036	0.102	0.135	0.151	
0.602	3.396	0.02	2.929	11.567	-0.130	0.035	0.102	0.134	0.150	
0.601	3.957	0.02	2.928	11.567	-0.128	0.037	0.103	0.135	0.153	
0.603	5.940	0.01	2.941	11.553	-0.115	0.048	0.112	0.145	0.171	
0.602	7.831	0.02	2.932	11.564	-0.101	0.061	0.122	0.155	0.185	
0.600	3.393	8.01	2.921	11.580	-0.145	0.034	0.111	0.139	0.153	
0.602	3.403	4.02	2.934	11.561	-0.137	0.031	0.103	0.128	0.141	

Table 7. Continued

(i) $\theta = 270^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—				
					0.951	0.963	0.974	0.986	
1.201	1.459	0.01	6.135	6.078	-0.361	-0.276	-0.205	-0.139	
1.201	1.985	0.02	6.136	6.074	-0.361	-0.496	-0.301	-0.215	
1.201	2.960	0.00	6.136	6.076	-0.361	-0.544	-0.314	-0.236	
1.201	2.938	0.01	6.135	6.076	-0.360	-0.544	-0.314	-0.235	
1.201	3.442	0.01	6.136	6.076	-0.360	-0.548	-0.316	-0.238	
1.201	3.913	0.01	6.136	6.074	-0.359	-0.536	-0.316	-0.237	
1.202	5.876	0.03	6.138	6.071	-0.359	-0.362	-0.283	-0.189	
1.201	7.896	0.04	6.137	6.075	-0.360	-0.285	-0.238	-0.137	
1.200	3.427	8.05	6.132	6.084	-0.371	-0.568	-0.345	-0.265	
1.200	3.413	4.00	6.133	6.083	-0.383	-0.441	-0.260	-0.195	
0.900	1.480	0.00	4.947	8.726	-0.067	0.014	0.076	0.121	
0.899	1.971	0.02	4.943	8.732	-0.083	0.002	0.067	0.115	
0.900	2.955	0.04	4.950	8.721	-0.094	-0.006	0.060	0.111	
0.901	3.389	0.01	4.955	8.713	-0.093	-0.005	0.060	0.110	
0.902	3.929	0.00	4.957	8.709	-0.093	-0.005	0.061	0.112	
0.901	5.880	0.00	4.952	8.715	-0.077	0.008	0.073	0.119	
0.901	7.835	0.02	4.952	8.717	-0.034	0.019	0.070	0.124	
0.900	3.431	8.03	4.950	8.723	-0.126	0.001	0.074	0.125	
0.902	3.386	4.00	4.959	8.707	-0.103	-0.011	0.055	0.101	
0.800	1.474	-0.01	4.336	9.687	-0.057	0.025	0.084	0.135	
0.802	1.966	0.02	4.351	9.664	-0.078	0.018	0.081	0.132	
0.802	2.940	0.03	4.350	9.665	-0.087	0.015	0.079	0.129	
0.802	3.399	0.04	4.352	9.662	-0.086	0.014	0.078	0.129	
0.800	3.904	0.02	4.340	9.681	-0.082	0.015	0.078	0.130	
0.802	5.884	0.03	4.348	9.668	-0.048	0.016	0.073	0.131	
0.801	7.775	0.01	4.346	9.671	-0.018	0.027	0.079	0.137	
0.800	3.404	8.03	4.340	9.682	-0.125	0.017	0.094	0.142	
0.800	3.400	4.02	4.335	9.687	-0.099	0.009	0.079	0.122	
0.601	1.471	-0.01	2.923	11.573	-0.084	0.040	0.113	0.147	
0.600	1.967	0.00	2.919	11.579	-0.099	0.030	0.106	0.140	
0.601	2.956	0.01	2.929	11.566	-0.105	0.025	0.102	0.136	
0.602	3.396	0.02	2.929	11.567	-0.106	0.025	0.103	0.135	
0.601	3.957	0.02	2.928	11.567	-0.105	0.026	0.104	0.136	
0.603	5.940	0.01	2.941	11.553	-0.088	0.036	0.112	0.145	
0.602	7.831	0.02	2.932	11.564	-0.070	0.049	0.121	0.153	
0.600	3.393	8.01	2.921	11.580	-0.132	0.020	0.110	0.141	
0.602	3.403	4.02	2.934	11.561	-0.118	0.015	0.101	0.129	

Table 7. Continued

(j) $\theta = 285^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—							
					0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.201	1.459	0.01	6.135	6.078	-0.147	-0.226	-0.495	-0.240	-0.262	-0.242	-0.159	
1.201	1.985	0.02	6.136	6.074	-0.147	-0.226	-0.496	-0.584	-0.342	-0.210	-0.099	
1.201	2.960	0.00	6.136	6.076	-0.149	-0.228	-0.498	-0.588	-0.363	-0.233	-0.115	
1.201	2.938	0.01	6.135	6.076	-0.148	-0.228	-0.497	-0.588	-0.363	-0.232	-0.114	
1.201	3.442	0.01	6.136	6.076	-0.149	-0.229	-0.498	-0.589	-0.367	-0.237	-0.115	
1.201	3.913	0.01	6.136	6.074	-0.152	-0.231	-0.499	-0.589	-0.367	-0.236	-0.110	
1.202	5.876	0.03	6.138	6.071	-0.151	-0.231	-0.499	-0.519	-0.320	-0.187	-0.073	
1.201	7.896	0.04	6.137	6.075	-0.154	-0.234	-0.499	-0.348	-0.261	-0.157	-0.049	
1.200	3.427	8.05	6.132	6.084	-0.206	-0.265	-0.502	-0.596	-0.371	-0.285	-0.179	
1.200	3.413	4.00	6.133	6.083	-0.219	-0.284	-0.511	-0.301	-0.318	-0.305	-0.218	
0.900	1.480	0.00	4.947	8.726	-0.258	-0.179	-0.047	-0.007	0.040	0.114	0.171	
0.899	1.971	0.02	4.943	8.732	-0.277	-0.201	-0.060	-0.017	0.032	0.109	0.165	
0.900	2.955	0.04	4.950	8.721	-0.292	-0.221	-0.062	-0.027	0.021	0.099	0.161	
0.901	3.389	-0.01	4.955	8.713	-0.291	-0.221	-0.059	-0.028	0.018	0.099	0.162	
0.902	3.929	0.00	4.957	8.709	-0.291	-0.220	-0.056	-0.027	0.019	0.100	0.163	
0.901	5.880	0.00	4.952	8.715	-0.274	-0.195	-0.050	-0.013	0.036	0.110	0.170	
0.901	7.835	0.02	4.952	8.717	-0.241	-0.159	-0.061	0.017	0.067	0.125	0.176	
0.900	3.431	8.03	4.950	8.723	-0.292	-0.237	-0.182	0.002	0.071	0.116	0.146	
0.902	3.386	4.00	4.959	8.707	-0.291	-0.211	-0.075	-0.034	0.026	0.097	0.151	
0.800	1.474	0.01	4.336	9.687	-0.177	-0.161	-0.108	0.022	0.081	0.138	0.181	
0.802	1.966	0.02	4.351	9.664	-0.190	-0.178	-0.121	0.011	0.074	0.133	0.175	
0.802	2.940	0.03	4.350	9.665	-0.198	-0.188	-0.132	0.007	0.072	0.132	0.173	
0.802	3.399	0.04	4.352	9.662	-0.198	-0.188	-0.130	0.005	0.071	0.132	0.173	
0.800	3.904	0.02	4.340	9.681	-0.197	-0.186	-0.131	0.008	0.073	0.132	0.173	
0.800	5.884	0.03	4.348	9.668	-0.176	-0.158	-0.112	0.022	0.083	0.136	0.177	
0.801	7.775	0.01	4.346	9.671	-0.151	-0.126	-0.083	0.040	0.092	0.141	0.182	
0.800	3.404	8.03	4.340	9.682	-0.203	-0.214	-0.195	0.015	0.088	0.140	0.168	
0.800	3.400	4.02	4.335	9.687	-0.195	-0.177	-0.118	-0.004	0.060	0.124	0.163	
0.601	1.471	-0.01	2.923	11.573	-0.146	-0.156	-0.143	0.035	0.103	0.156	0.179	
0.600	1.967	0.00	2.919	11.579	-0.155	-0.170	-0.163	0.026	0.095	0.146	0.165	
0.601	2.956	0.01	2.929	11.566	-0.160	-0.176	-0.169	0.021	0.092	0.143	0.161	
0.602	3.396	0.02	2.929	11.567	-0.162	-0.177	-0.172	0.021	0.091	0.142	0.161	
0.601	3.957	0.02	2.928	11.567	-0.160	-0.175	-0.168	0.021	0.092	0.145	0.162	
0.603	5.940	0.01	2.941	11.553	-0.148	-0.159	-0.149	0.033	0.102	0.155	0.176	
0.602	7.831	0.02	2.932	11.564	-0.134	-0.138	-0.122	0.042	0.110	0.165	0.187	
0.600	3.393	8.01	2.921	11.580	-0.182	-0.199	-0.204	0.017	0.100	0.153	0.162	
0.602	3.403	4.02	2.934	11.561	-0.169	-0.174	-0.163	0.013	0.087	0.141	0.153	

Table 7. Continued

(k) $\theta = 300^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—									
					0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.201	1.459	0.01	6.135	6.078	-0.154	-0.160	-0.155	-0.208	-0.526	-0.311	-0.159	-0.090	-0.058	
1.201	1.985	0.02	6.136	6.074	-0.155	-0.159	-0.155	-0.208	-0.582	-0.285	-0.212	-0.153	-0.074	
1.201	2.960	0.00	6.136	6.076	-0.156	-0.160	-0.158	-0.209	-0.583	-0.308	-0.227	-0.171	-0.088	
1.201	2.938	0.01	6.135	6.076	-0.156	-0.160	-0.158	-0.209	-0.583	-0.308	-0.226	-0.171	-0.086	
1.201	3.442	0.01	6.136	6.076	-0.158	-0.160	-0.159	-0.211	-0.584	-0.312	-0.228	-0.176	-0.085	
1.201	3.913	0.01	6.136	6.074	-0.160	-0.161	-0.162	-0.213	-0.586	-0.306	-0.225	-0.178	-0.079	
1.202	5.876	0.03	6.138	6.071	-0.160	-0.160	-0.163	-0.213	-0.559	-0.263	-0.189	-0.152	-0.046	
1.201	7.896	0.04	6.137	6.075	-0.162	-0.162	-0.166	-0.215	-0.357	-0.245	-0.161	-0.116	-0.013	
1.200	3.427	8.05	6.132	6.084	-0.232	-0.254	-0.216	-0.246	-0.581	-0.361	-0.274	-0.217	-0.114	
1.200	3.413	4.00	6.133	6.083	-0.228	-0.233	-0.225	-0.270	-0.601	-0.340	-0.221	-0.158	-0.106	
0.900	1.480	0.00	4.947	8.726	-0.334	-0.319	-0.302	-0.236	0.027	0.081	0.119	0.154	0.115	
0.899	1.971	0.02	4.943	8.732	-0.338	-0.326	-0.317	-0.261	0.017	0.073	0.113	0.148	0.106	
0.900	2.955	0.04	4.950	8.721	-0.338	-0.328	-0.330	-0.288	0.008	0.065	0.105	0.140	0.102	
0.901	3.389	-0.01	4.955	8.713	-0.335	-0.326	-0.328	-0.289	0.009	0.065	0.105	0.140	0.102	
0.902	3.929	0.00	4.957	8.709	-0.335	-0.326	-0.327	-0.288	0.009	0.066	0.106	0.141	0.103	
0.901	5.880	0.00	4.952	8.715	-0.335	-0.323	-0.315	-0.259	0.022	0.077	0.116	0.151	0.115	
0.901	7.835	0.02	4.952	8.717	-0.333	-0.313	-0.288	-0.213	0.033	0.075	0.121	0.161	0.126	
0.900	3.431	8.03	4.950	8.723	-0.407	-0.391	-0.331	-0.250	0.016	0.055	0.082	0.114	0.092	
0.902	3.386	4.00	4.959	8.707	-0.363	-0.350	-0.331	-0.266	0.004	0.055	0.090	0.129	0.093	
0.800	1.474	0.01	4.336	9.687	-0.317	-0.241	-0.198	-0.193	0.038	0.085	0.129	0.178	0.130	
0.802	1.966	0.02	4.351	9.664	-0.330	-0.254	-0.211	-0.208	0.029	0.082	0.125	0.172	0.119	
0.802	2.940	0.03	4.350	9.665	-0.337	-0.262	-0.219	-0.216	0.025	0.077	0.121	0.167	0.115	
0.802	3.399	0.04	4.352	9.662	-0.338	-0.263	-0.220	-0.217	0.025	0.078	0.121	0.168	0.115	
0.800	3.904	0.02	4.340	9.681	-0.334	-0.261	-0.219	-0.215	0.025	0.076	0.121	0.169	0.117	
0.802	5.884	0.03	4.348	9.668	-0.320	-0.244	-0.199	-0.192	0.032	0.074	0.123	0.176	0.128	
0.801	7.775	0.01	4.346	9.671	-0.302	-0.224	-0.176	-0.162	0.044	0.080	0.128	0.185	0.137	
0.800	3.404	8.03	4.340	9.682	-0.384	-0.273	-0.217	-0.232	0.024	0.085	0.117	0.152	0.103	
0.800	3.400	4.02	4.335	9.687	-0.351	-0.266	-0.213	-0.205	0.017	0.071	0.108	0.157	0.105	
0.601	1.471	-0.01	2.923	11.573	-0.224	-0.178	-0.156	-0.175	0.041	0.100	0.129	0.183	0.123	
0.600	1.967	0.00	2.919	11.579	-0.230	-0.185	-0.164	-0.187	0.029	0.091	0.120	0.169	0.101	
0.601	2.956	0.01	2.929	11.566	-0.234	-0.189	-0.170	-0.193	0.024	0.087	0.115	0.164	0.095	
0.602	3.396	0.02	2.929	11.567	-0.237	-0.191	-0.172	-0.194	0.023	0.086	0.113	0.163	0.094	
0.601	3.957	0.02	2.928	11.567	-0.235	-0.190	-0.170	-0.191	0.025	0.086	0.115	0.167	0.098	
0.603	5.940	0.01	2.941	11.553	-0.226	-0.180	-0.158	-0.177	0.039	0.097	0.124	0.184	0.117	
0.602	7.831	0.02	2.932	11.564	-0.216	-0.170	-0.144	-0.159	0.052	0.105	0.132	0.197	0.128	
0.600	3.393	8.01	2.921	11.580	-0.283	-0.223	-0.191	-0.218	0.017	0.084	0.107	0.158	0.082	
0.602	3.403	4.02	2.934	11.561	-0.256	-0.207	-0.176	-0.191	0.013	0.074	0.097	0.157	0.085	

Table 7. Continued

(1) $\theta = 315^\circ$

M	NPR	α , deg	q_∞ , psi	p_∞ , psi	C_p at values of x/L of—										
					0.883	0.898	0.912	0.927	0.941	0.951	0.963	0.974	0.986	0.997	
1.201	1.459	0.01	6.135	6.078	-0.095	-0.116	-0.169	-0.165	-0.251	-0.381	-0.282	-0.248	-0.188	-0.141	
1.201	1.985	0.02	6.136	6.074	-0.095	-0.115	-0.168	-0.164	-0.251	-0.380	-0.526	-0.363	-0.207	-0.151	
1.201	2.960	0.00	6.136	6.076	-0.095	-0.114	-0.168	-0.165	-0.251	-0.381	-0.527	-0.470	-0.222	-0.169	
1.201	2.938	0.01	6.135	6.076	-0.095	-0.114	-0.167	-0.164	-0.251	-0.381	-0.527	-0.468	-0.223	-0.169	
1.201	3.442	0.01	6.136	6.076	-0.096	-0.113	-0.168	-0.164	-0.251	-0.381	-0.528	-0.475	-0.226	-0.173	
1.201	3.913	0.01	6.136	6.074	-0.098	-0.112	-0.168	-0.165	-0.252	-0.382	-0.529	-0.463	-0.227	-0.175	
1.202	5.876	0.03	6.138	6.071	-0.098	-0.110	-0.166	-0.164	-0.252	-0.382	-0.528	-0.294	-0.196	-0.159	
1.201	7.896	0.04	6.137	6.075	-0.100	-0.111	-0.168	-0.165	-0.253	-0.383	-0.494	-0.243	-0.167	-0.137	
1.200	3.427	8.05	6.132	6.084	-0.169	-0.189	-0.256	-0.246	-0.308	-0.430	-0.566	-0.341	-0.247	-0.225	
1.200	3.413	4.00	6.133	6.083	-0.169	-0.193	-0.240	-0.236	-0.315	-0.433	-0.347	-0.325	-0.257	-0.203	
0.900	1.480	0.00	4.947	8.726	-0.236	-0.298	-0.337	-0.348	-0.279	-0.150	0.027	0.066	0.108	0.149	
0.899	1.971	0.02	4.943	8.732	-0.238	-0.301	-0.342	-0.360	-0.308	-0.164	0.017	0.059	0.100	0.142	
0.900	2.955	0.04	4.950	8.721	-0.237	-0.300	-0.344	-0.369	-0.343	-0.168	0.009	0.050	0.093	0.134	
0.901	3.389	-0.01	4.955	8.713	-0.234	-0.297	-0.341	-0.367	-0.344	-0.168	0.008	0.048	0.093	0.133	
0.902	3.929	0.00	4.957	8.709	-0.235	-0.298	-0.341	-0.367	-0.344	-0.165	0.009	0.048	0.093	0.132	
0.901	5.880	0.00	4.952	8.715	-0.235	-0.298	-0.339	-0.358	-0.309	-0.154	0.021	0.059	0.103	0.139	
0.901	7.835	0.02	4.952	8.717	-0.235	-0.297	-0.333	-0.336	-0.258	-0.145	0.040	0.082	0.119	0.153	
0.900	3.431	8.03	4.950	8.723	-0.307	-0.363	-0.417	-0.370	-0.265	-0.209	0.004	0.050	0.080	0.114	
0.902	3.386	4.00	4.959	8.707	-0.267	-0.328	-0.367	-0.377	-0.308	-0.156	0.004	0.047	0.084	0.118	
0.800	1.474	0.01	4.336	9.687	-0.291	-0.304	-0.261	-0.225	-0.227	-0.208	0.023	0.090	0.130	0.169	
0.802	1.966	0.02	4.351	9.664	-0.299	-0.317	-0.274	-0.237	-0.240	-0.223	0.014	0.083	0.124	0.163	
0.802	2.940	0.03	4.350	9.665	-0.303	-0.323	-0.282	-0.244	-0.249	-0.232	0.008	0.080	0.121	0.160	
0.802	3.399	0.04	4.352	9.662	-0.303	-0.324	-0.283	-0.246	-0.246	-0.232	0.008	0.079	0.121	0.159	
0.800	3.904	0.02	4.340	9.681	-0.300	-0.321	-0.280	-0.244	-0.245	-0.231	0.008	0.080	0.122	0.159	
0.802	5.884	0.03	4.348	9.668	-0.294	-0.308	-0.264	-0.226	-0.225	-0.210	0.019	0.089	0.130	0.167	
0.801	7.775	0.01	4.346	9.671	-0.283	-0.293	-0.245	-0.206	-0.201	-0.182	0.039	0.100	0.138	0.175	
0.800	3.404	8.03	4.340	9.682	-0.385	-0.379	-0.293	-0.241	-0.254	-0.266	-0.002	0.078	0.118	0.146	
0.800	3.400	4.02	4.335	9.687	-0.337	-0.346	-0.286	-0.240	-0.233	-0.206	0.004	0.067	0.112	0.148	
0.601	1.471	-0.01	2.923	11.573	-0.217	-0.224	-0.191	-0.185	-0.205	-0.208	0.007	0.095	0.146	0.178	
0.600	1.967	0.00	2.919	11.579	-0.222	-0.230	-0.198	-0.194	-0.216	-0.224	-0.006	0.084	0.134	0.163	
0.601	2.956	0.01	2.929	11.566	-0.225	-0.235	-0.203	-0.198	-0.223	-0.231	-0.012	0.079	0.130	0.157	
0.602	3.396	0.02	2.929	11.567	-0.227	-0.238	-0.205	-0.201	-0.225	-0.233	-0.013	0.077	0.129	0.157	
0.601	3.957	0.02	2.928	11.567	-0.226	-0.237	-0.203	-0.199	-0.222	-0.231	-0.011	0.079	0.132	0.160	
0.603	5.940	0.01	2.941	11.553	-0.221	-0.231	-0.196	-0.188	-0.210	-0.215	0.003	0.090	0.144	0.177	
0.602	7.831	0.02	2.932	11.564	-0.213	-0.223	-0.186	-0.177	-0.194	-0.193	0.020	0.101	0.155	0.191	
0.600	3.393	8.01	2.921	11.580	-0.290	-0.292	-0.242	-0.227	-0.253	-0.267	-0.022	0.075	0.135	0.160	
0.602	3.403	4.02	2.934	11.561	-0.257	-0.268	-0.224	-0.212	-0.226	-0.218	-0.014	0.070	0.127	0.155	

Table 7. Continued

(m) Top interfairing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—				
					0.709	0.723	0.898	0.941	
1.201	1.459	0.01	6.135	6.078	-0.045	-0.040	-0.146	-0.233	
1.201	1.985	0.02	6.136	6.074	-0.045	-0.040	-0.146	-0.233	
1.201	2.960	0.00	6.136	6.076	-0.046	-0.039	-0.148	-0.234	
1.201	2.938	0.01	6.135	6.076	-0.047	-0.040	-0.148	-0.234	
1.201	3.442	0.01	6.136	6.076	-0.048	-0.040	-0.150	-0.234	
1.201	3.913	0.01	6.136	6.074	-0.050	-0.040	-0.152	-0.235	
1.202	5.876	0.03	6.138	6.071	-0.051	-0.040	-0.152	-0.235	
1.201	7.896	0.04	6.137	6.075	-0.053	-0.040	-0.155	-0.236	
1.200	3.427	8.05	6.132	6.084	-0.179	-0.169	-0.222	-0.264	
1.200	3.413	4.00	6.133	6.083	-0.127	-0.125	-0.221	-0.286	
0.900	1.480	0.00	4.947	8.726	-0.054	-0.035	-0.319	-0.129	
0.899	1.971	0.02	4.943	8.732	-0.056	-0.037	-0.323	-0.149	
0.900	2.955	0.04	4.950	8.721	-0.062	-0.040	-0.323	-0.165	
0.901	3.389	-0.01	4.955	8.713	-0.056	-0.036	-0.320	-0.164	
0.902	3.929	0.00	4.957	8.709	-0.058	-0.037	-0.320	-0.162	
0.901	5.880	0.00	4.952	8.715	-0.061	-0.040	-0.320	-0.142	
0.901	7.835	0.02	4.952	8.717	-0.058	-0.038	-0.318	-0.108	
0.900	3.431	8.03	4.950	8.723	-0.117	-0.097	-0.390	-0.204	
0.902	3.386	4.00	4.959	8.707	-0.055	-0.046	-0.347	-0.164	
0.800	1.474	0.01	4.336	9.687	-0.048	-0.041	-0.304	-0.128	
0.802	1.966	0.02	4.351	9.664	-0.049	-0.043	-0.317	-0.145	
0.802	2.940	0.03	4.350	9.665	-0.049	-0.043	-0.323	-0.155	
0.802	3.399	0.04	4.352	9.662	-0.049	-0.042	-0.324	-0.155	
0.800	3.904	0.02	4.340	9.681	-0.046	-0.042	-0.321	-0.154	
0.802	5.884	0.03	4.348	9.668	-0.051	-0.044	-0.306	-0.124	
0.801	7.775	0.01	4.346	9.671	-0.050	-0.043	-0.289	-0.090	
0.800	3.404	8.03	4.340	9.682	-0.095	-0.081	-0.368	-0.184	
0.800	3.400	4.02	4.335	9.687	-0.065	-0.057	-0.338	-0.148	
0.601	1.471	-0.01	2.923	11.573	-0.036	-0.041	-0.216	-0.135	
0.600	1.967	0.00	2.919	11.579	-0.036	-0.041	-0.223	-0.148	
0.601	2.956	0.01	2.929	11.566	-0.036	-0.041	-0.226	-0.155	
0.602	3.396	0.02	2.929	11.567	-0.038	-0.043	-0.229	-0.157	
0.601	3.957	0.02	2.928	11.567	-0.038	-0.043	-0.227	-0.155	
0.603	5.940	0.01	2.941	11.553	-0.038	-0.043	-0.219	-0.139	
0.602	7.831	0.02	2.932	11.564	-0.036	-0.043	-0.208	-0.119	
0.600	3.393	8.01	2.921	11.580	-0.072	-0.076	-0.273	-0.174	
0.602	3.403	4.02	2.934	11.561	-0.052	-0.059	-0.247	-0.155	

Table 7. Concluded

(n) Bottom interfringing

M	NPR	α , deg	q_{∞} , psi	p_{∞} , psi	C_p at values of x/L of—									
					0.709	0.723	0.738	0.752	0.767	0.781	0.869	0.883	0.941	
1.201	1.459	0.01	6.135	6.078	-0.027	-0.050	-0.075	-0.089	-0.099	-0.102	-0.231	-0.245	-0.141	
1.201	1.985	0.02	6.136	6.074	-0.027	-0.050	-0.075	-0.089	-0.099	-0.103	-0.229	-0.244	-0.143	
1.201	2.960	0.00	6.136	6.076	-0.028	-0.051	-0.076	-0.089	-0.101	-0.105	-0.230	-0.244	-0.144	
1.201	2.938	0.01	6.135	6.076	-0.027	-0.050	-0.074	-0.088	-0.100	-0.104	-0.230	-0.245	-0.144	
1.201	3.442	0.01	6.136	6.076	-0.026	-0.049	-0.074	-0.088	-0.101	-0.106	-0.229	-0.244	-0.144	
1.201	3.913	0.01	6.136	6.074	-0.025	-0.048	-0.074	-0.088	-0.102	-0.108	-0.228	-0.243	-0.145	
1.202	5.876	0.03	6.138	6.071	-0.023	-0.045	-0.072	-0.085	-0.102	-0.108	-0.227	-0.242	-0.146	
1.201	7.896	0.04	6.137	6.075	-0.021	-0.043	-0.071	-0.084	-0.101	-0.108	-0.228	-0.243	-0.146	
1.200	3.427	8.05	6.132	6.084	0.177	0.145	0.113	0.088	0.059	0.021	-0.181	-0.193	-0.120	
1.200	3.413	4.00	6.133	6.083	0.061	0.046	0.031	0.027	0.009	-0.020	-0.227	-0.252	-0.133	
0.900	1.480	0.00	4.947	8.726	-0.057	-0.059	-0.056	-0.052	-0.057	-0.077	-0.082	-0.094	-0.149	
0.899	1.971	0.02	4.943	8.732	-0.058	-0.060	-0.057	-0.052	-0.057	-0.077	-0.084	-0.097	-0.159	
0.900	2.955	0.04	4.950	8.721	-0.061	-0.062	-0.059	-0.054	-0.059	-0.079	-0.083	-0.096	-0.158	
0.901	3.389	-0.01	4.955	8.713	-0.060	-0.060	-0.058	-0.052	-0.057	-0.076	-0.083	-0.096	-0.157	
0.902	3.929	0.00	4.957	8.709	-0.059	-0.059	-0.056	-0.051	-0.056	-0.076	-0.082	-0.094	-0.143	
0.901	5.880	0.00	4.952	8.715	-0.061	-0.060	-0.057	-0.051	-0.056	-0.076	-0.080	-0.090	-0.118	
0.901	7.835	0.02	4.952	8.717	-0.061	-0.060	-0.057	-0.052	-0.056	-0.076	-0.072	-0.075	-0.195	
0.900	3.431	8.03	4.950	8.723	-0.045	-0.079	-0.095	-0.102	-0.113	-0.142	-0.072	-0.075	-0.173	
0.902	3.386	4.00	4.959	8.707	-0.049	-0.065	-0.068	-0.067	-0.072	-0.095	-0.077	-0.087	-0.173	
0.800	1.474	0.01	4.336	9.687	-0.047	-0.058	-0.060	-0.059	-0.065	-0.084	-0.081	-0.086	-0.120	
0.802	1.966	0.02	4.351	9.664	-0.046	-0.058	-0.060	-0.060	-0.065	-0.085	-0.082	-0.087	-0.135	
0.802	2.940	0.03	4.350	9.665	-0.046	-0.058	-0.061	-0.060	-0.065	-0.085	-0.083	-0.088	-0.142	
0.802	3.399	0.04	4.352	9.662	-0.047	-0.058	-0.061	-0.060	-0.066	-0.085	-0.083	-0.089	-0.142	
0.800	3.904	0.02	4.340	9.681	-0.047	-0.058	-0.061	-0.061	-0.065	-0.084	-0.083	-0.089	-0.141	
0.802	5.884	0.03	4.348	9.668	-0.048	-0.059	-0.061	-0.061	-0.065	-0.084	-0.082	-0.087	-0.119	
0.801	7.775	0.01	4.346	9.671	-0.049	-0.059	-0.061	-0.061	-0.066	-0.085	-0.080	-0.084	-0.094	
0.800	3.404	8.03	4.340	9.682	-0.008	-0.031	-0.042	-0.048	-0.056	-0.081	-0.062	-0.065	-0.174	
0.800	3.400	4.02	4.335	9.687	-0.022	-0.040	-0.046	-0.048	-0.054	-0.075	-0.073	-0.079	-0.153	
0.601	1.471	-0.01	2.923	11.573	-0.040	-0.053	-0.056	-0.056	-0.057	-0.073	-0.073	-0.075	-0.113	
0.600	1.967	0.00	2.919	11.579	-0.040	-0.052	-0.055	-0.055	-0.056	-0.072	-0.074	-0.076	-0.123	
0.601	2.956	0.01	2.929	11.566	-0.042	-0.054	-0.056	-0.056	-0.057	-0.073	-0.076	-0.078	-0.129	
0.602	3.396	0.02	2.929	11.567	-0.043	-0.055	-0.058	-0.058	-0.058	-0.074	-0.077	-0.080	-0.130	
0.601	3.957	0.02	2.928	11.567	-0.043	-0.056	-0.058	-0.059	-0.058	-0.074	-0.077	-0.080	-0.129	
0.603	5.940	0.01	2.941	11.553	-0.043	-0.055	-0.057	-0.058	-0.057	-0.073	-0.076	-0.077	-0.116	
0.602	7.831	0.02	2.932	11.564	-0.045	-0.058	-0.058	-0.060	-0.058	-0.073	-0.076	-0.076	-0.101	
0.600	3.393	8.01	2.921	11.580	0.008	-0.011	-0.018	-0.023	-0.023	-0.042	-0.050	-0.054	-0.150	
0.602	3.403	4.02	2.934	11.561	-0.018	-0.033	-0.037	-0.040	-0.038	-0.054	-0.066	-0.070	-0.140	

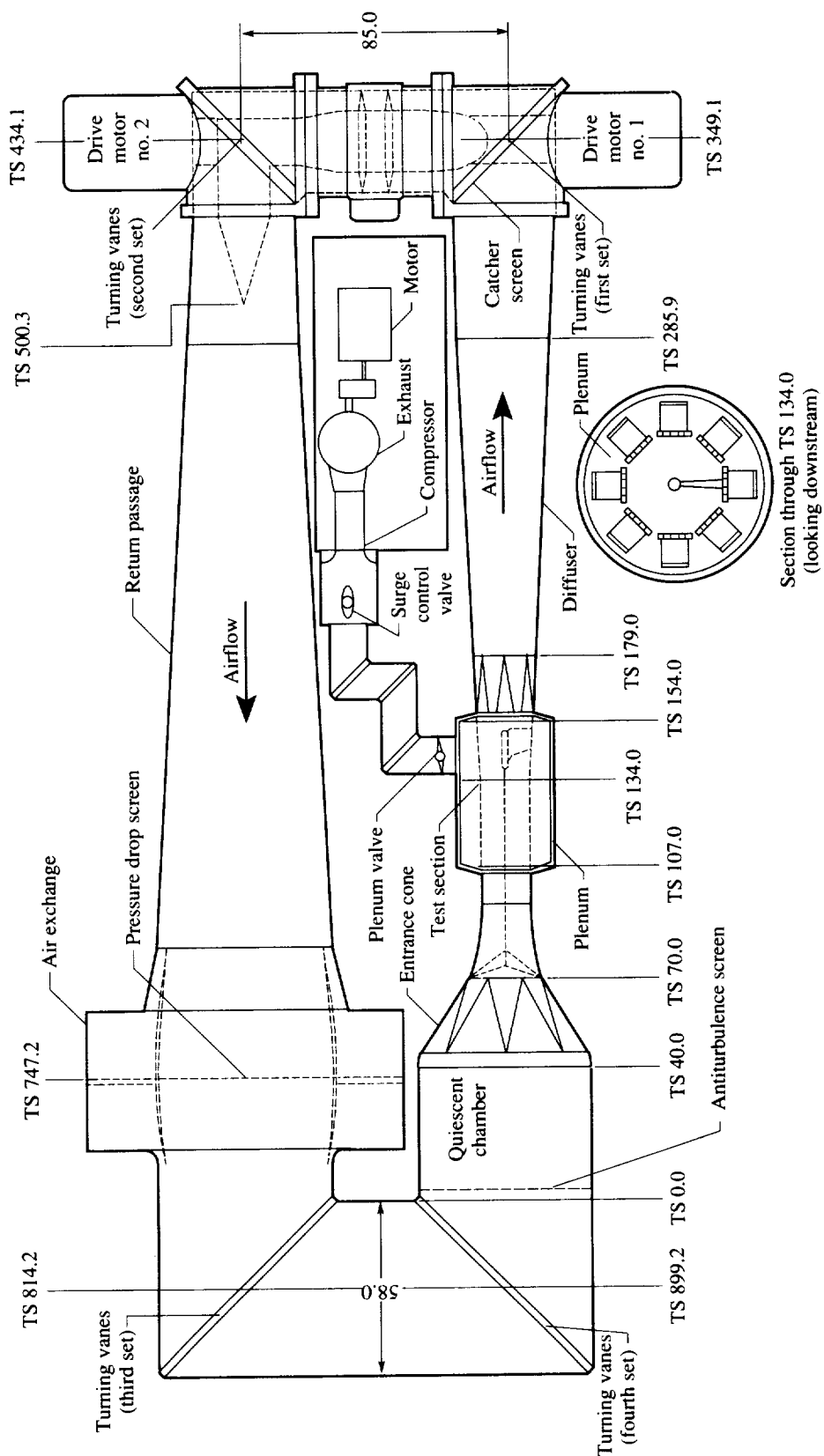
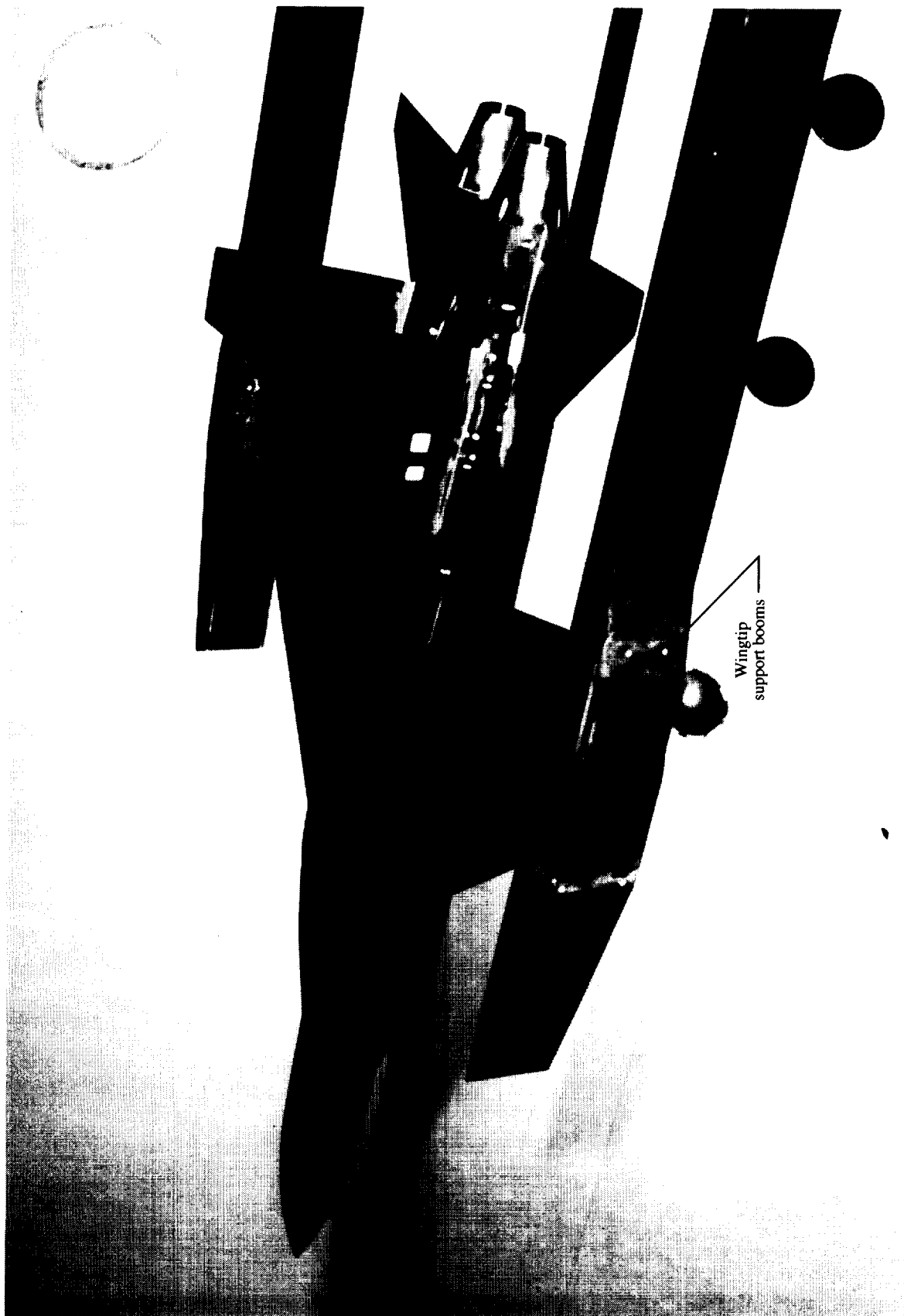
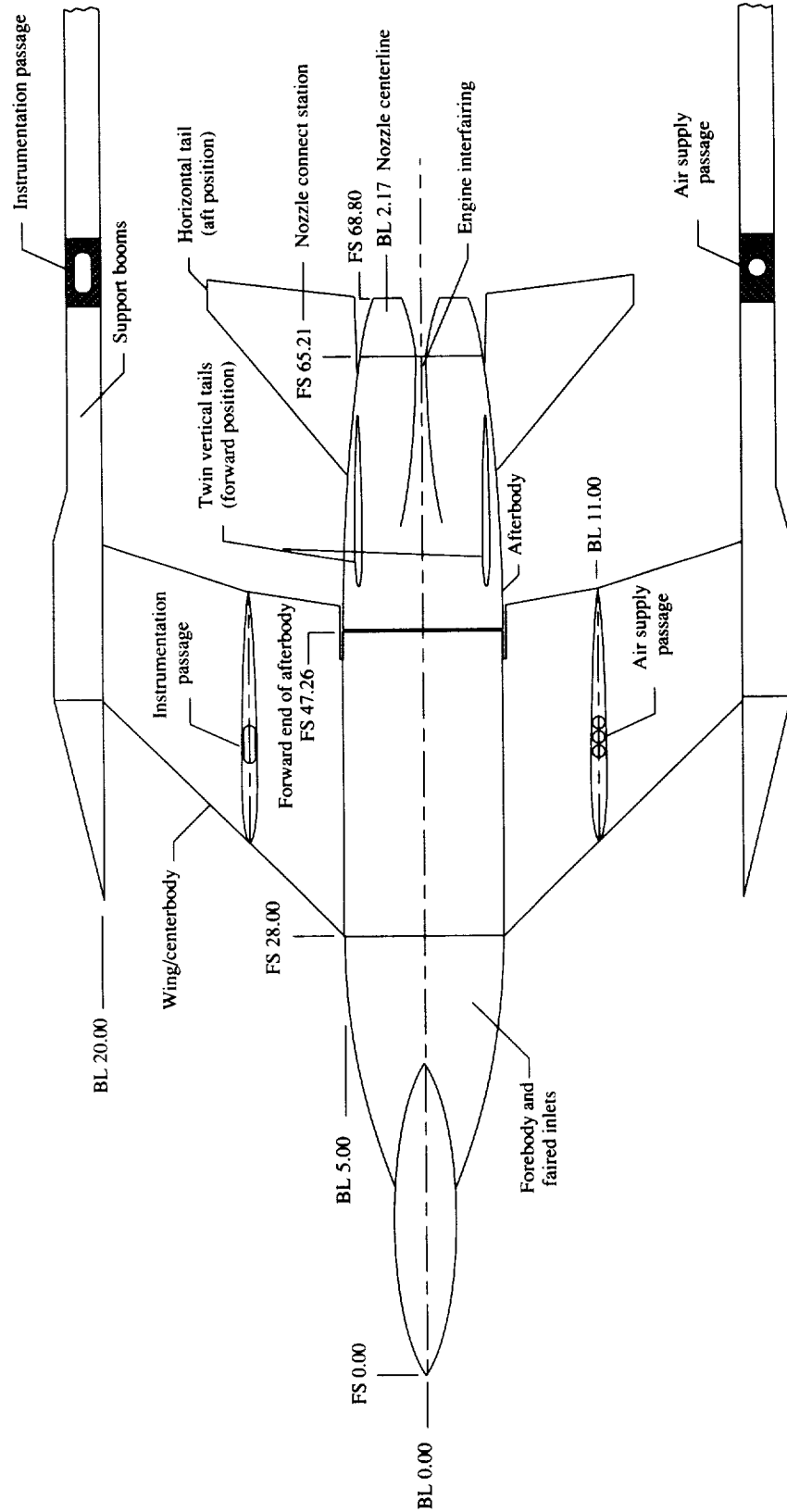


Figure 1. The Langley 16-Foot Transonic Tunnel. Dimensions are given in feet.



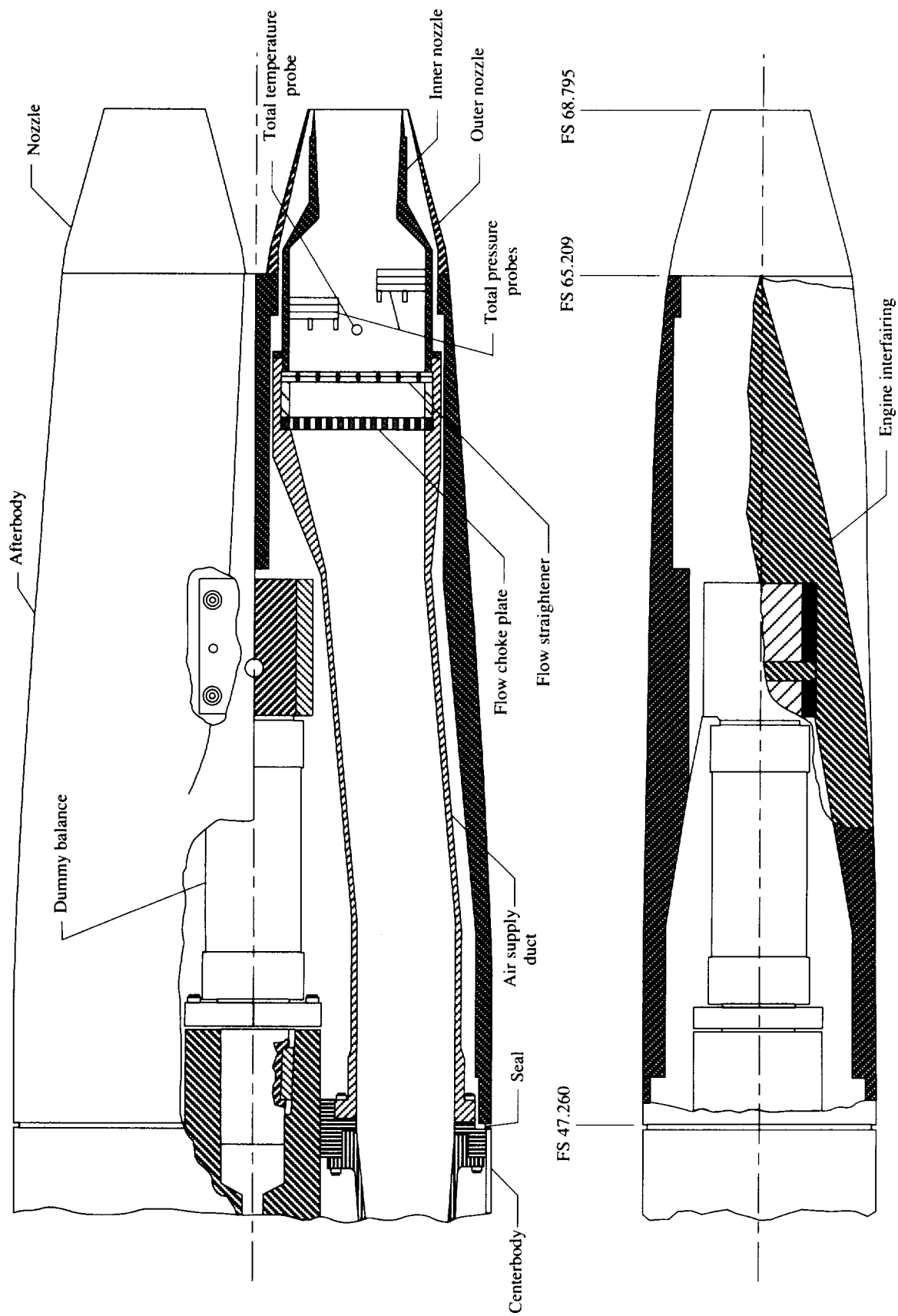
L-91-06848

Figure 2. Photograph of model installed in the Langley 16-Foot Transonic Tunnel.



(a) Model and wingtip support system.

Figure 3. Model sketches. All dimensions are given in inches unless otherwise noted.

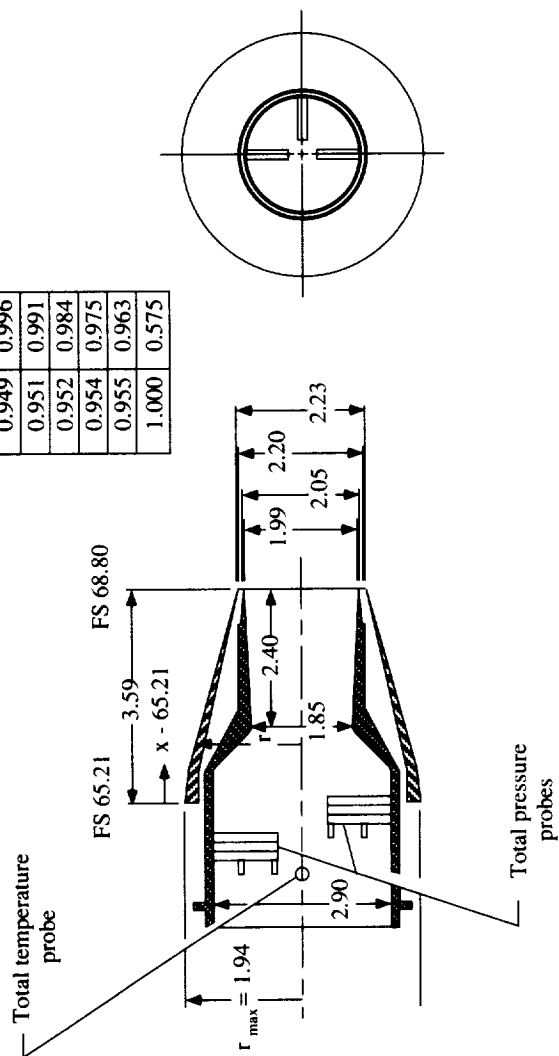


(b) Internal details of afterbody support system and flow path.

Figure 3. Continued.

Nozzle external coordinates:
 $L = 68.80$ in.

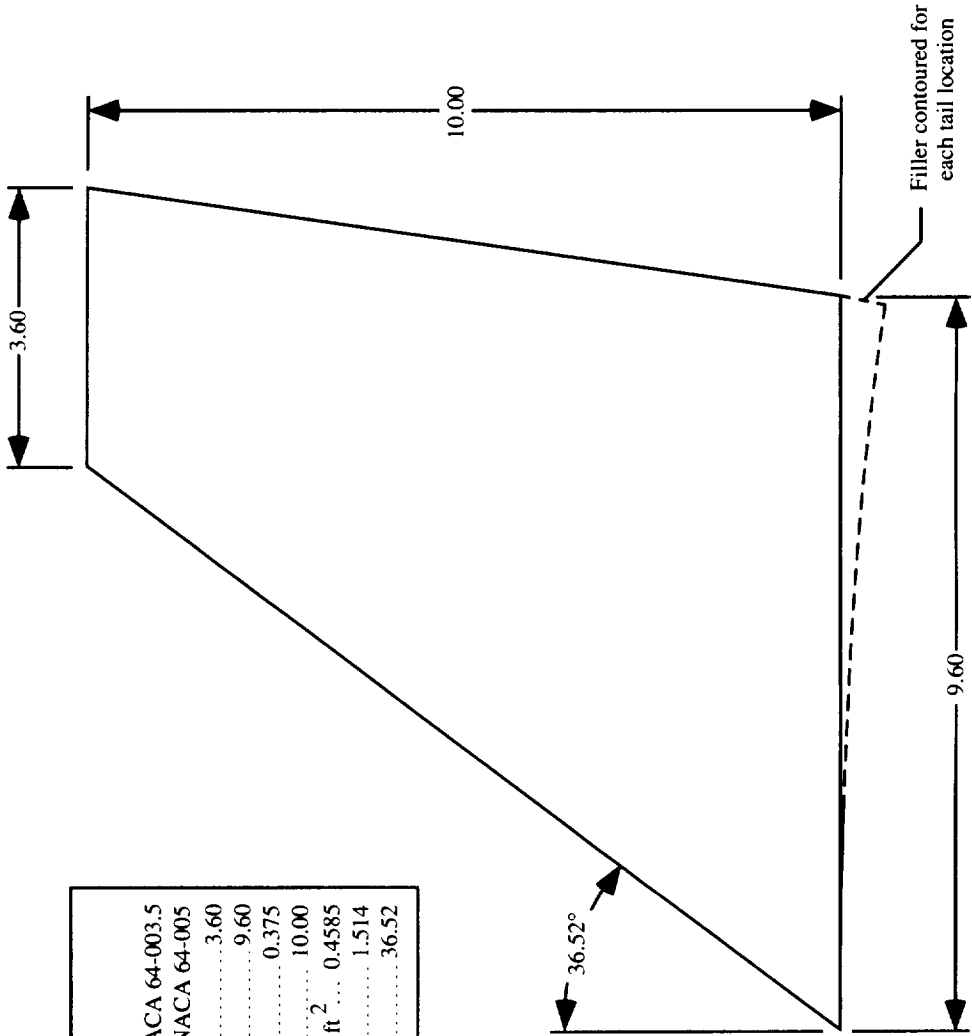
x/L	$r/r_{\max}$
0.948	1.000
0.949	0.996
0.951	0.991
0.952	0.984
0.954	0.975
0.955	0.963
1.000	0.575



(c) Nozzle geometric details.

Figure 3. Continued.

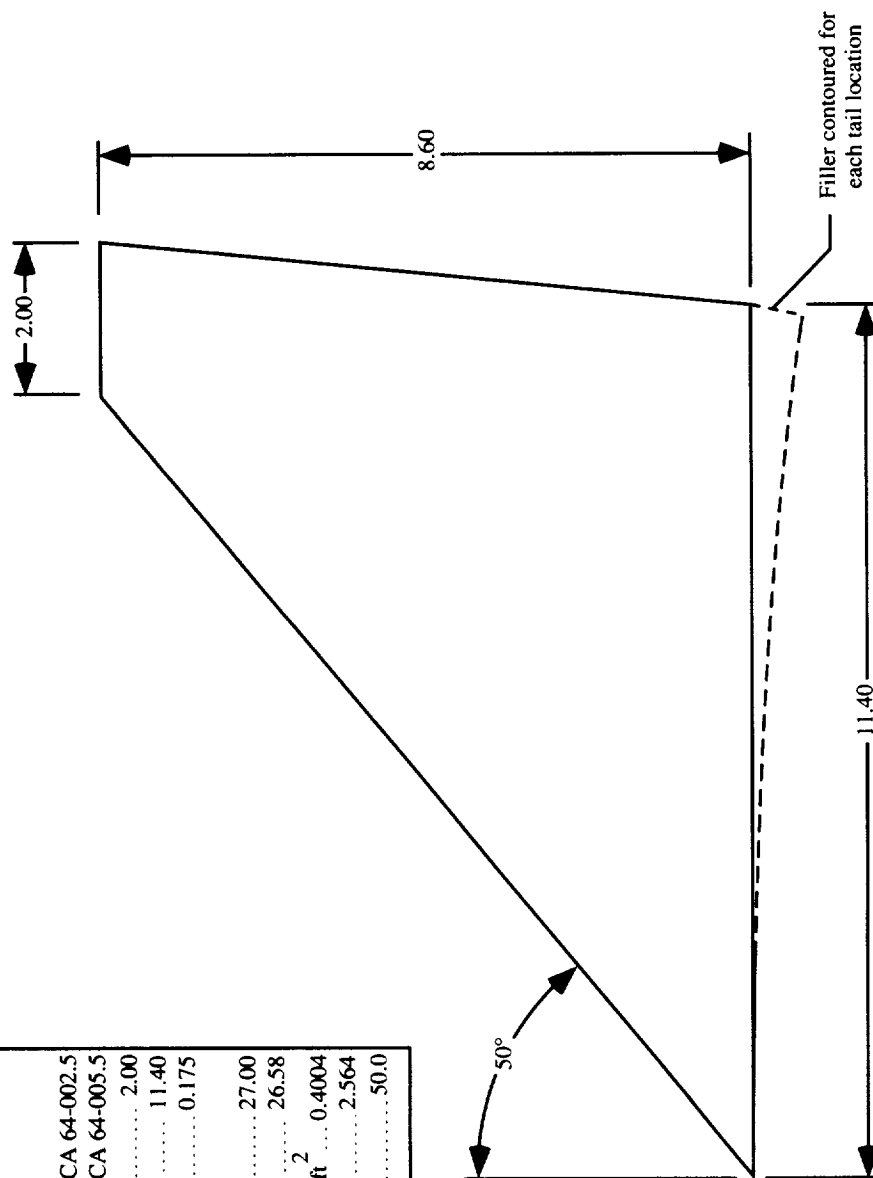
Vertical-Tail Geometry		
Airfoil section:		
Tip	NACA 64-003.5	
Root	NACA 64-005	
Tip chord, in.	3.60	
Root chord, in.	9.60	
Taper ratio	0.375	
Tail height (root to tip), in.	10.00	
Planform area (one side exposed, filler excluded), ft ² ...	0.4585	
Aspect ratio (exposed, filler excluded)	1.514	
Λ_{le} , deg	36.52	



(d) Vertical-tail geometry.

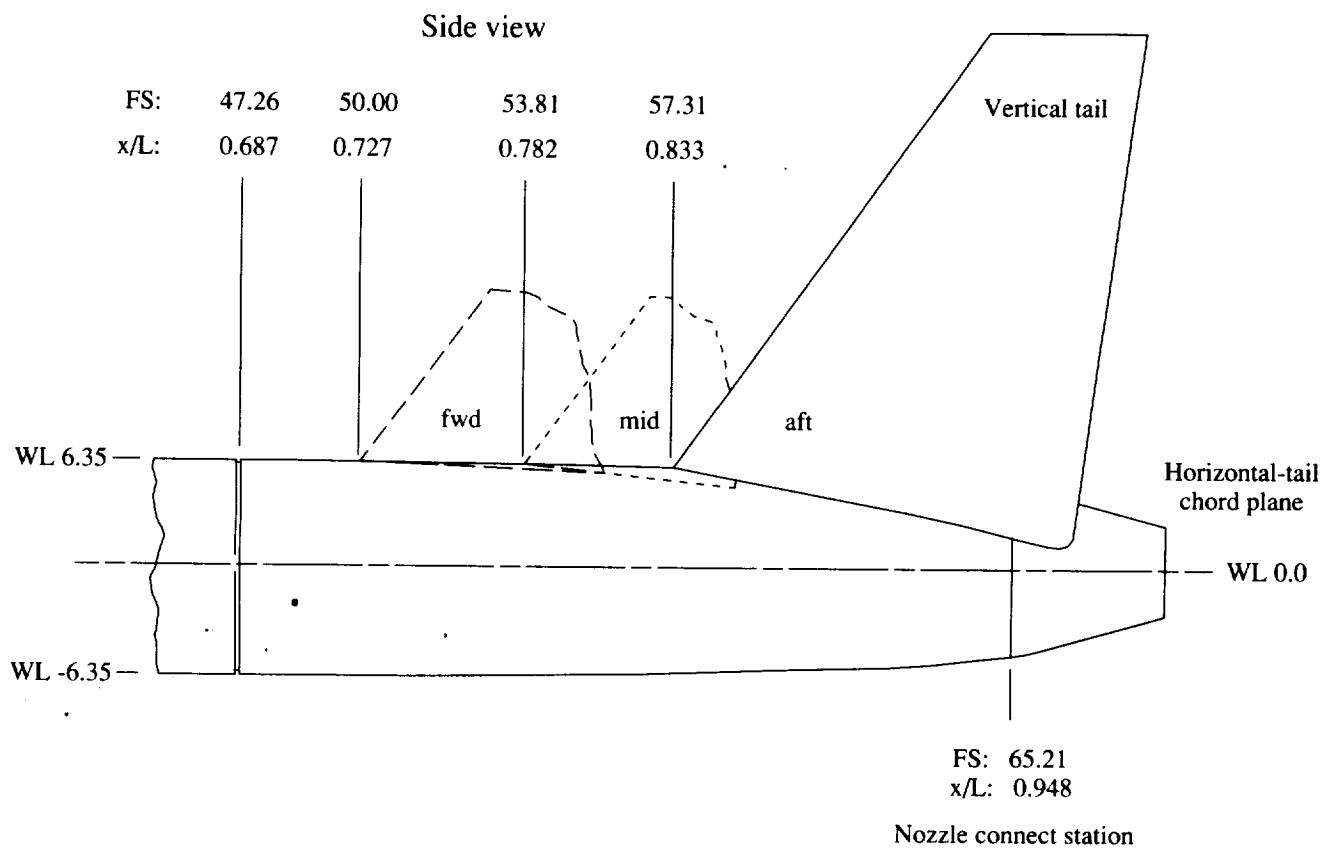
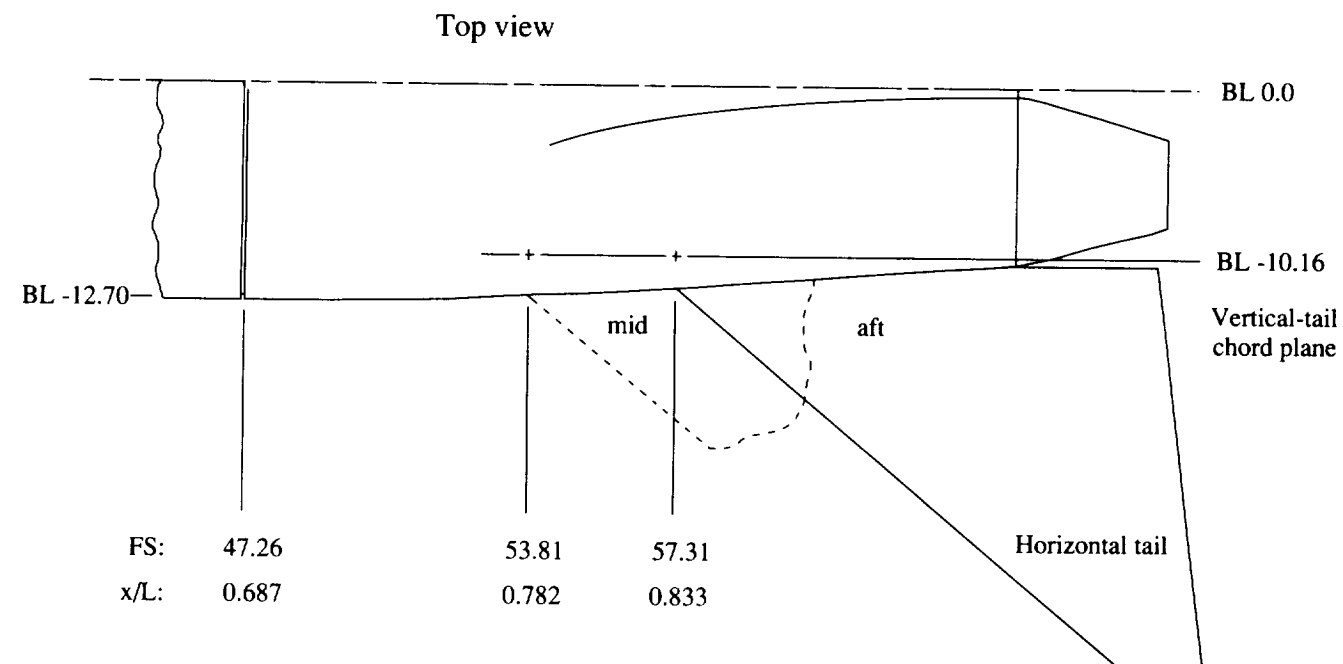
Figure 3. Continued.

Horizontal-Tail Geometry	
Airfoil section:	
Tip.....	NACA 64-002.5
Root.....	NACA 64-005.5
Tip chord, in.....	2.00
Root chord, in.....	11.40
Taper ratio.....	0.175
Span, in.:	
Mid on body.....	27.00
Aft on body.....	26.58
Planform area (one side exposed, filler excluded), ft ²	0.4004
Aspect ratio (exposed, filler excluded).....	2.564
Λ_{le} , deg.....	50.0



(e) Horizontal-tail geometry.

Figure 3. Continued.



(f) Sketch showing horizontal and vertical-tail positions on afterbody.

Figure 3. Concluded.

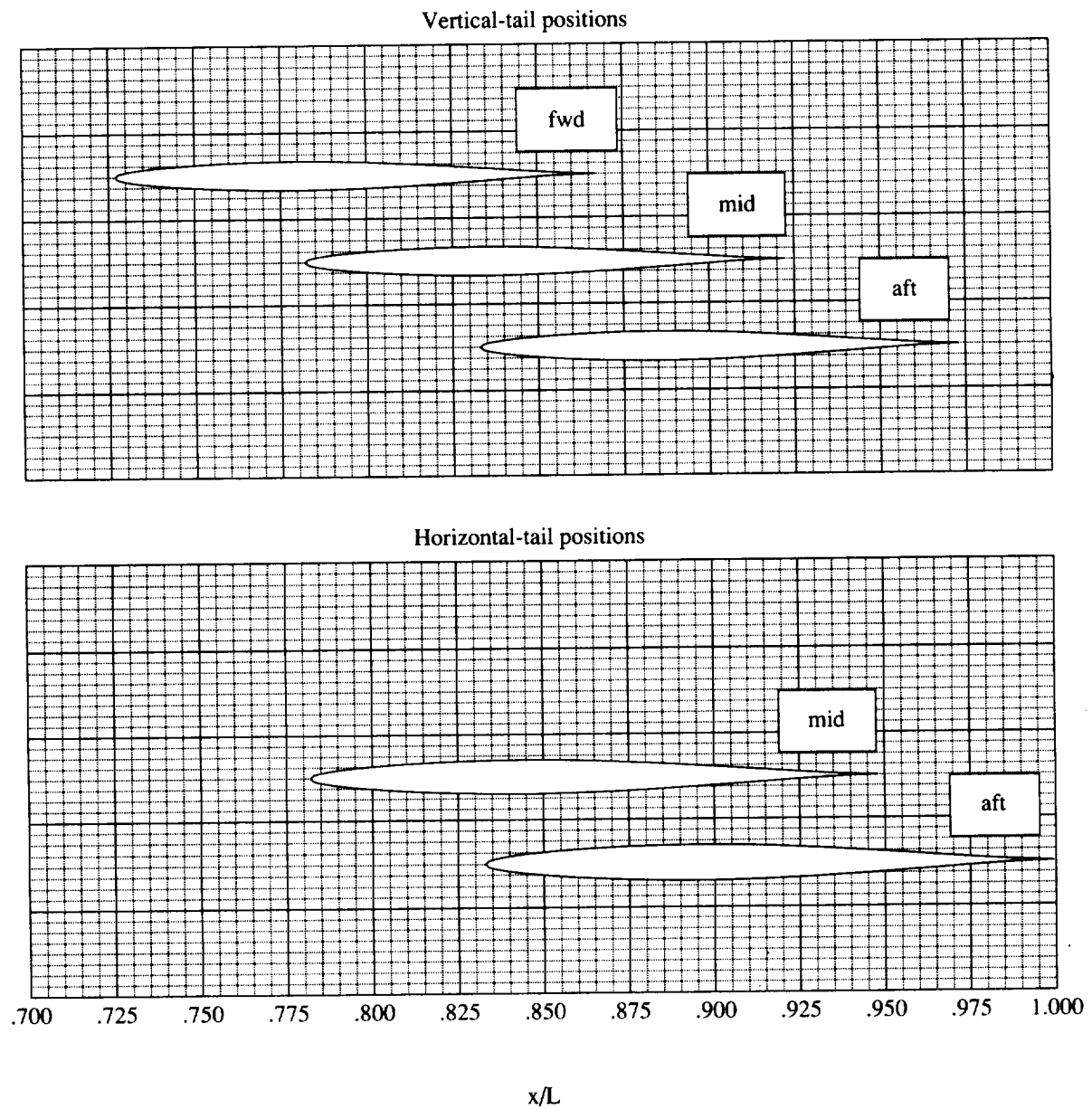


Figure 4. Positions of vertical and horizontal tails shown on typical plotting grid. Afterbody begins at $x/L = 0.687$.

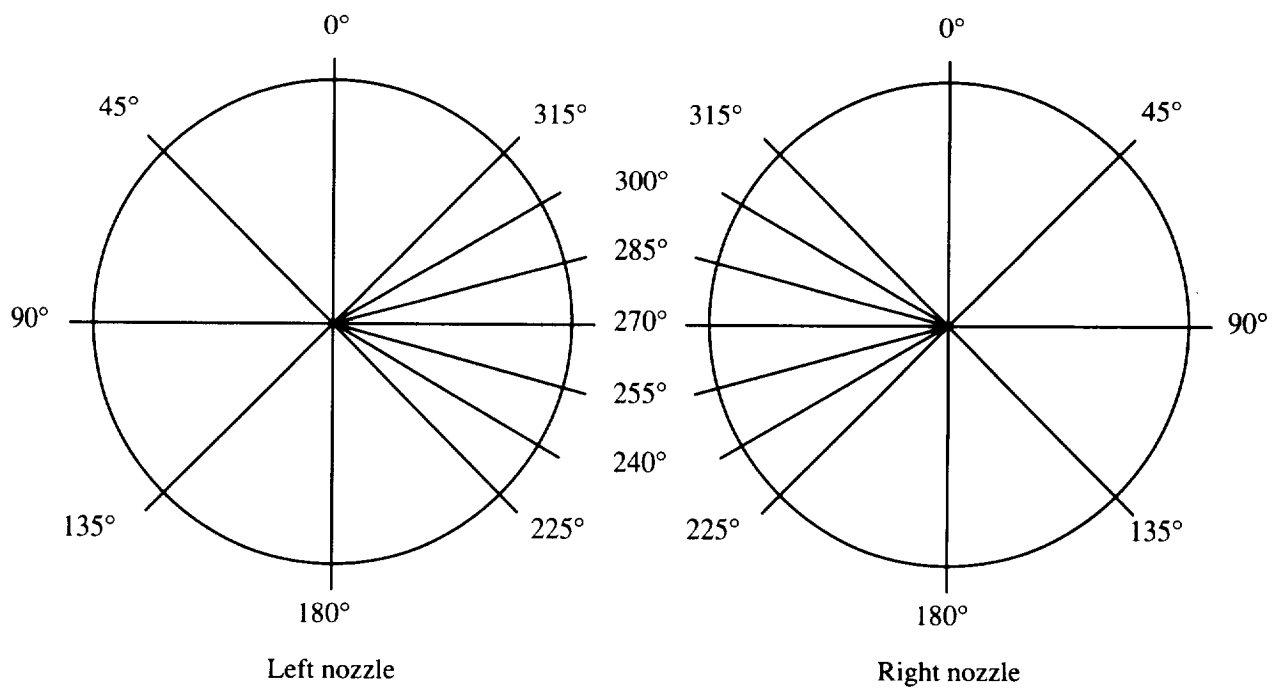


Figure 5. Circumferential positions of pressure orifice rows as defined for left and right nozzles. See table 1 for distribution of rows between nozzles as tested on model.

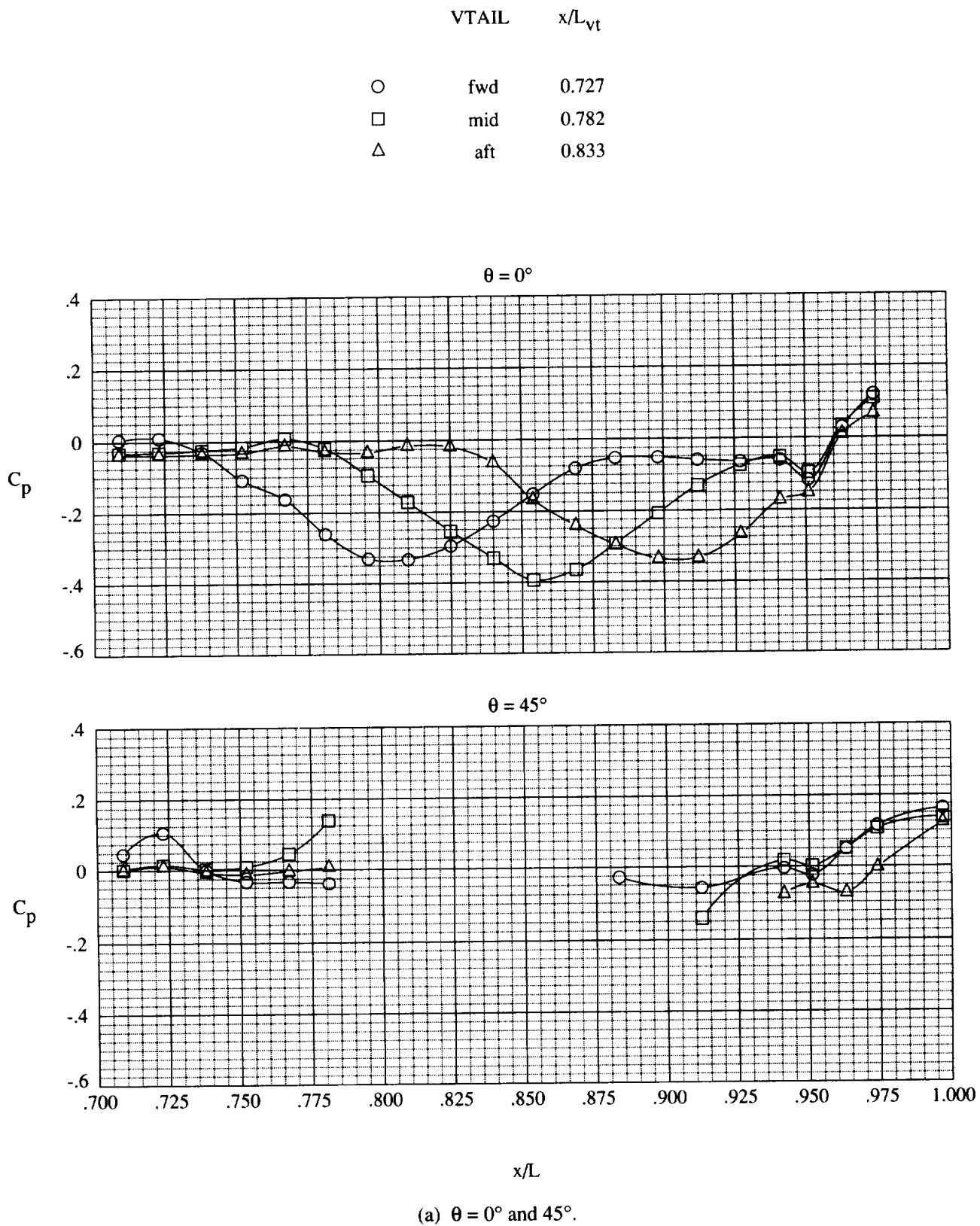
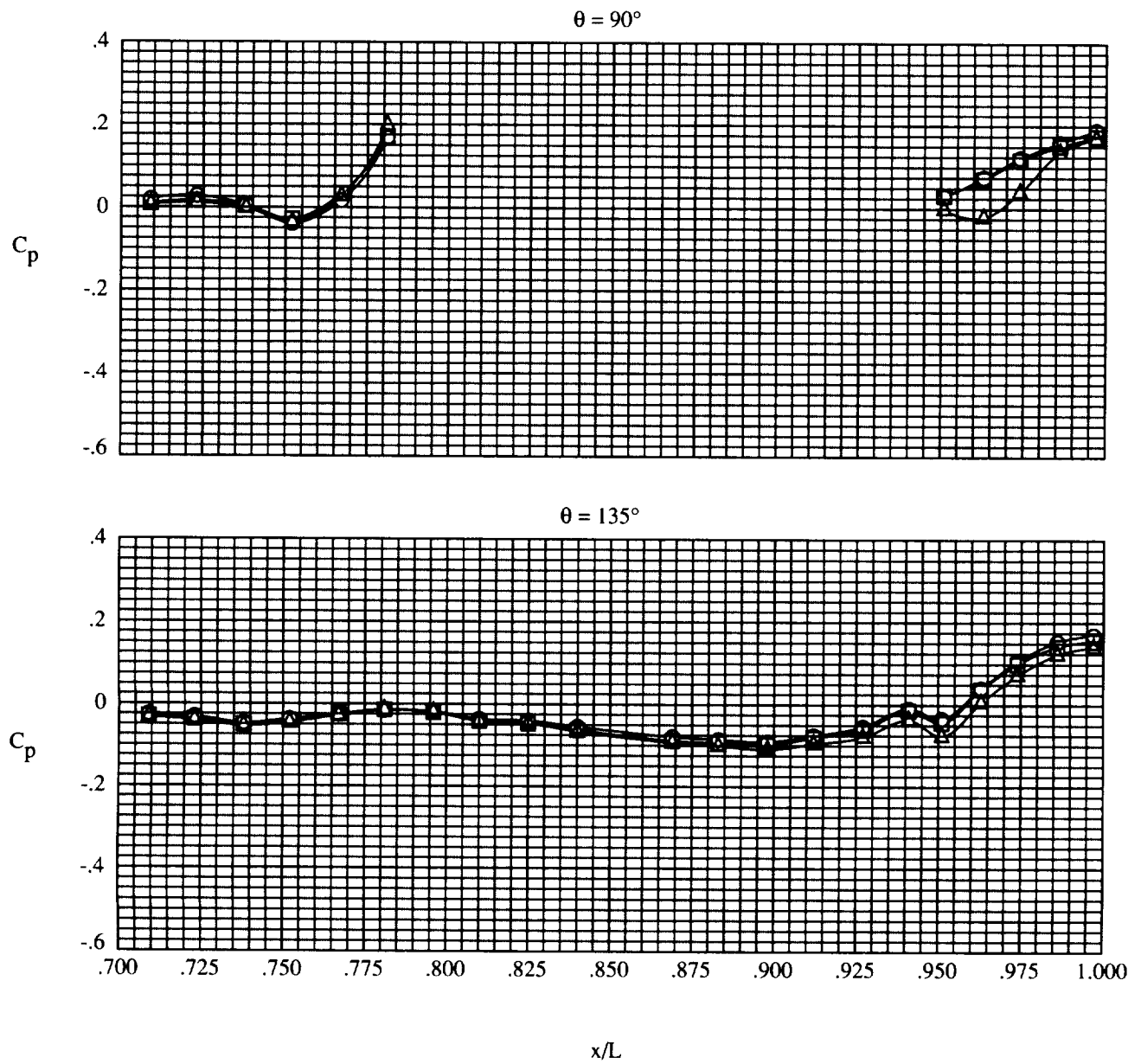


Figure 6. Effect of vertical-tail position on pressure distributions for $M = 0.8$, $\alpha = 0^\circ$, $NPR = 3.4$, and HTAIL = mid position.

	VTAIL	x/L_{vt}
○	fwd	0.727
□	mid	0.782
△	aft	0.833

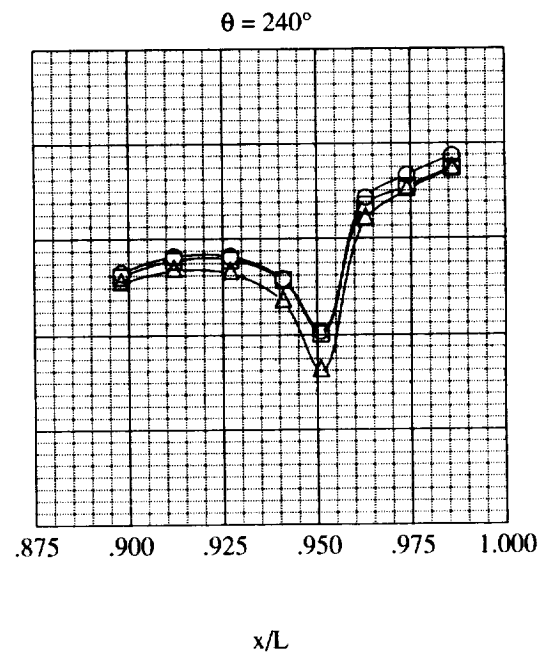
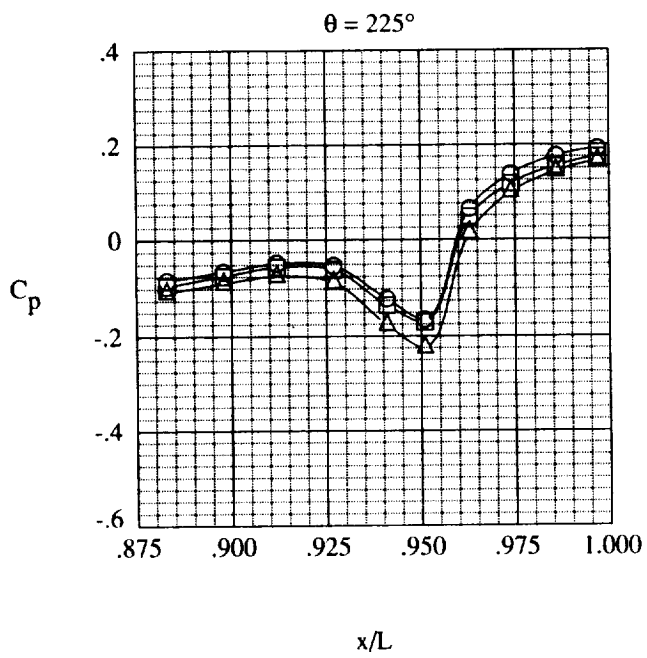
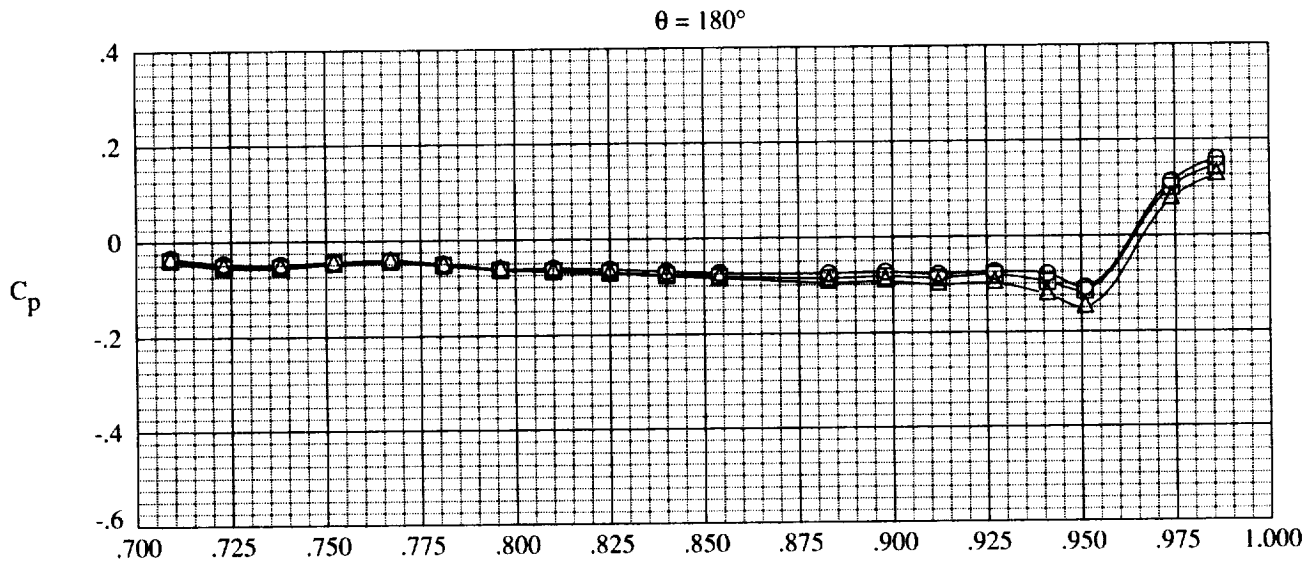


(b) $\theta = 90^\circ$ and 135° .

Figure 6. Continued.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 6. Continued.

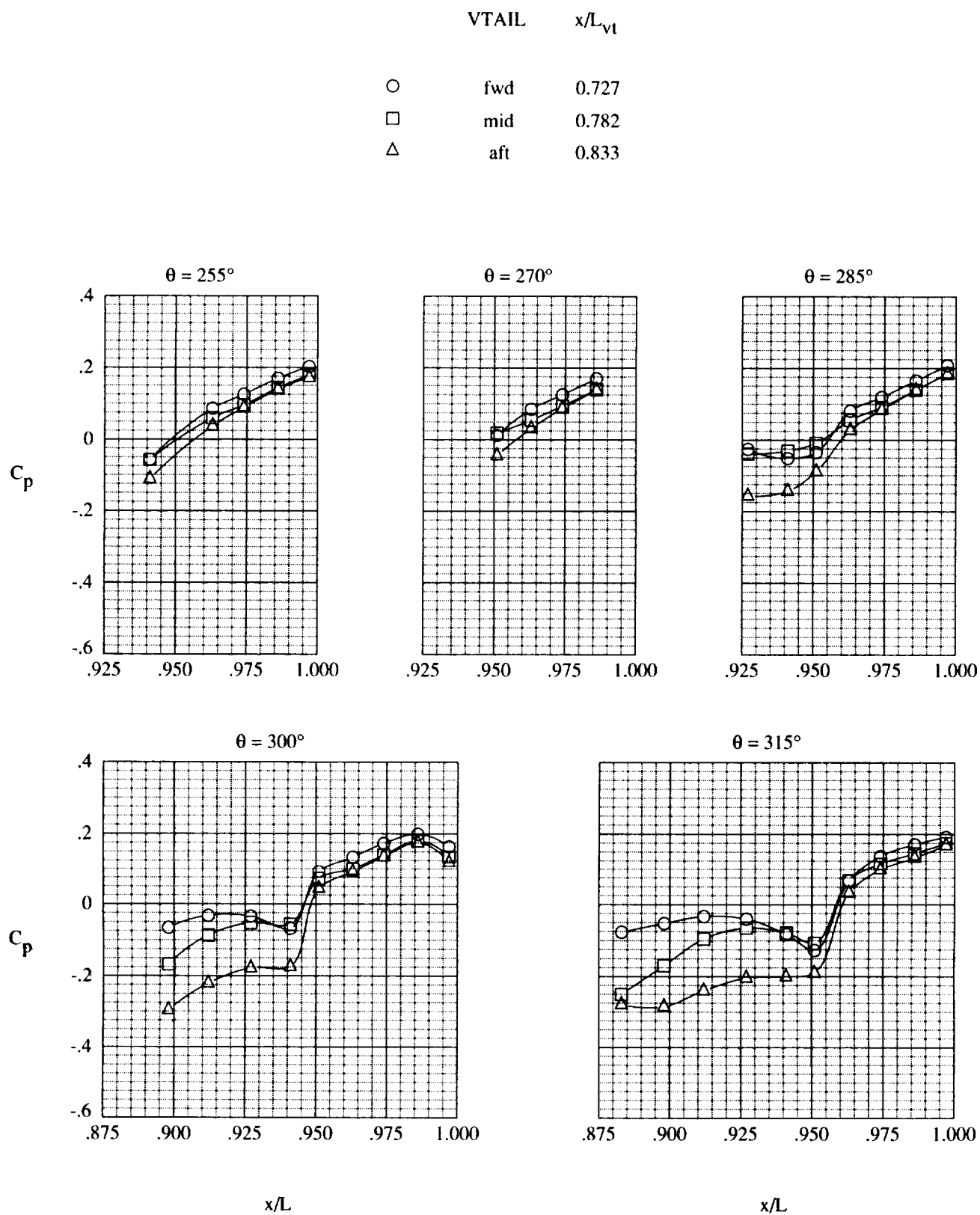
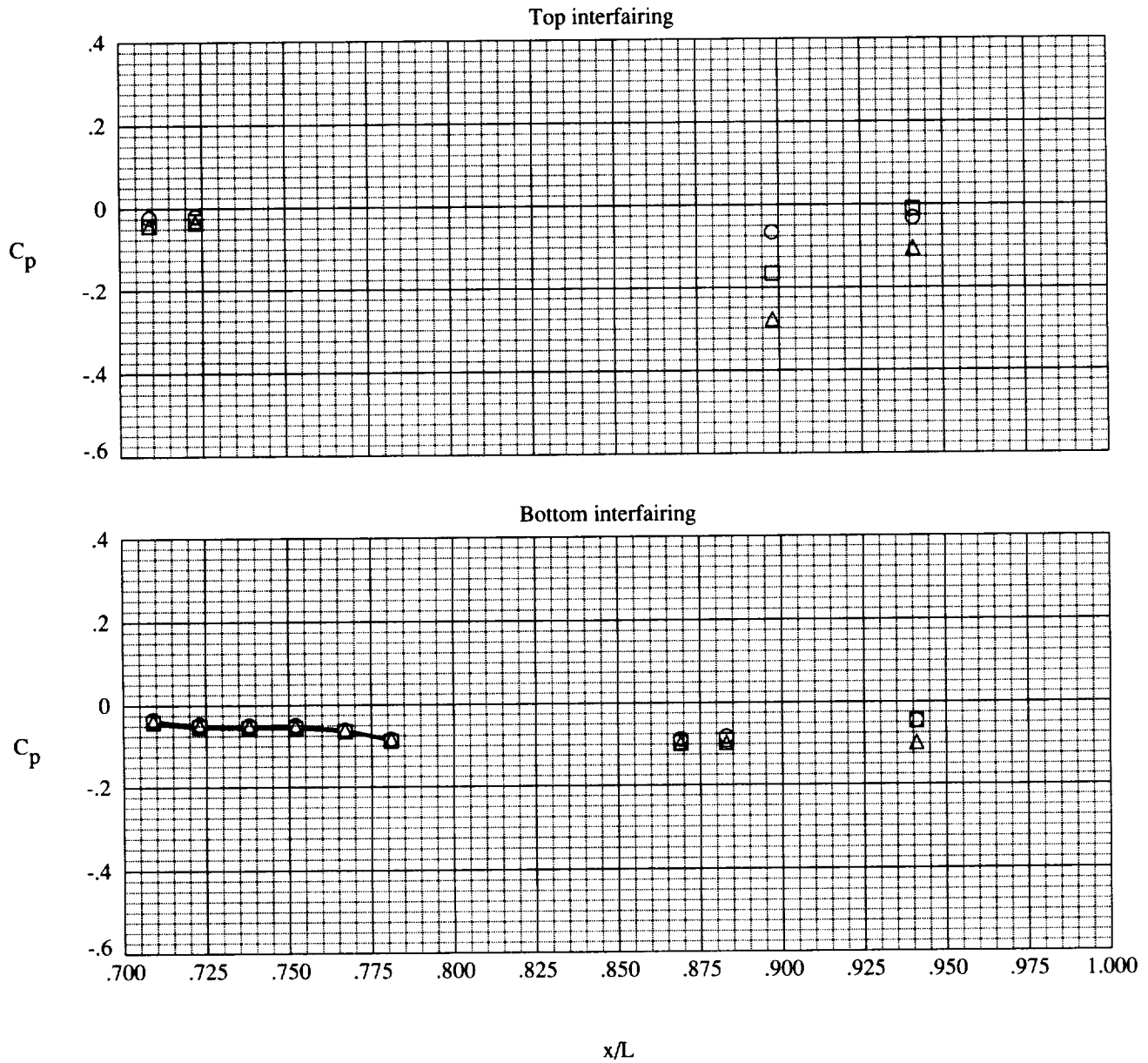


Figure 6. Continued.

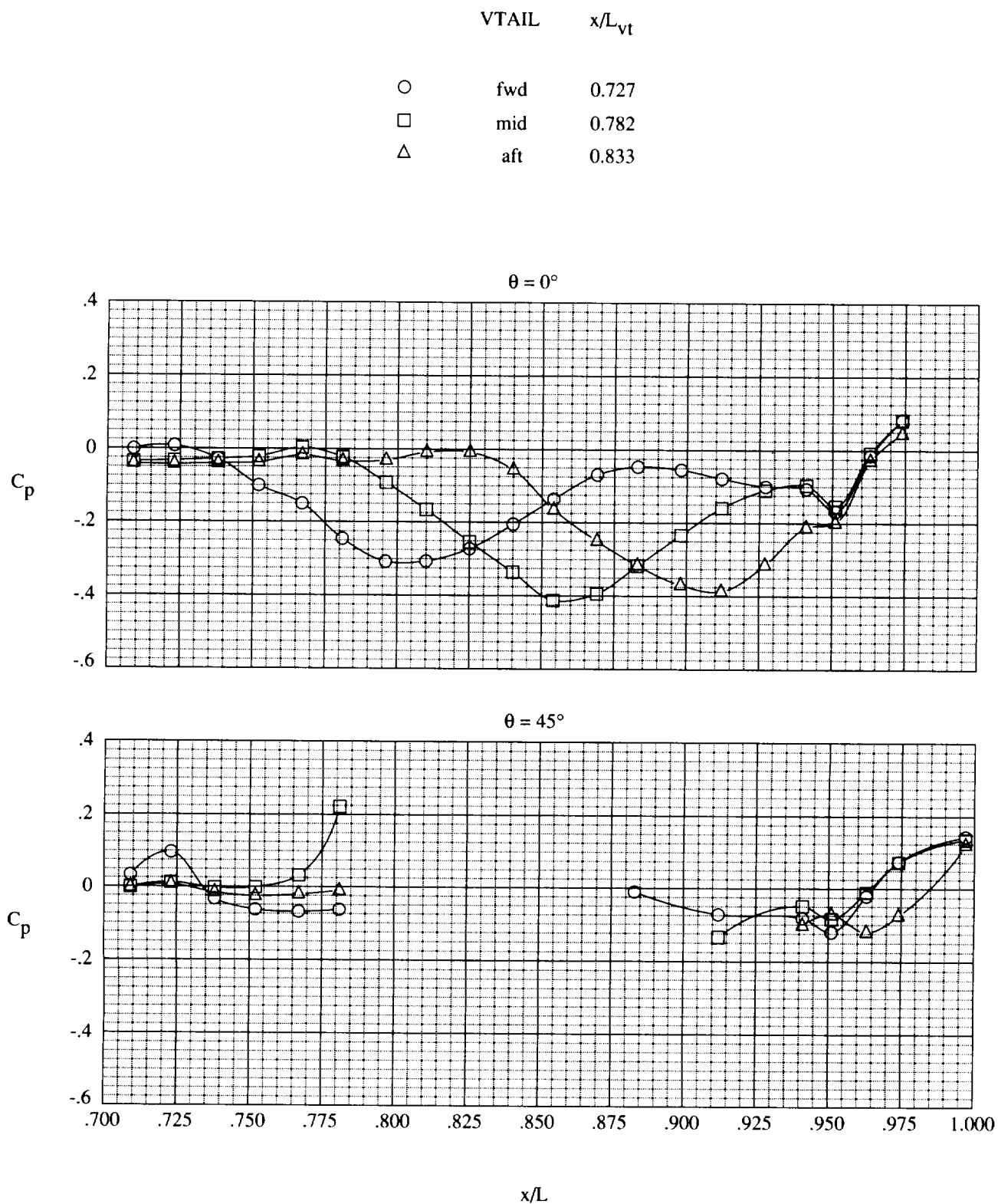
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



(e) Top and bottom interfairings.

Figure 6. Concluded.

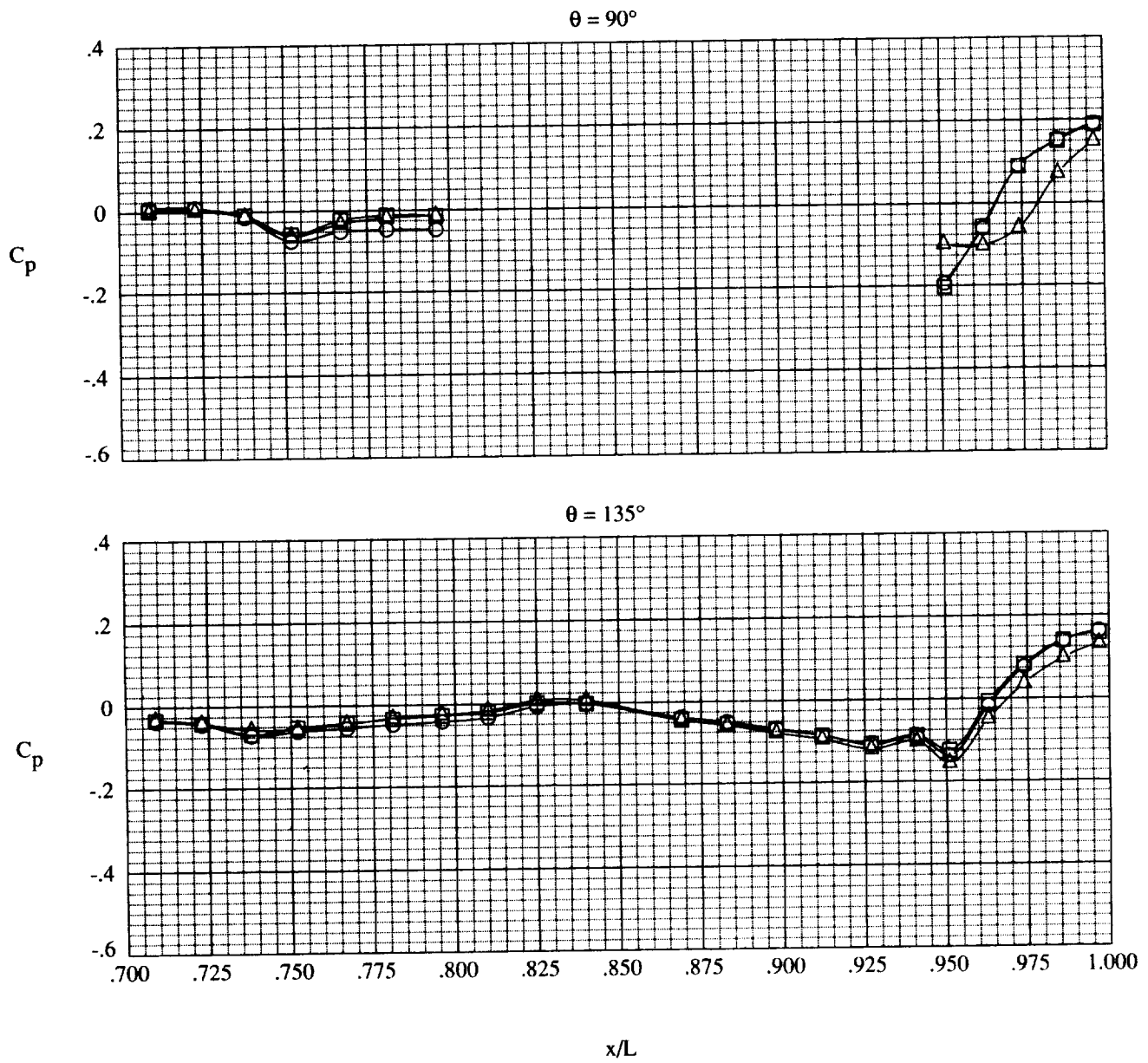


(a) $\theta = 0^\circ$ and 45° .

Figure 7. Effect of vertical-tail position on pressure distributions for $M = 0.8$, $\alpha = 0^\circ$, $NPR = 3.4$, and HTAIL = aft position.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

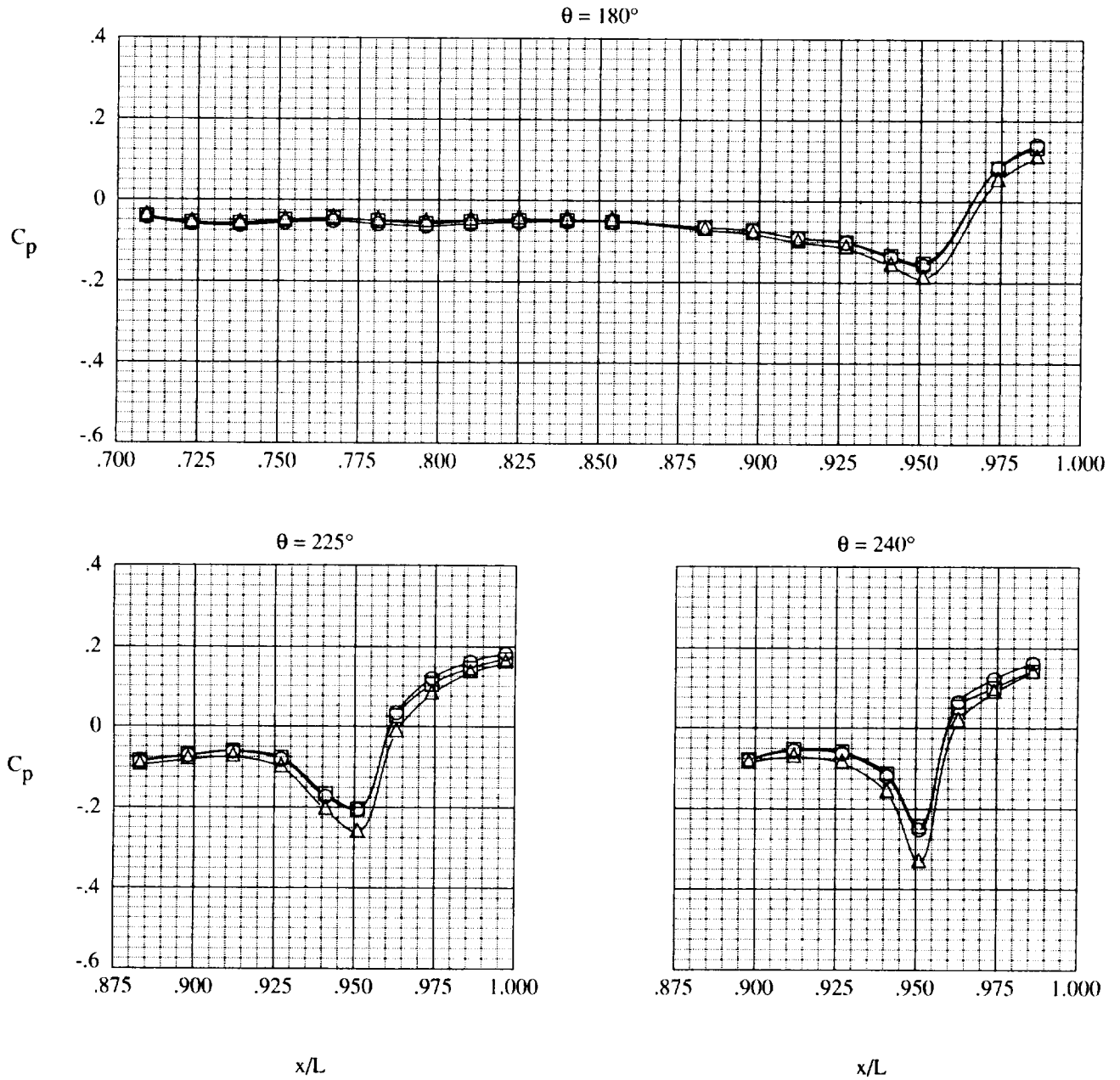


(b) $\theta = 90^\circ$ and 135° .

Figure 7. Continued.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

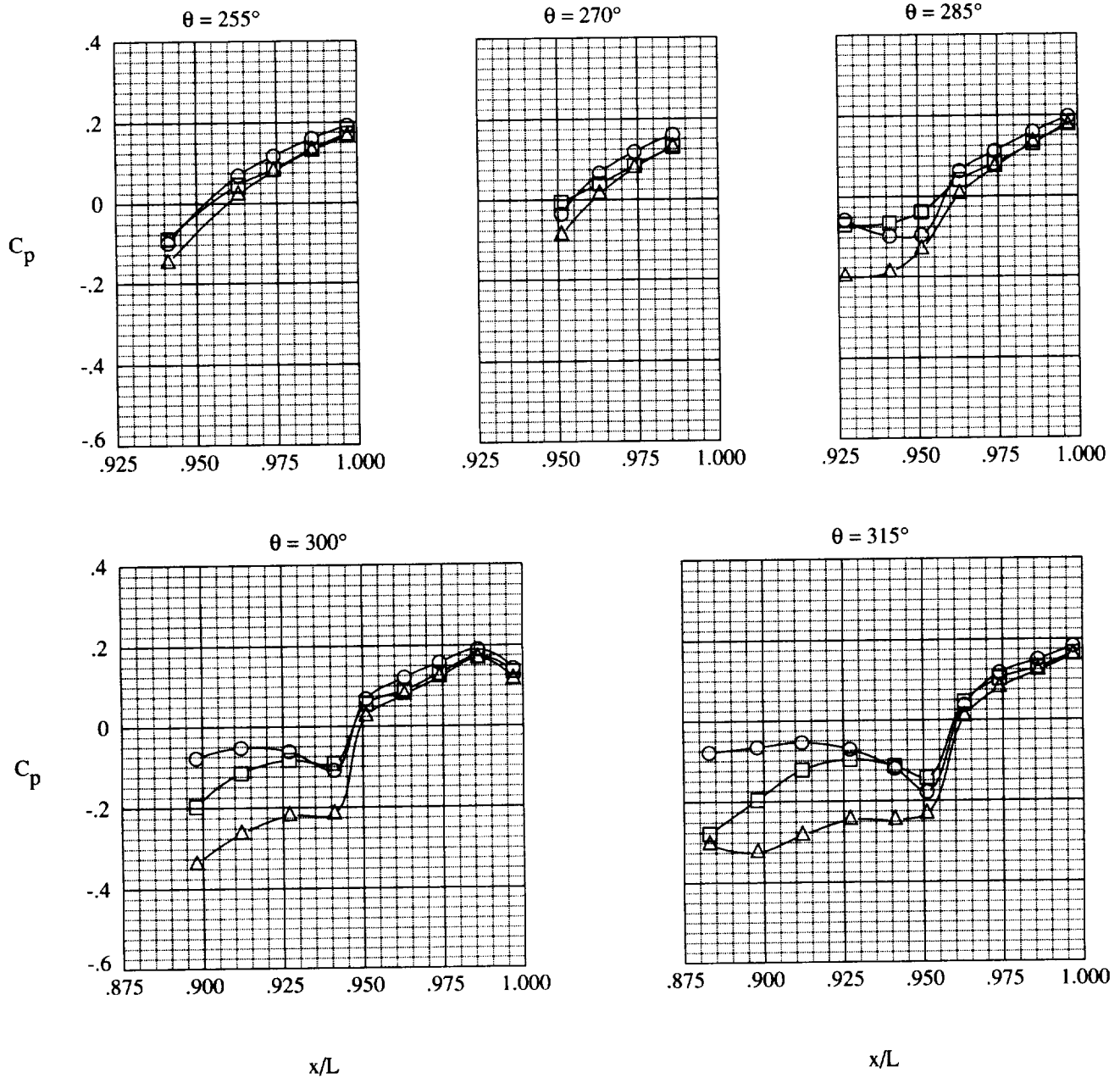


(c) $\theta = 180^\circ$, 225° , and 240° .

Figure 7. Continued.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

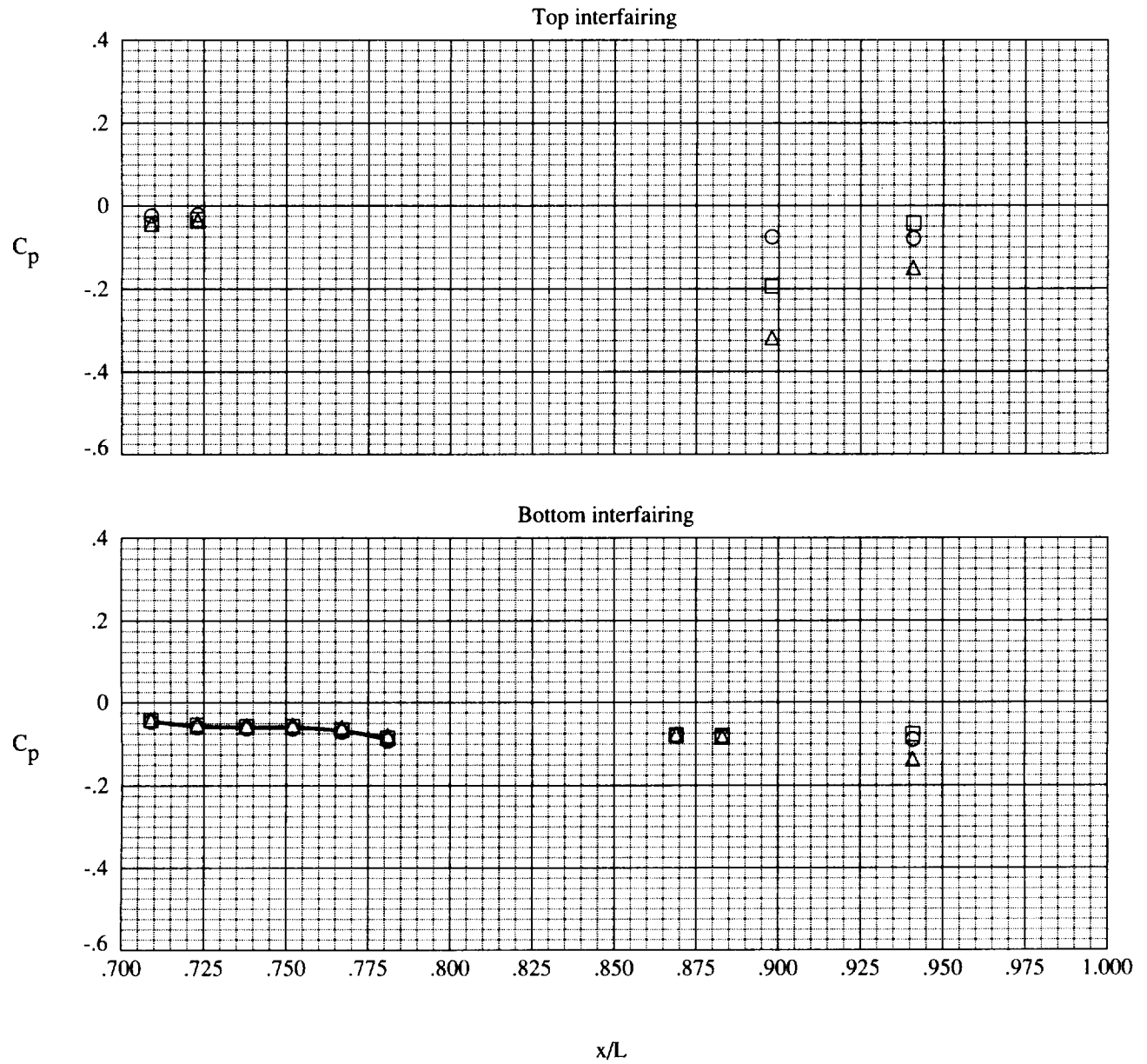


(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 7. Continued.

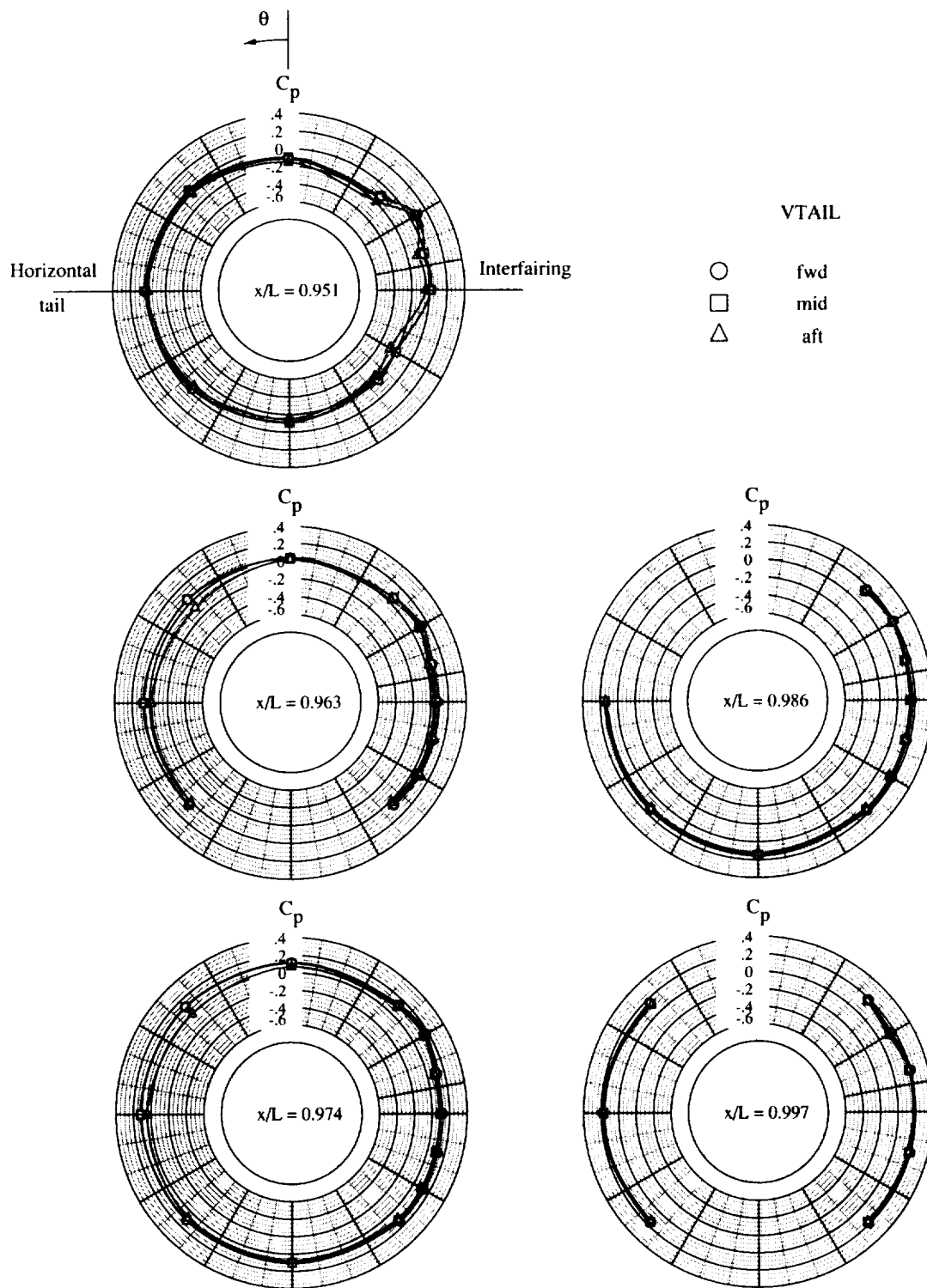
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



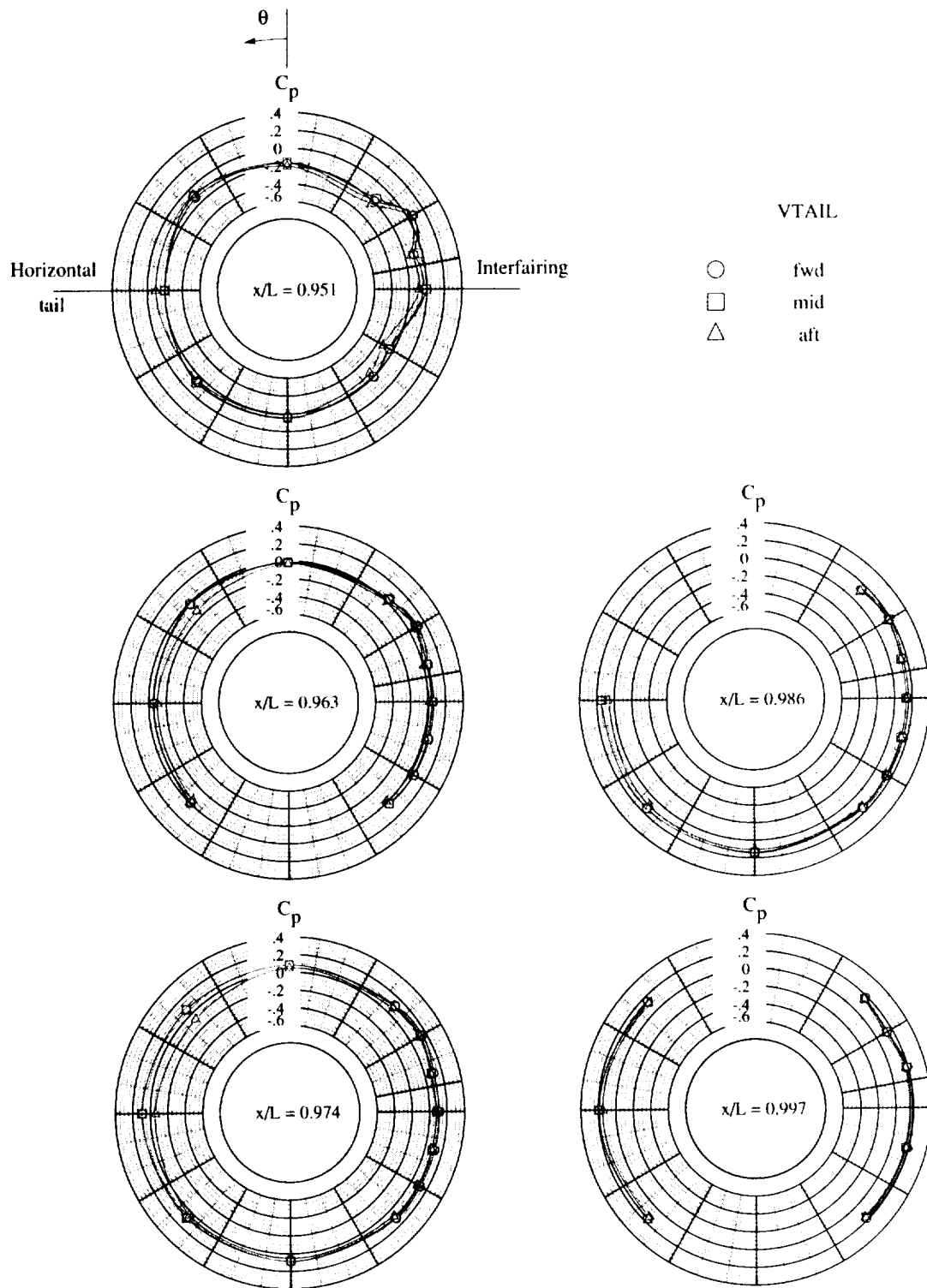
(e) Top and bottom interfairings.

Figure 7. Concluded.



(a) HTAIL = mid position.

Figure 8. Effect of vertical-tail position on static pressure distribution around left nozzle for $M = 0.8$, $\alpha = 0^\circ$, and $NPR = 3.4$.

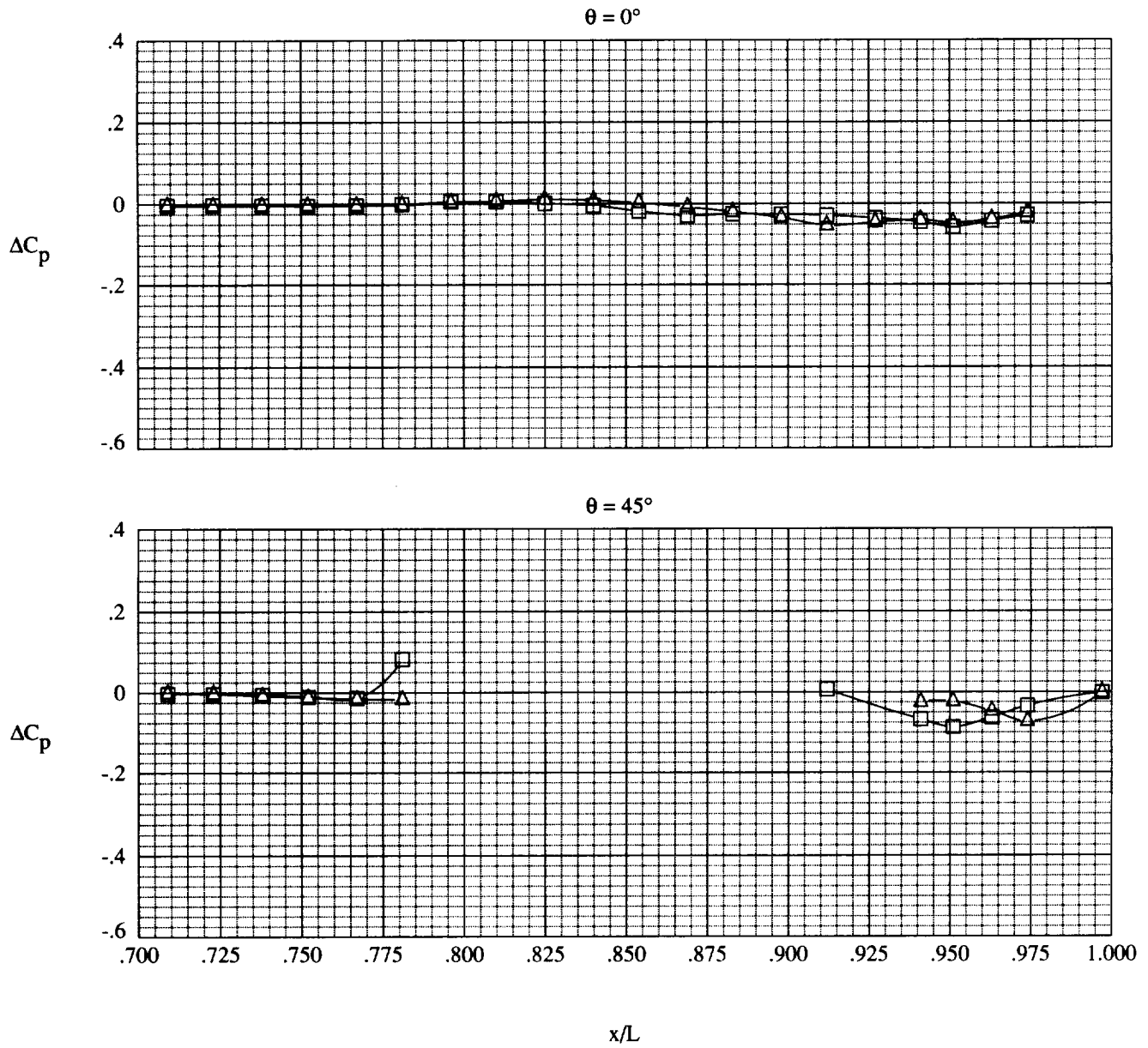


(b) HTAIL = aft position.

Figure 8. Concluded.

VTAIL x/L_{vt}

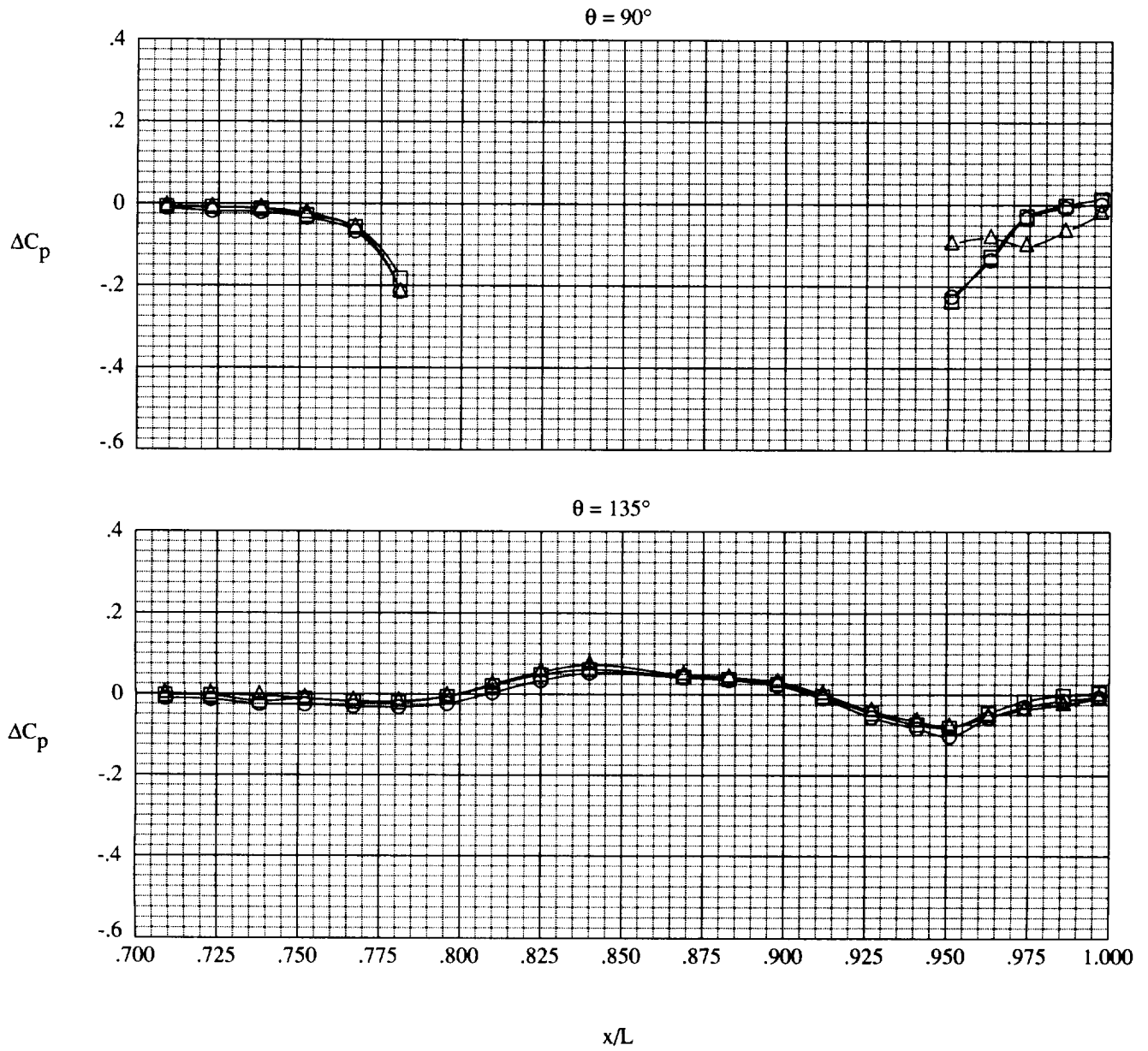
□	mid	0.782
△	aft	0.833



(a) $\theta = 0^\circ$ and 45° .

Figure 9. Change in static pressure coefficients from moving horizontal tails from mid to aft position for $M = 0.8$, $\alpha = 0^\circ$, and $NPR = 3.4$.

	VTAIL	x/L_{vt}
○	fwd	0.727
□	mid	0.782
△	aft	0.833

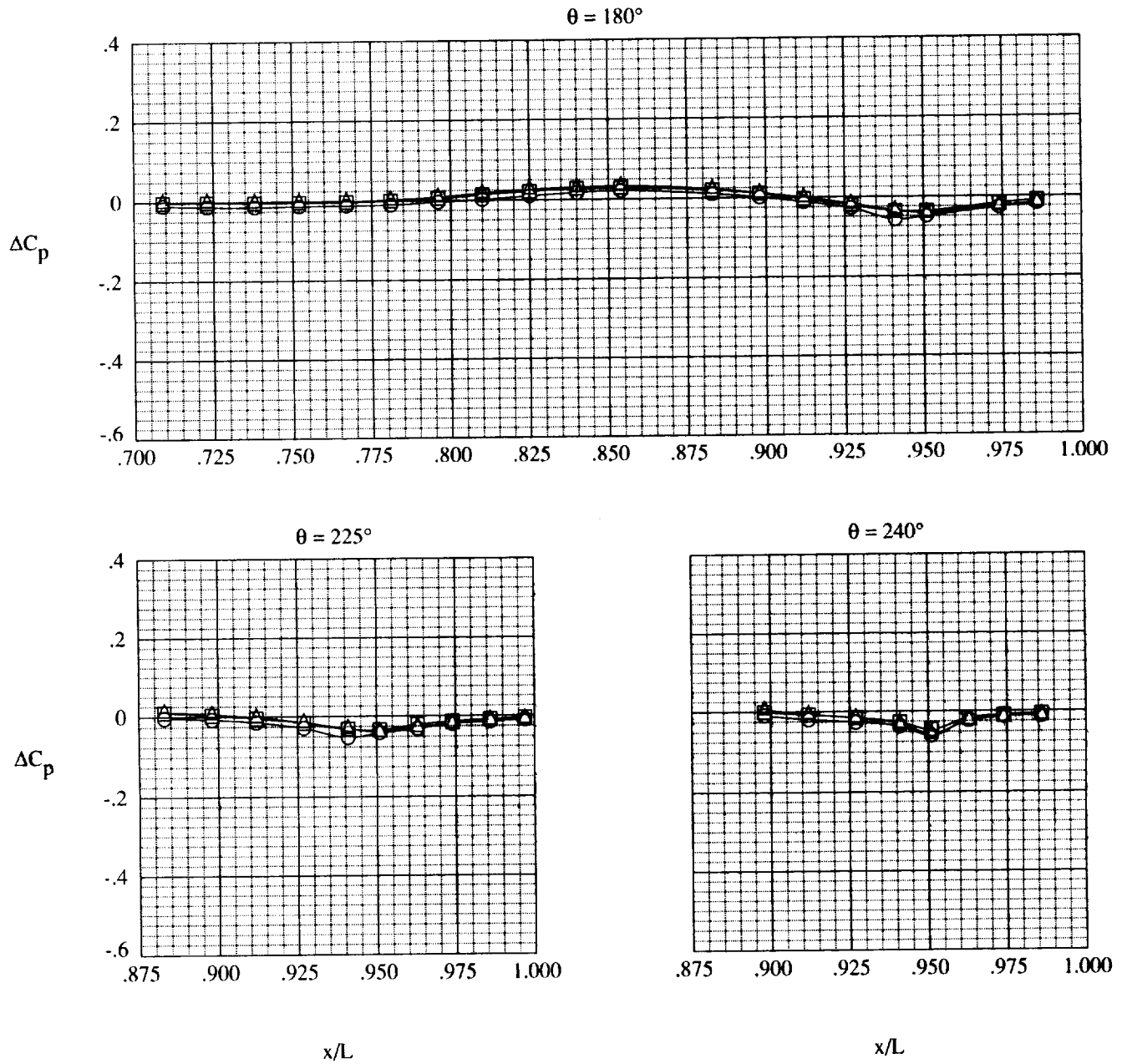


(b) $\theta = 90^\circ$ and 135° .

Figure 9. Continued.

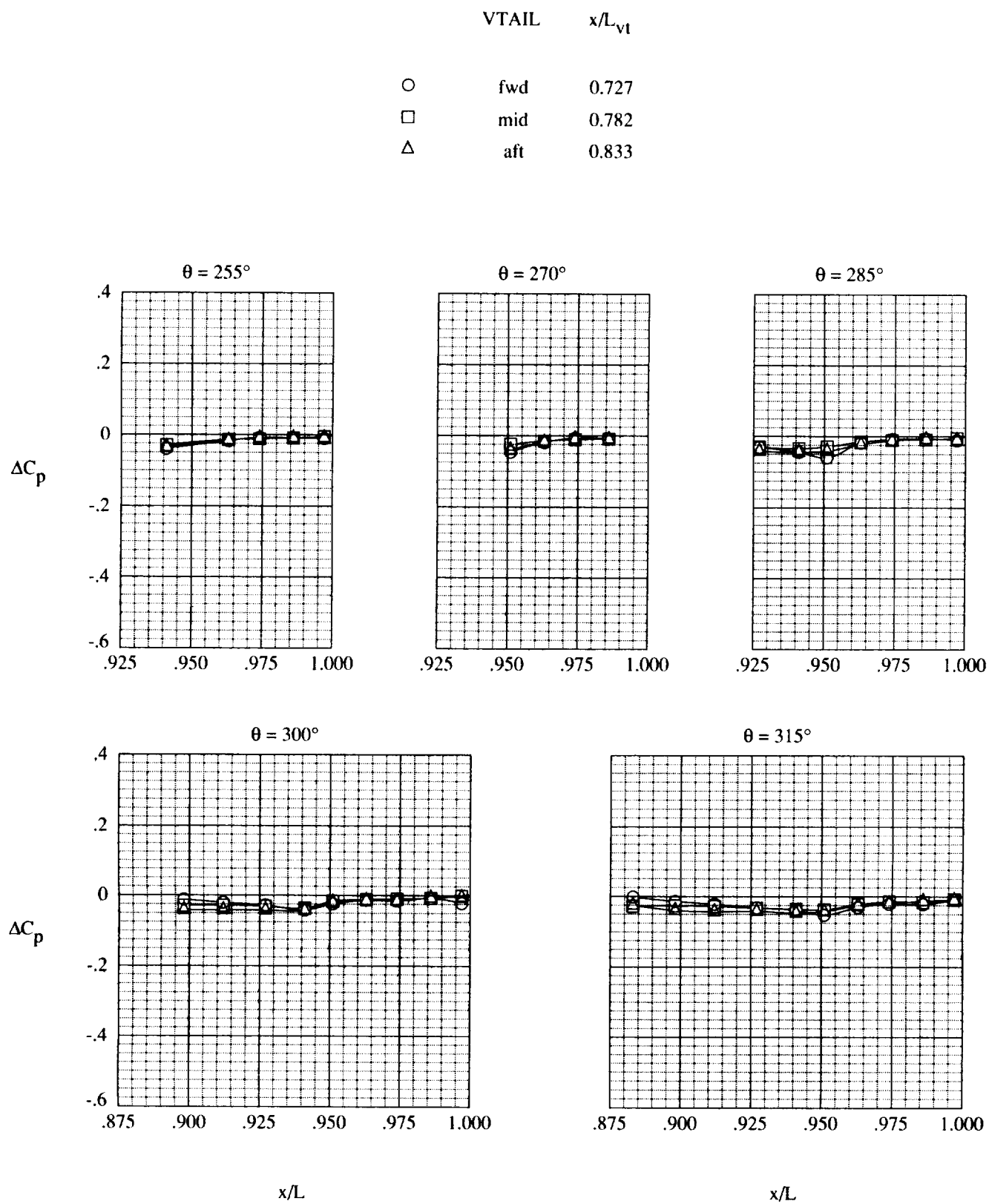
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



(c) $\theta = 180^\circ$, 225° , and 240° .

Figure 9. Continued.

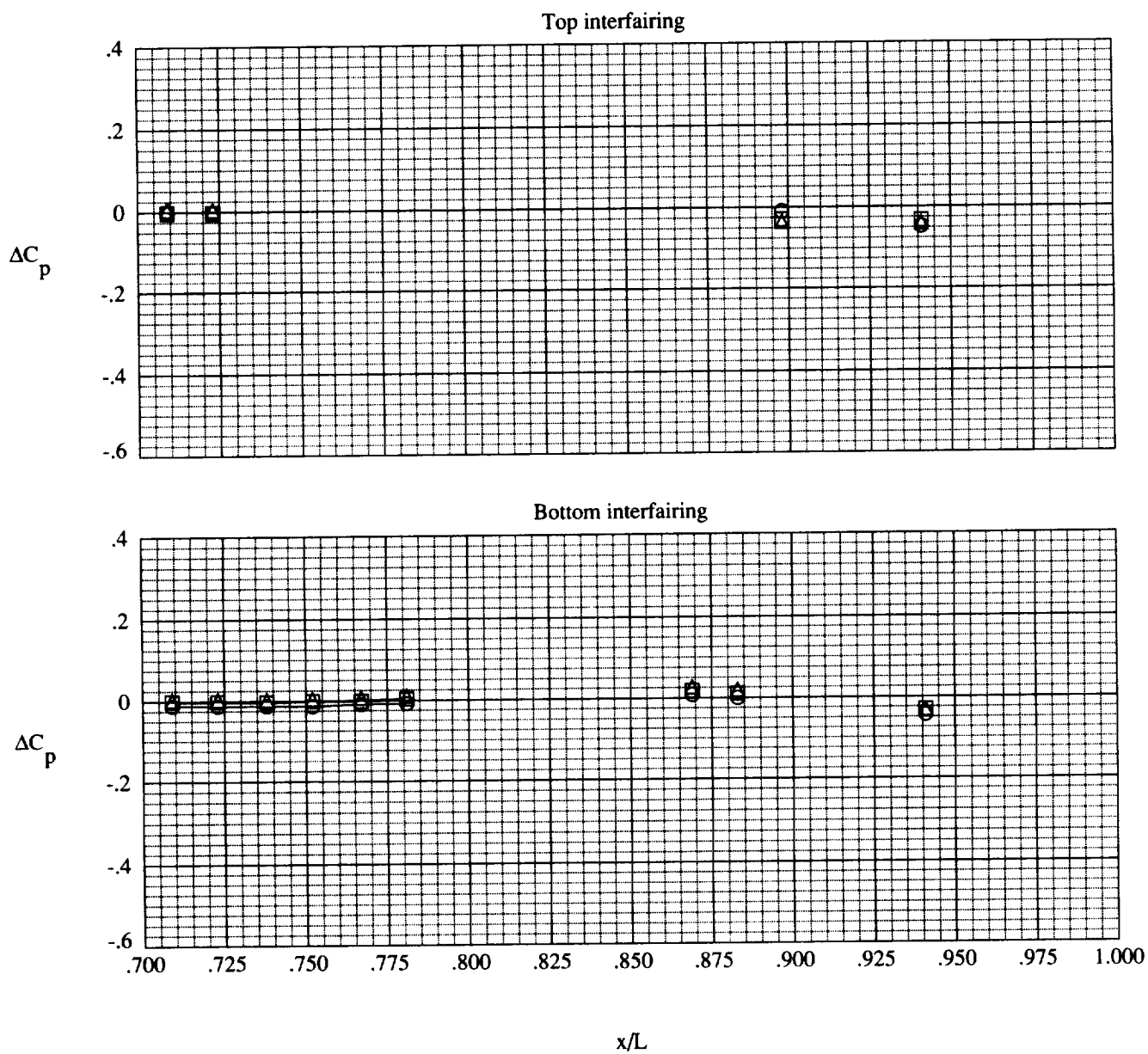


(d) $\theta = 255^\circ$, 270° , 285° , 300° , and 315° .

Figure 9. Continued.

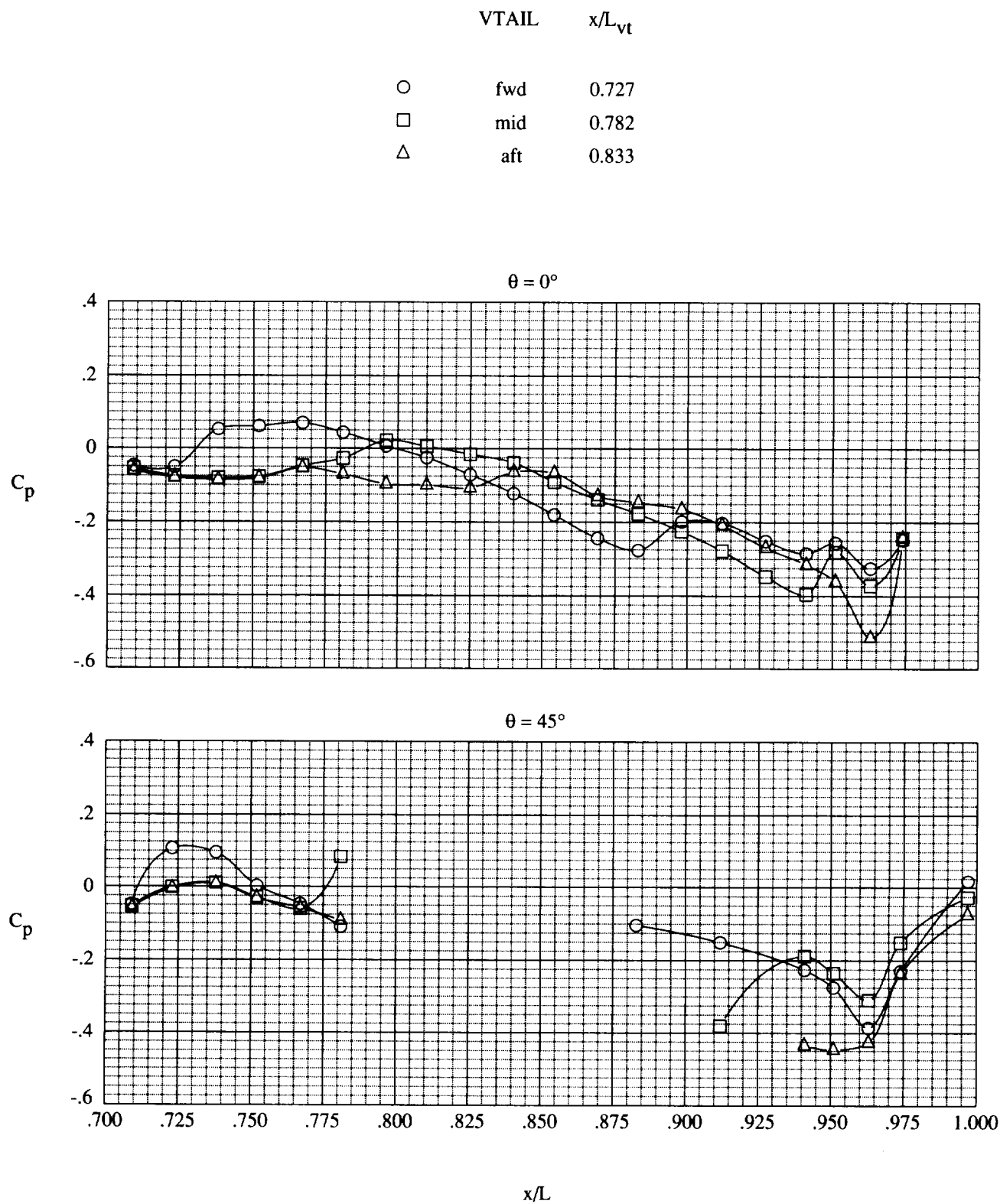
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



(e) Top and bottom interfairings.

Figure 9. Concluded.

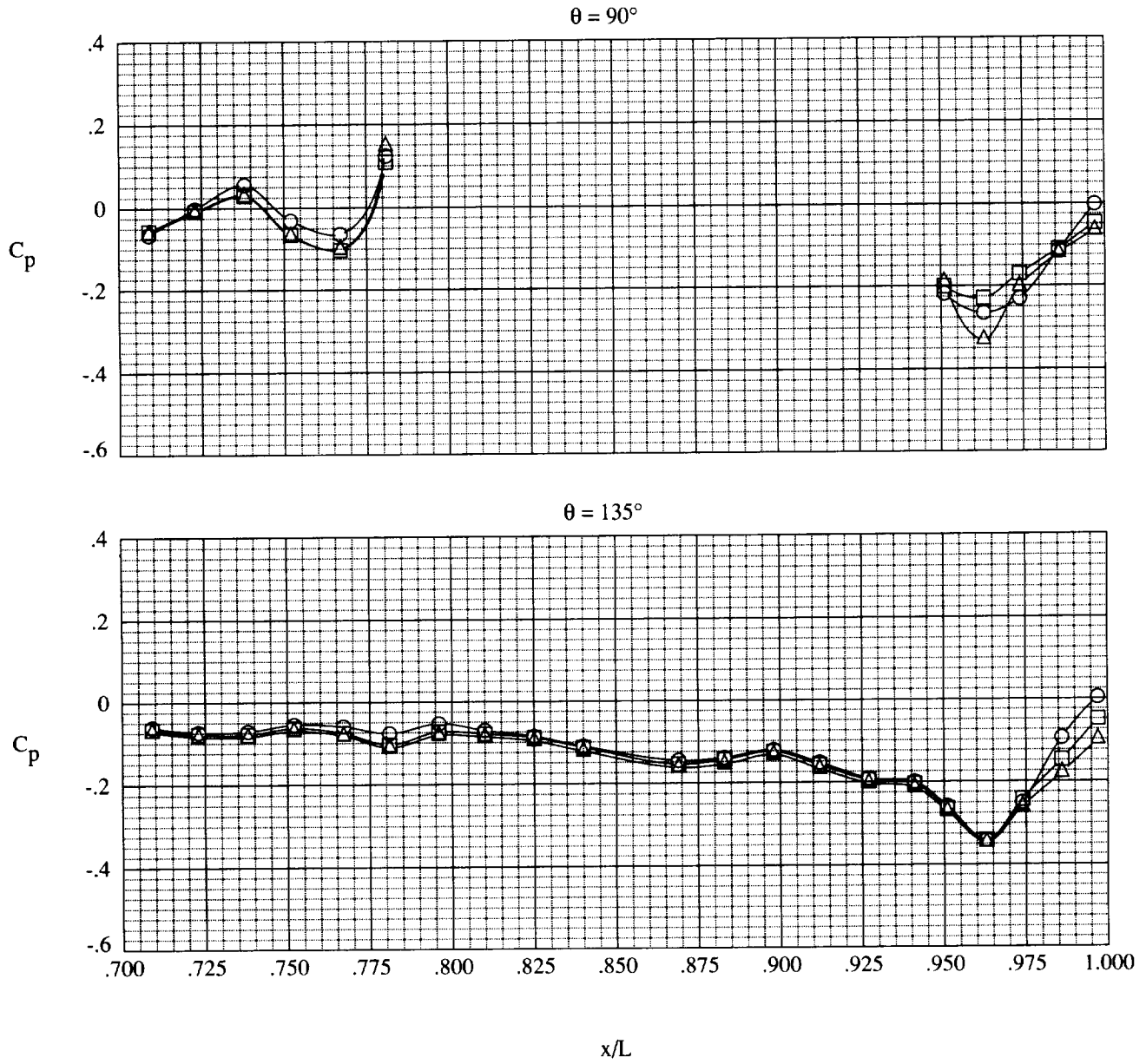


(a) $\theta = 0^\circ$ and 45° .

Figure 10. Effect of vertical-tail position on pressure distributions for $M = 1.2$, $\alpha = 0^\circ$, $NPR = 3.4$, and HTAIL = mid position.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

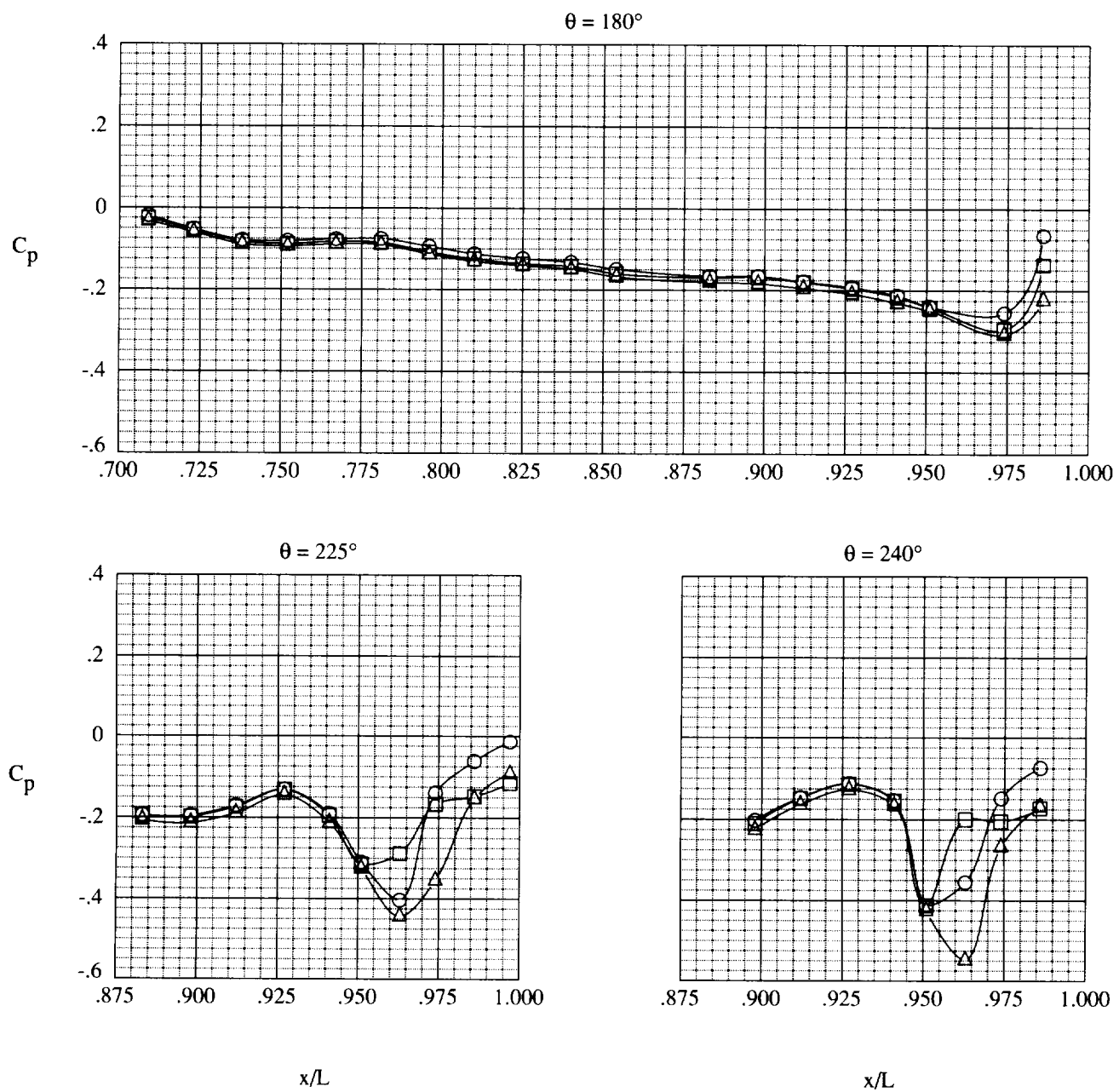


(b) $\theta = 90^\circ$ and 135° .

Figure 10. Continued.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

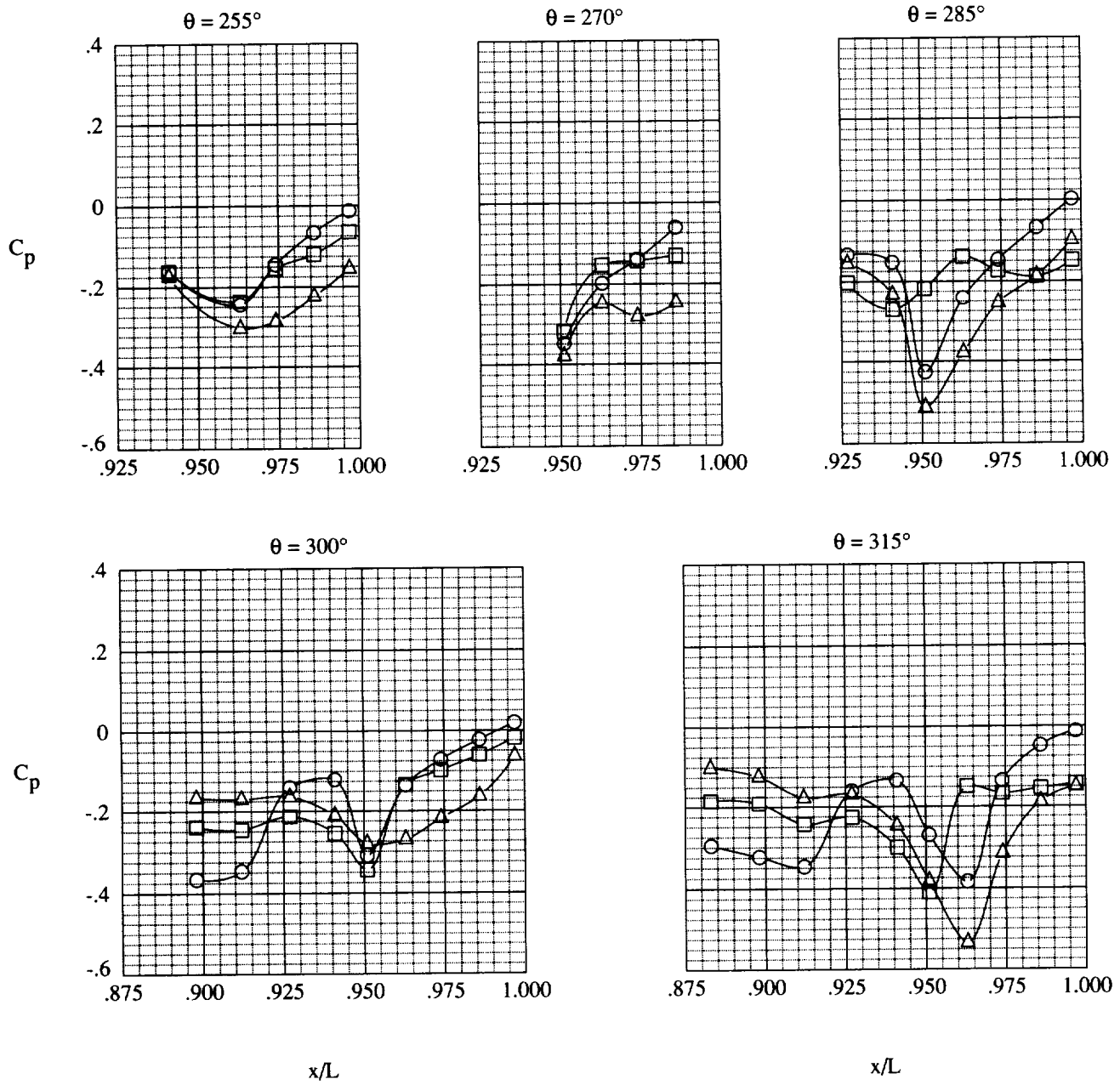


(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 10. Continued.

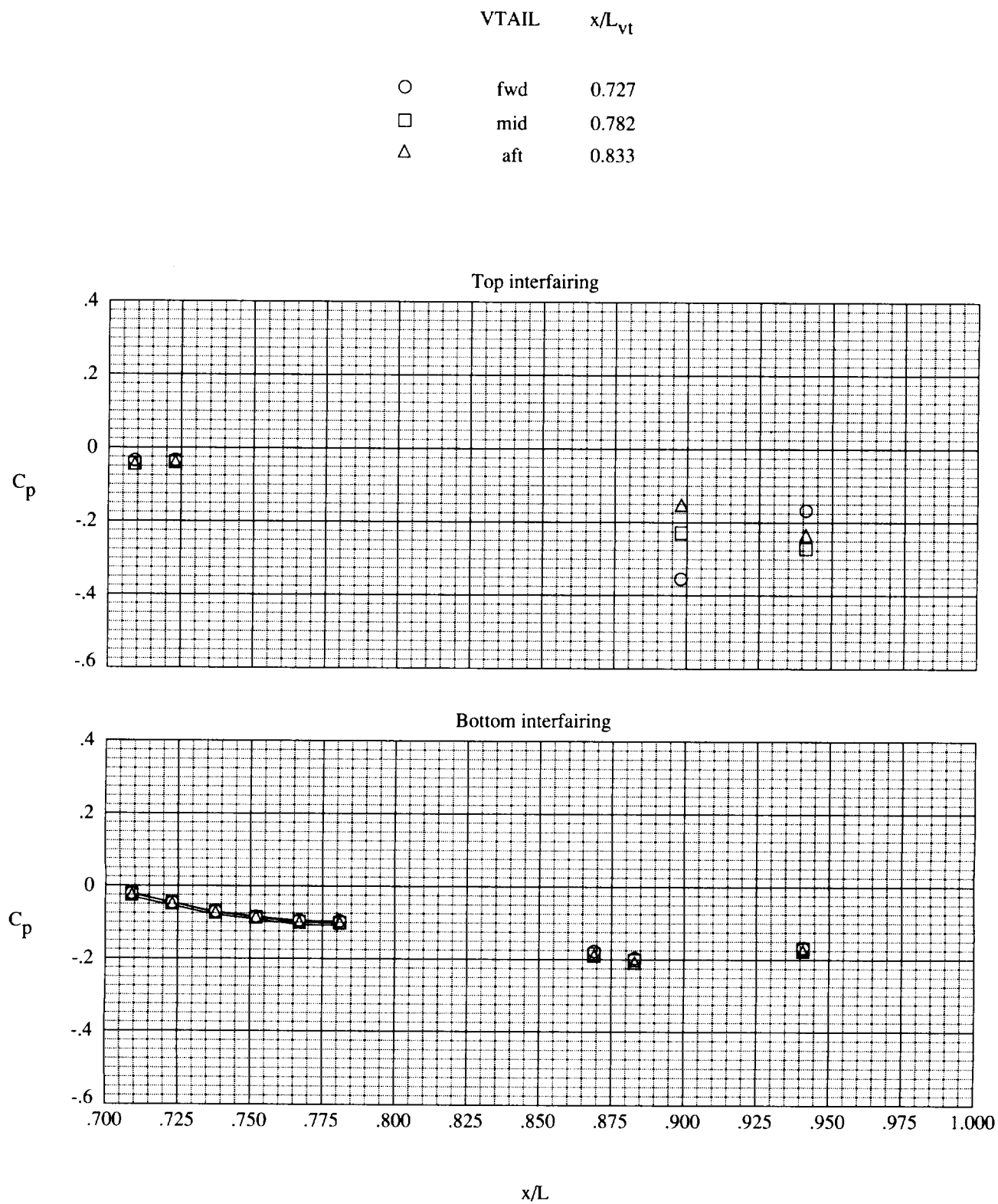
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



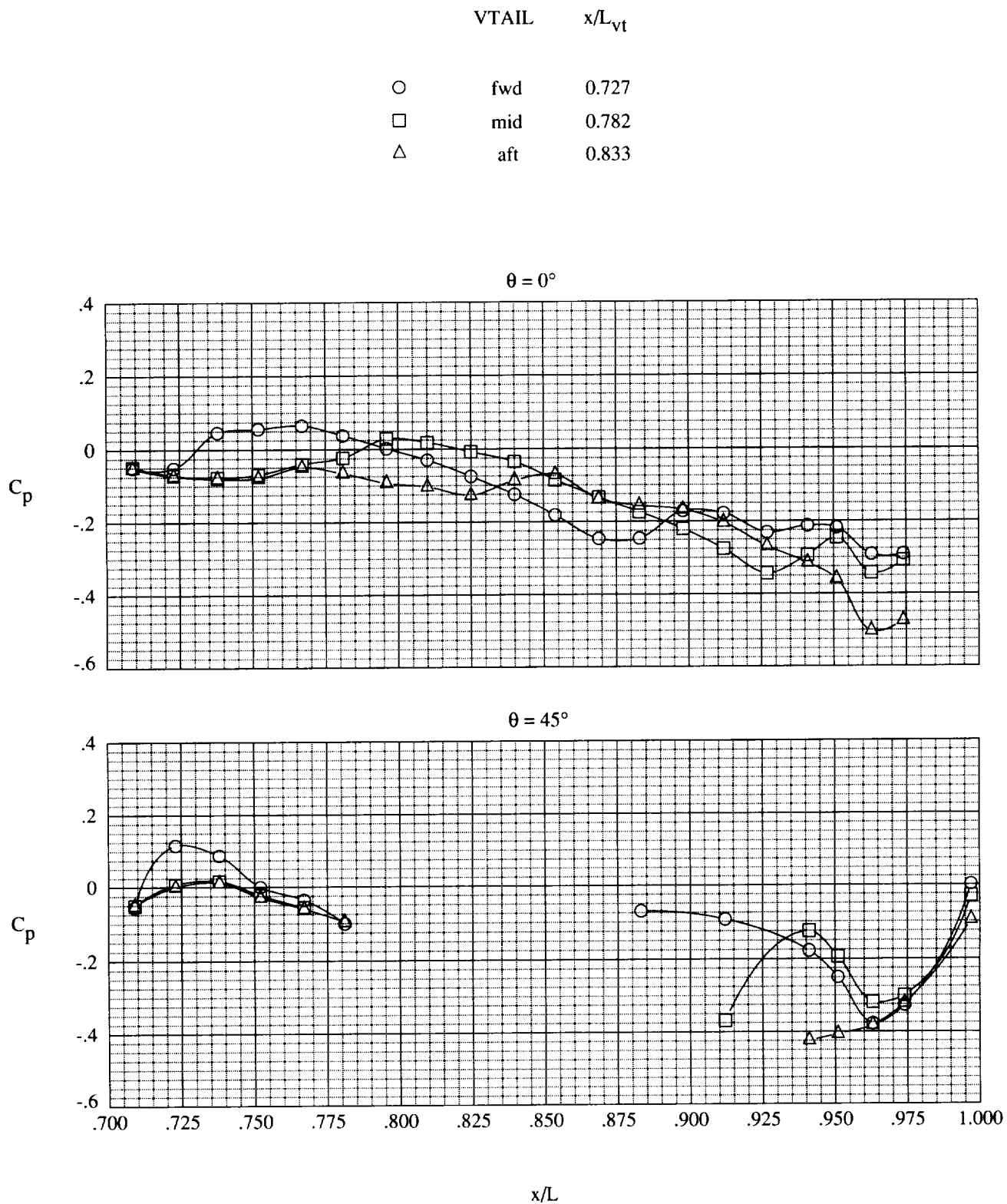
(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 10. Continued.



(e) Top and bottom interfairings.

Figure 10. Concluded.

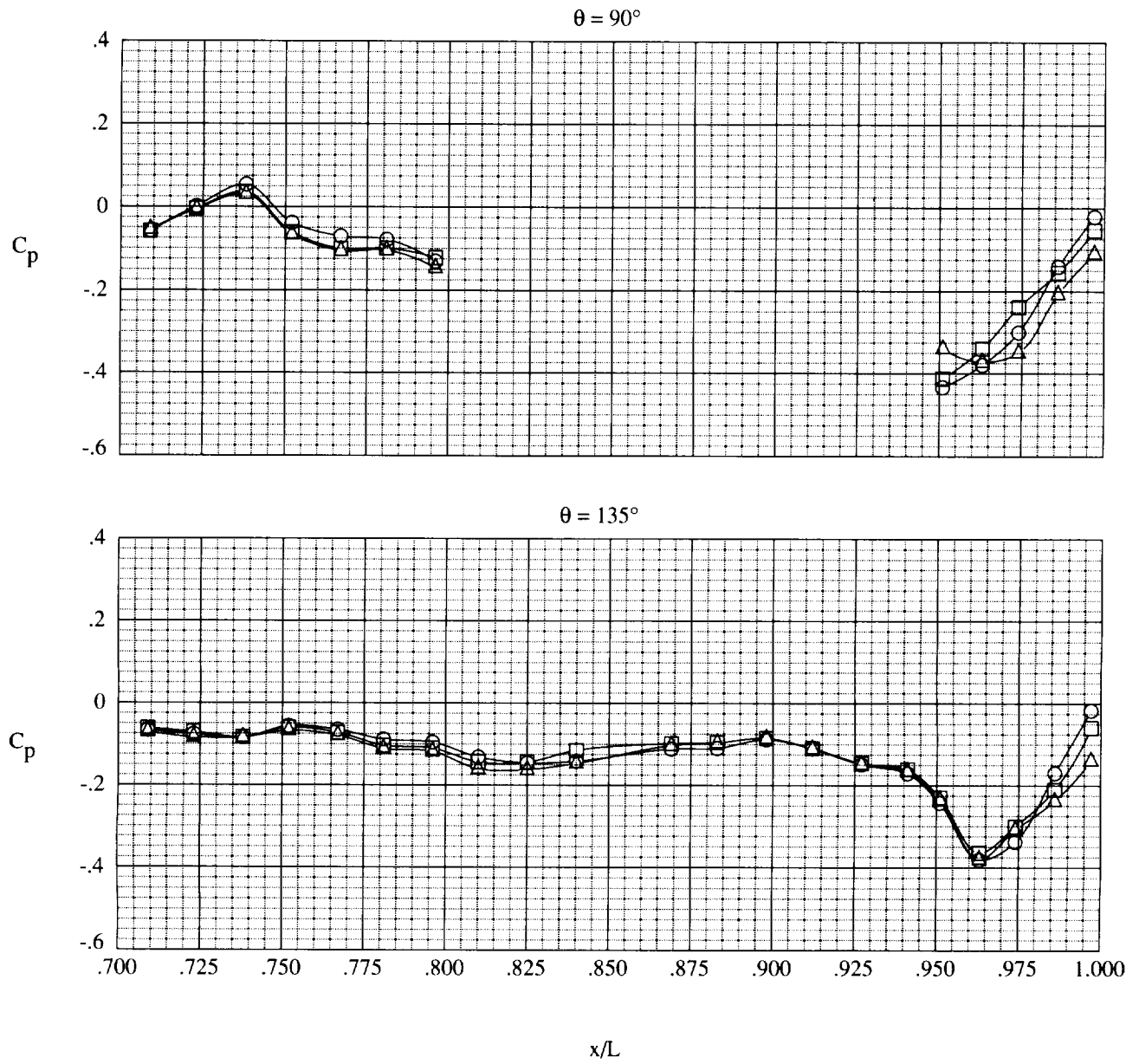


(a) $\theta = 0^\circ$ and 45° .

Figure 11. Effect of vertical-tail position on pressure distributions for $M = 1.2$, $\alpha = 0^\circ$, $NPR = 3.4$, and HTAIL = aft position.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

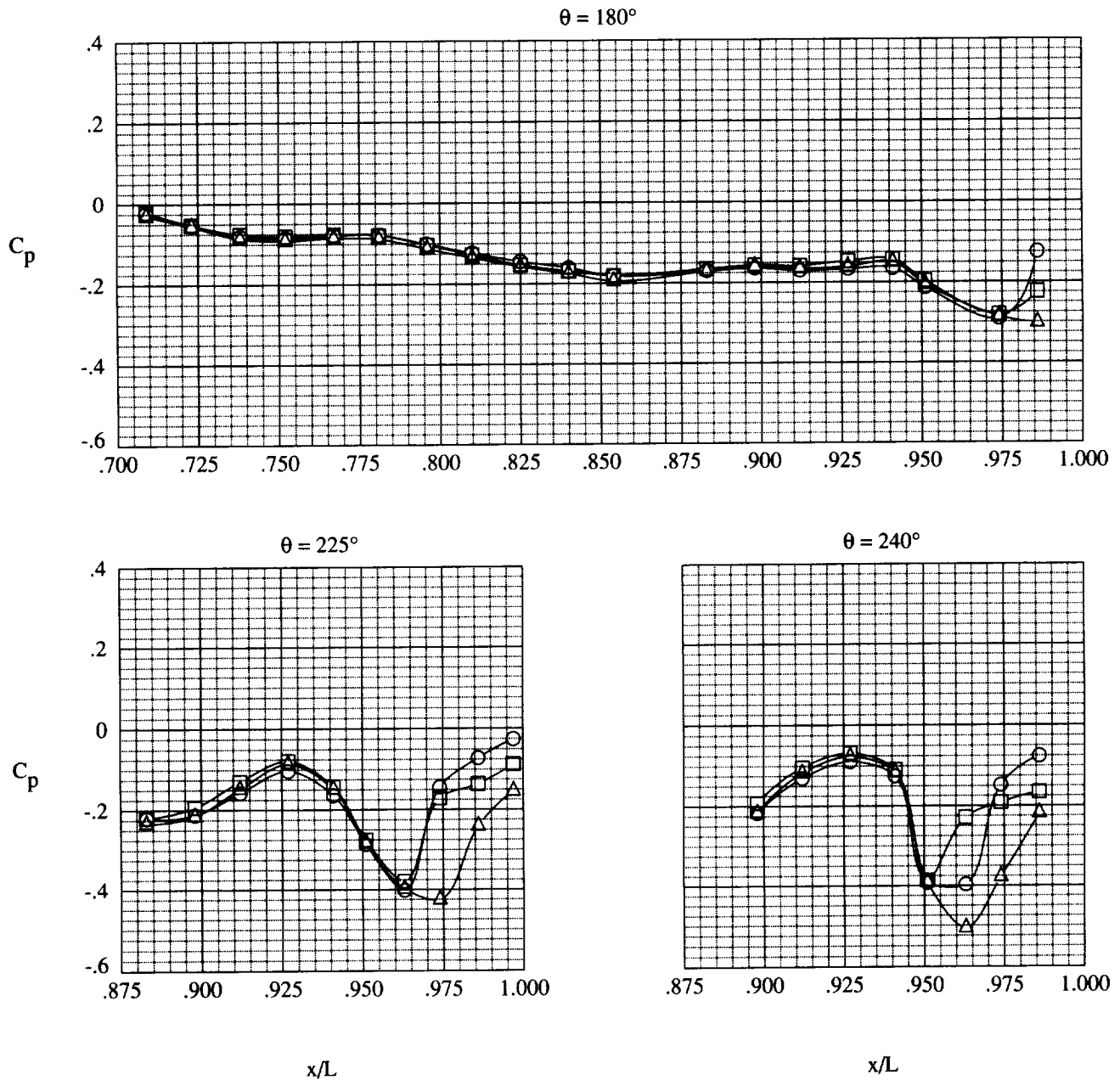


(b) $\theta = 90^\circ$ and 135° .

Figure 11. Continued.

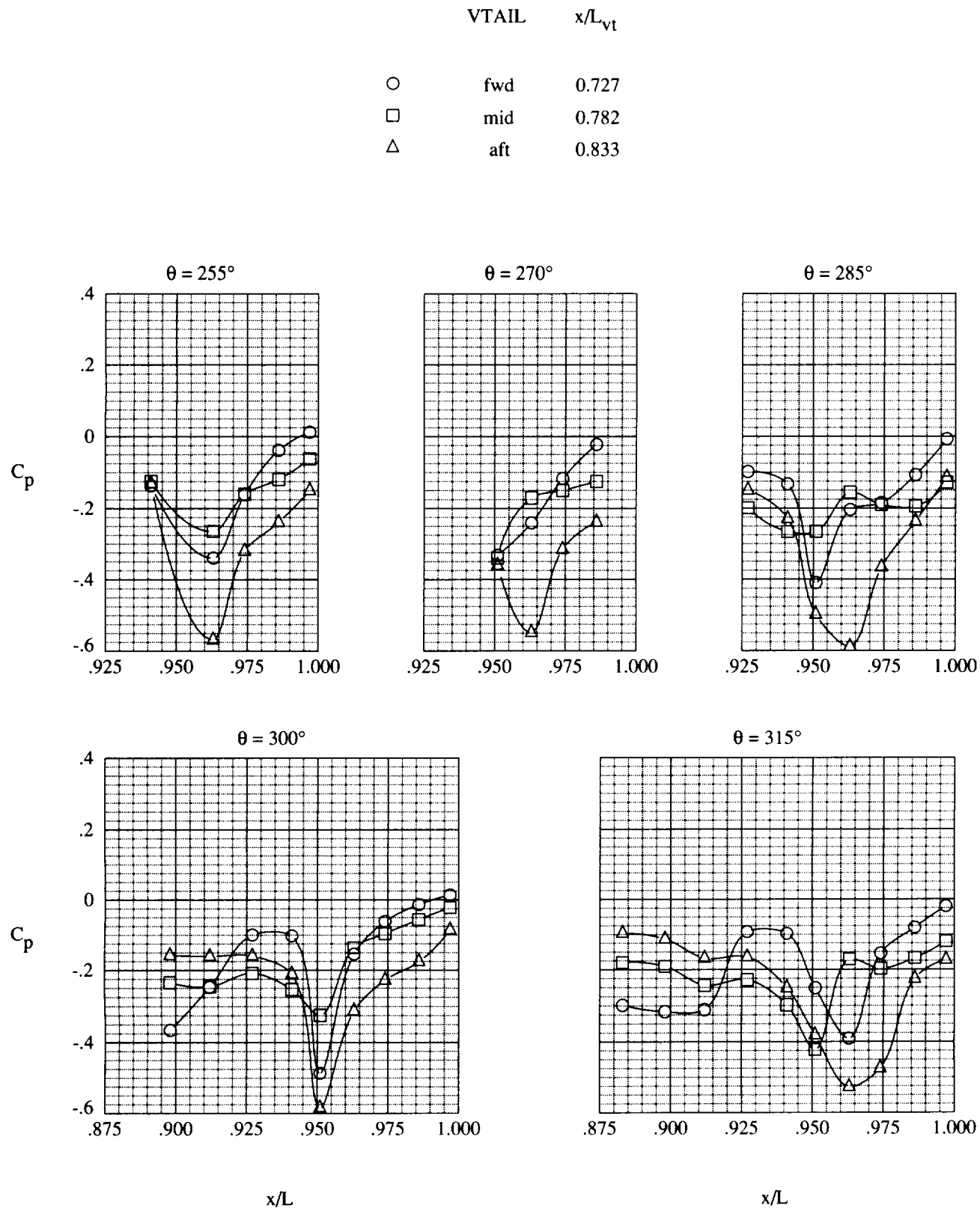
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 11. Continued.

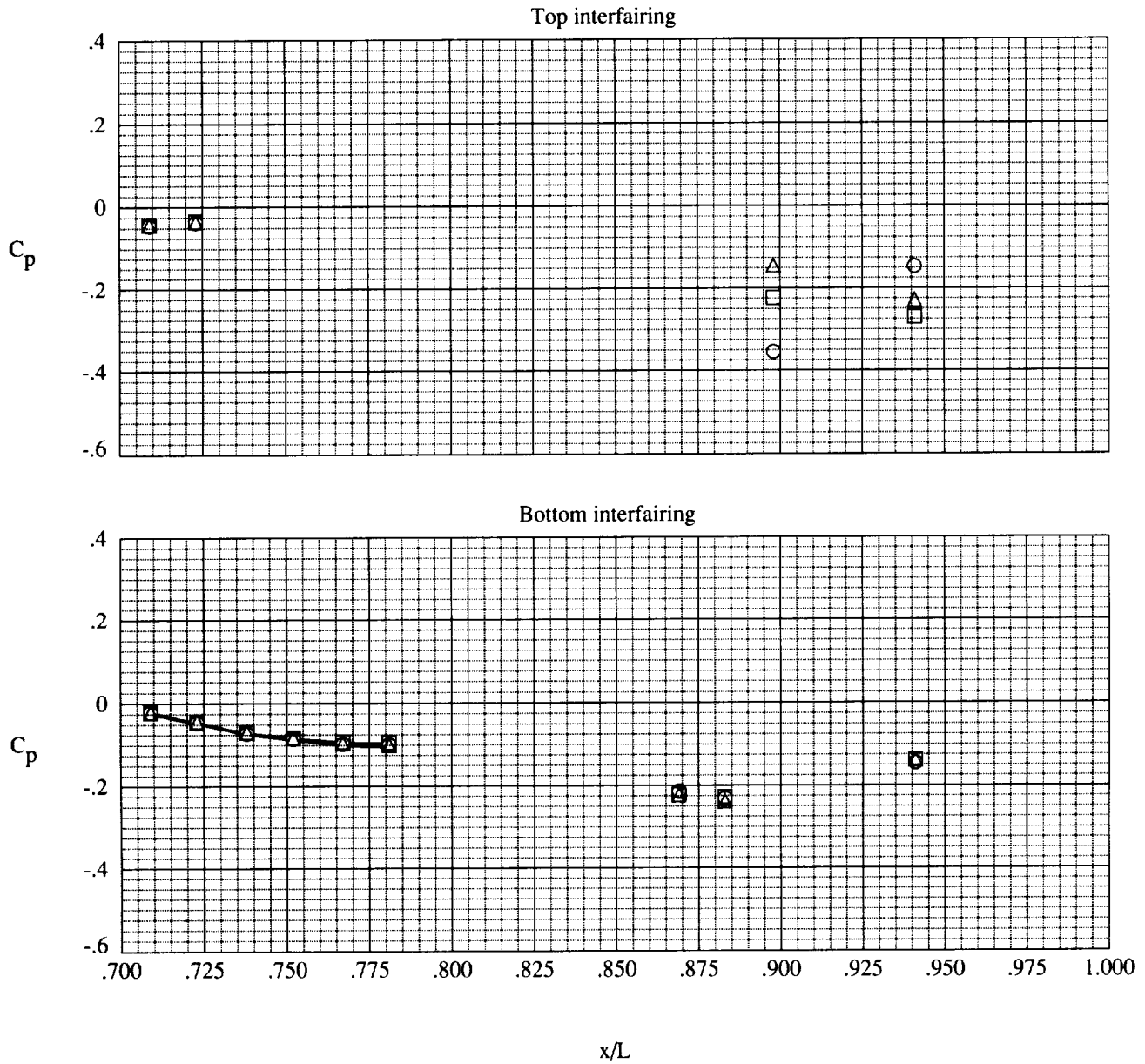


(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 11. Continued.

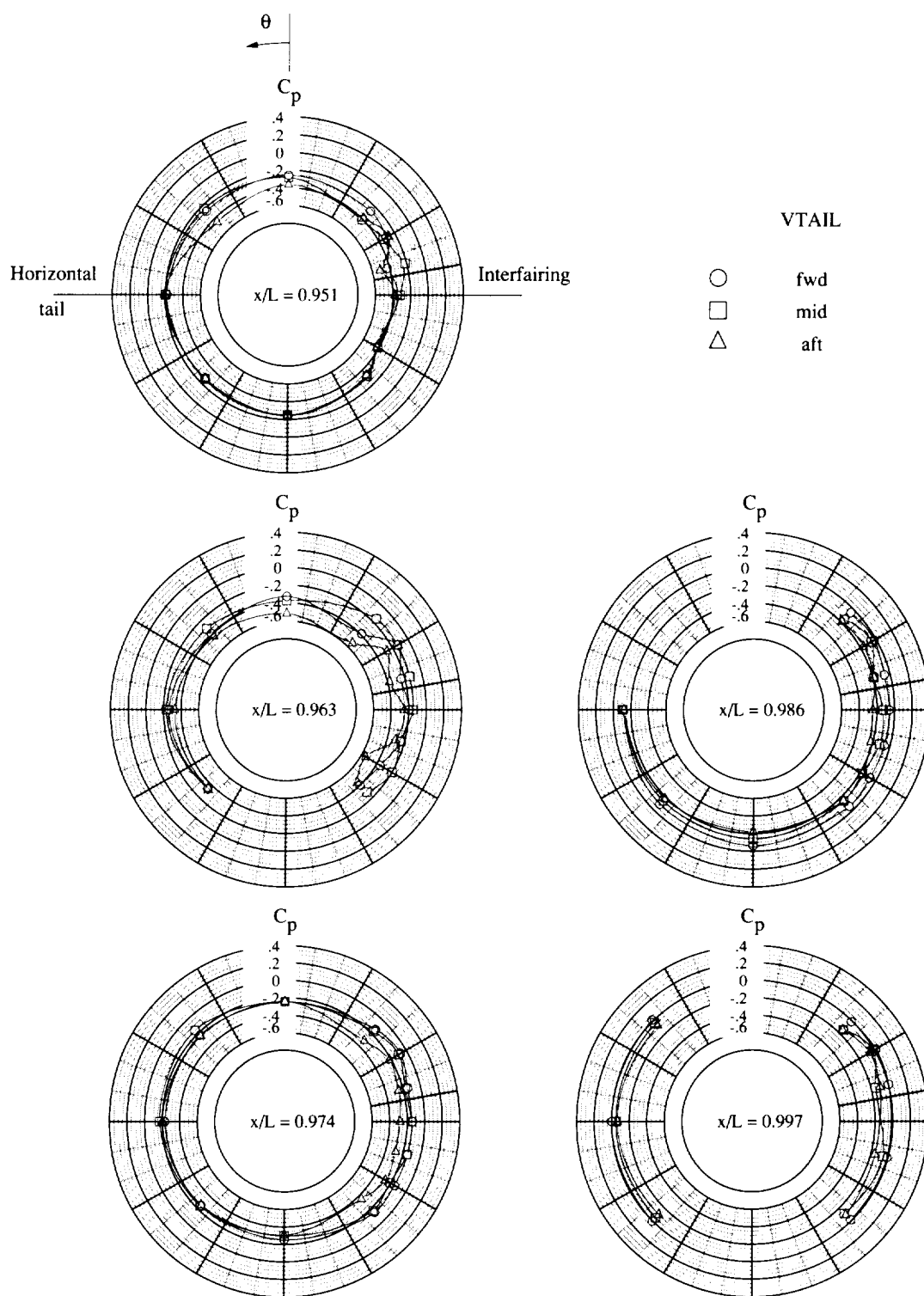
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



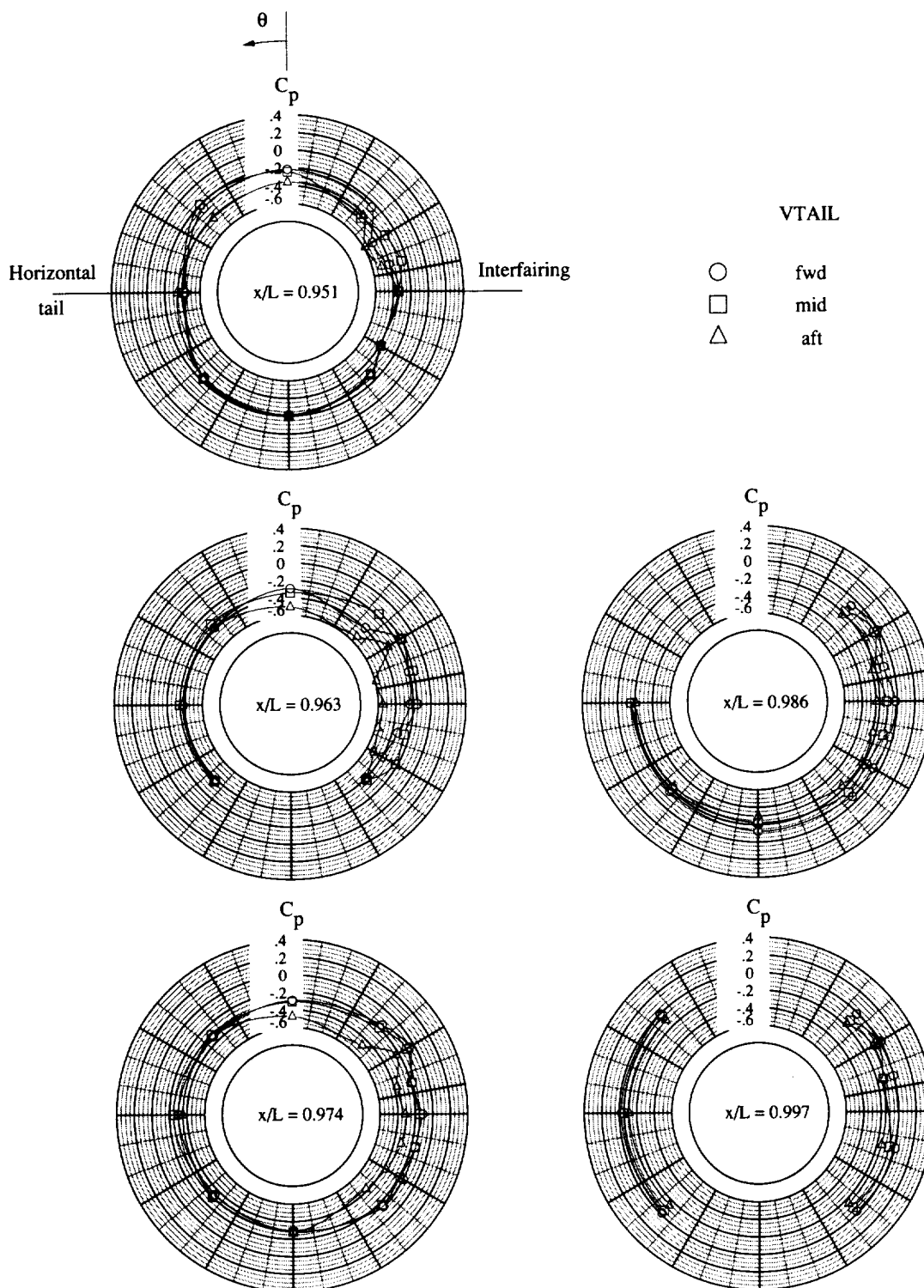
(e) Top and bottom interfairings.

Figure 11. Concluded.



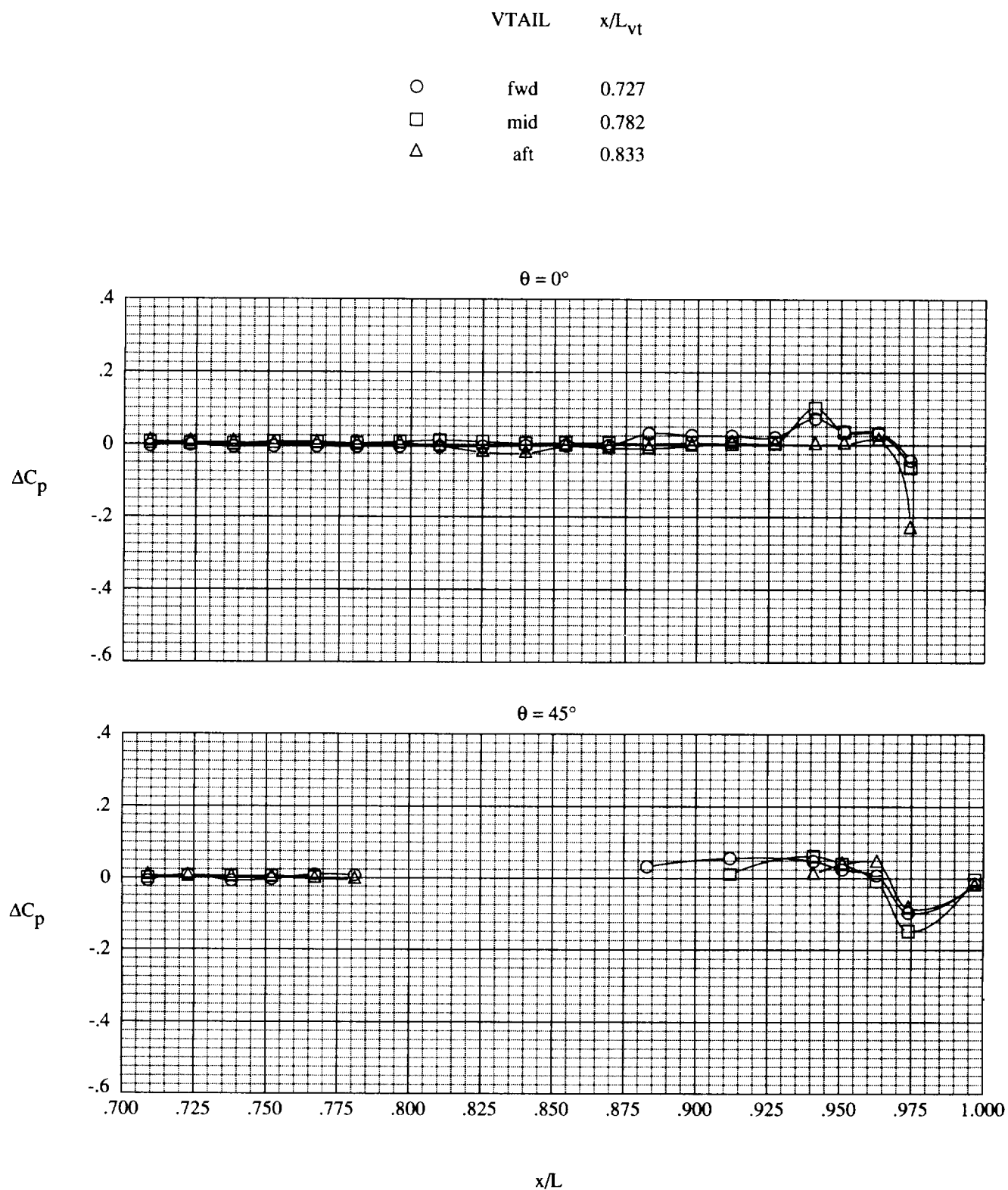
(a) HTAIL = mid position.

Figure 12. Effect of vertical-tail position on static pressure distribution around left nozzle for $M = 1.2$, $\alpha = 0^\circ$, and $NPR = 3.4$.



(b) HTAIL = aft position.

Figure 12. Concluded.

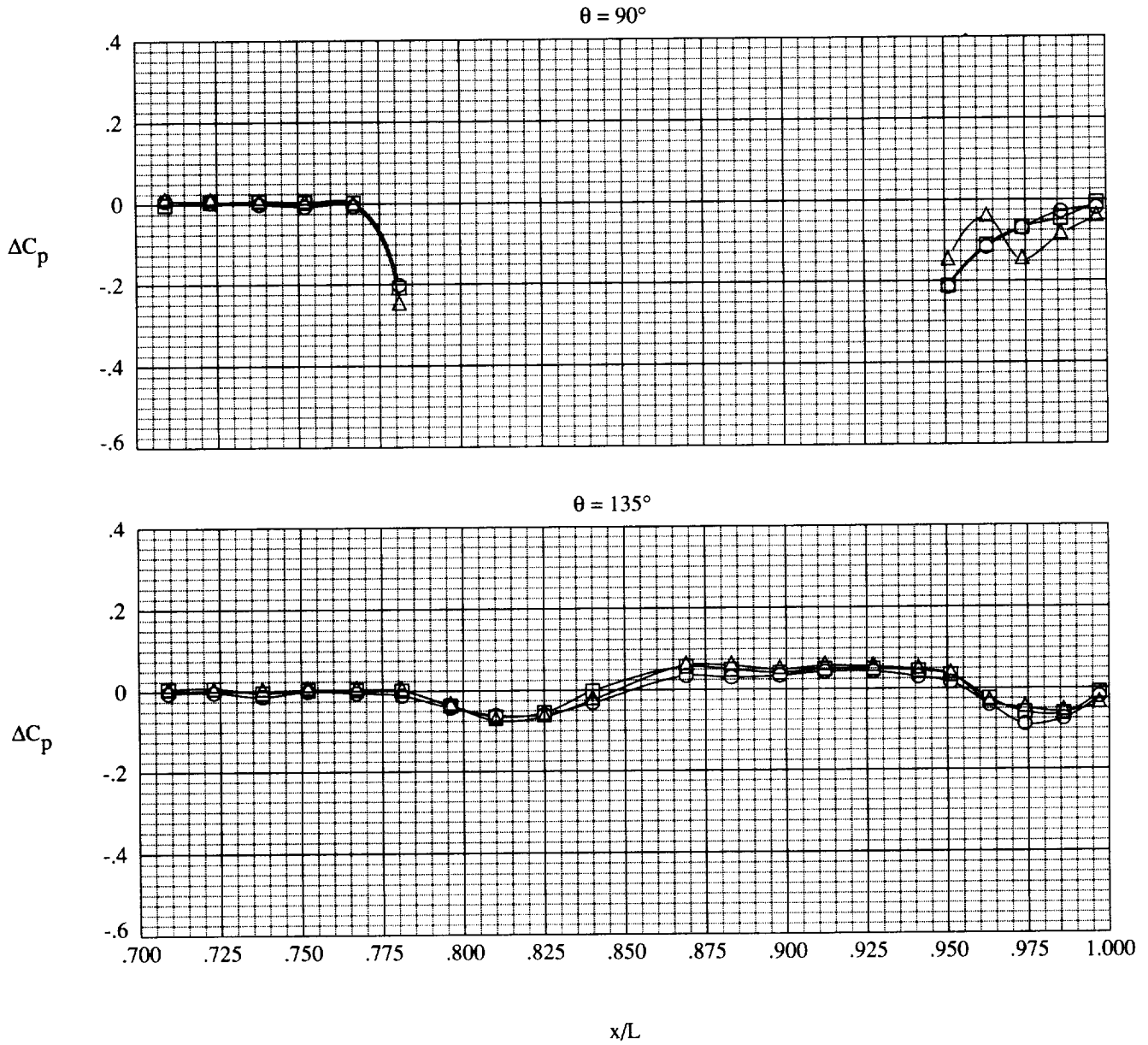


(a) $\theta = 0^\circ$ and 45° .

Figure 13. Change in static pressure coefficients from moving horizontal tails from mid to aft position for $M = 1.2$, $\alpha = 0^\circ$, and $NPR = 3.4$.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

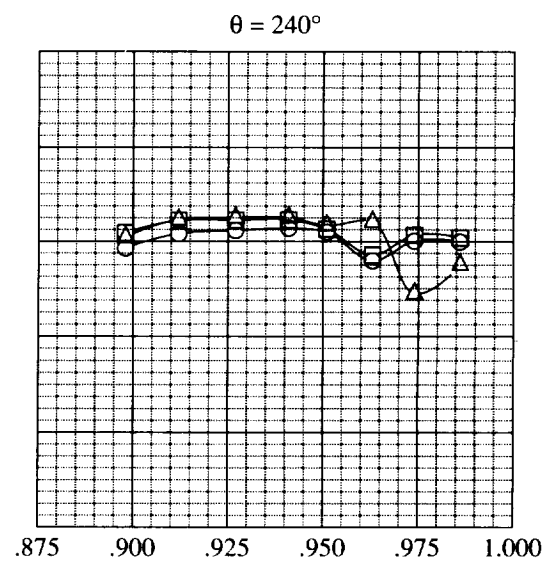
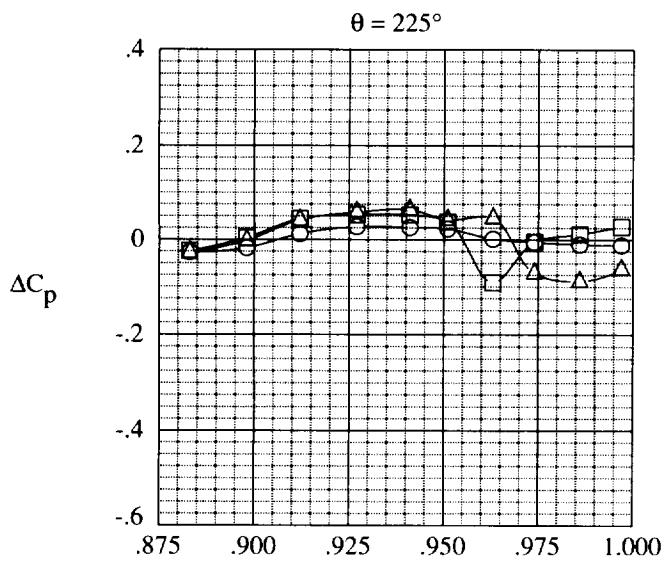
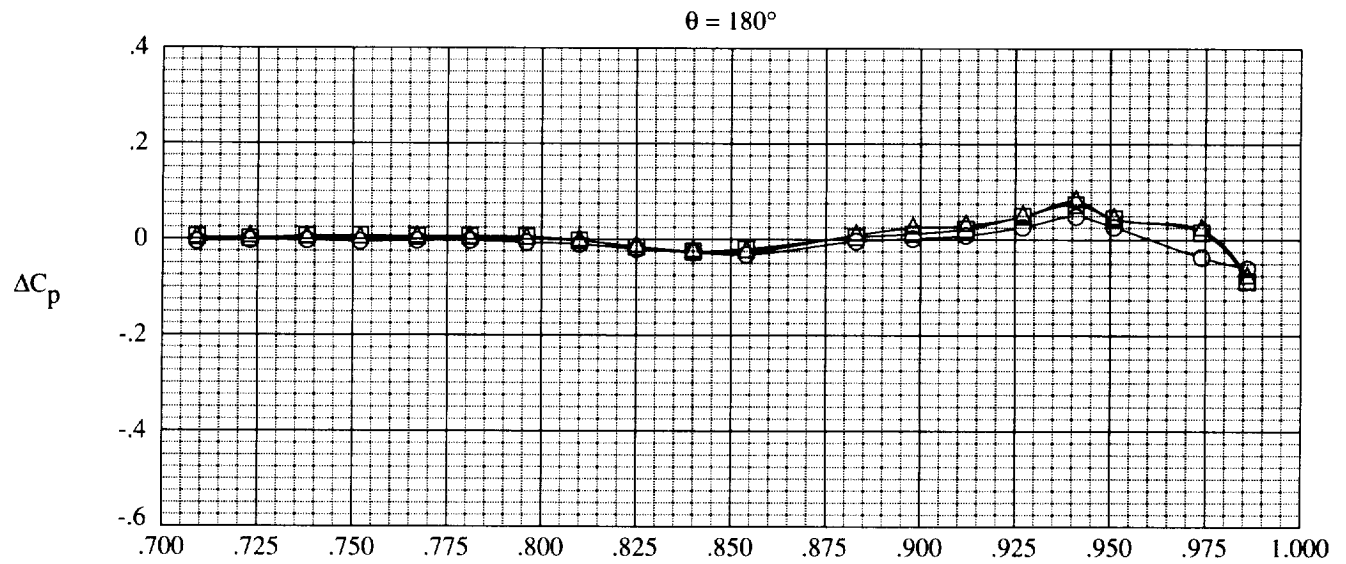


(b) $\theta = 90^\circ$ and 135° .

Figure 13. Continued.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

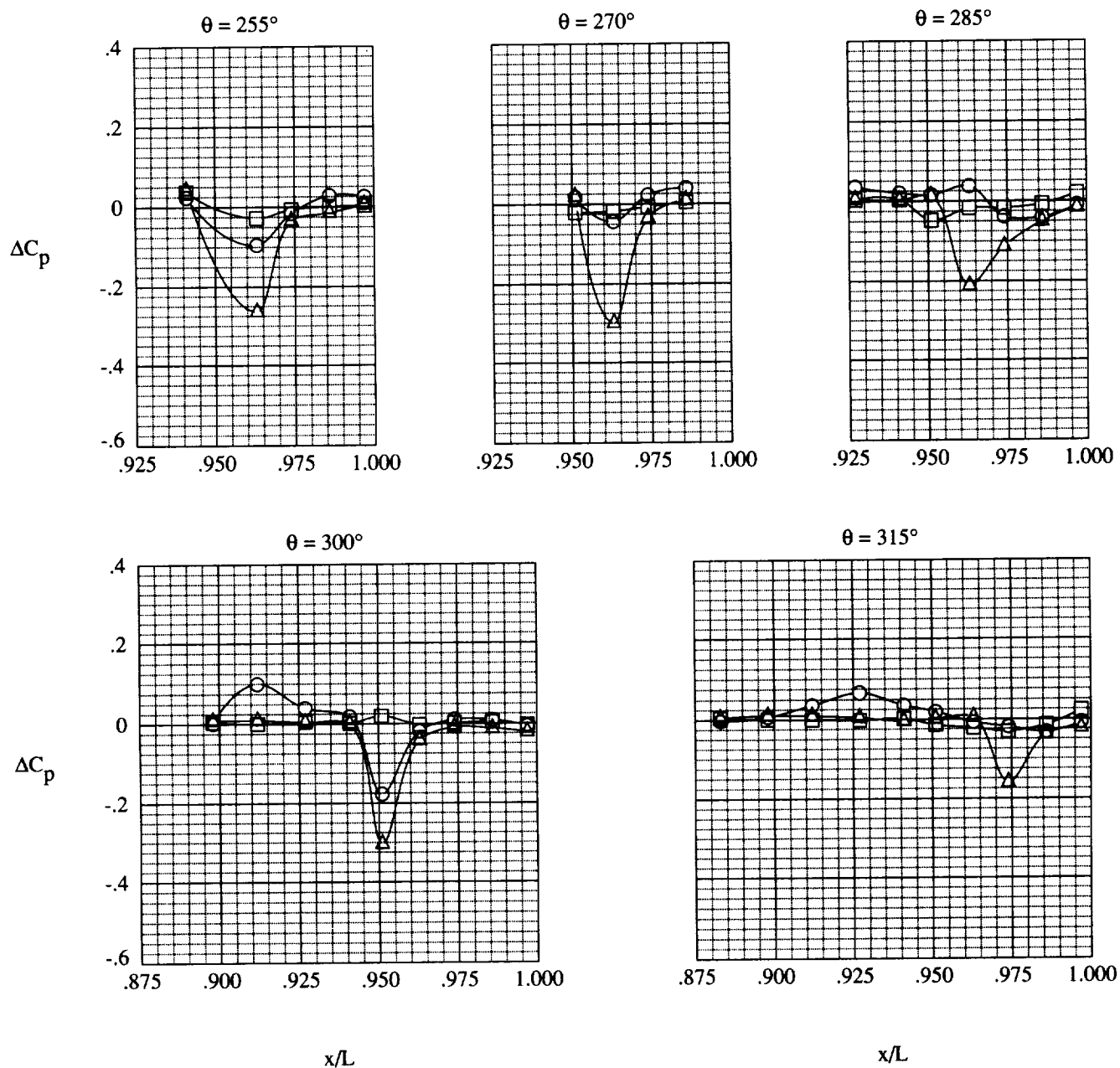


(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 13. Continued.

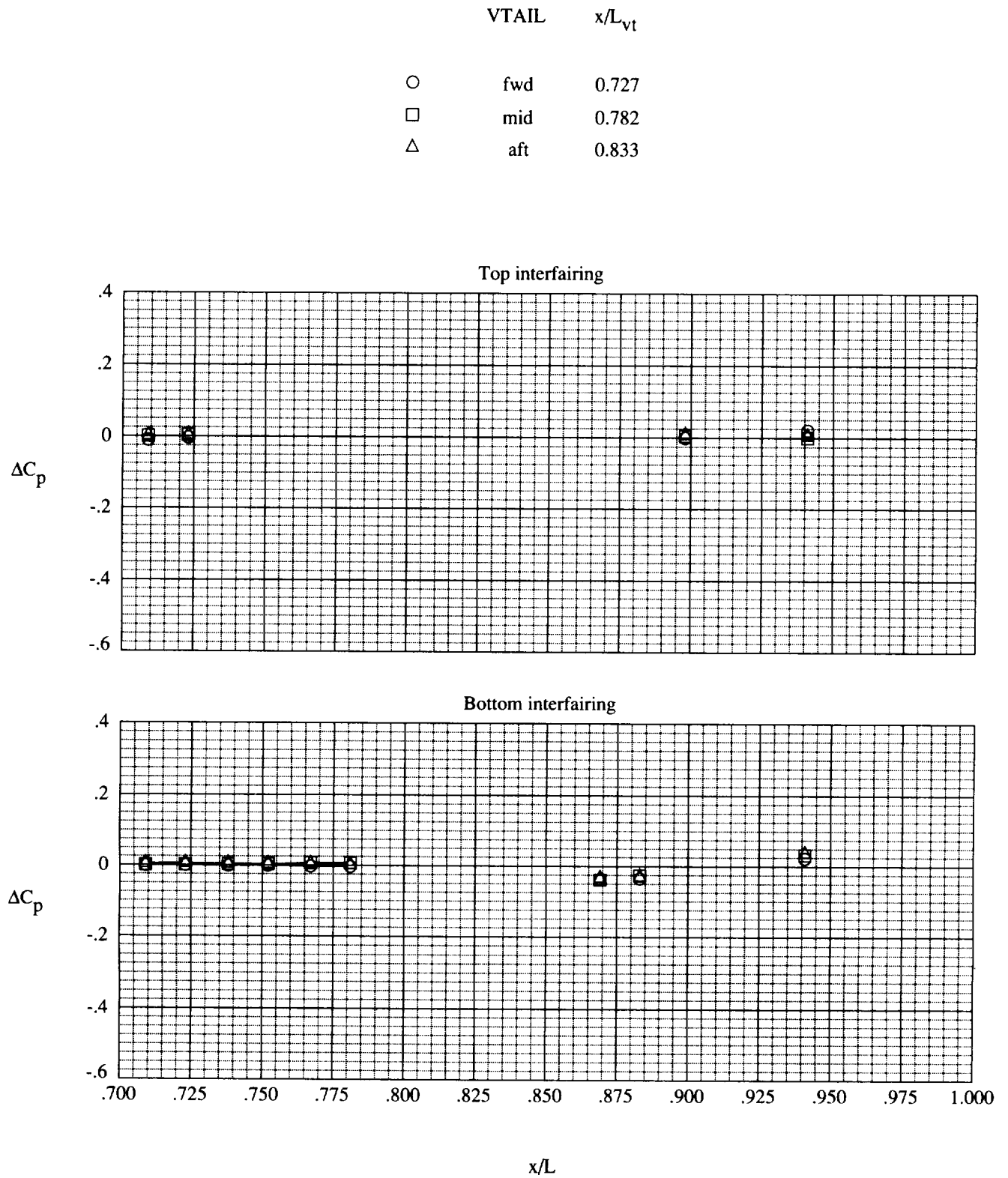
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



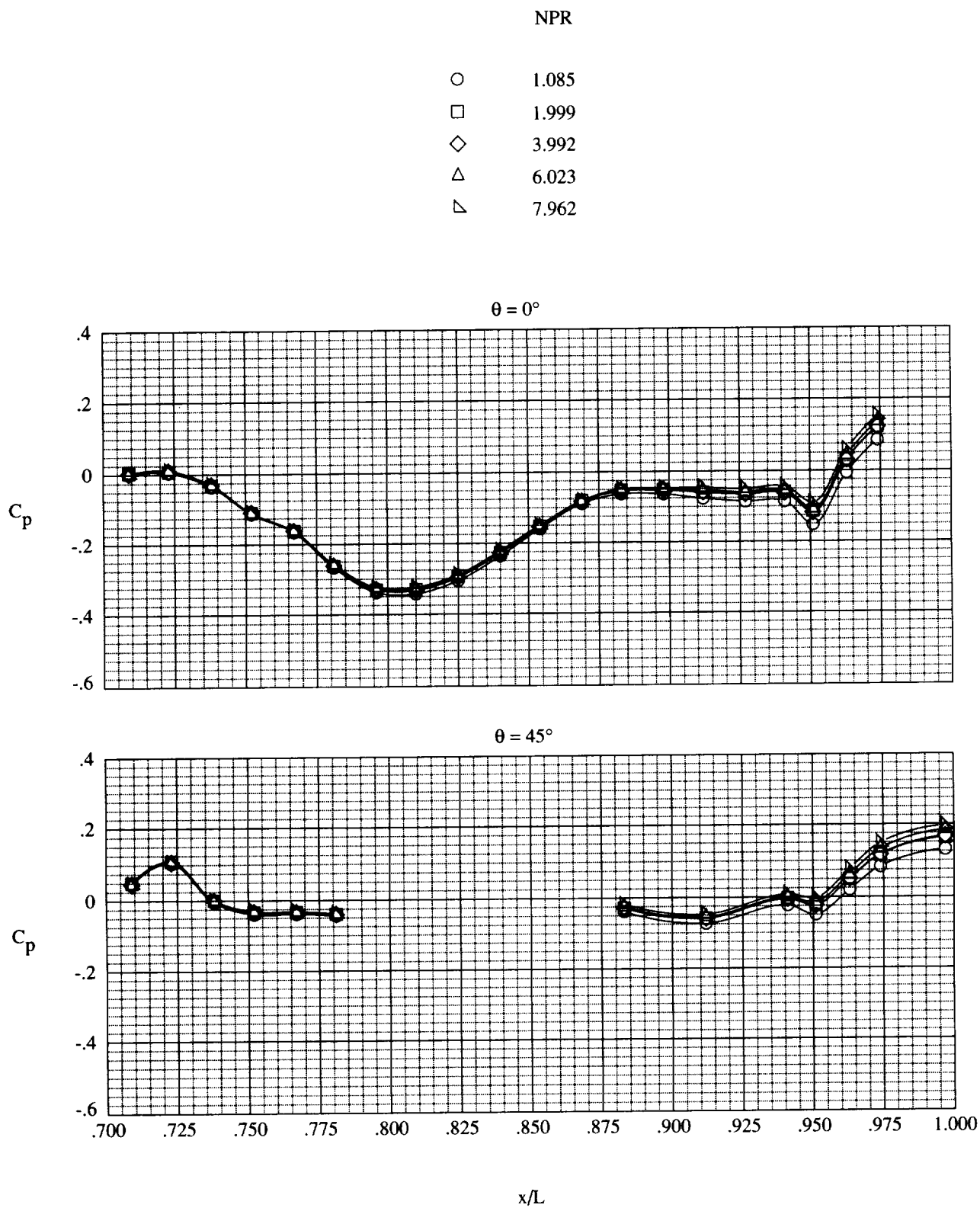
(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 13. Continued.



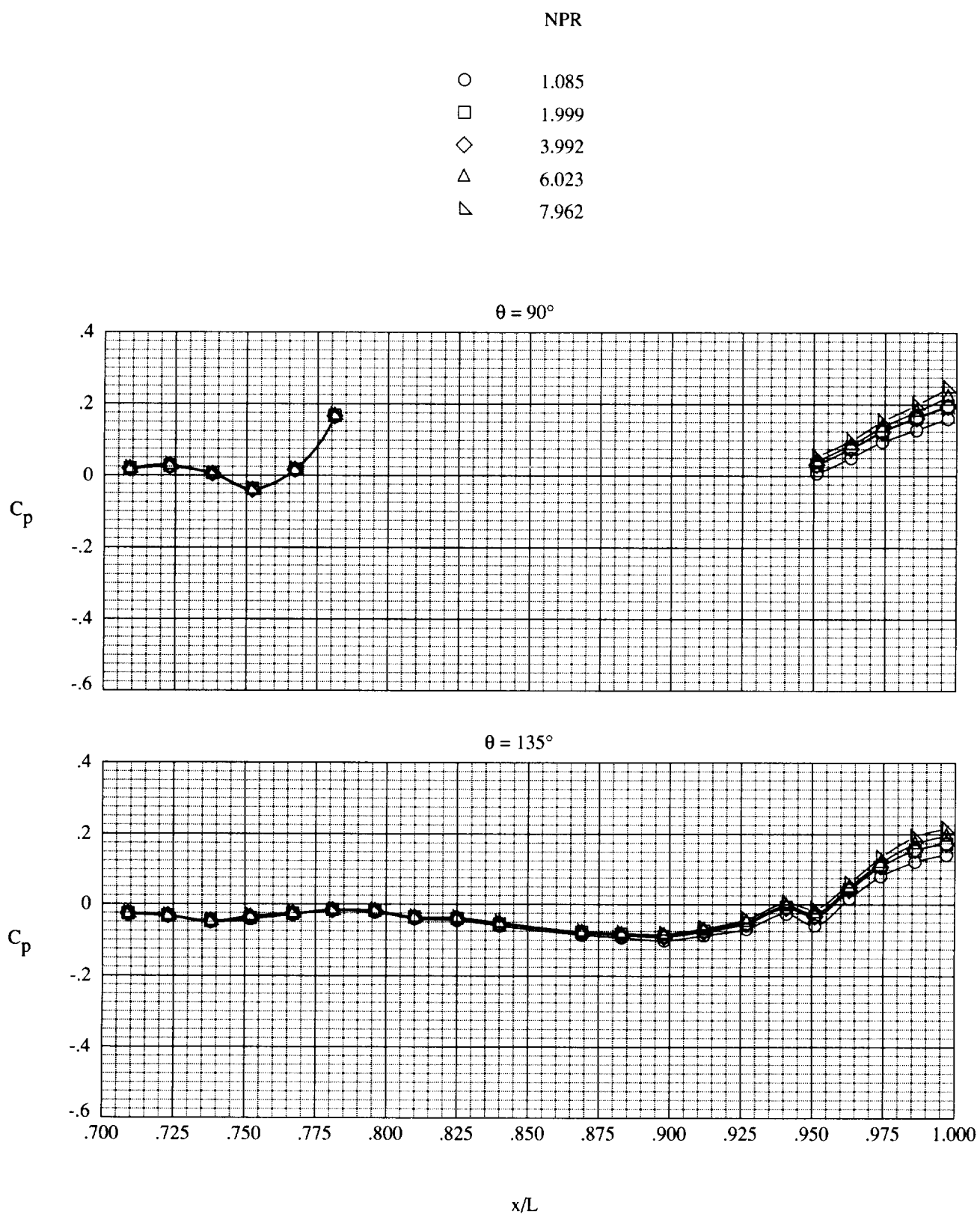
(e) Top and bottom interfairings.

Figure 13. Concluded.



(a) $\theta = 0^\circ$ and 45° .

Figure 14. Effect of nozzle pressure ratio on pressure distributions for $M = 0.8$, $\alpha = 0^\circ$, HTAIL = mid position, and VTAIL = fwd position.

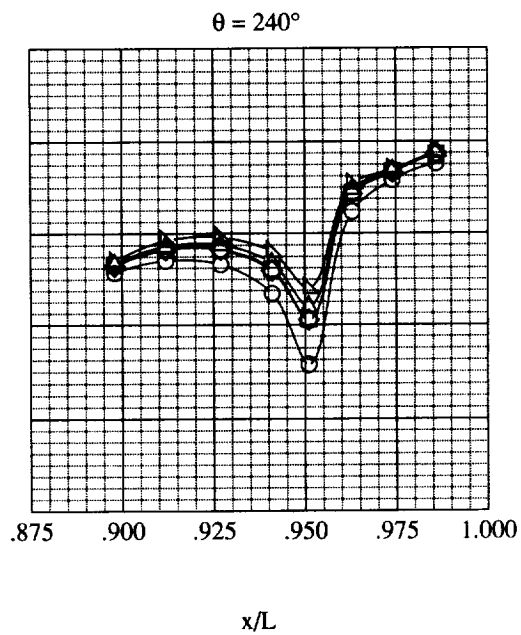
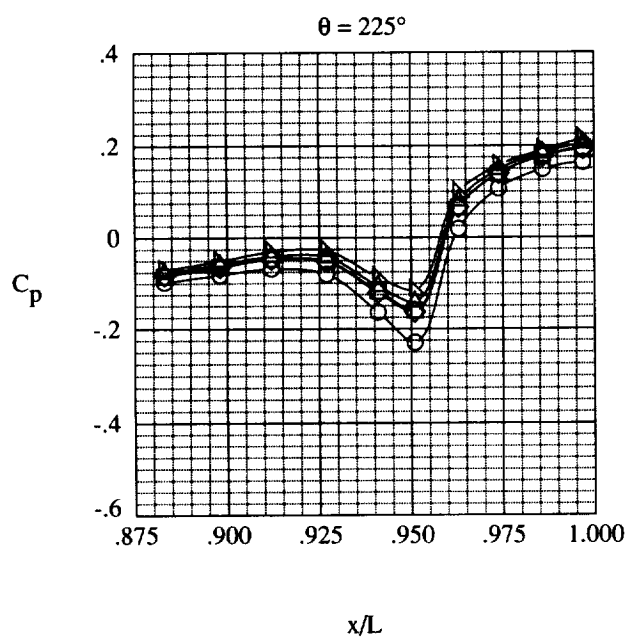
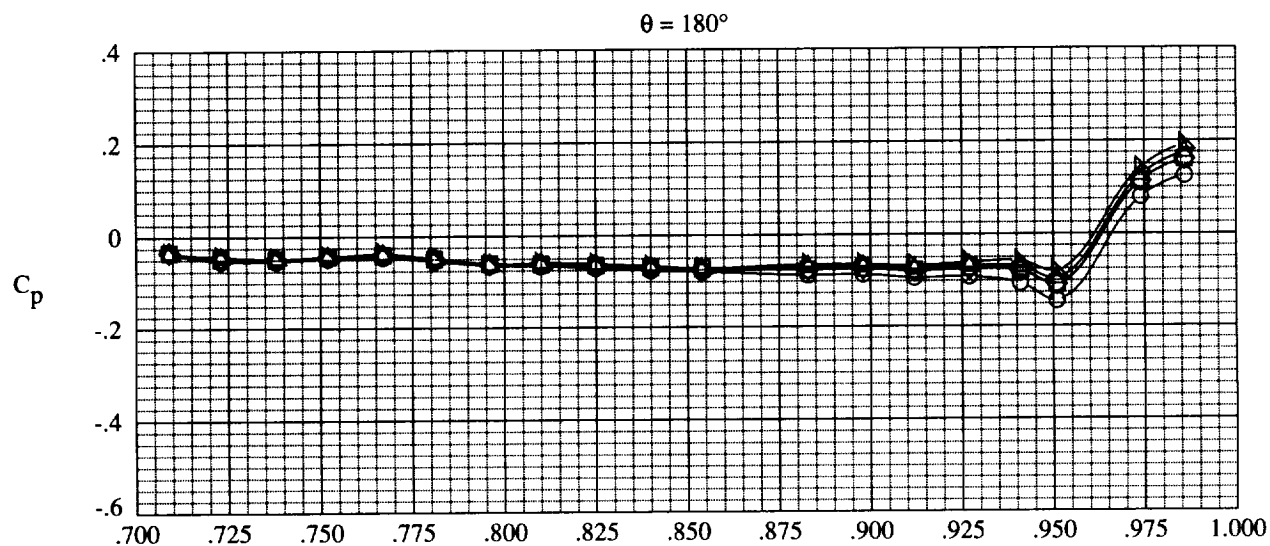


(b) $\theta = 90^\circ$ and 135° .

Figure 14. Continued.

NPR

○	1.085
□	1.999
◇	3.992
△	6.023
▽	7.962



(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 14. Continued.

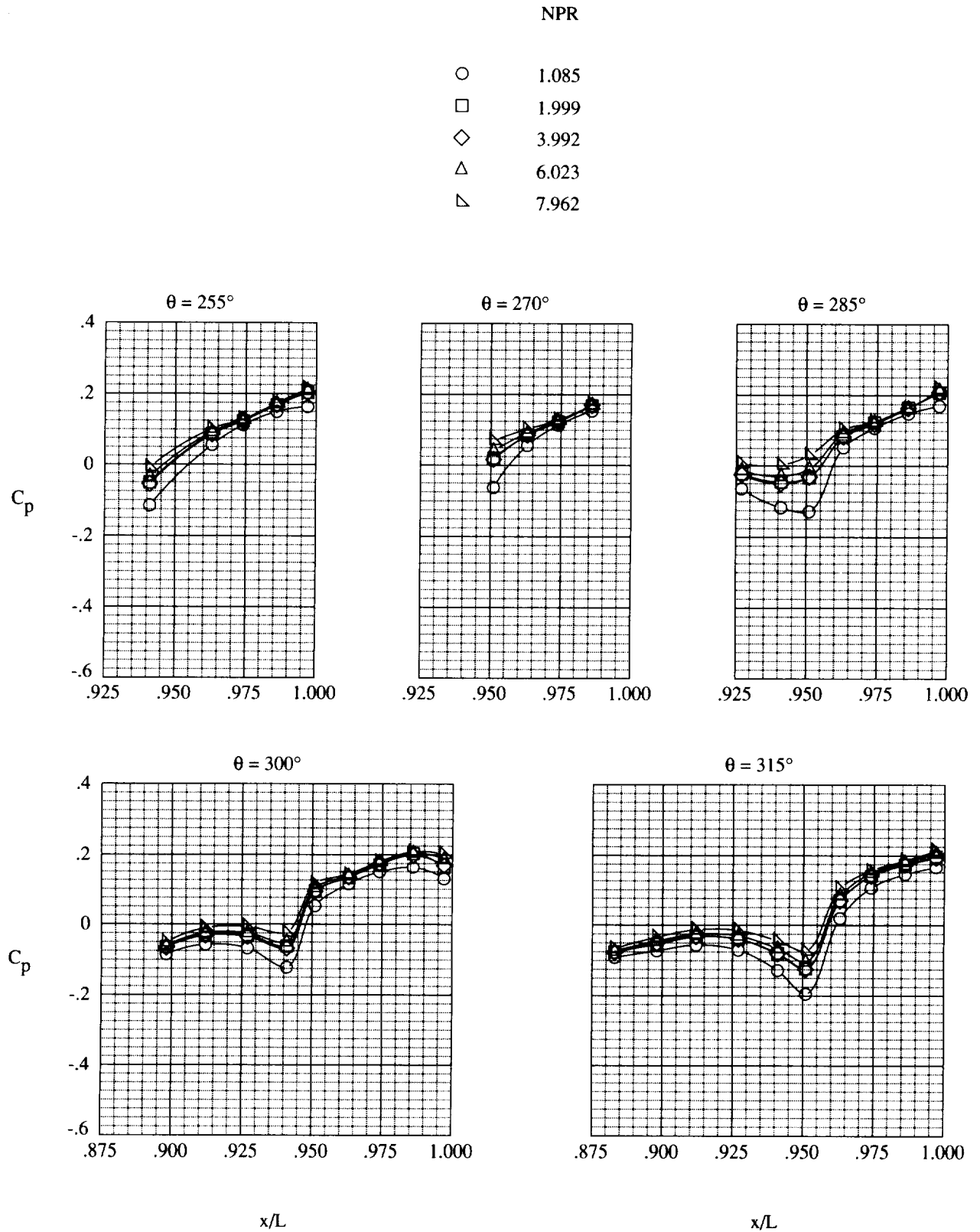
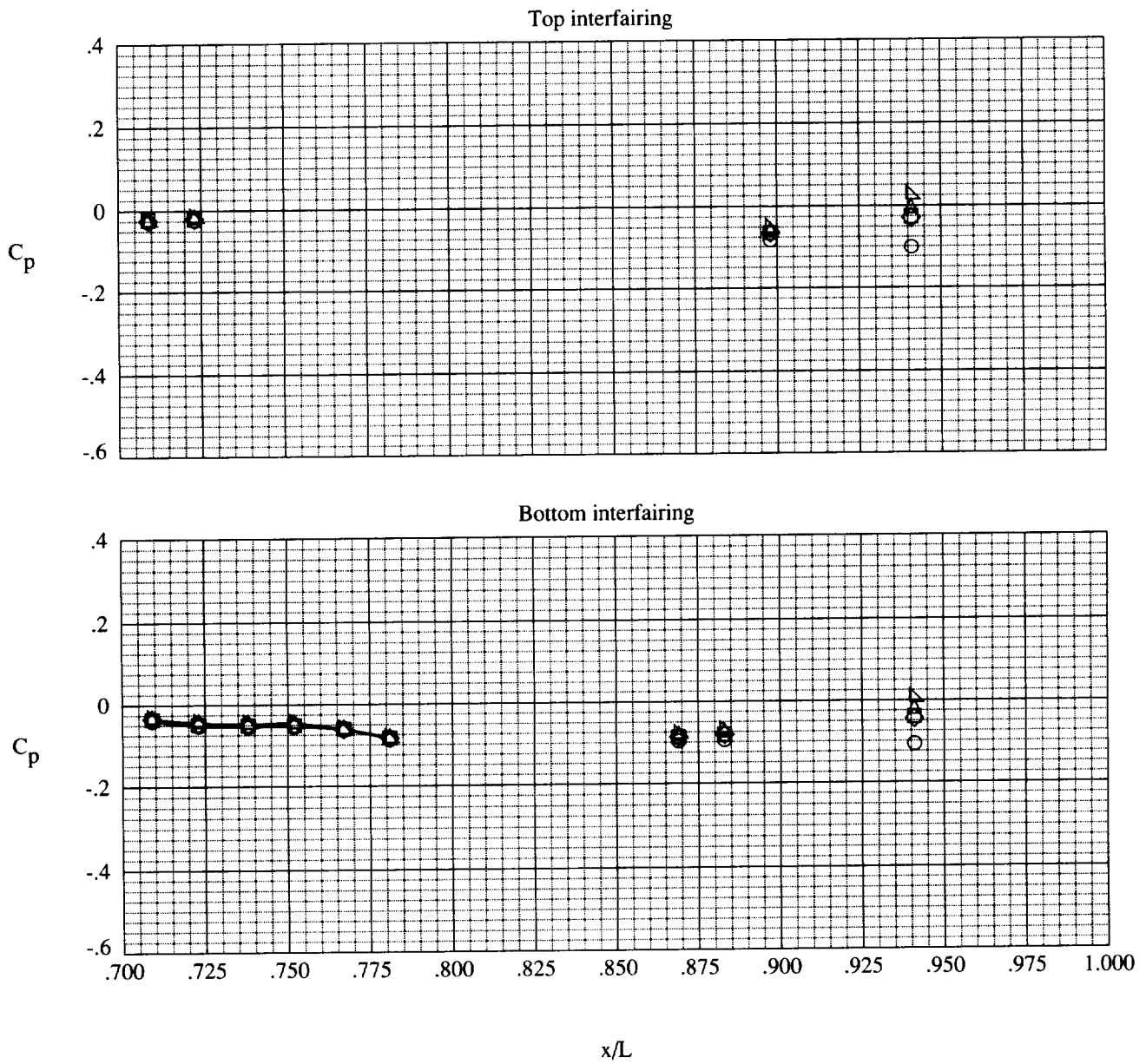


Figure 14. Continued.

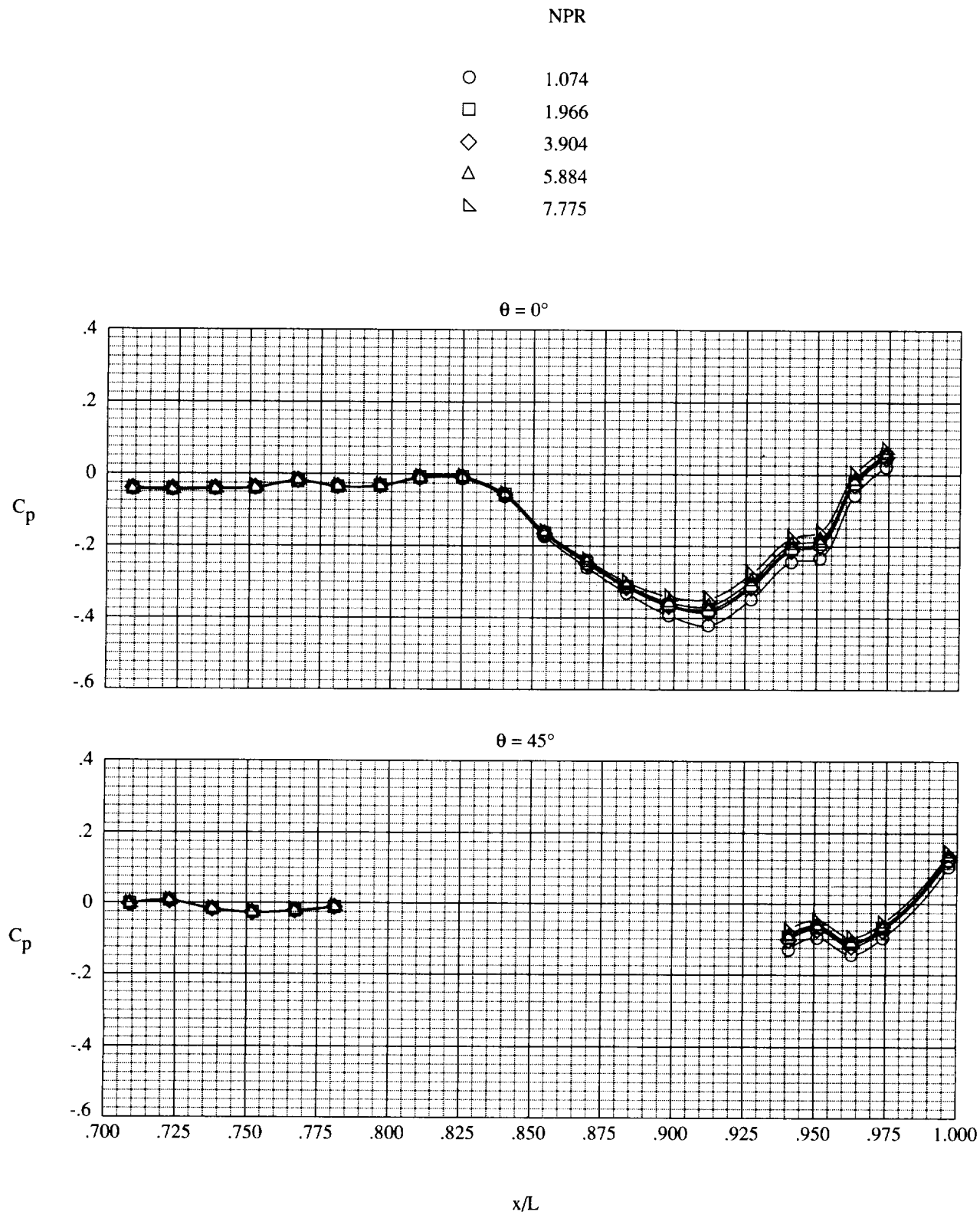
NPR

○	1.085
□	1.999
◇	3.992
△	6.023
▴	7.962



(e) Top and bottom interfairings.

Figure 14. Concluded.

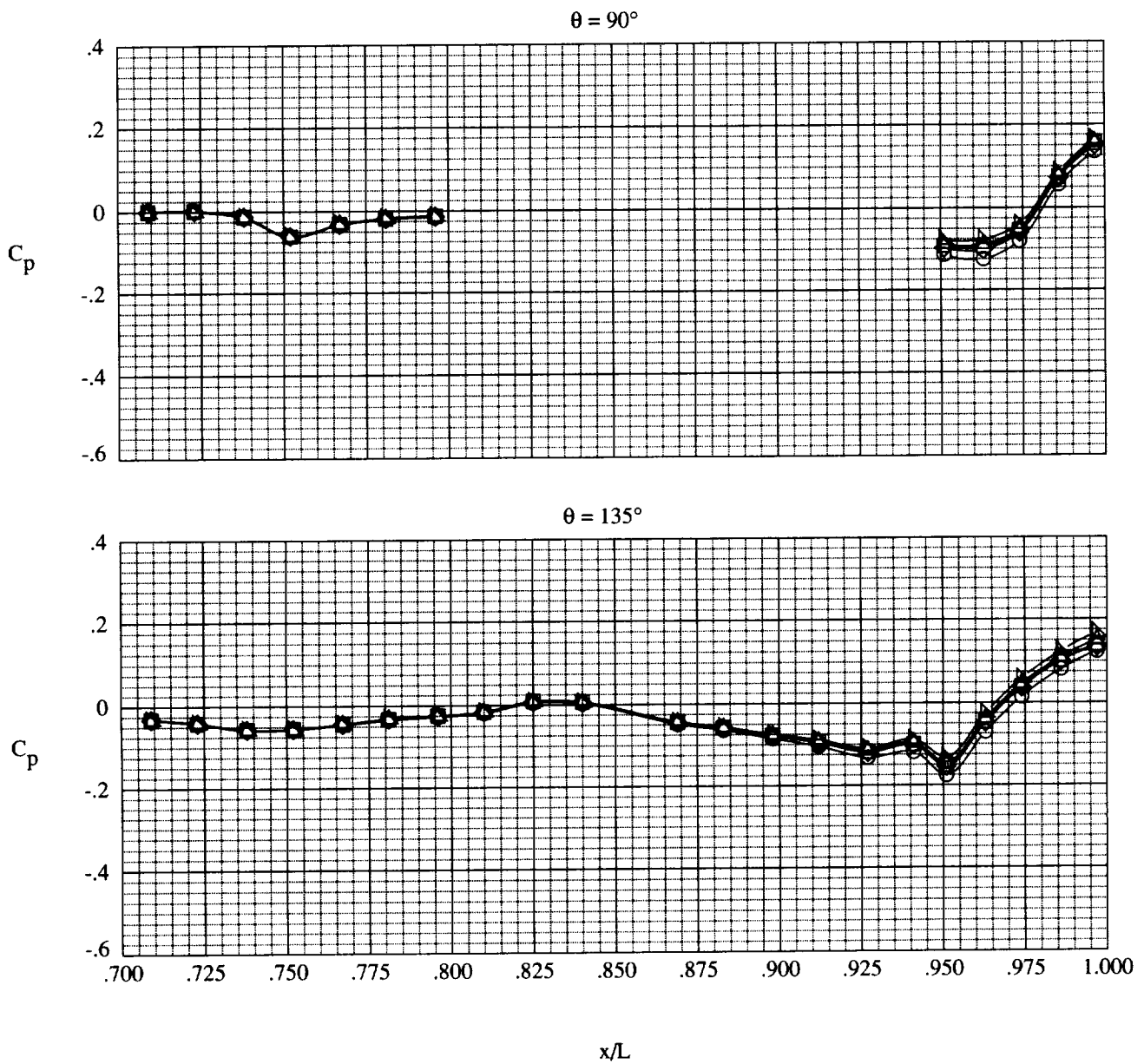


(a) $\theta = 0^\circ$ and 45° .

Figure 15. Effect of nozzle pressure ratio on pressure distributions for $M = 0.8$, $\alpha = 0^\circ$, HTAIL = aft position, and VTAIL = aft position.

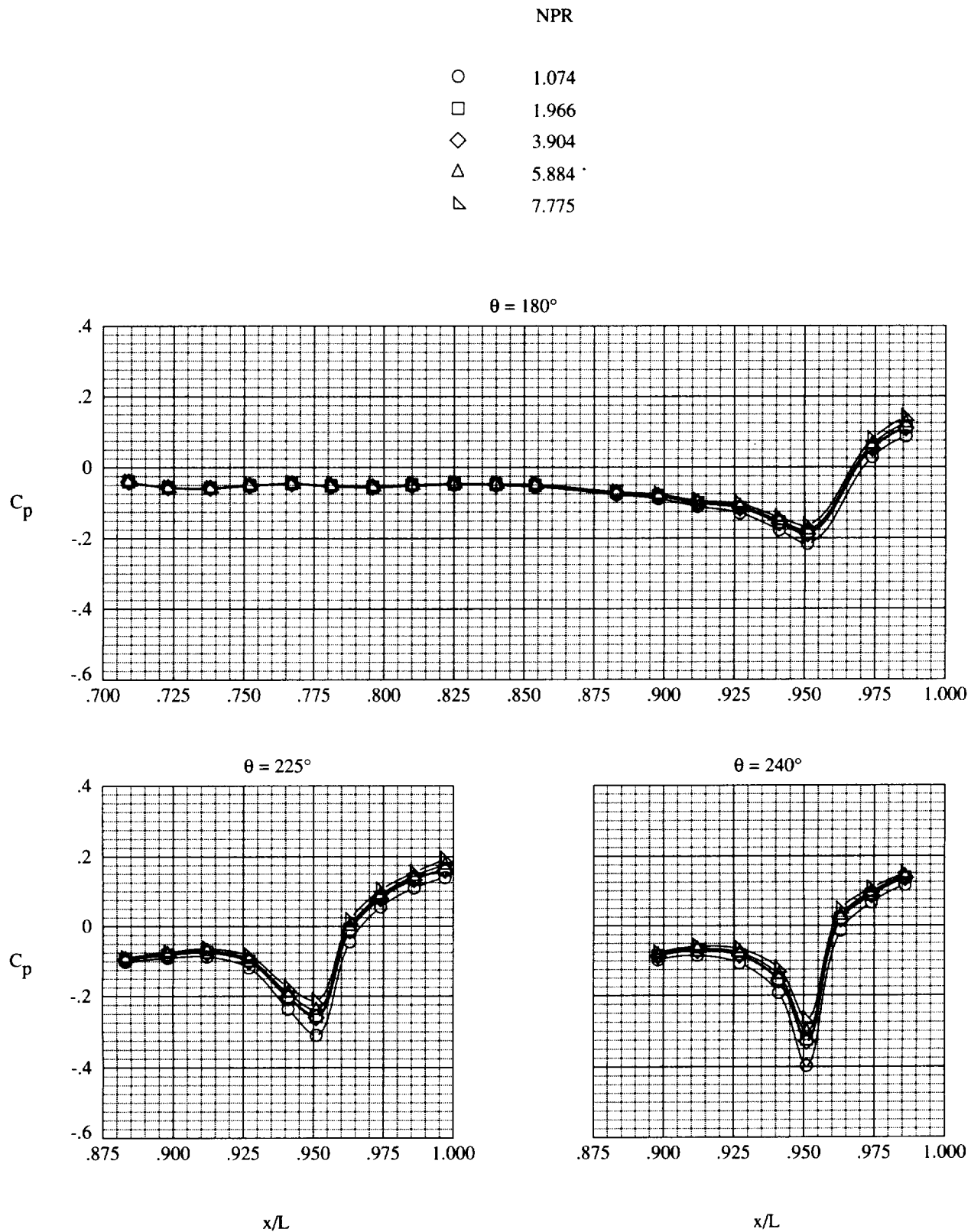
NPR

○	1.074
□	1.966
◇	3.904
△	5.884
▽	7.775



(b) $\theta = 90^\circ$ and 135° .

Figure 15. Continued.

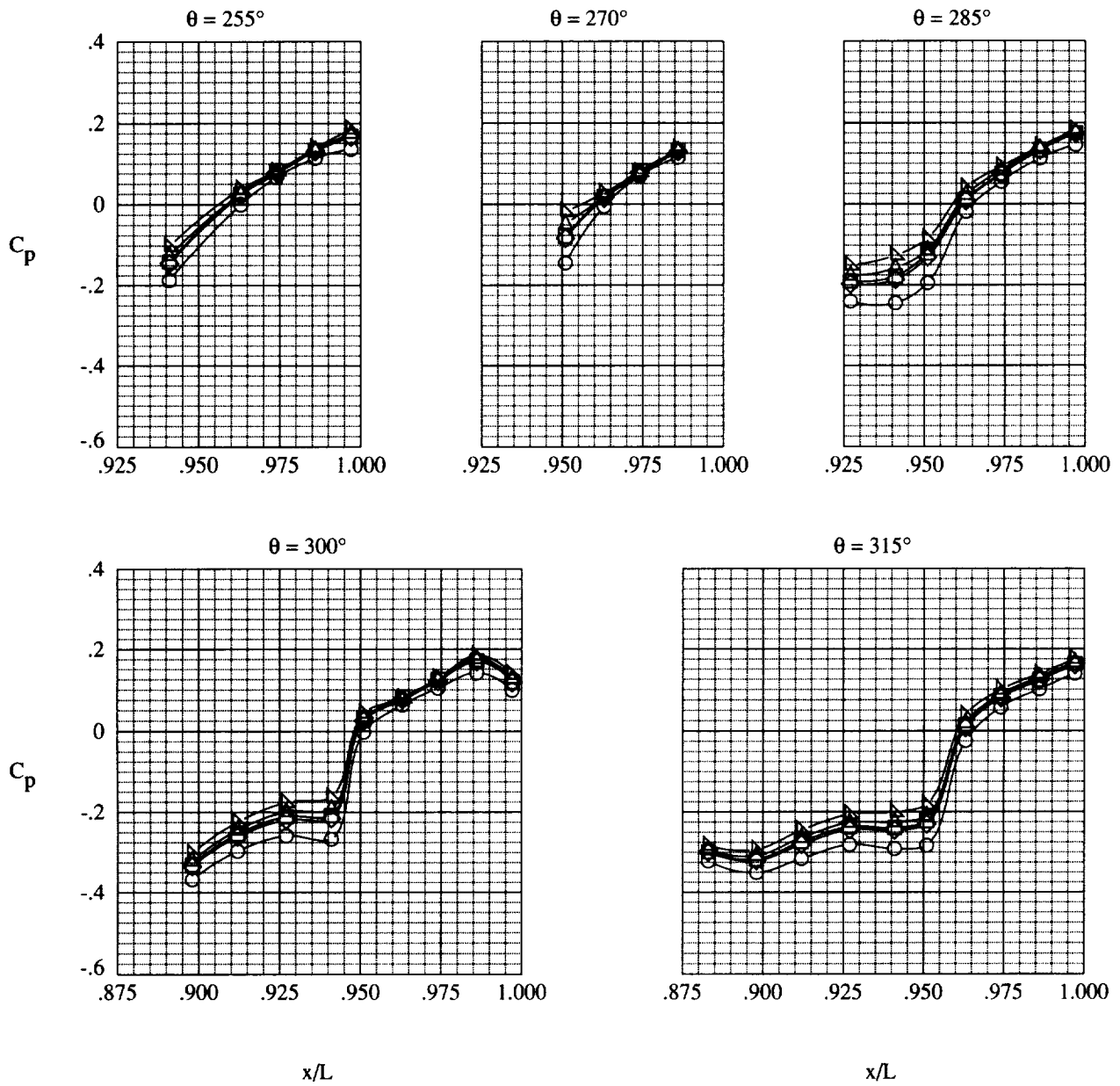


(c) $\theta = 180^\circ$, 225° , and 240° .

Figure 15. Continued.

NPR

○	1.074
□	1.966
◇	3.904
△	5.884
▴	7.775

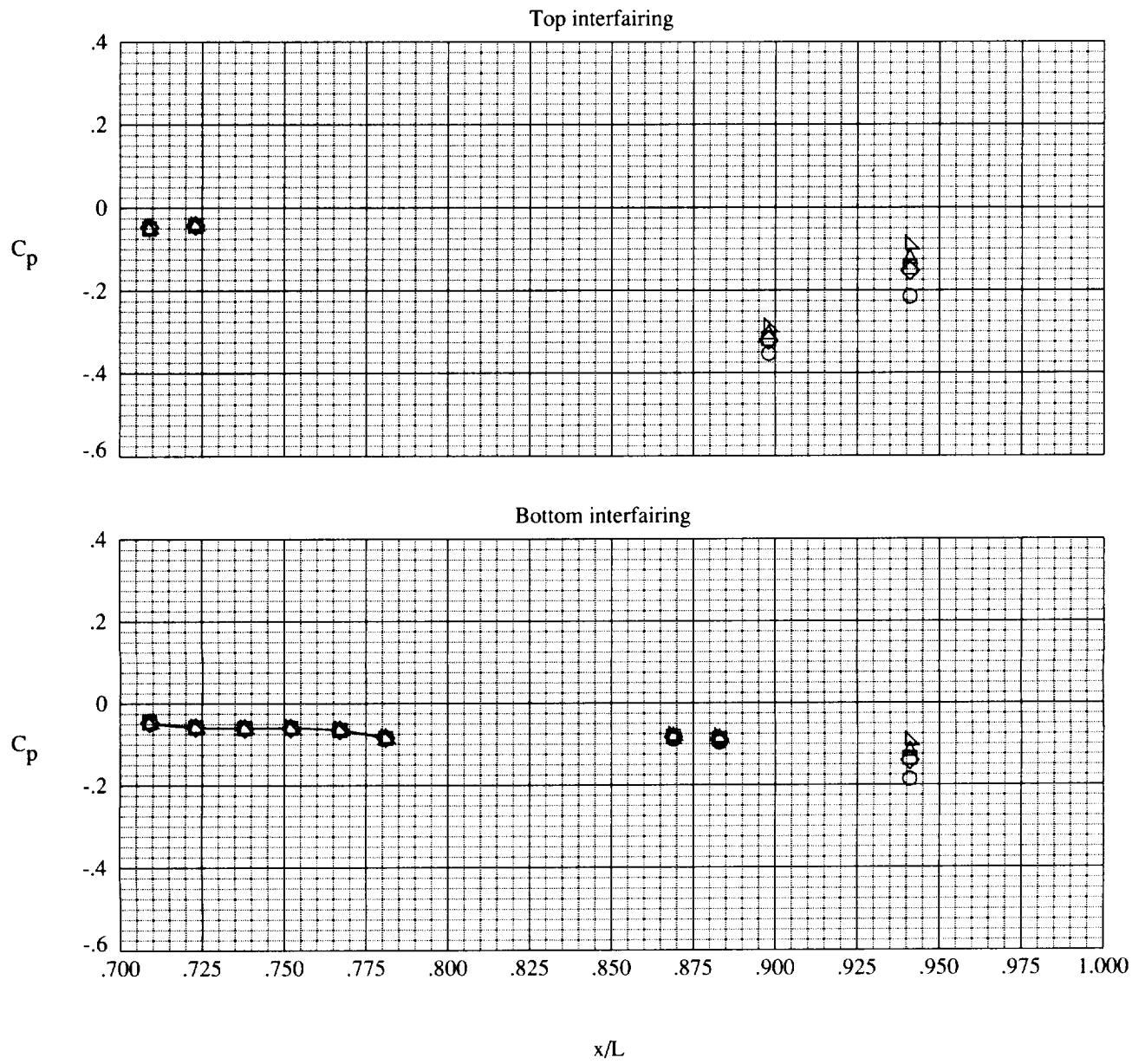


(d) $\theta = 255^\circ$, 270° , 285° , 300° , and 315° .

Figure 15. Continued.

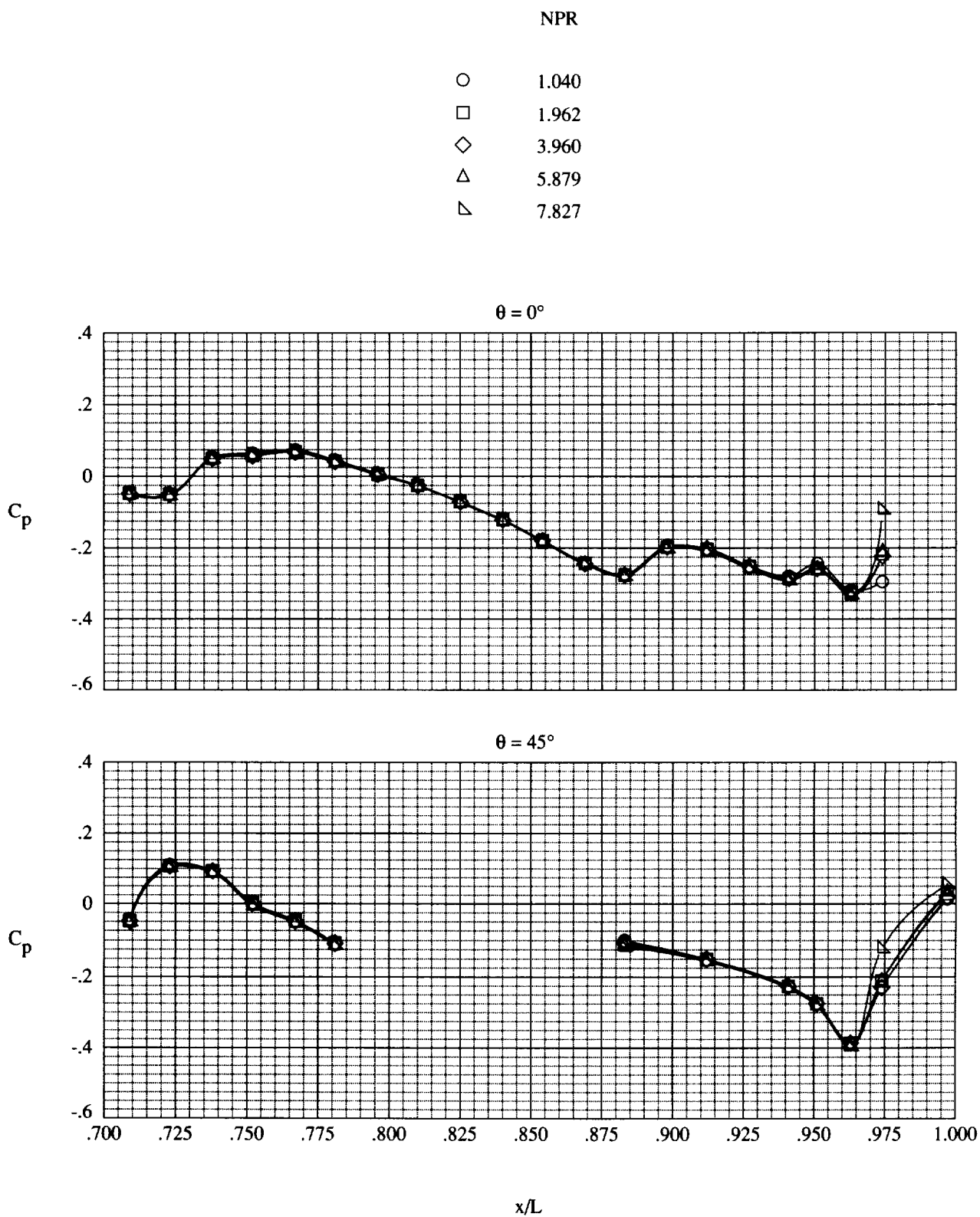
NPR

○	1.074
□	1.966
◇	3.904
△	5.884
▽	7.775



(e) Top and bottom interfairings.

Figure 15. Concluded.

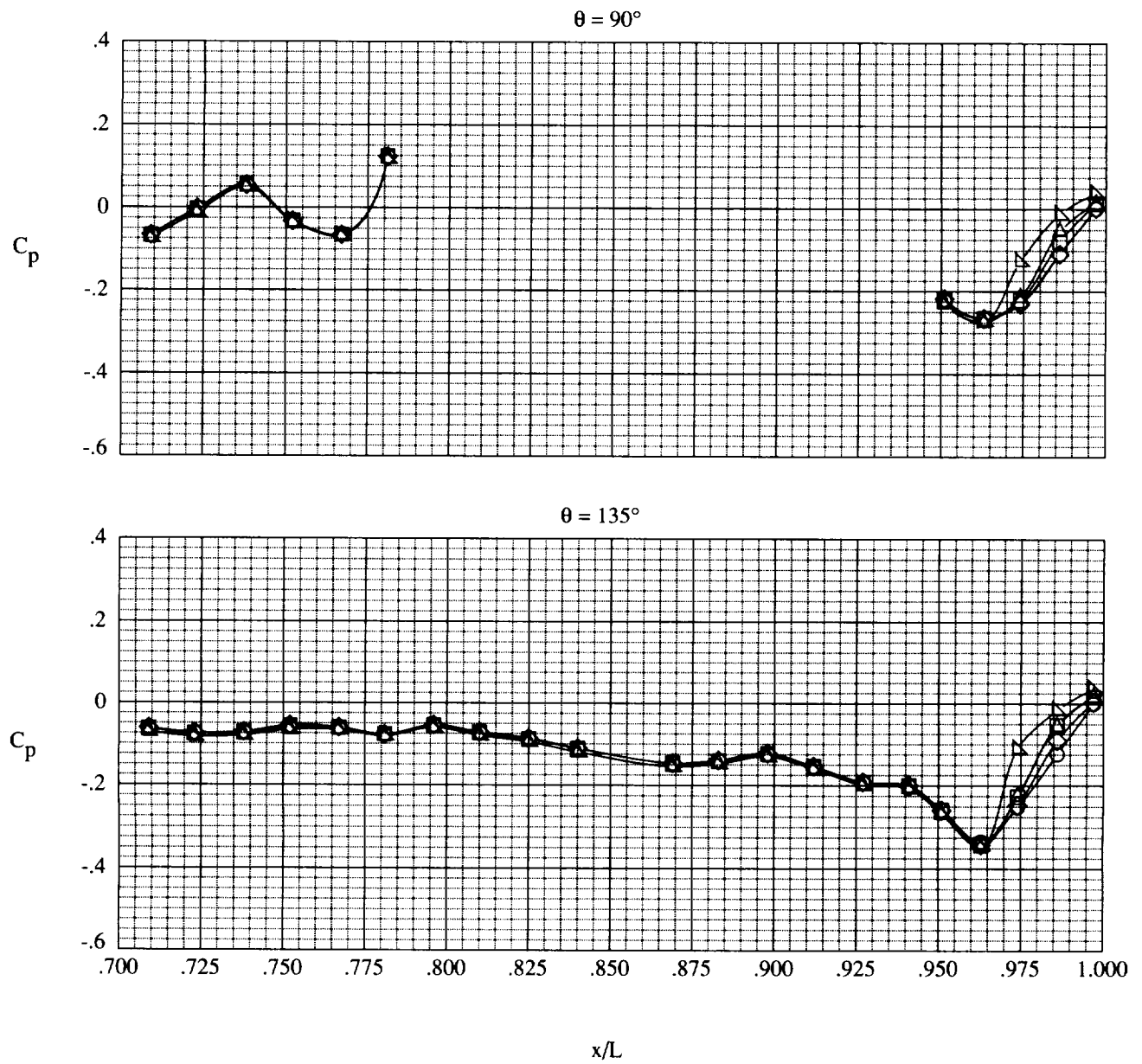


(a) $\theta = 0^\circ$ and 45° .

Figure 16. Effect of nozzle pressure ratio on pressure distributions for $M = 1.2$, $\alpha = 0^\circ$, HTAIL = mid position, and VTAIL = fwd position.

NPR

○	1.040
□	1.962
◇	3.960
△	5.879
▽	7.827

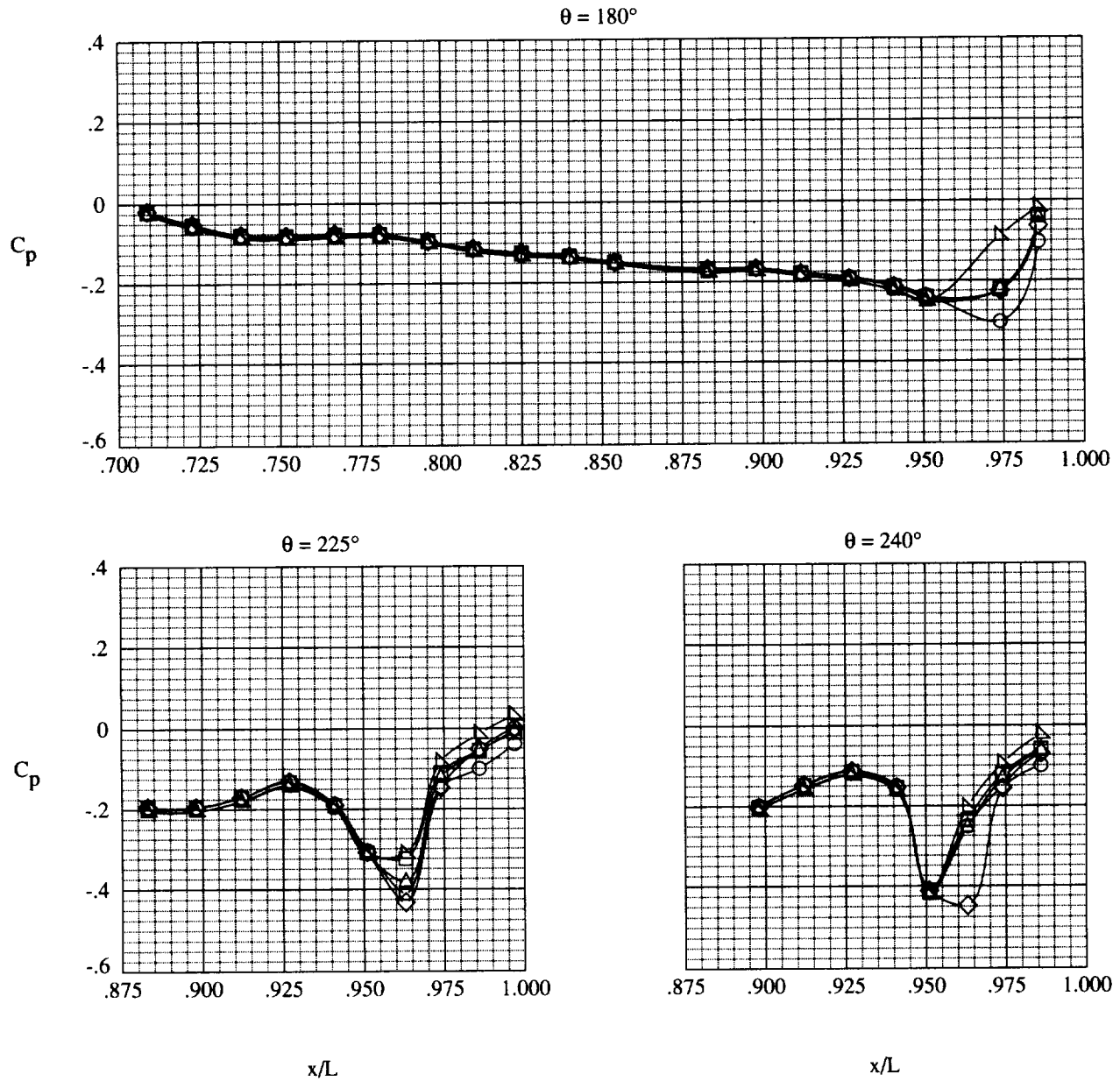


(b) $\theta = 90^\circ$ and 135° .

Figure 16. Continued.

NPR

○	1.040
□	1.962
◇	3.960
△	5.879
▴	7.827

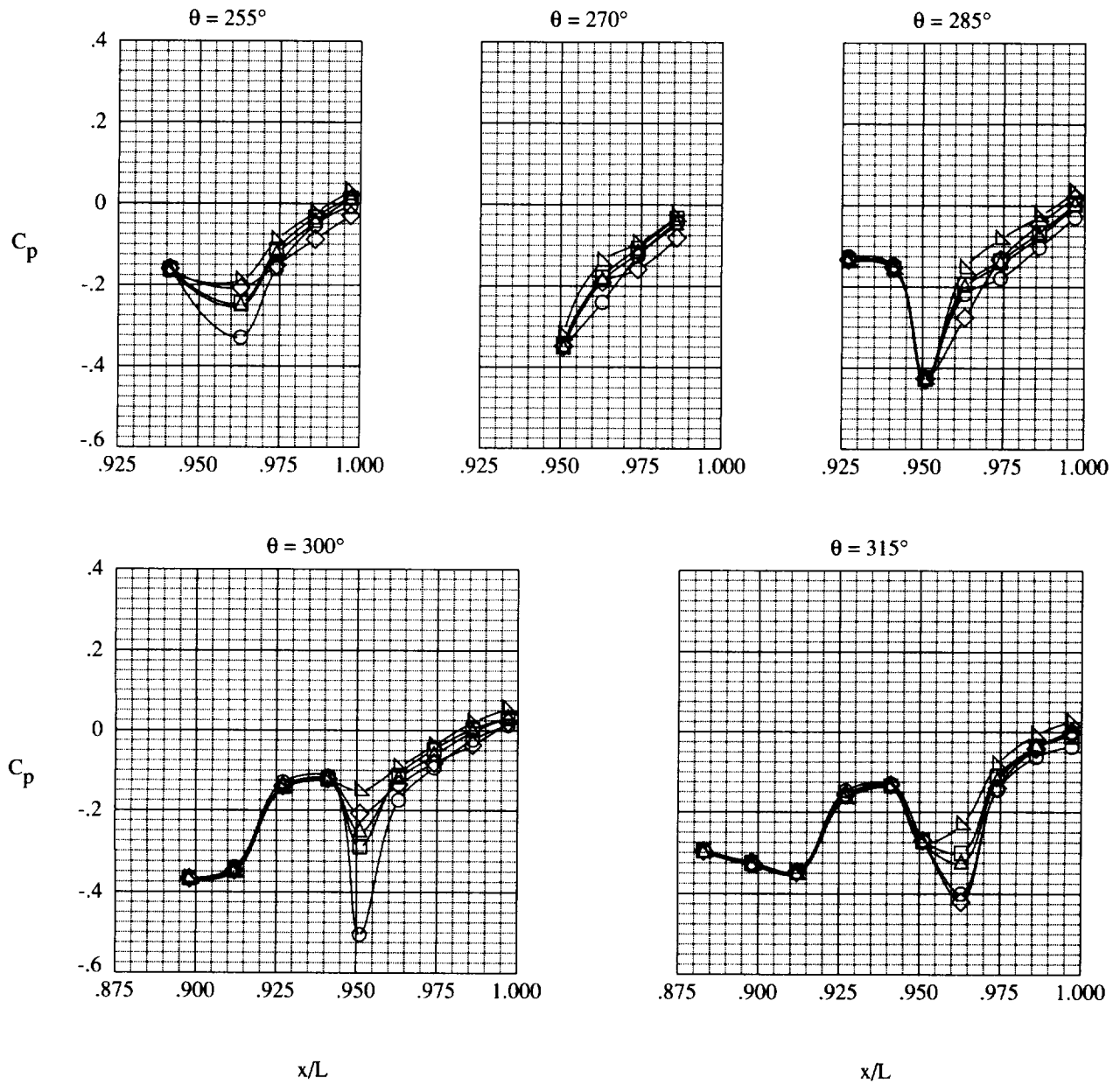


(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 16. Continued.

NPR

○	1.040
□	1.962
◇	3.960
△	5.879
▽	7.827

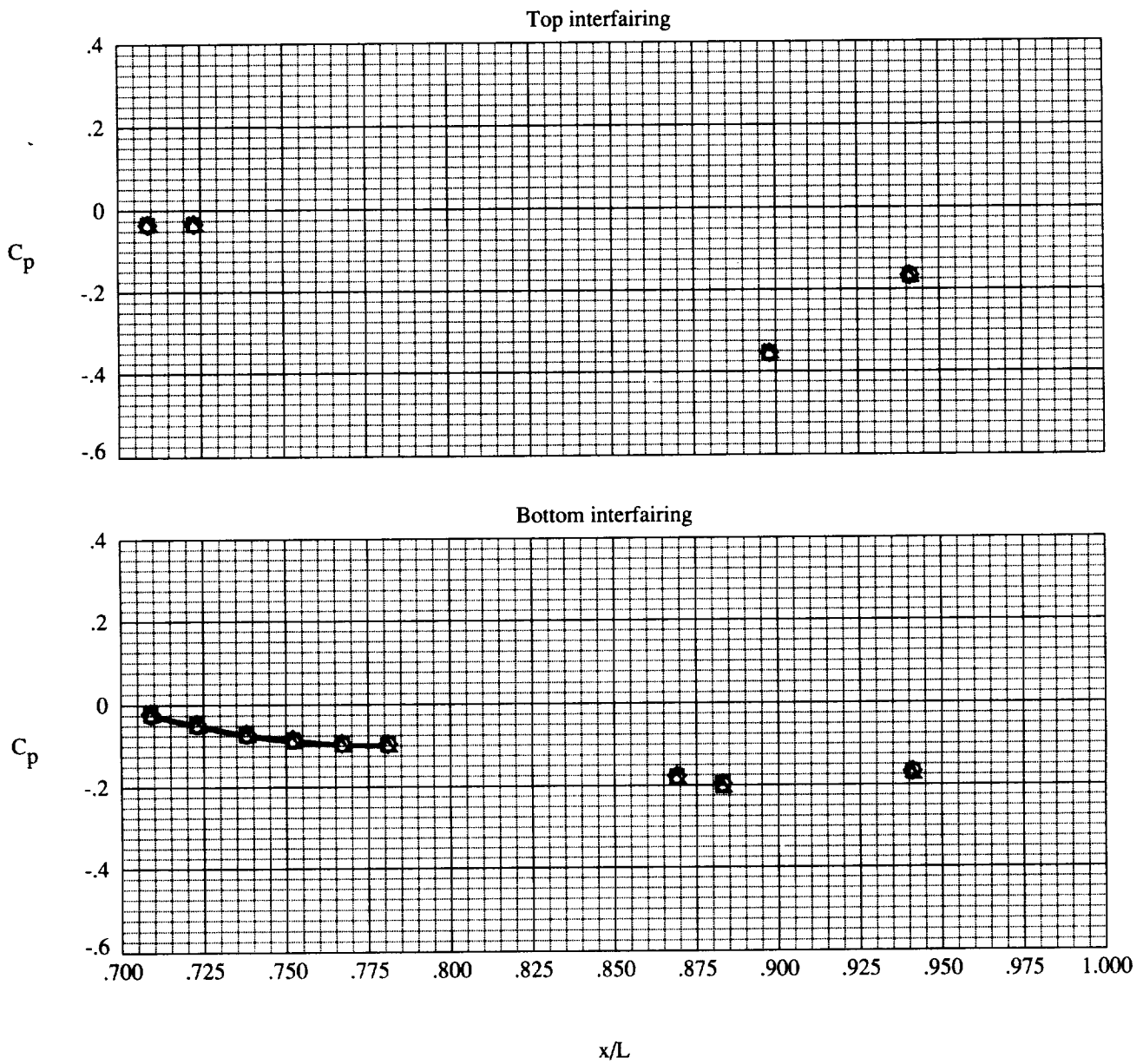


(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 16. Continued.

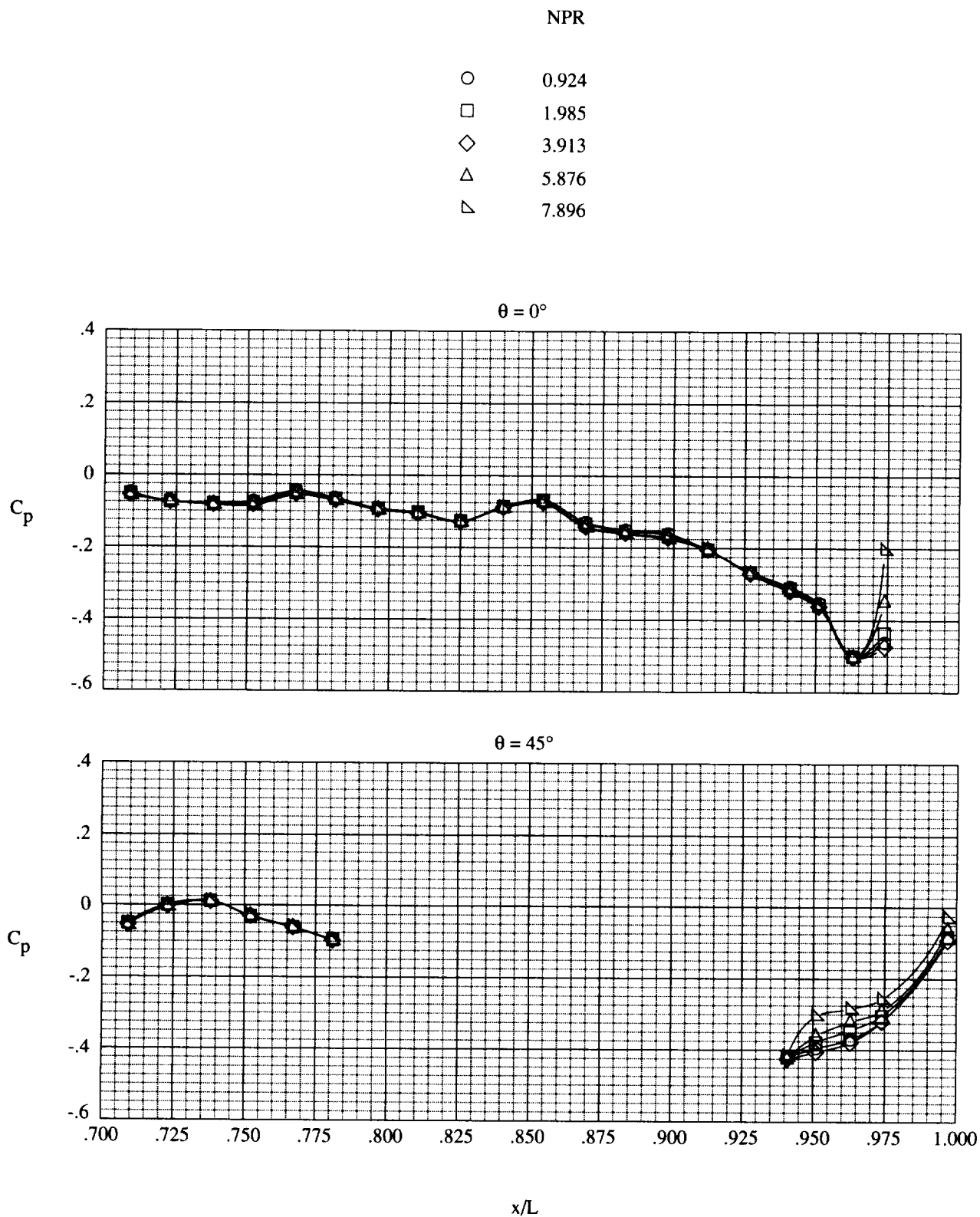
NPR

○	1.040
□	1.962
◇	3.960
△	5.879
▽	7.827



(e) Top and bottom interfairings.

Figure 16. Concluded.

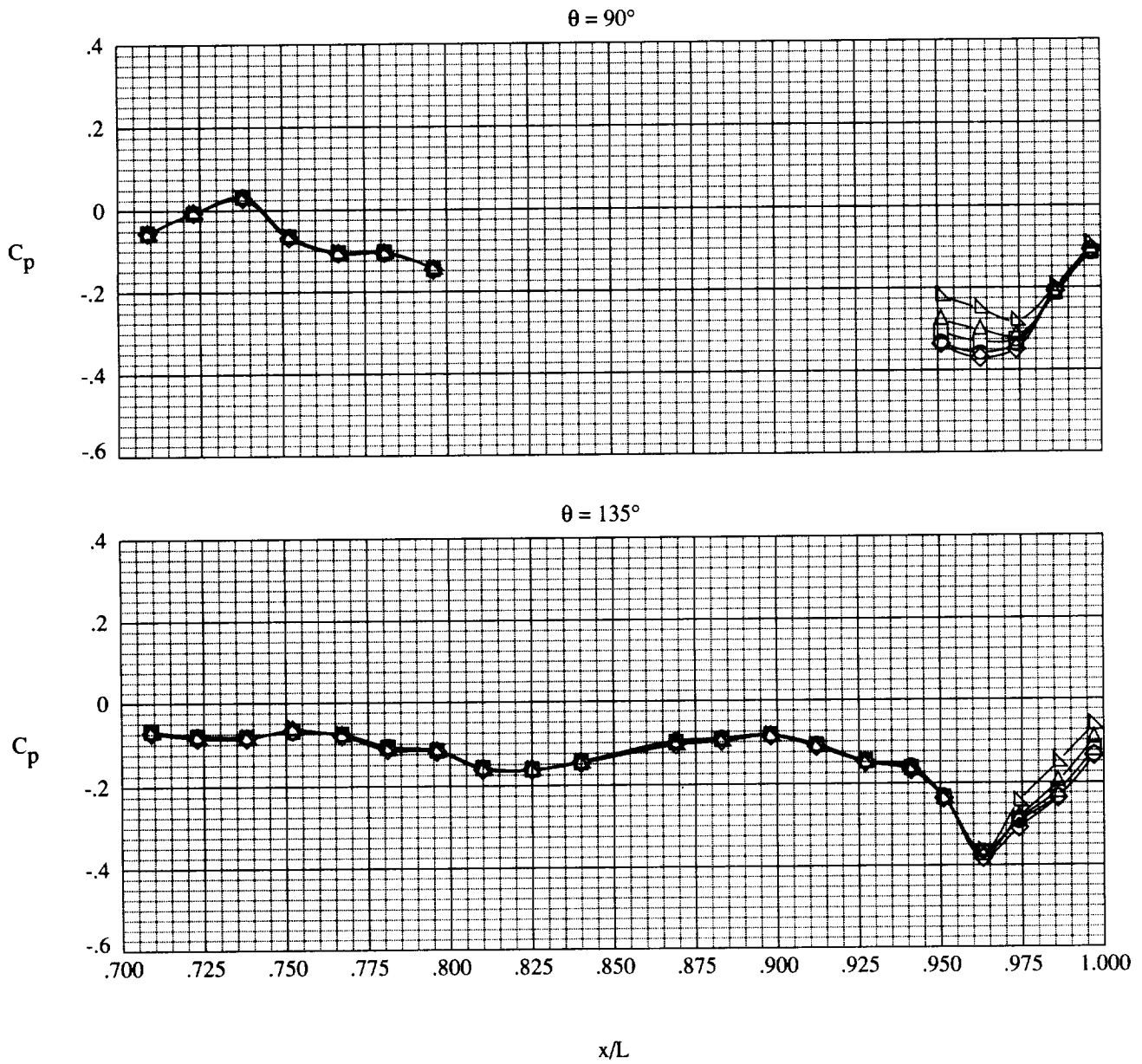


(a) $\theta = 0^\circ$ and 45° .

Figure 17. Effect of nozzle pressure ratio on pressure distributions for $M = 1.2$, $\alpha = 0^\circ$, HTAIL = aft position, and VTAIL = aft position.

NPR

○	0.924
□	1.985
◇	3.913
△	5.876
▽	7.896

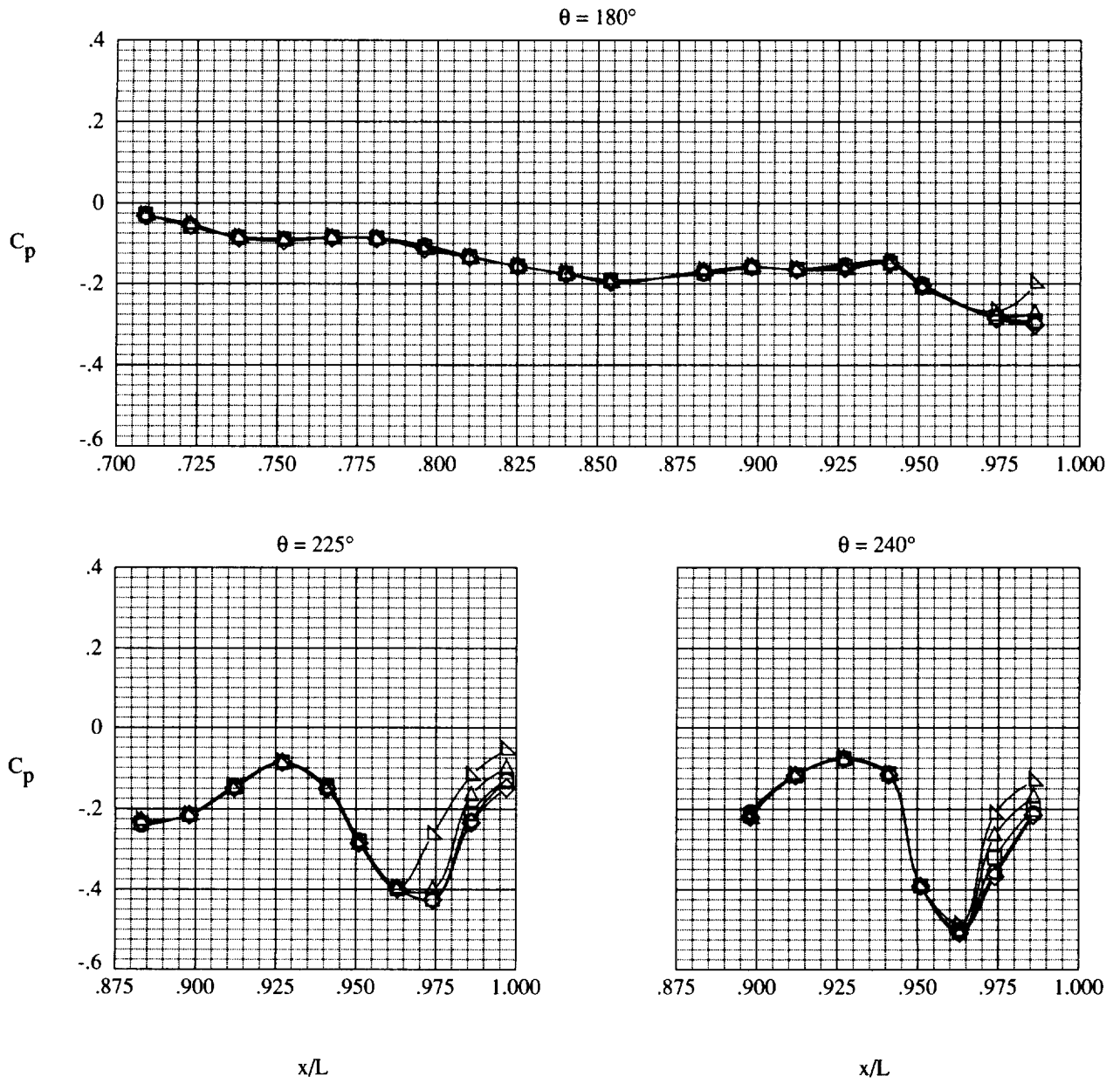


(b) $\theta = 90^\circ$ and 135° .

Figure 17. Continued.

NPR

○	0.924
□	1.985
◇	3.913
△	5.876
▽	7.896

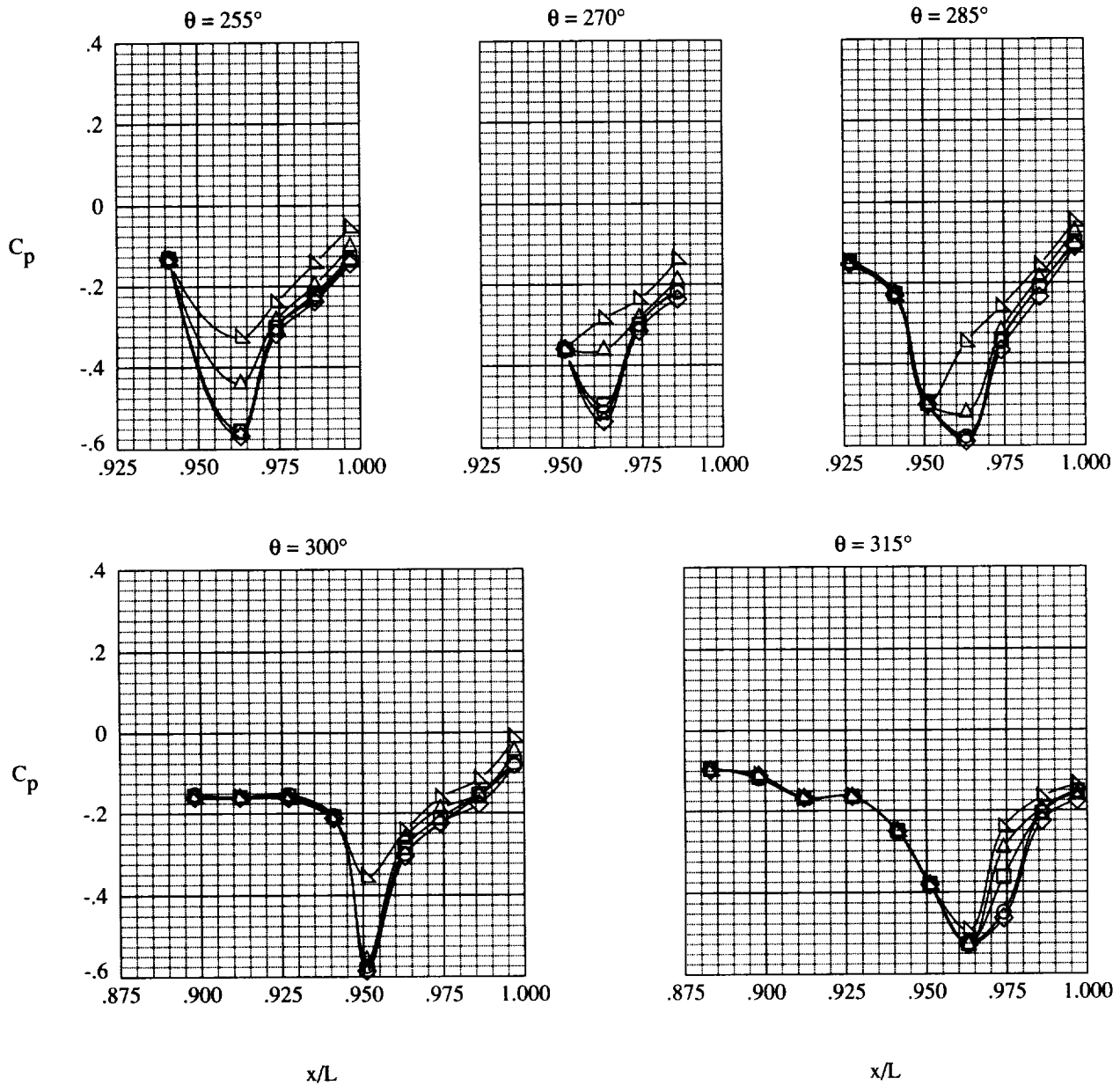


(c) $\theta = 180^\circ, 225^\circ, \text{ and } 240^\circ$.

Figure 17. Continued.

NPR

○	0.924
□	1.985
◇	3.913
△	5.876
▽	7.896

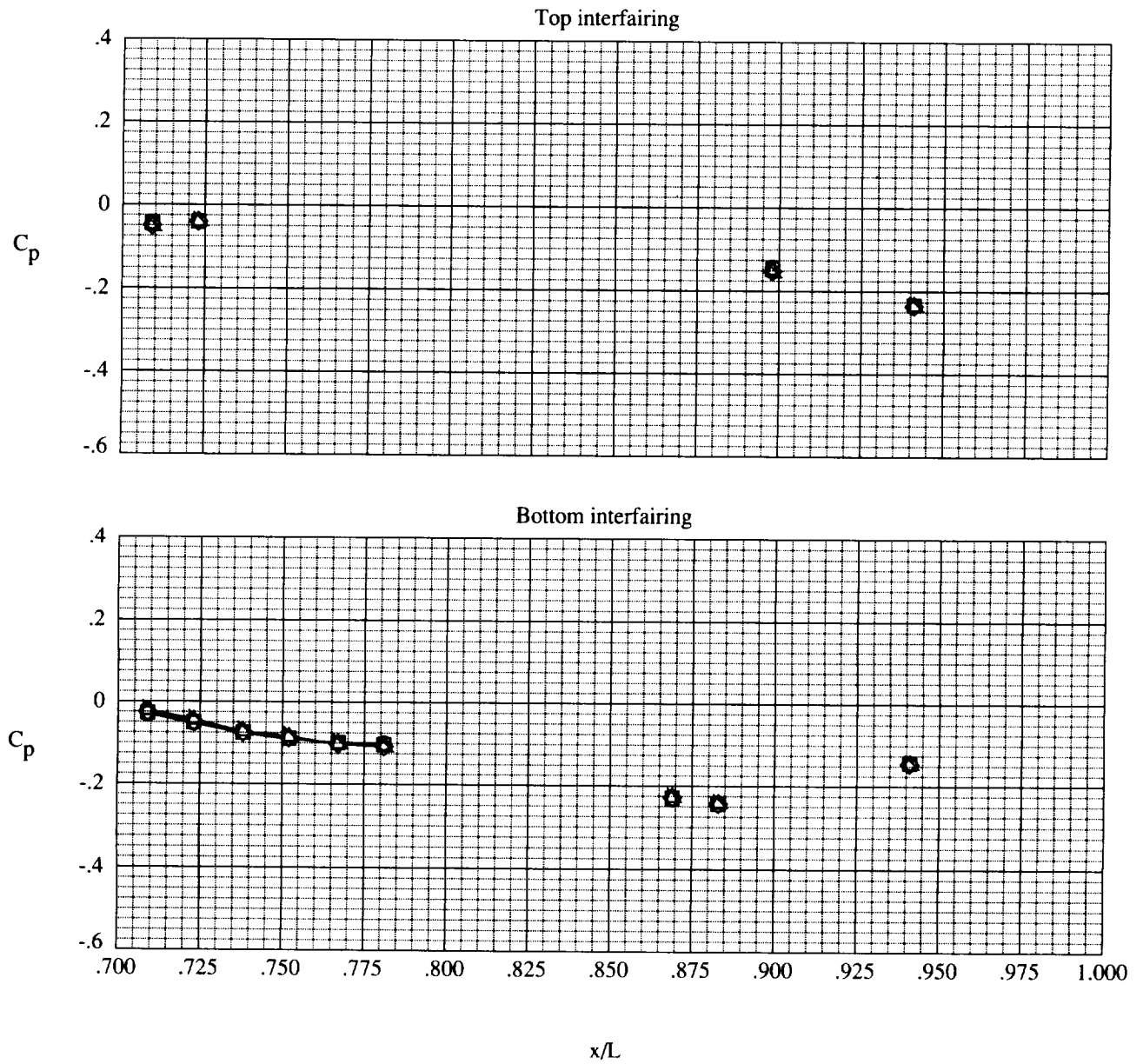


(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 17. Continued.

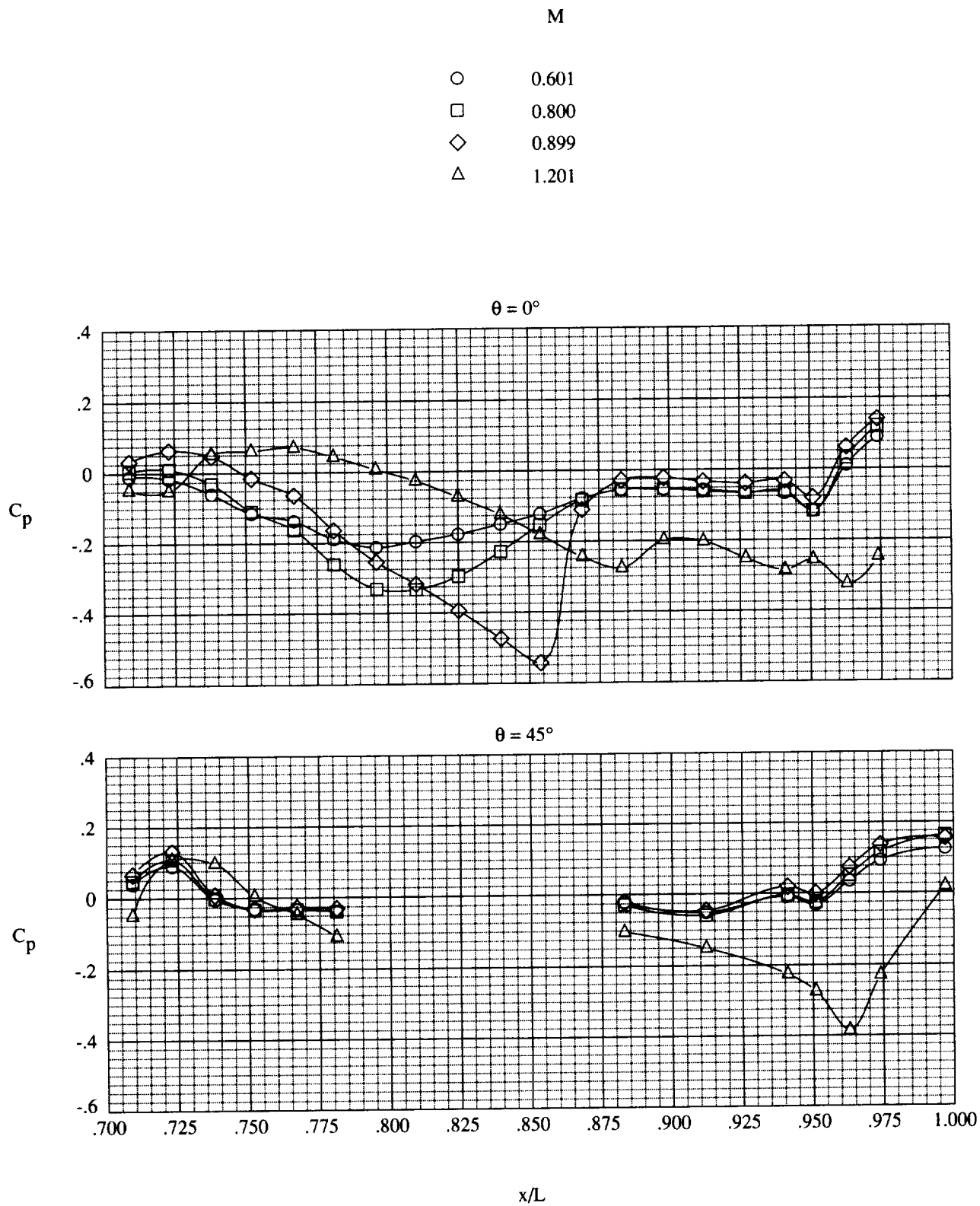
NPR

○	0.924
□	1.985
◇	3.913
△	5.876
▽	7.896



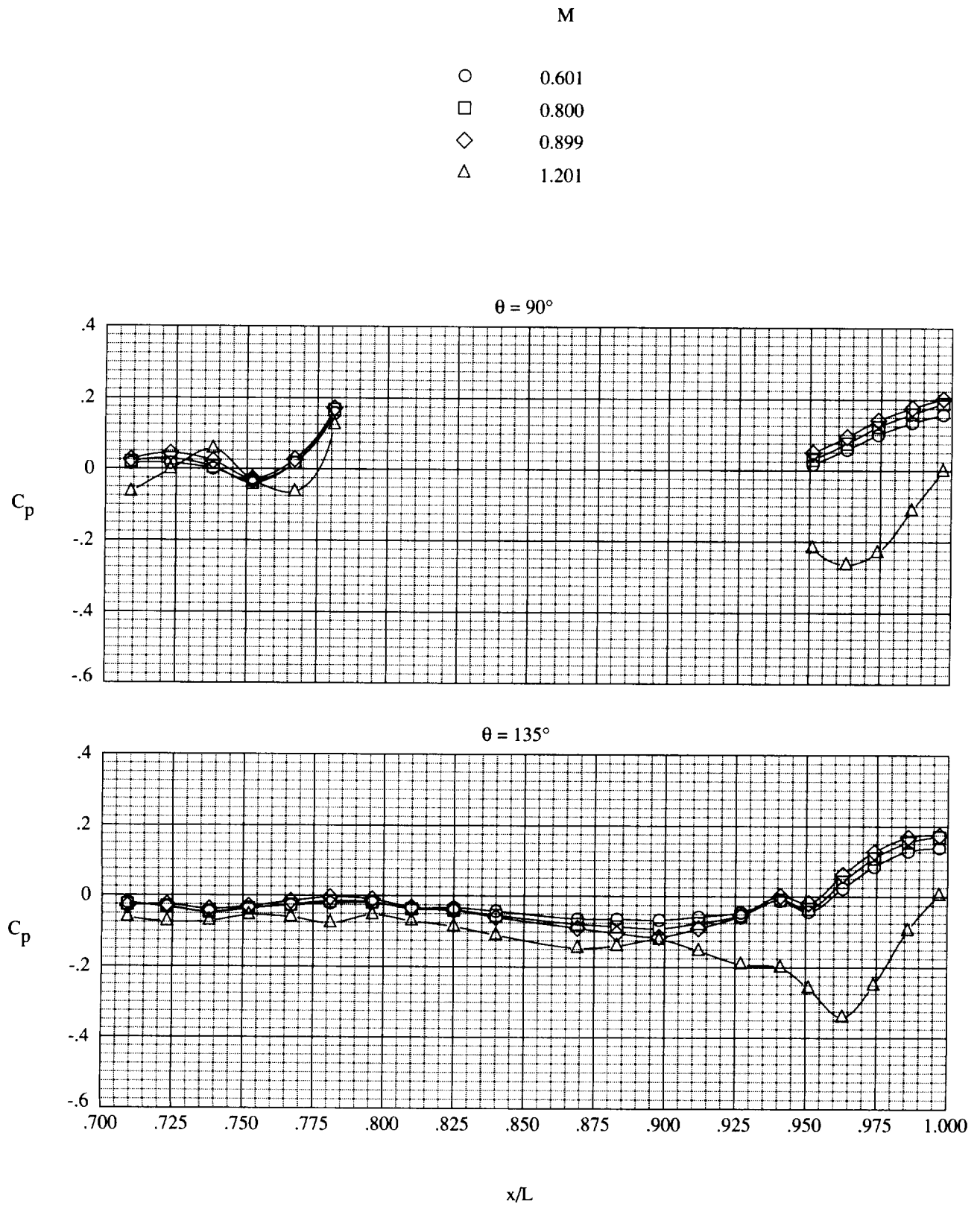
(e) Top and bottom interfairings.

Figure 17. Concluded.



(a) $\theta = 0^\circ$ and 45° .

Figure 18. Effect of Mach number on pressure distributions for $\alpha = 0^\circ$, NPR = 3.4, HTAIL = mid position, and VTAIL = fwd position.

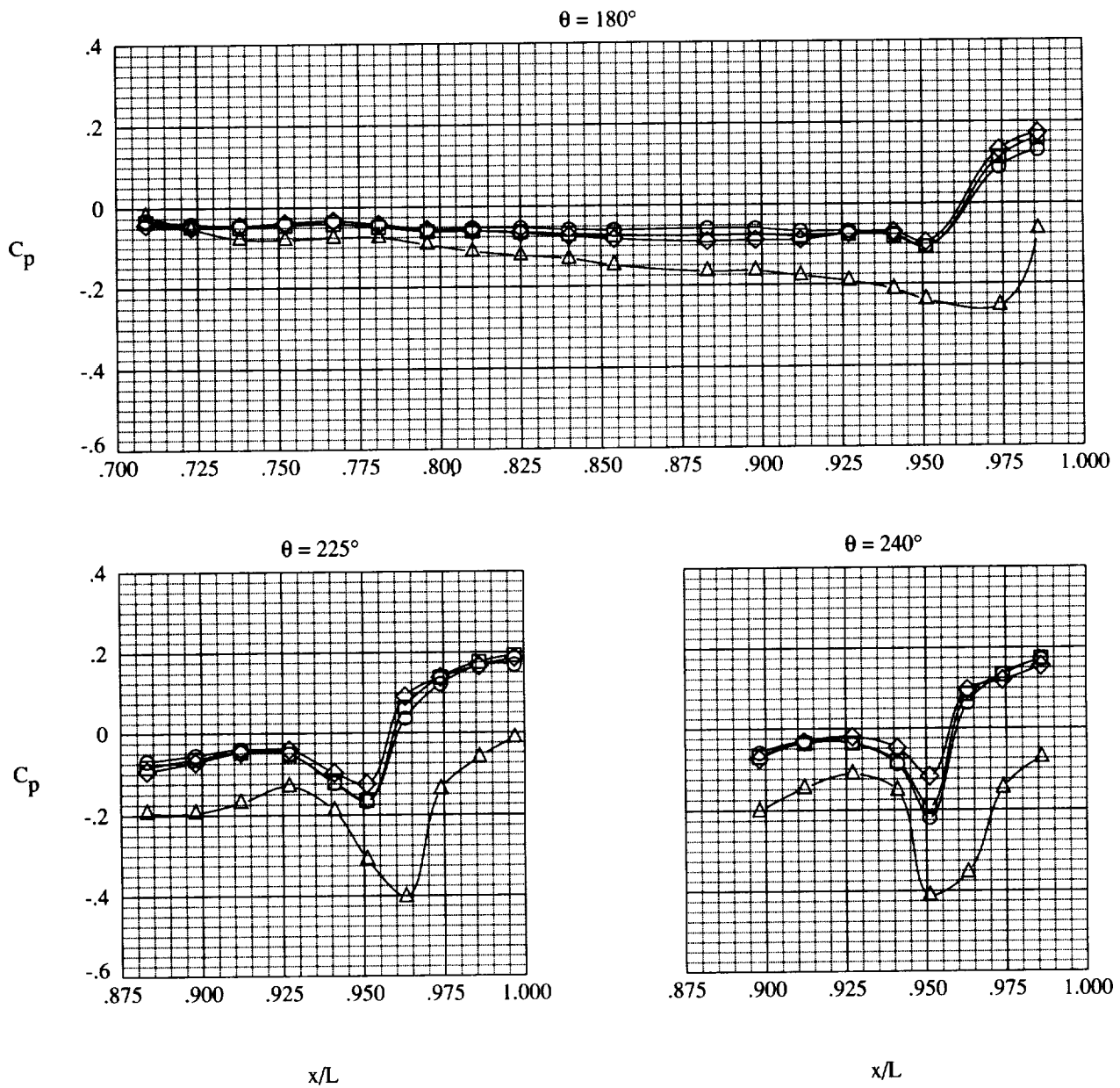


(b) $\theta = 90^\circ$ and 135° .

Figure 18. Continued.

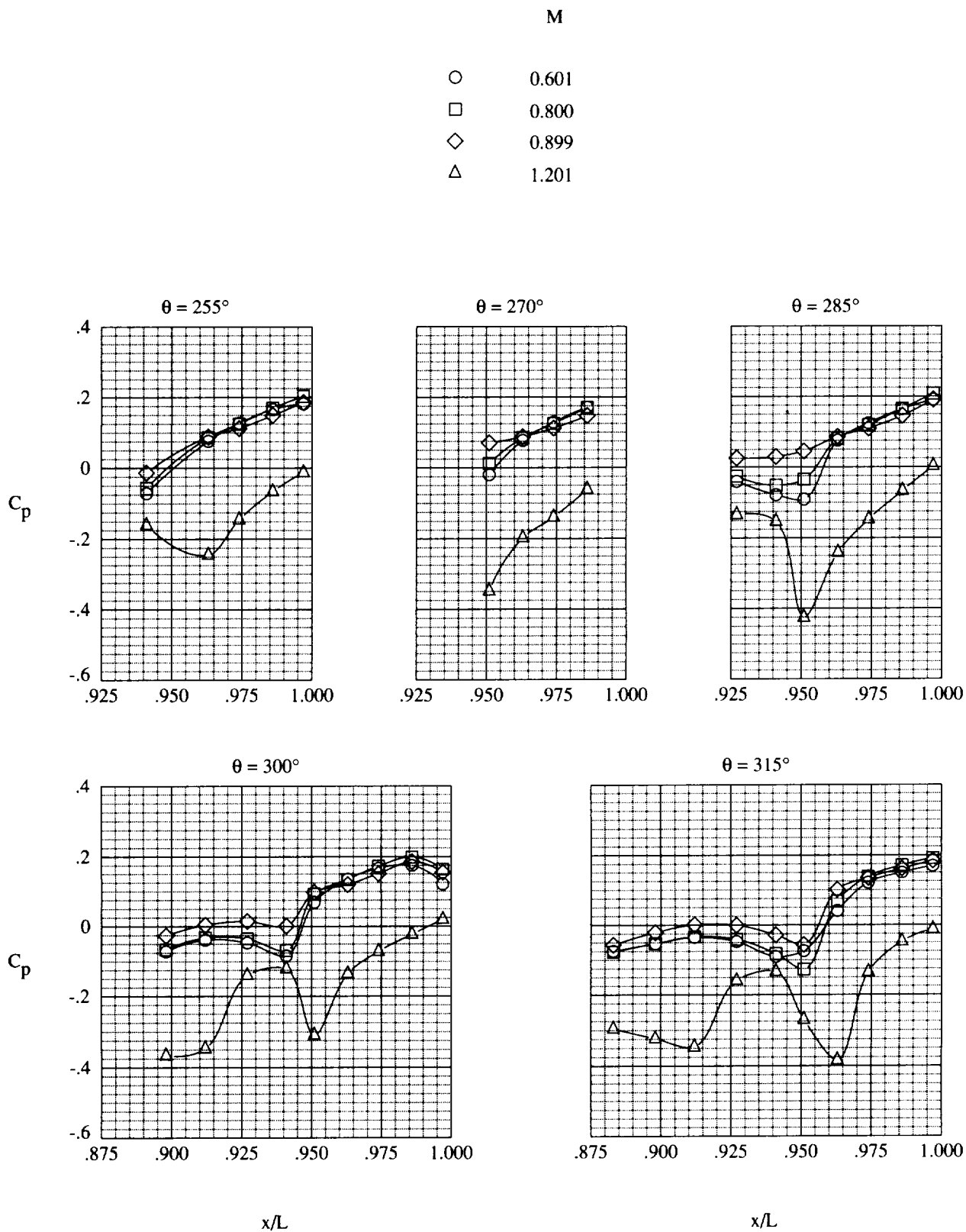
M

○	0.601
□	0.800
◇	0.899
△	1.201



(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 18. Continued.

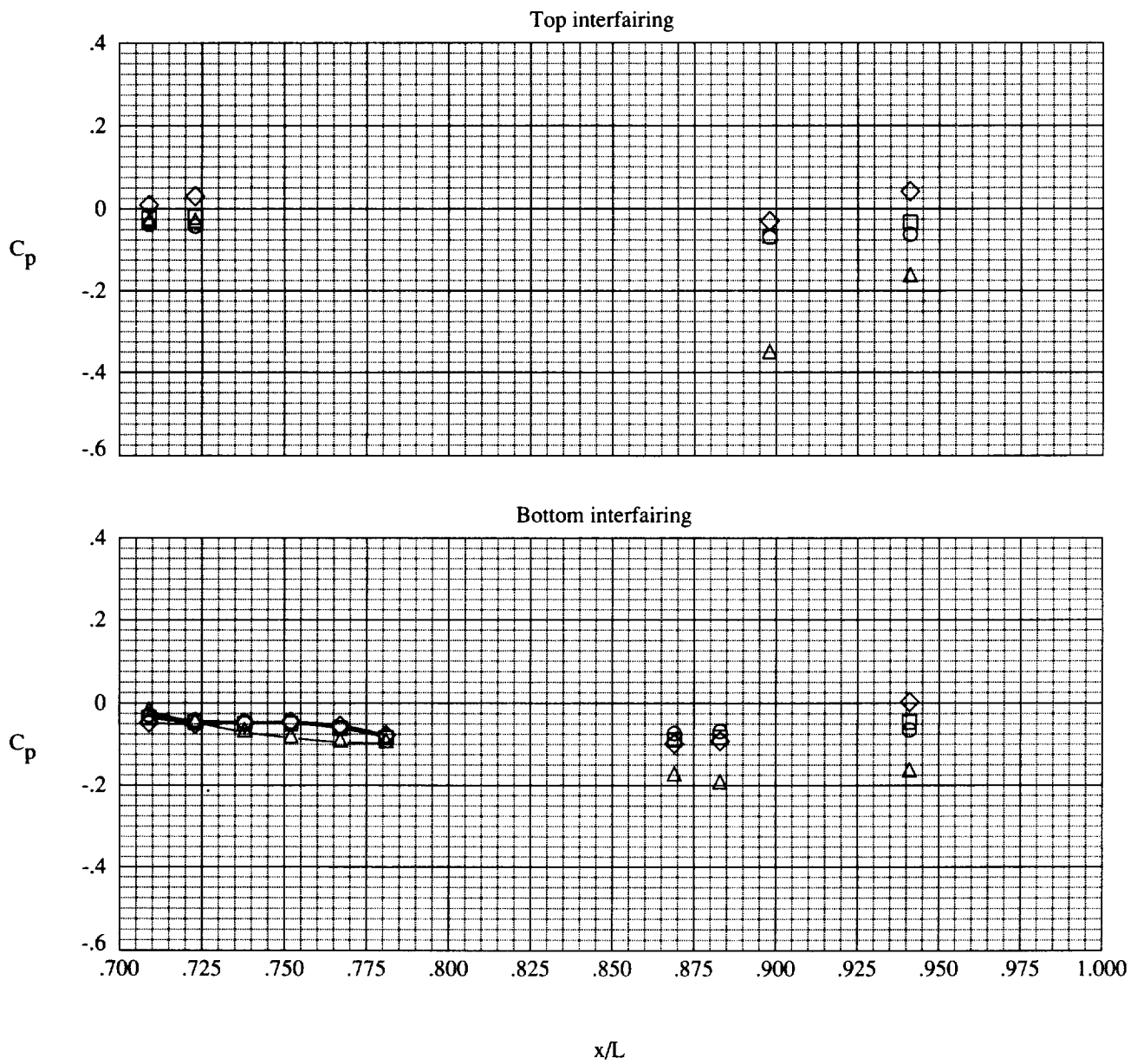


(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 18. Continued.

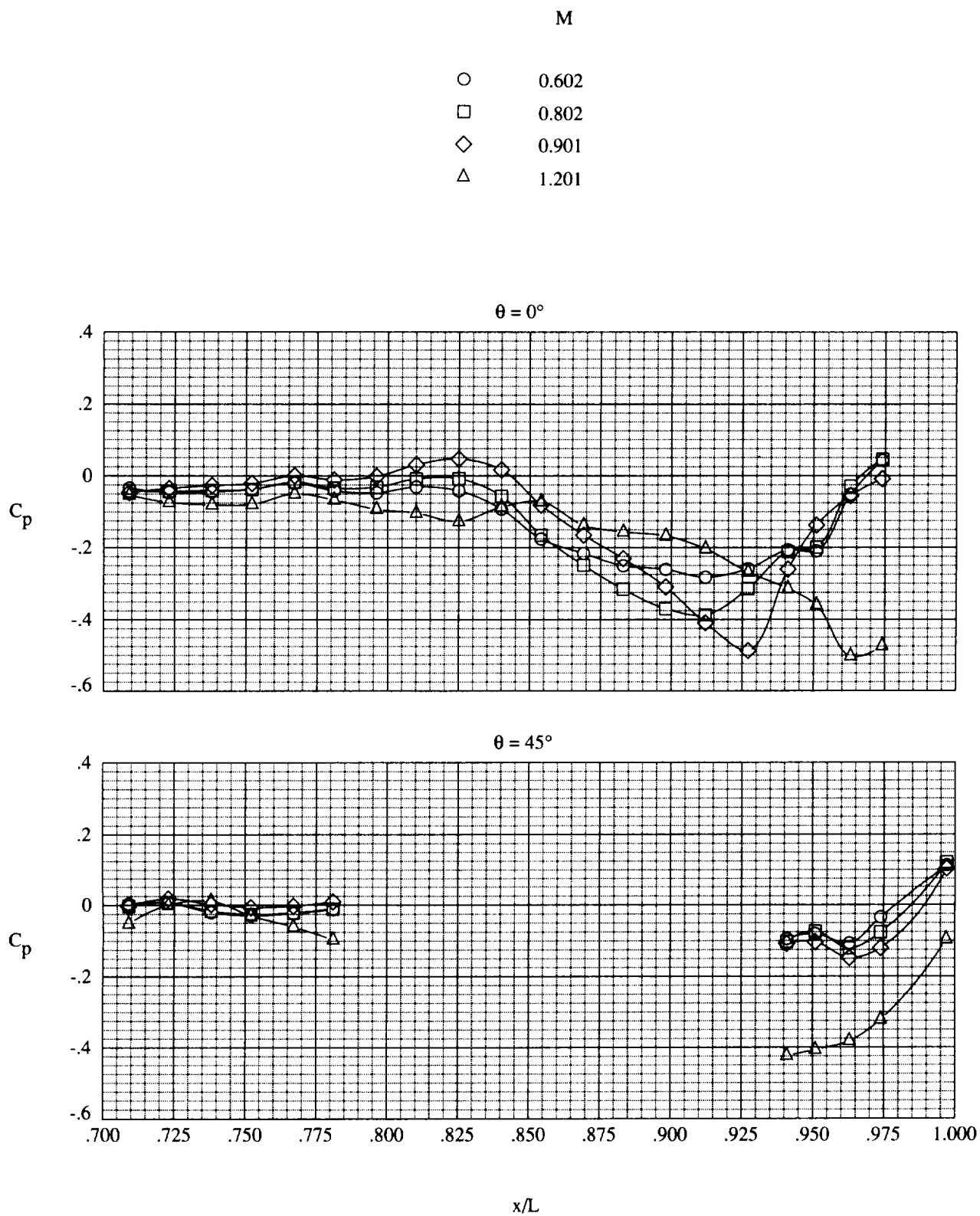
M

○	0.601
□	0.800
◇	0.899
△	1.201



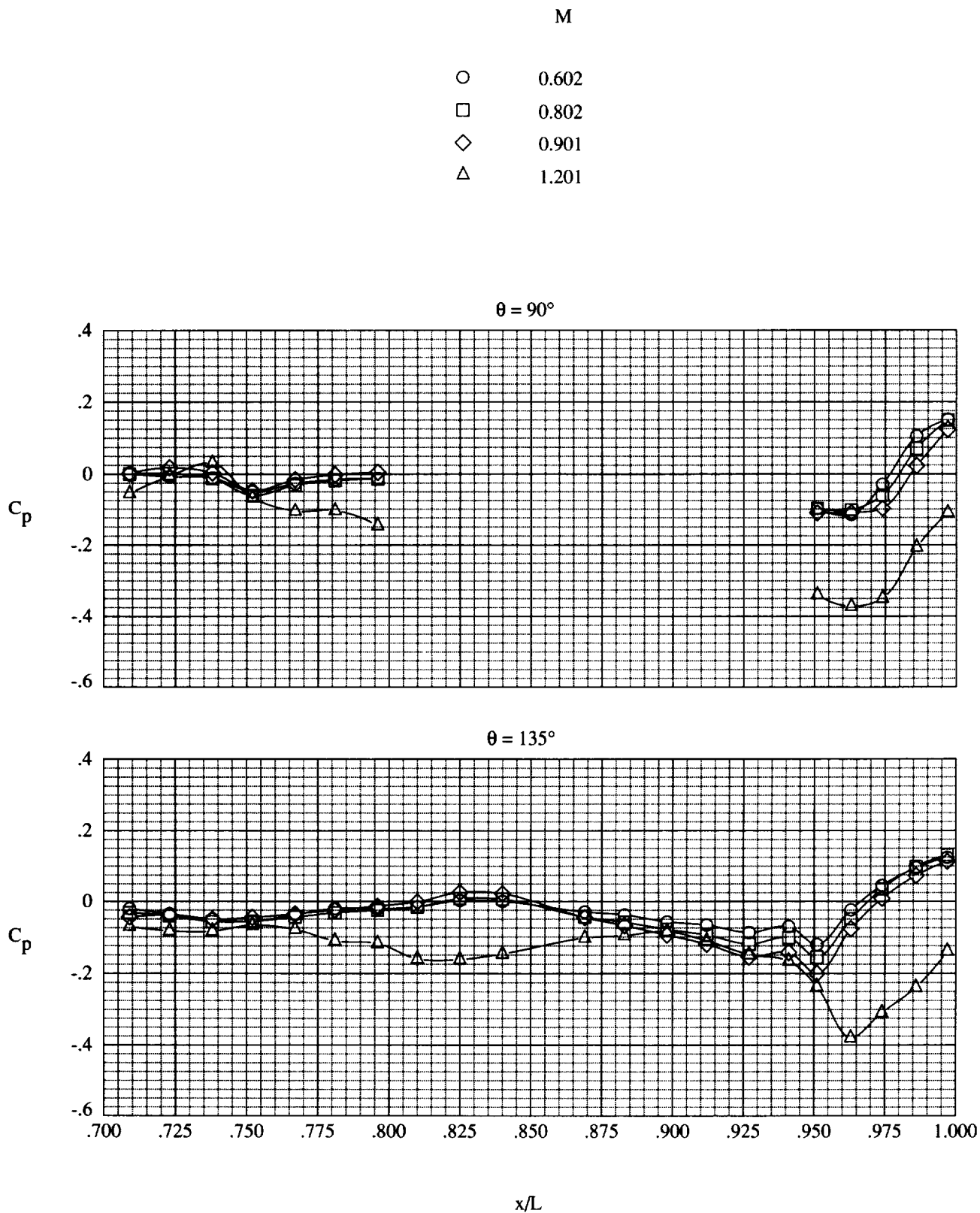
(e) Top and bottom interfairings.

Figure 18. Concluded.



(a) $\theta = 0^\circ$ and 45° .

Figure 19. Effect of Mach number on pressure distributions for $\alpha = 0^\circ$, NPR = 3.4, HTAIL = aft position, and VTAIL = aft position.

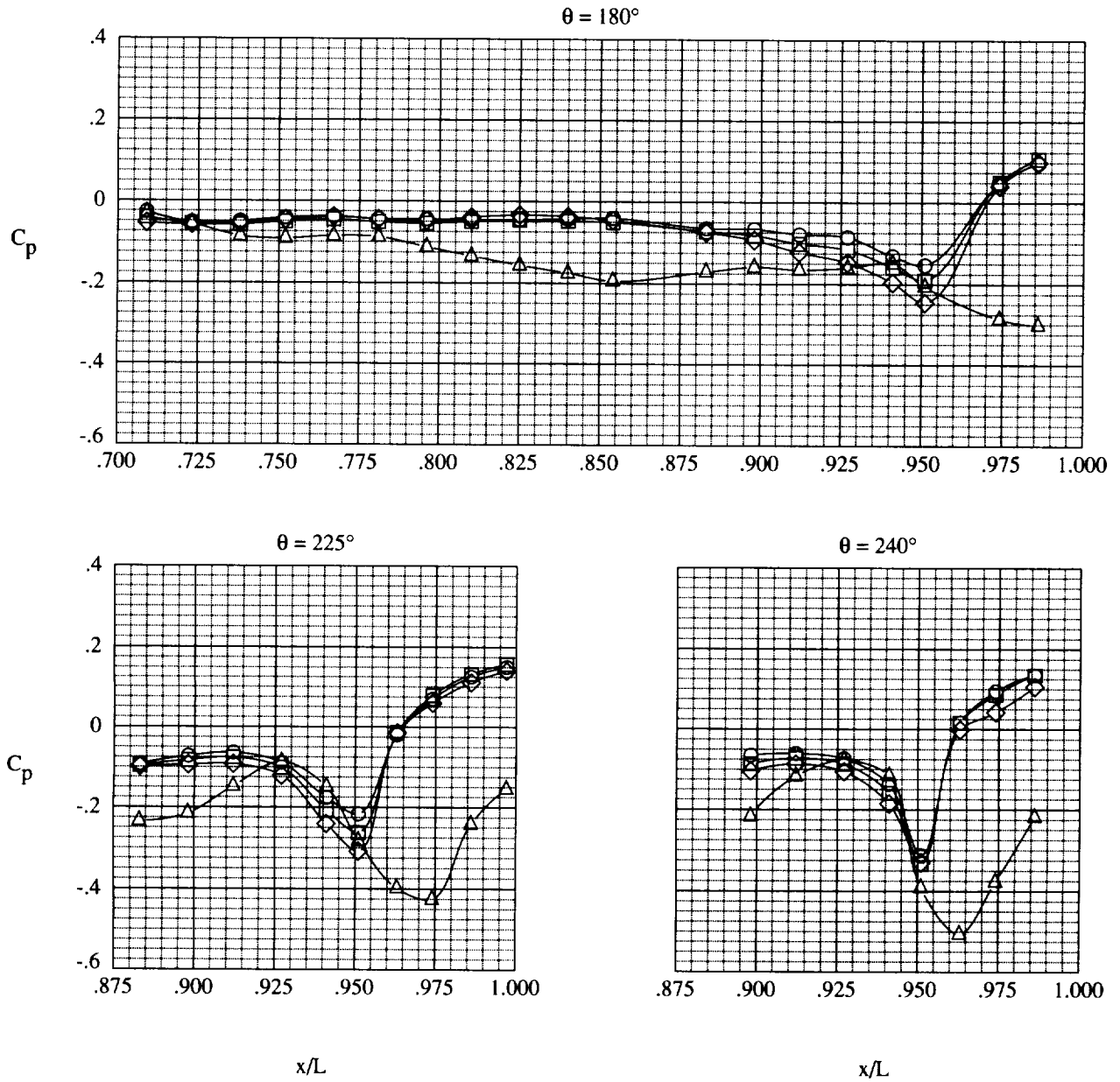


(b) $\theta = 90^\circ$ and 135° .

Figure 19. Continued.

M

○	0.602
□	0.802
◇	0.901
△	1.201

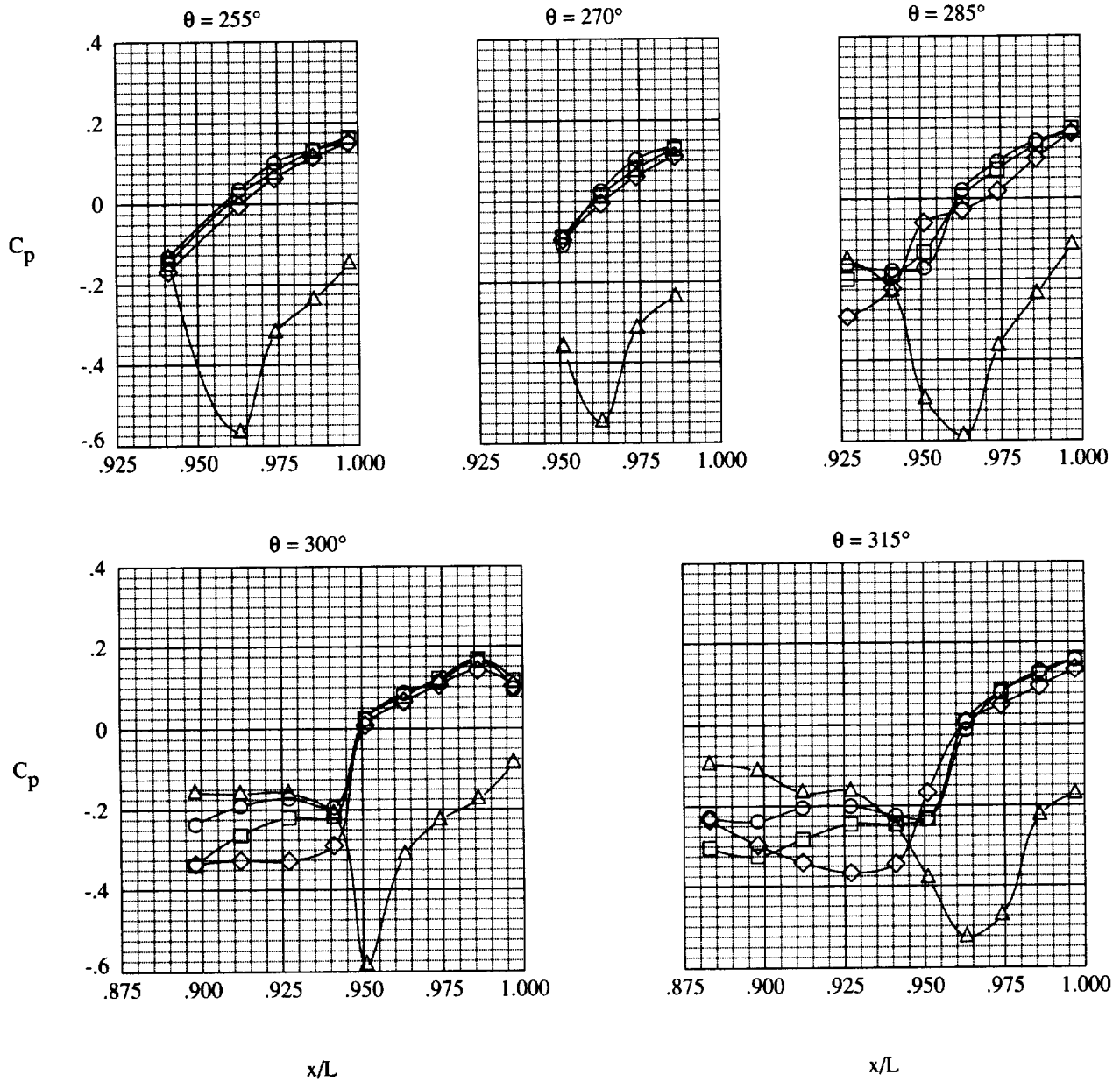


(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 19. Continued.

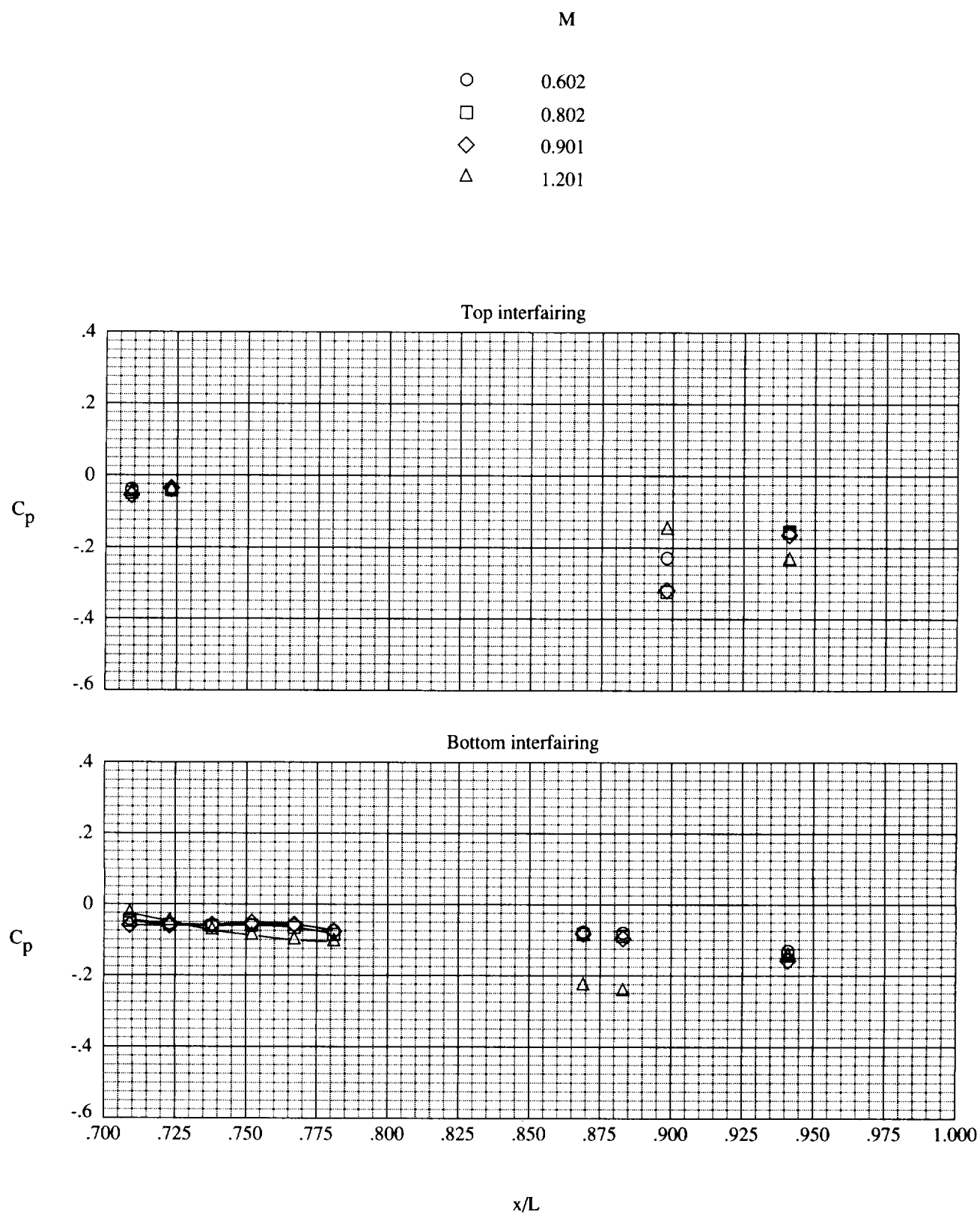
M

○	0.602
□	0.802
◇	0.901
△	1.201



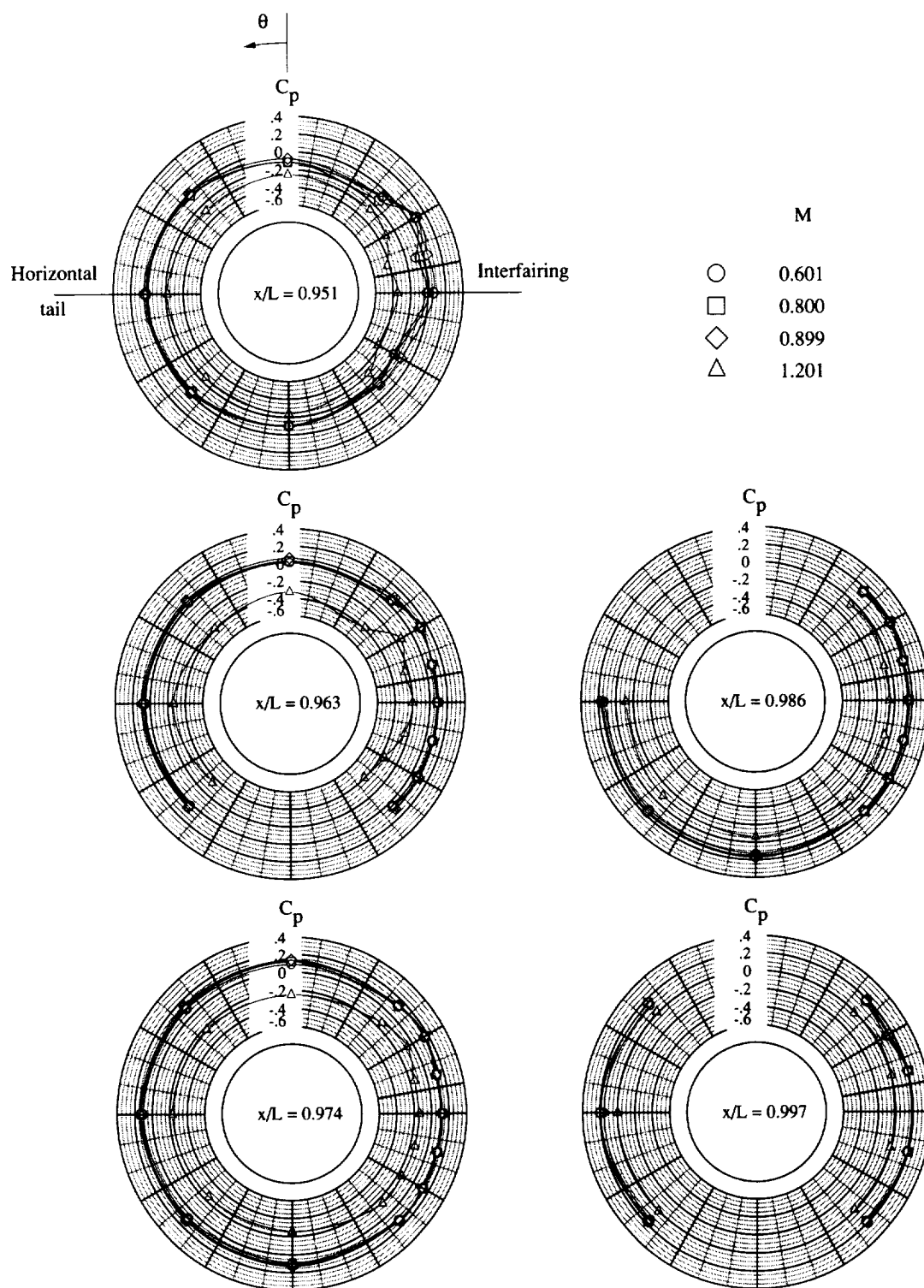
(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 19. Continued.



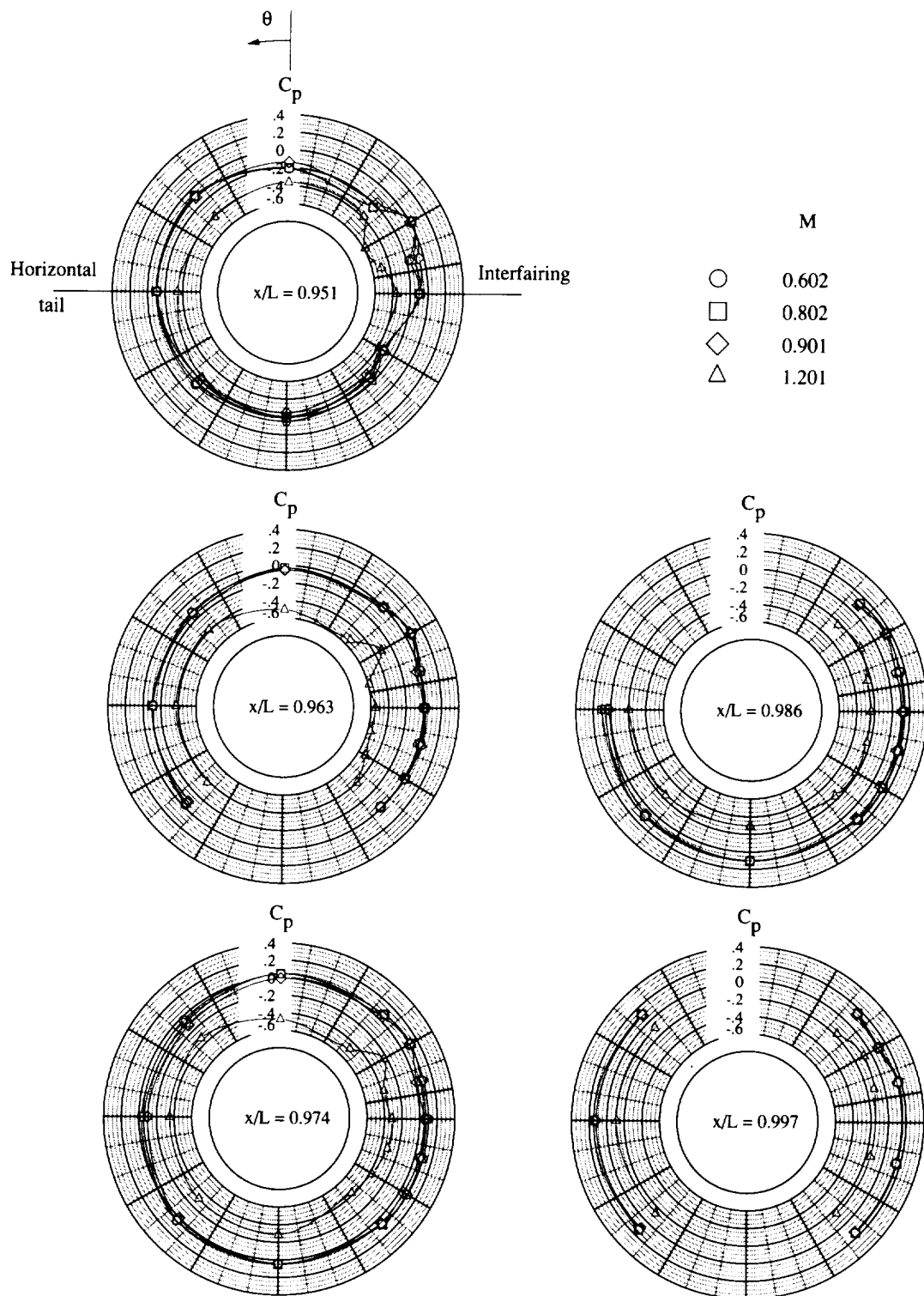
(e) Top and bottom interfairings.

Figure 19. Concluded.



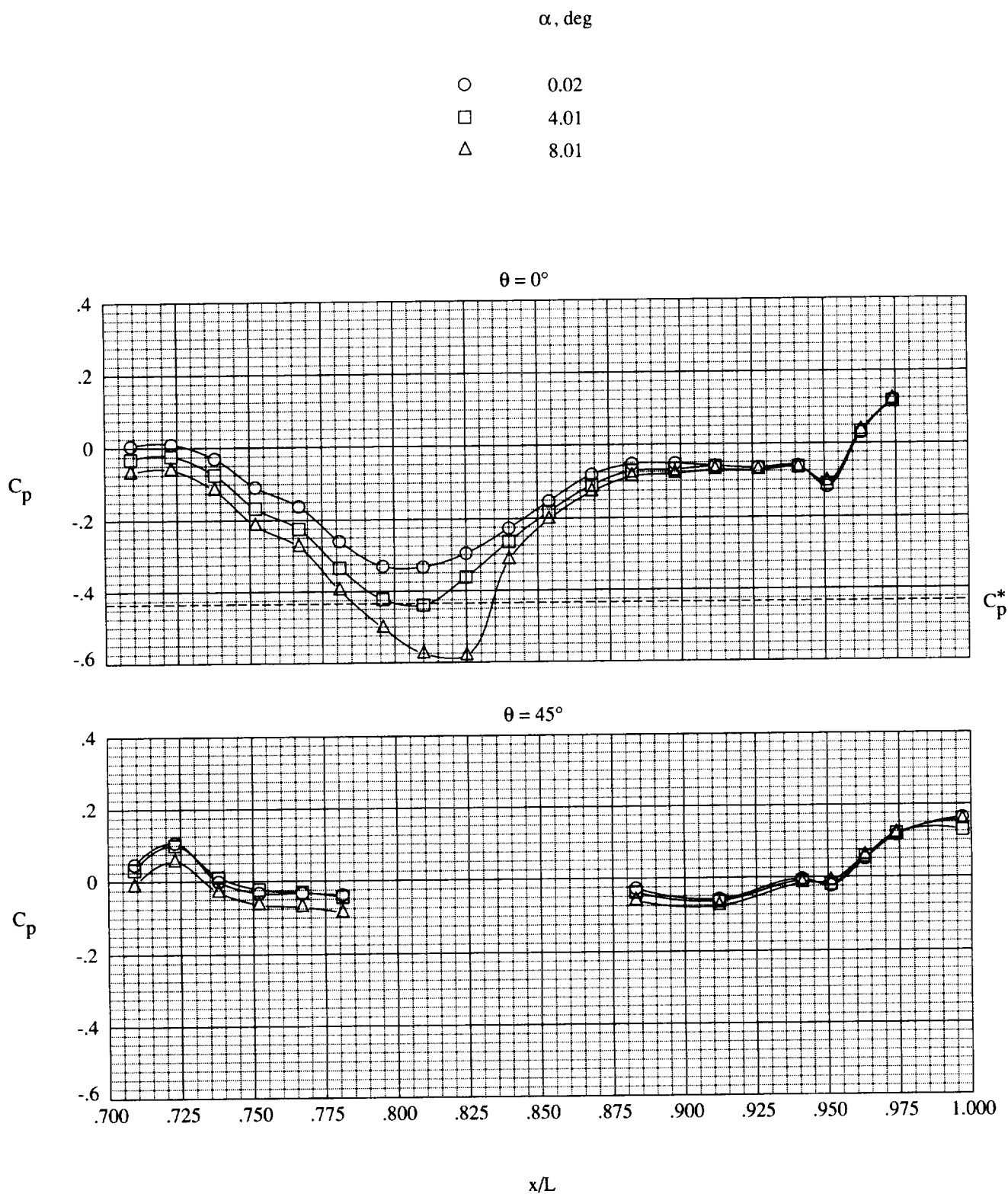
(a) HTAIL = mid position; VTAIL = fwd position.

Figure 20. Effect of Mach number on static pressure distribution around left nozzle for $\alpha = 0^\circ$ and NPR = 3.4.



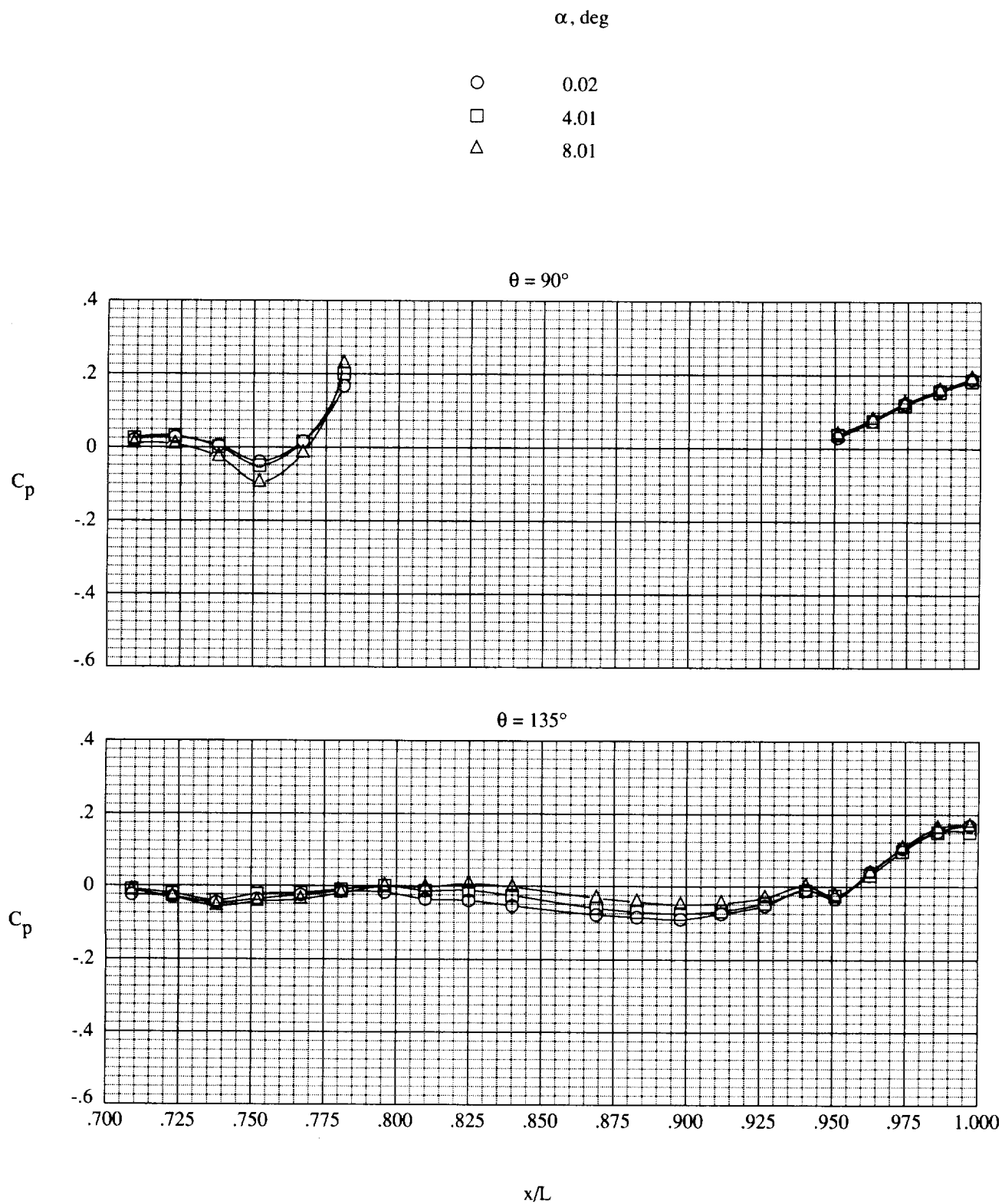
(b) HTAIL = aft position; VTAIL = aft position.

Figure 20. Concluded.



(a) $\theta = 0^\circ$ and 45° .

Figure 21. Effect of angle of attack on pressure distributions for $M = 0.8$, $NPR = 3.4$, HTAIL = mid position, and VTAIL = fwd position.

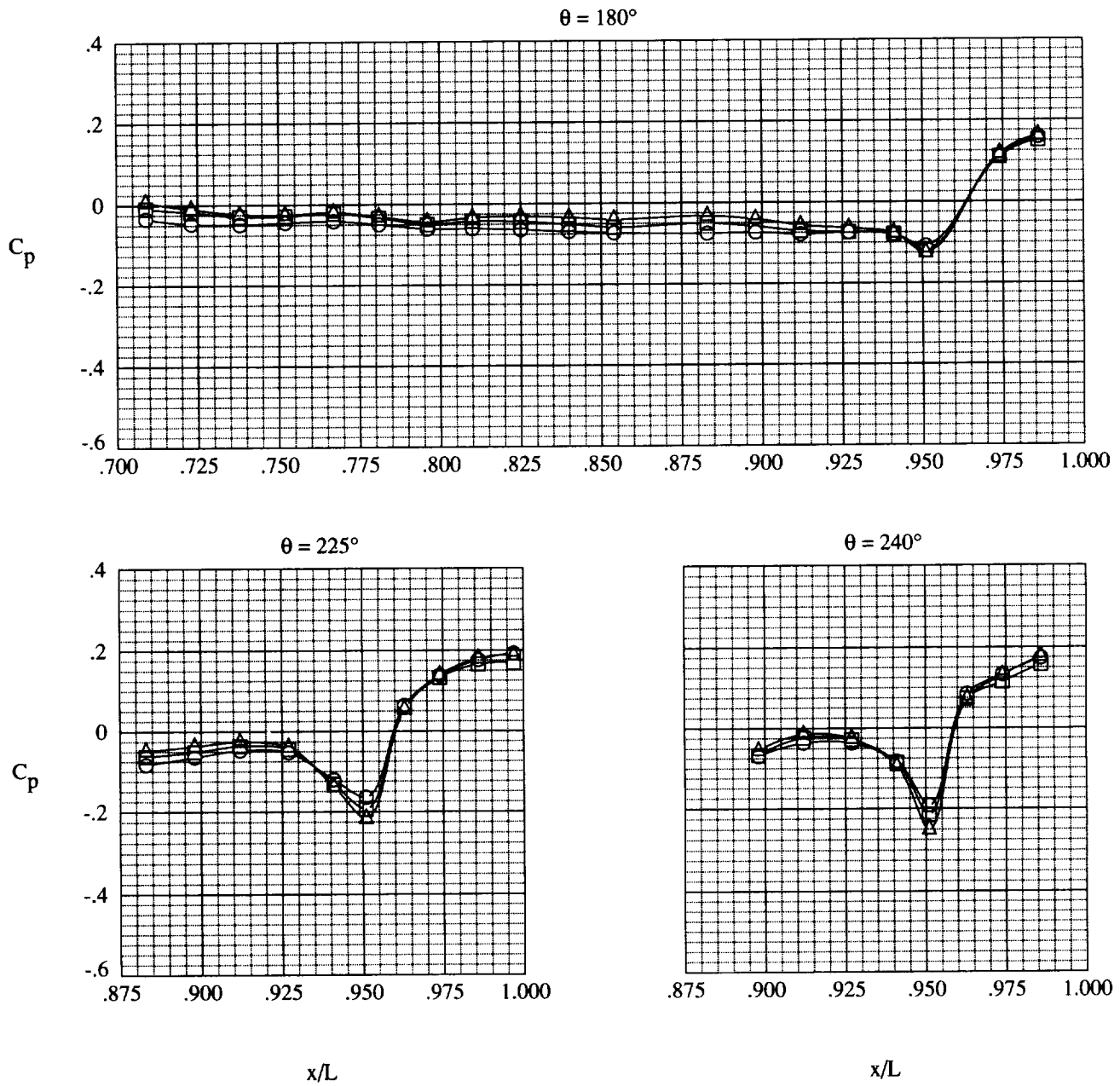


(b) $\theta = 90^\circ$ and 135° .

Figure 21. Continued.

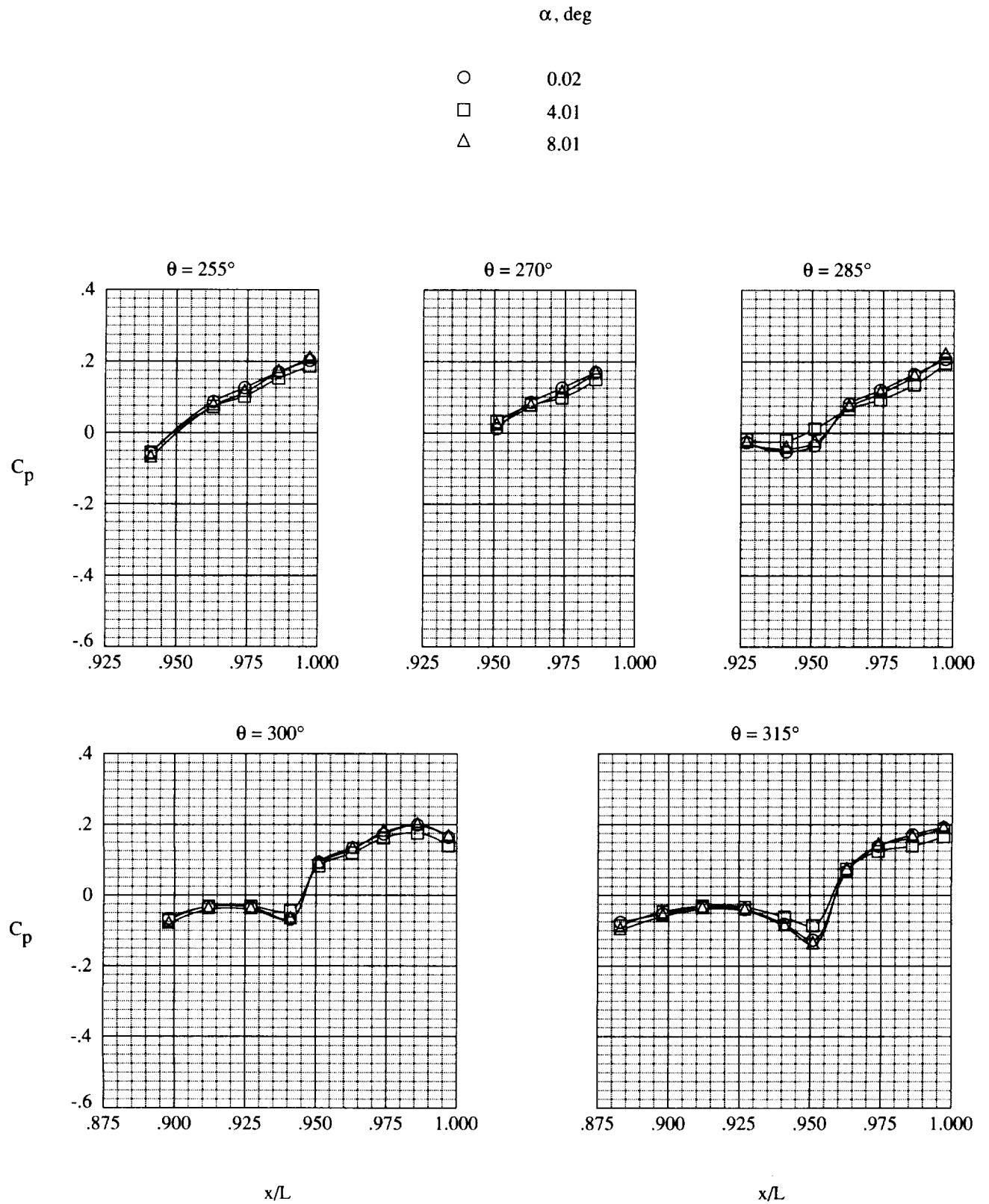
α , deg

- 0.02
- 4.01
- △ 8.01



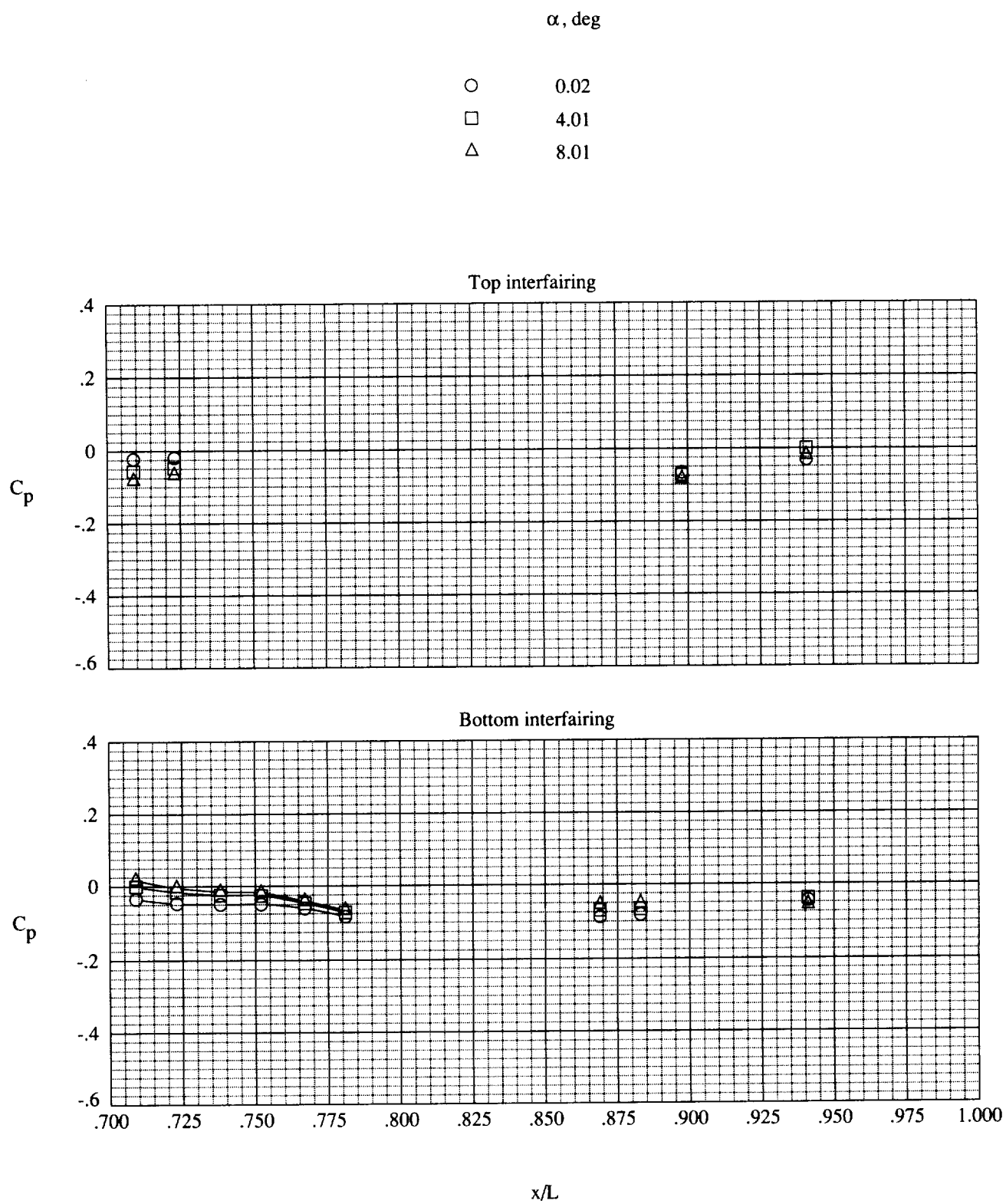
(c) $\theta = 180^\circ$, 225° , and 240° .

Figure 21. Continued.



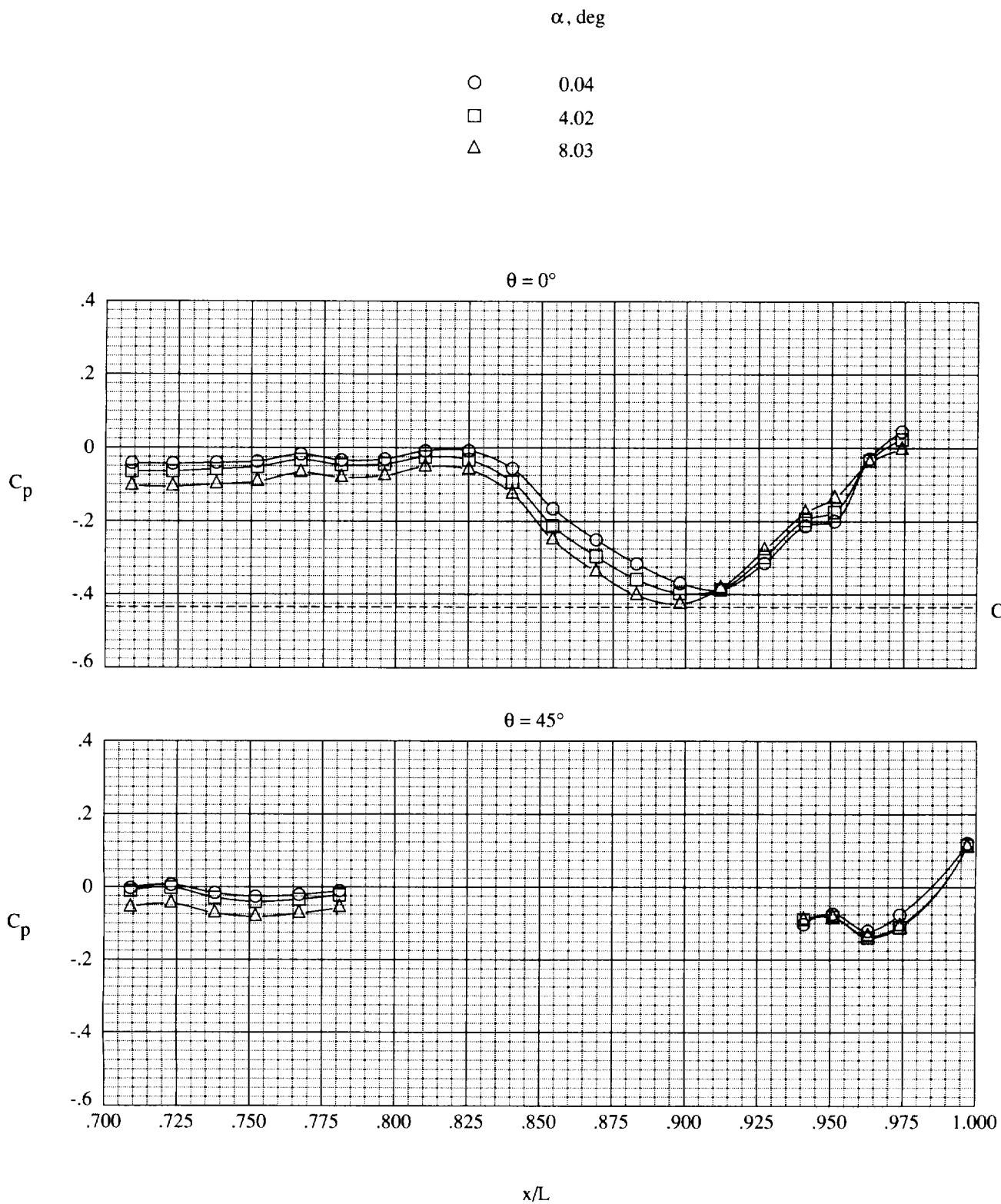
(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 21. Continued.



(e) Top and bottom interfairings.

Figure 21. Concluded.

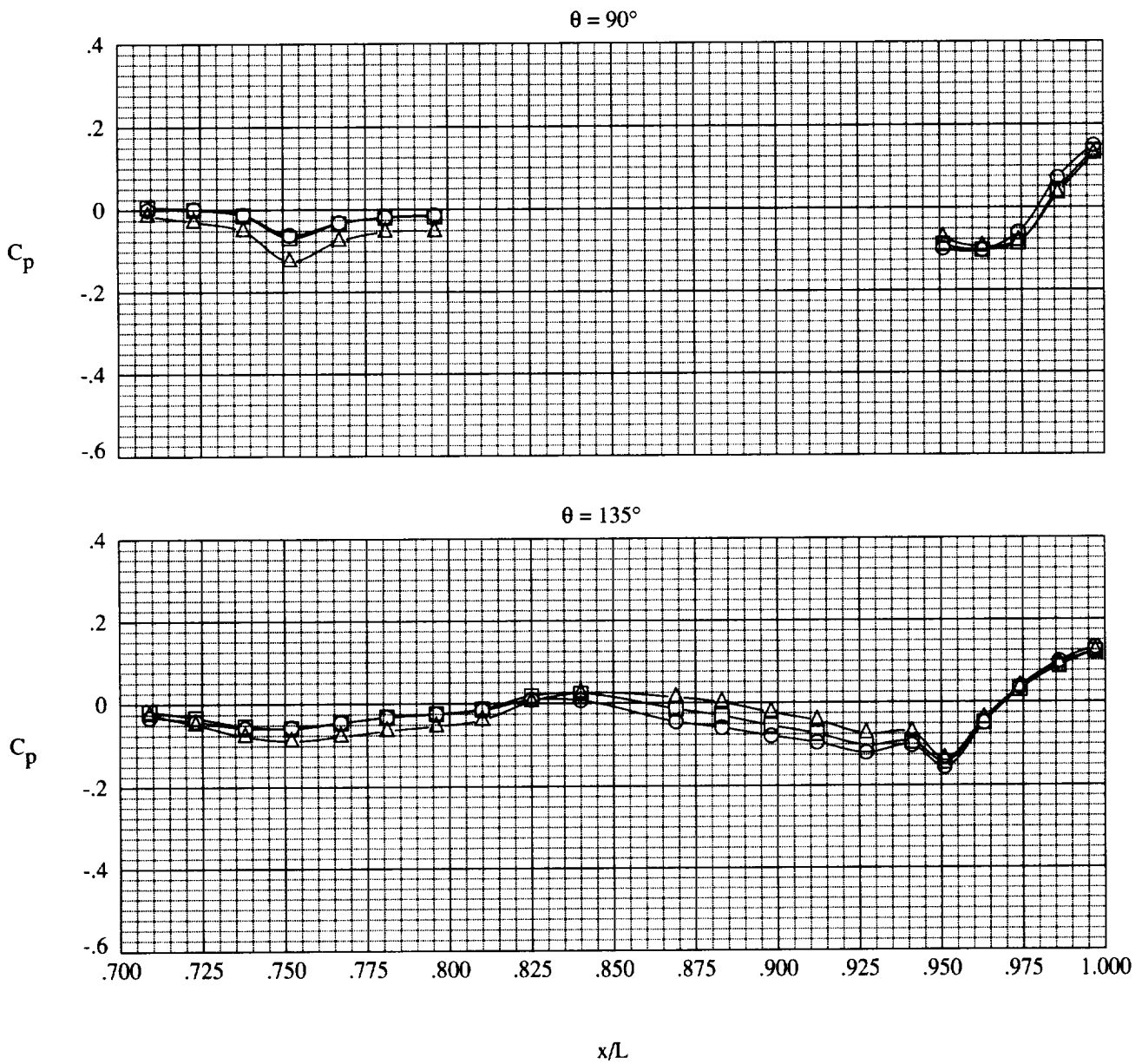


(a) $\theta = 0^\circ$ and 45° .

Figure 22. Effect of angle of attack on pressure distributions for $M = 0.8$, $NPR = 3.4$, HTAIL = aft position, and VTAIL = aft position.

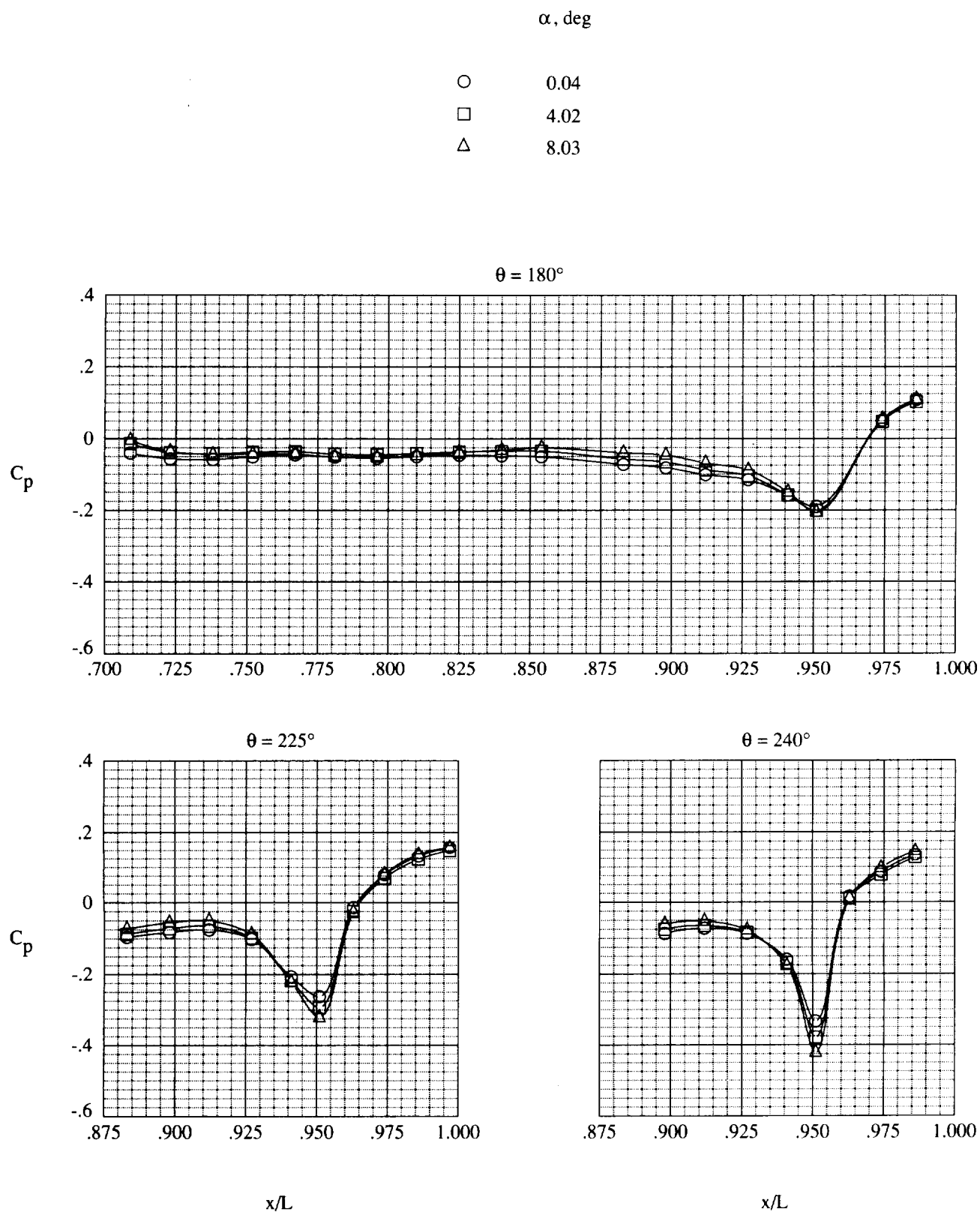
α , deg

- 0.04
- 4.02
- △ 8.03



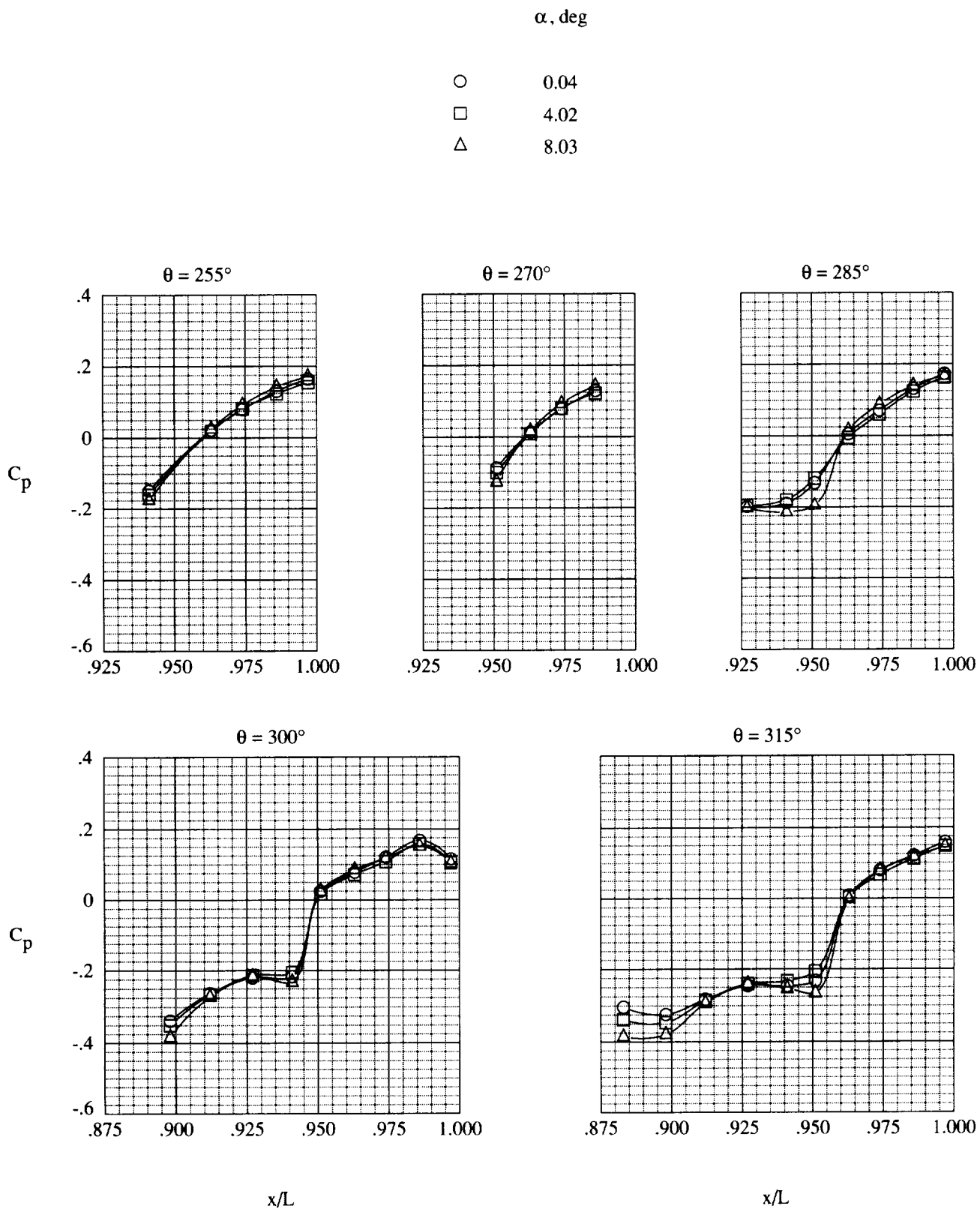
(b) $\theta = 90^\circ$ and 135° .

Figure 22. Continued.



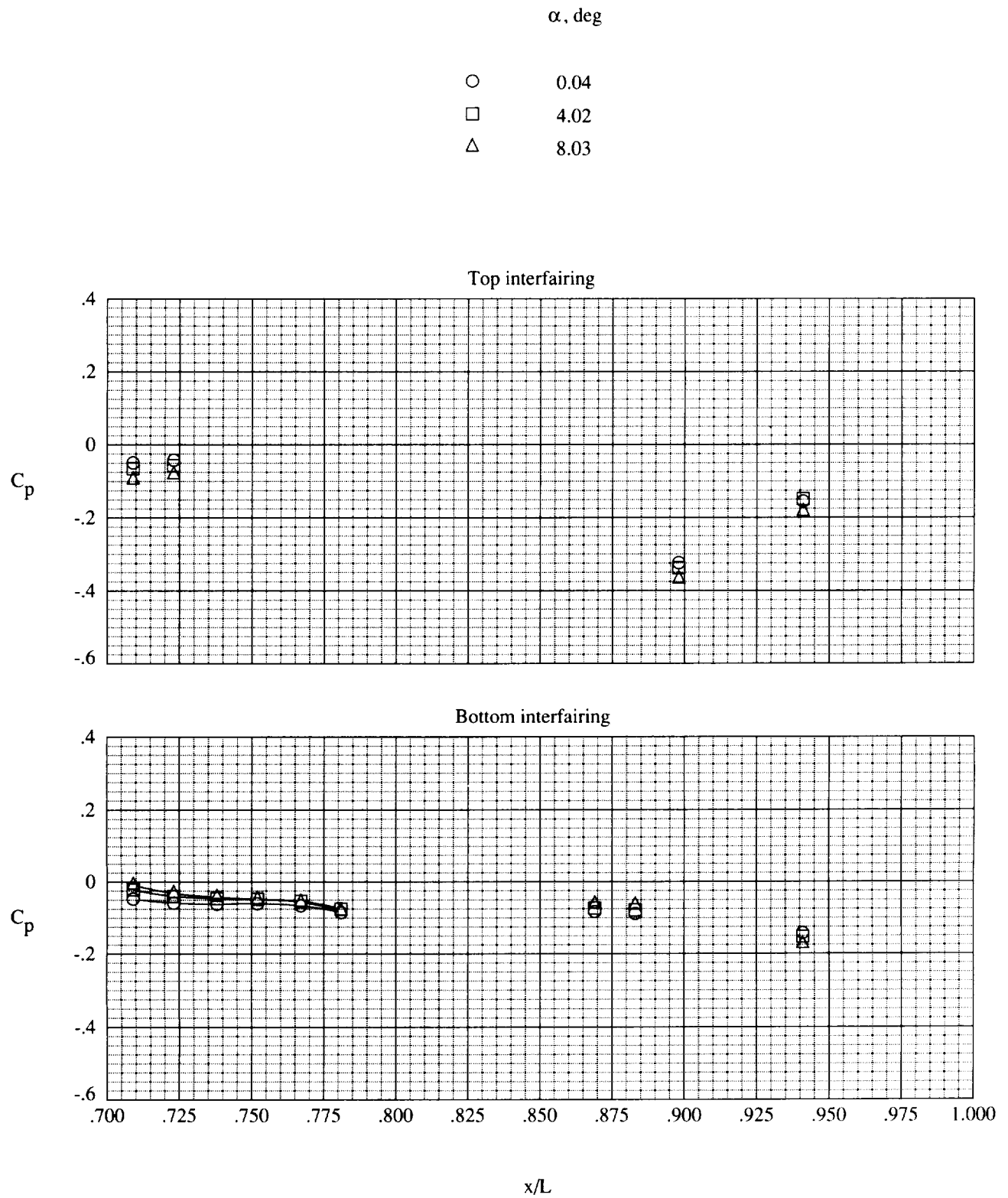
(c) $\theta = 180^\circ$, 225° , and 240° .

Figure 22. Continued.



(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 22. Continued.



(e) Top and bottom interfairings.

Figure 22. Concluded.

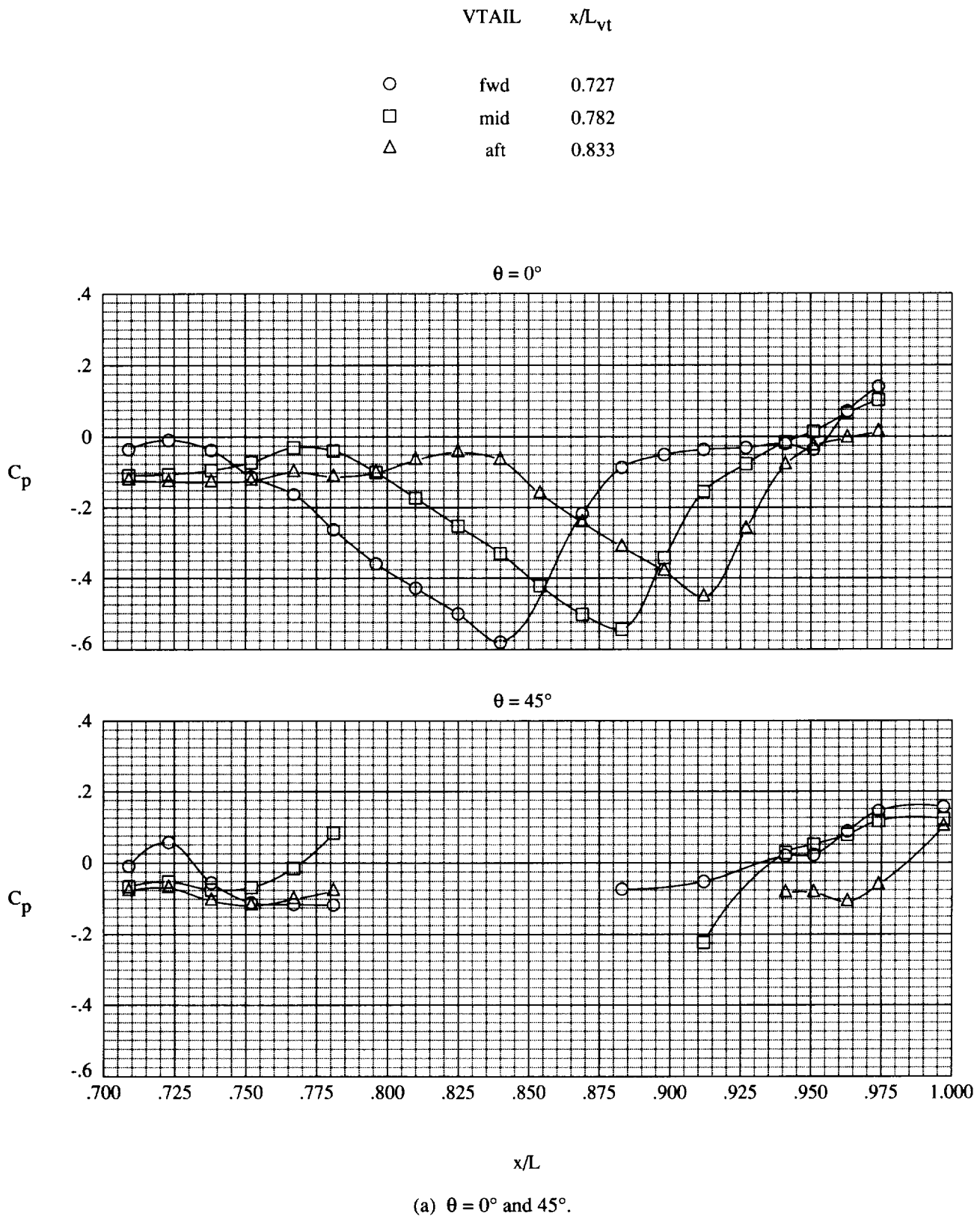
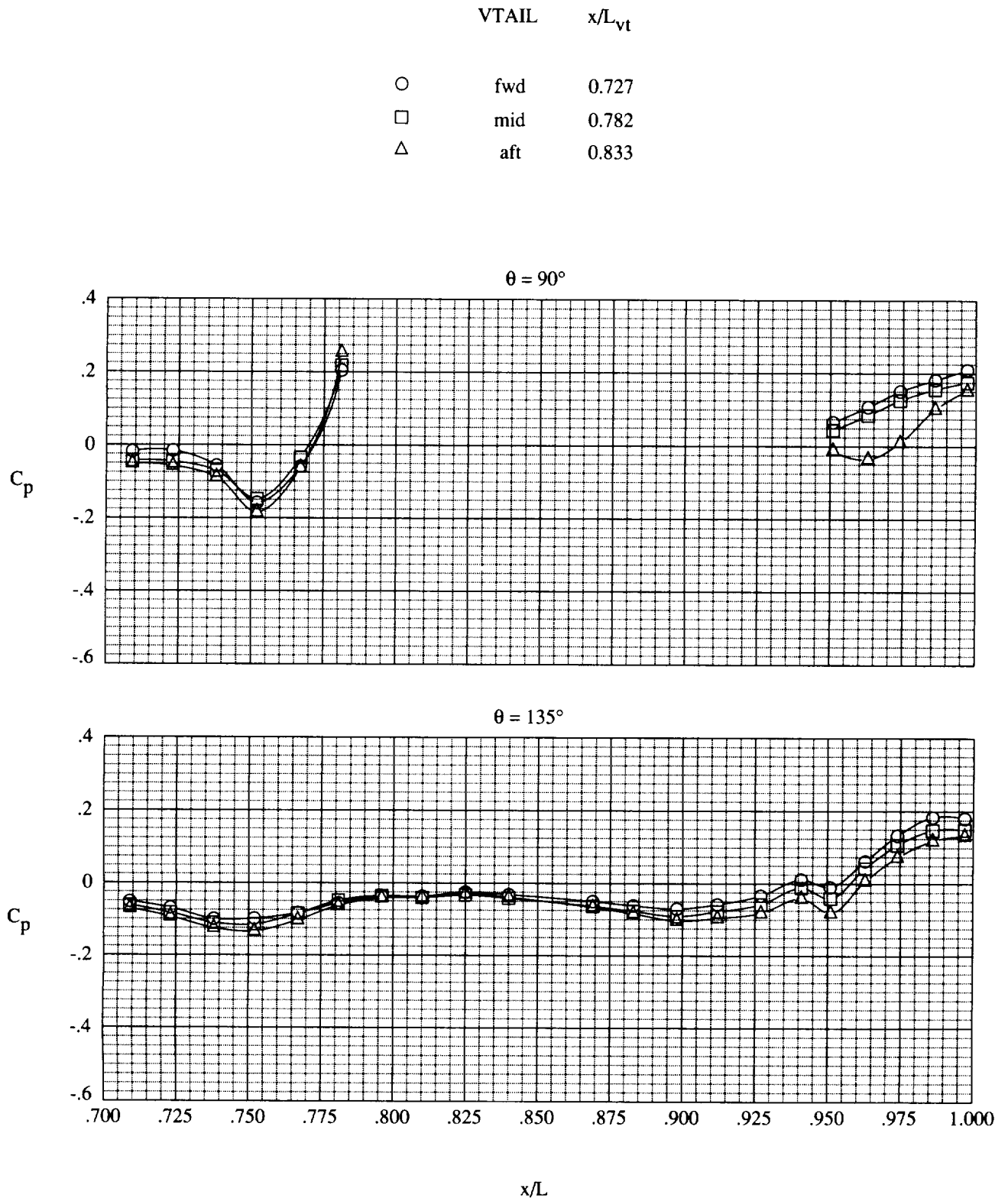


Figure 23. Effect of vertical-tail position on pressure distributions at positive angle of attack for $M = 0.9$, $\alpha = 8^\circ$, $NPR = 3.4$, and HTAIL = mid position.

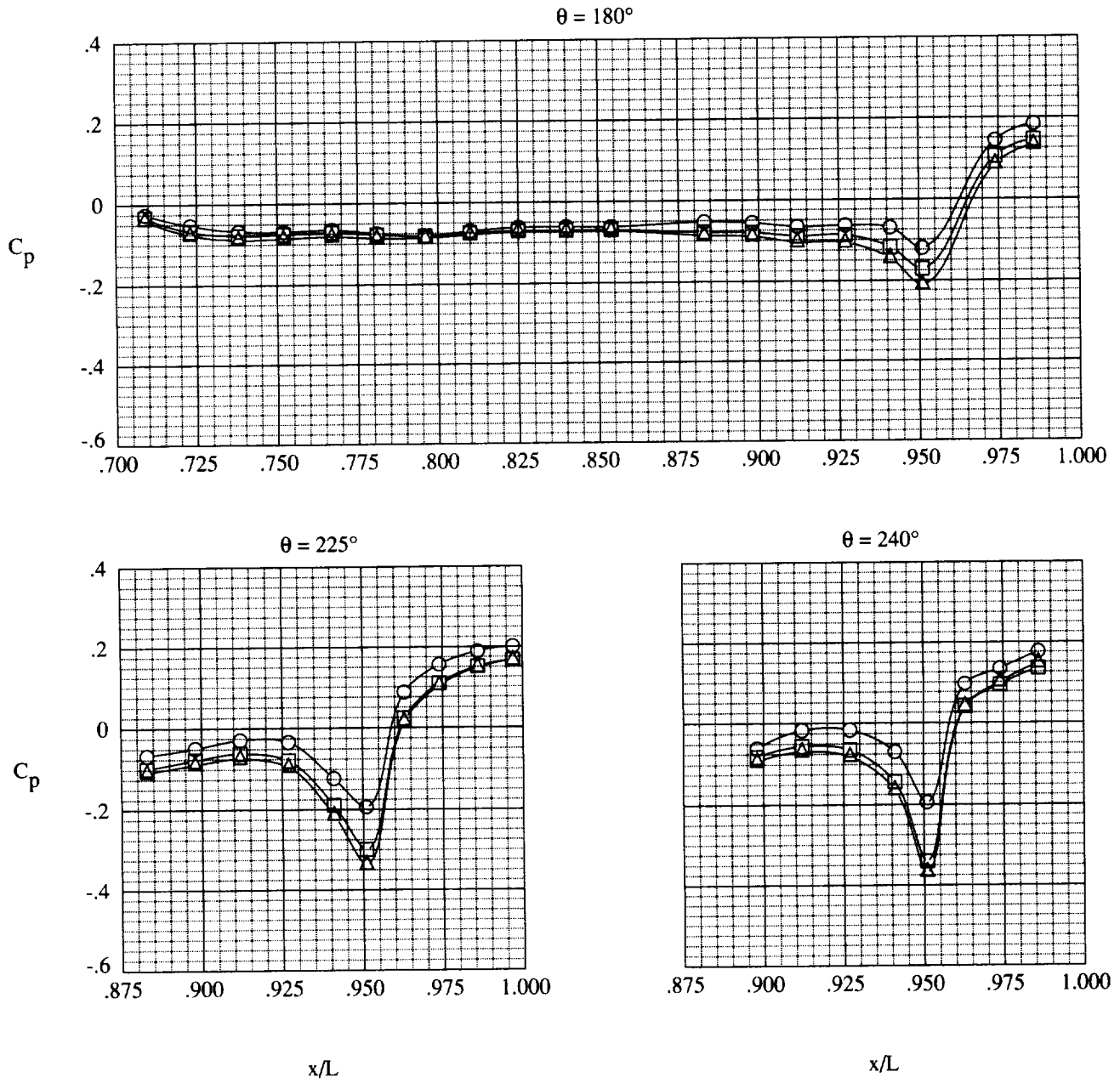


(b) $\theta = 90^\circ$ and 135° .

Figure 23. Continued.

VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833

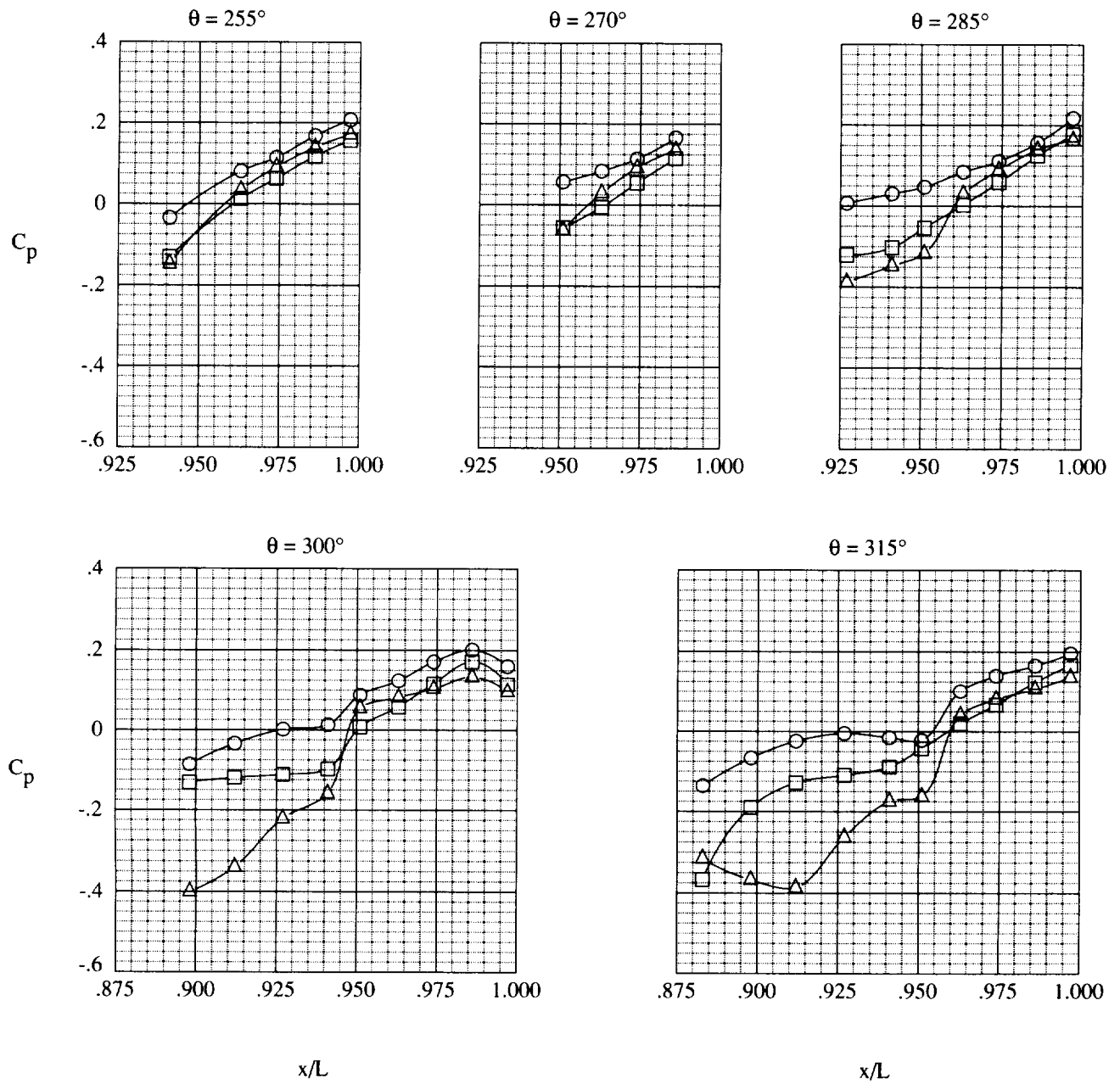


(c) $\theta = 180^\circ, 225^\circ$, and 240° .

Figure 23. Continued.

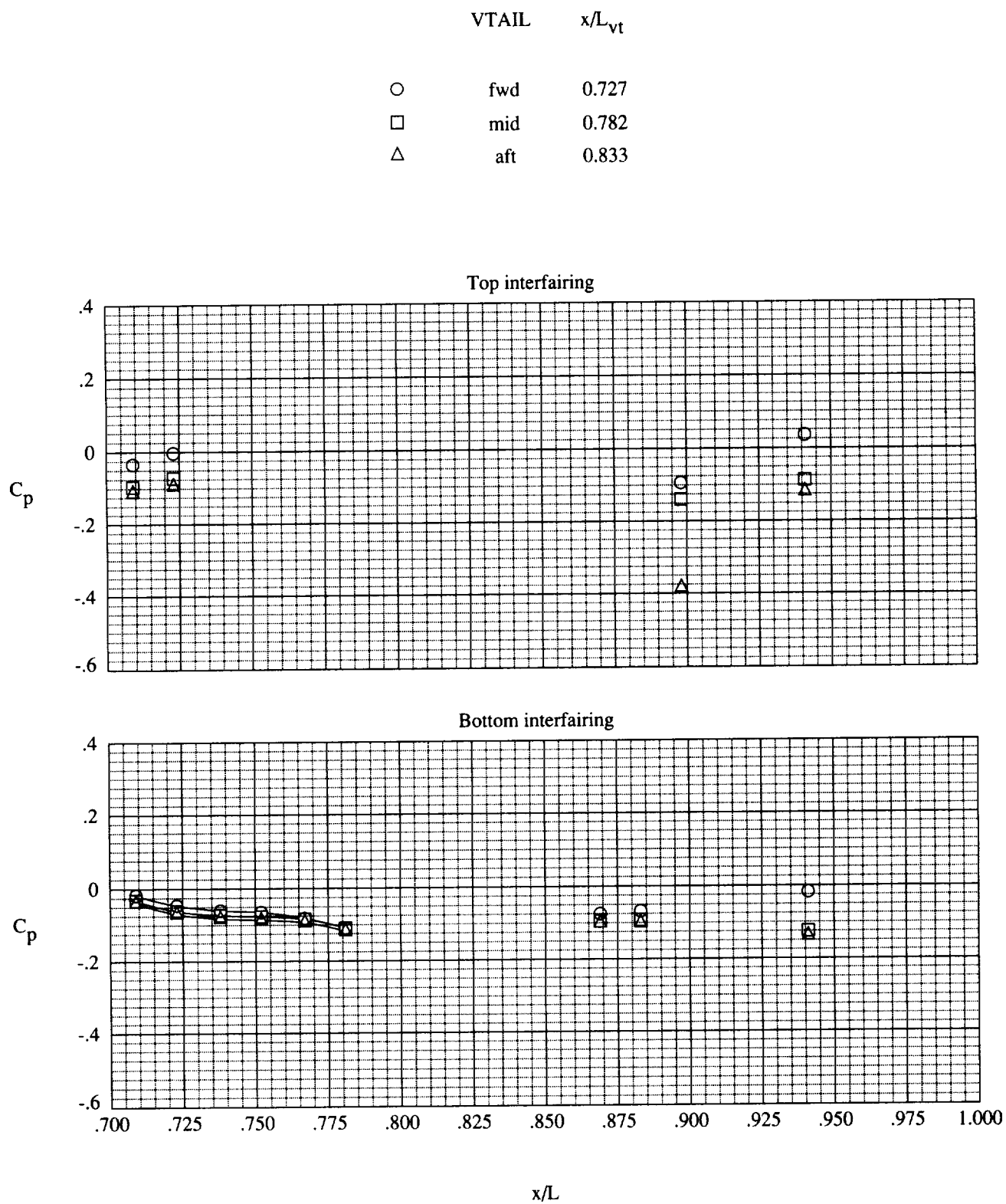
VTAIL x/L_{vt}

○	fwd	0.727
□	mid	0.782
△	aft	0.833



(d) $\theta = 255^\circ, 270^\circ, 285^\circ, 300^\circ$, and 315° .

Figure 23. Continued.



(e) Top and bottom interfairings.

Figure 23. Concluded.

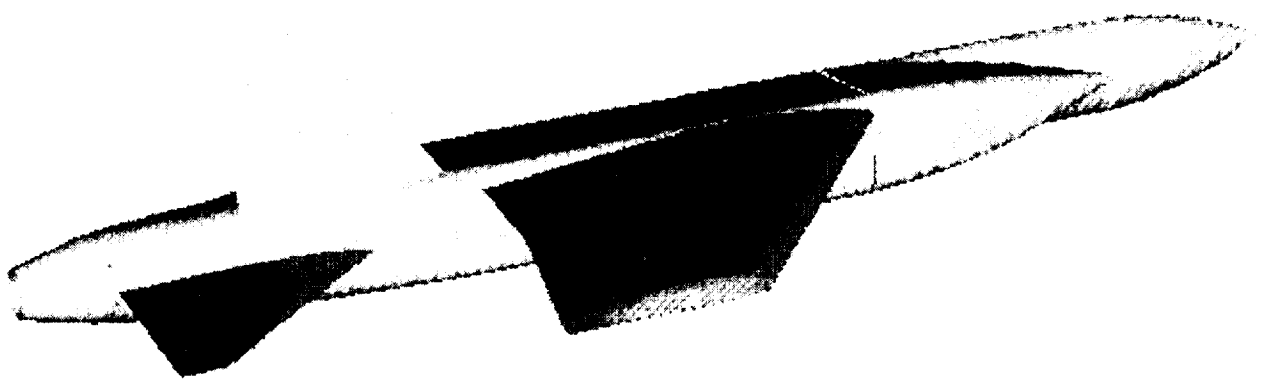
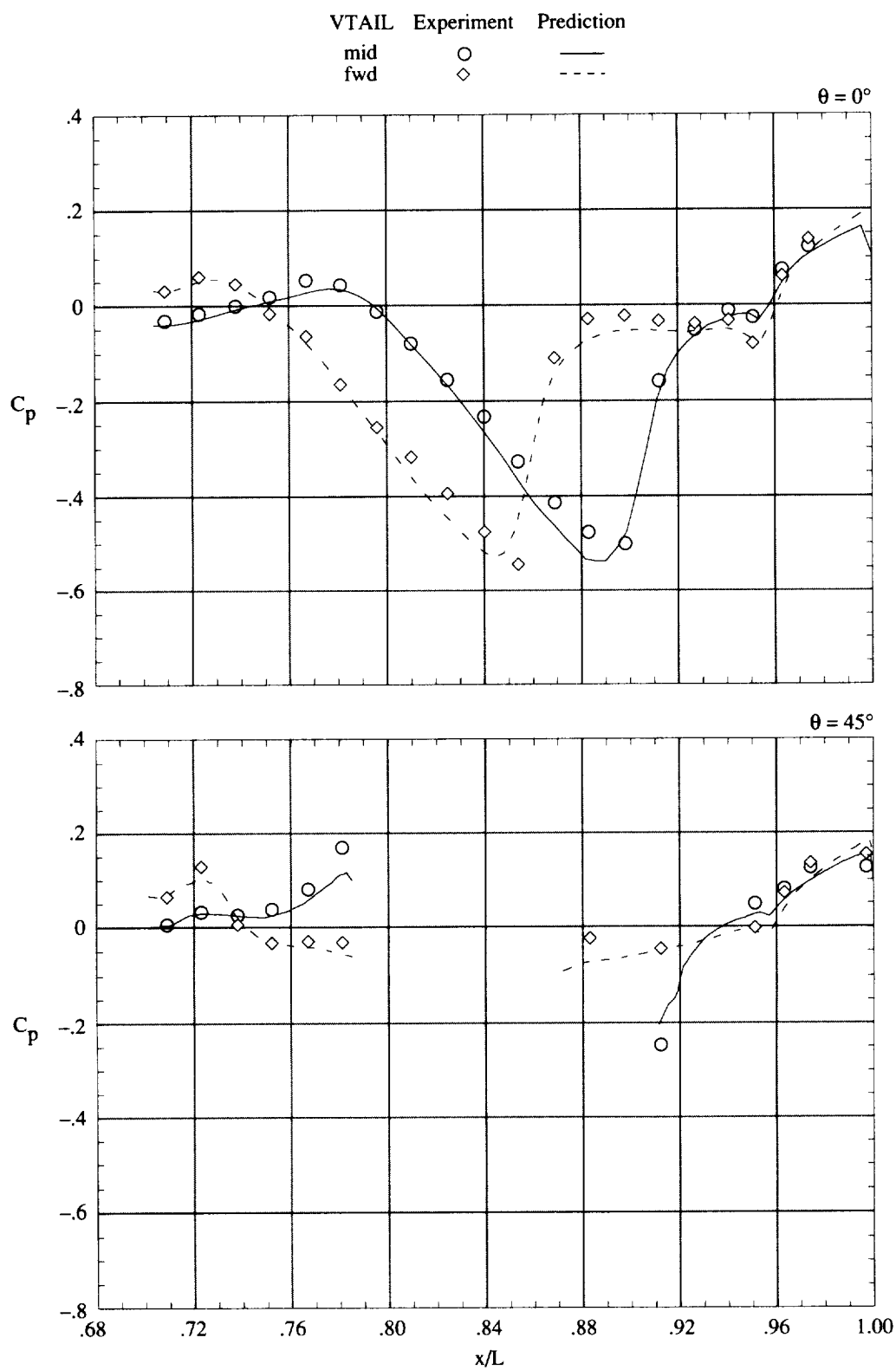
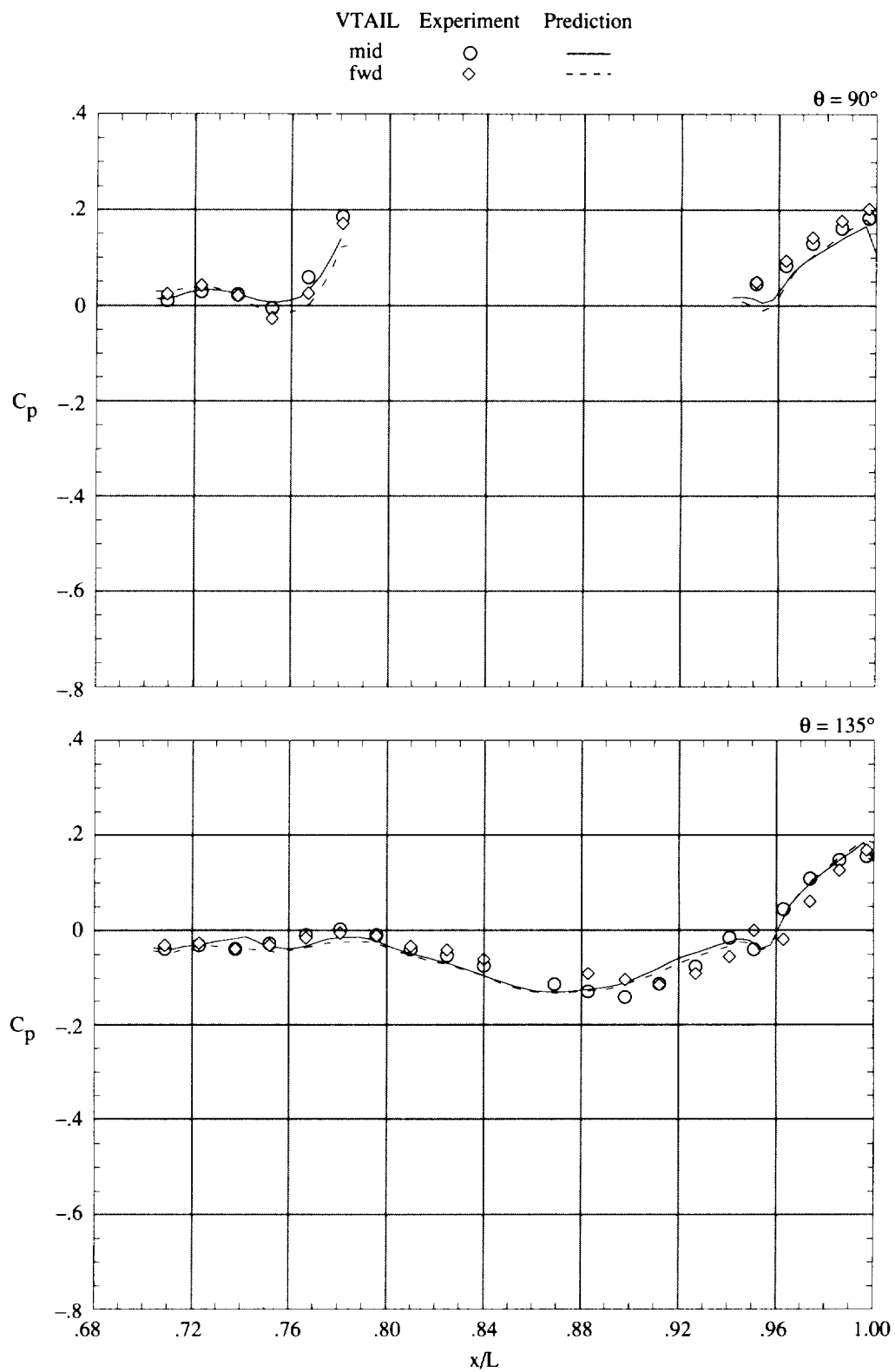


Figure 24. Computational model of twin-engine generic fighter. Vertical tail shown in forward position. Horizontal tail shown in mid position.



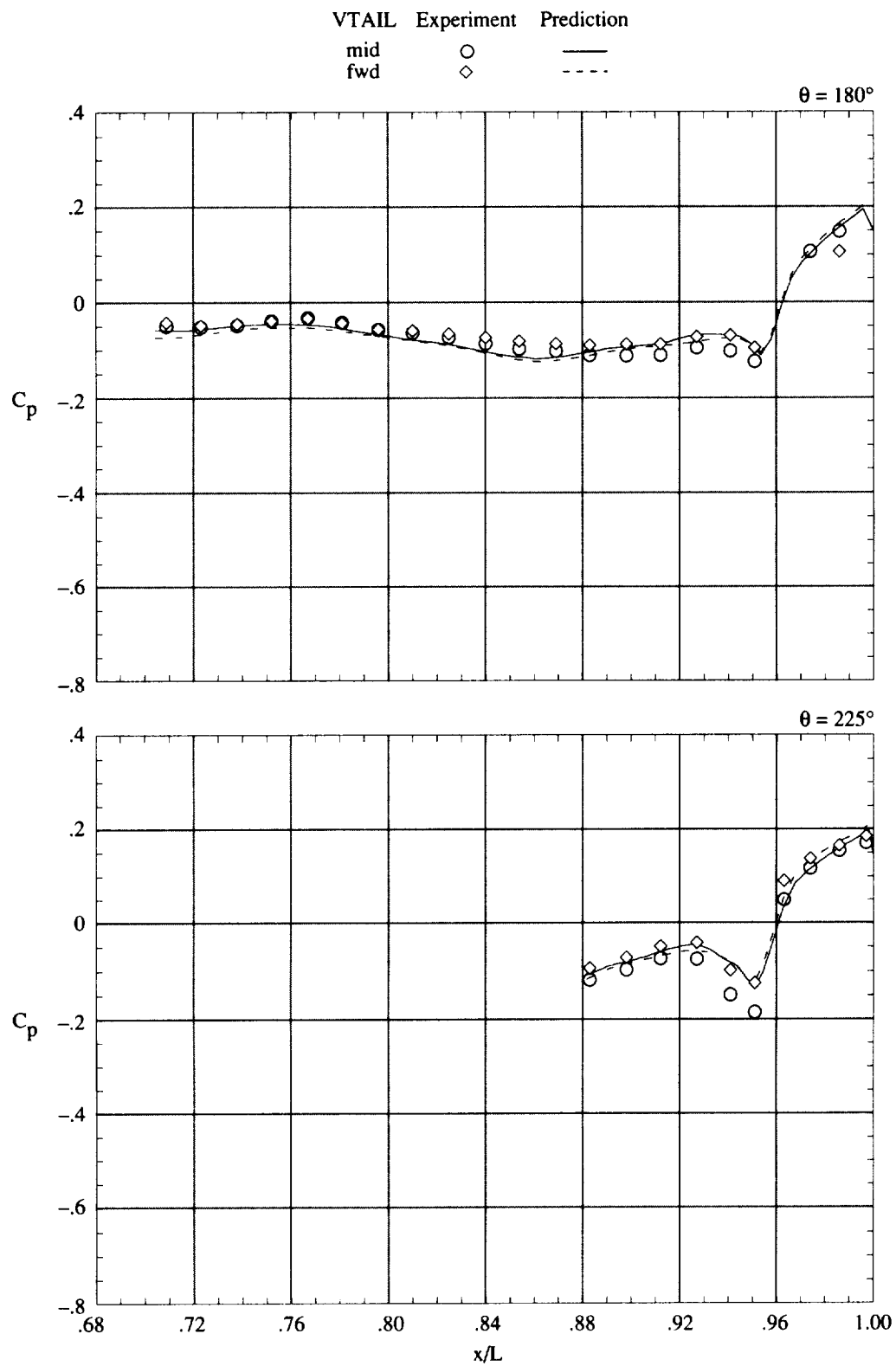
(a) $\theta = 0^\circ$ and 45° .

Figure 25. Predicted and experimental pressure coefficients on afterbody and nozzle surfaces for $M = 0.9$, $\alpha = 0^\circ$, and $NPR = 3.4$ with horizontal tails in mid position.



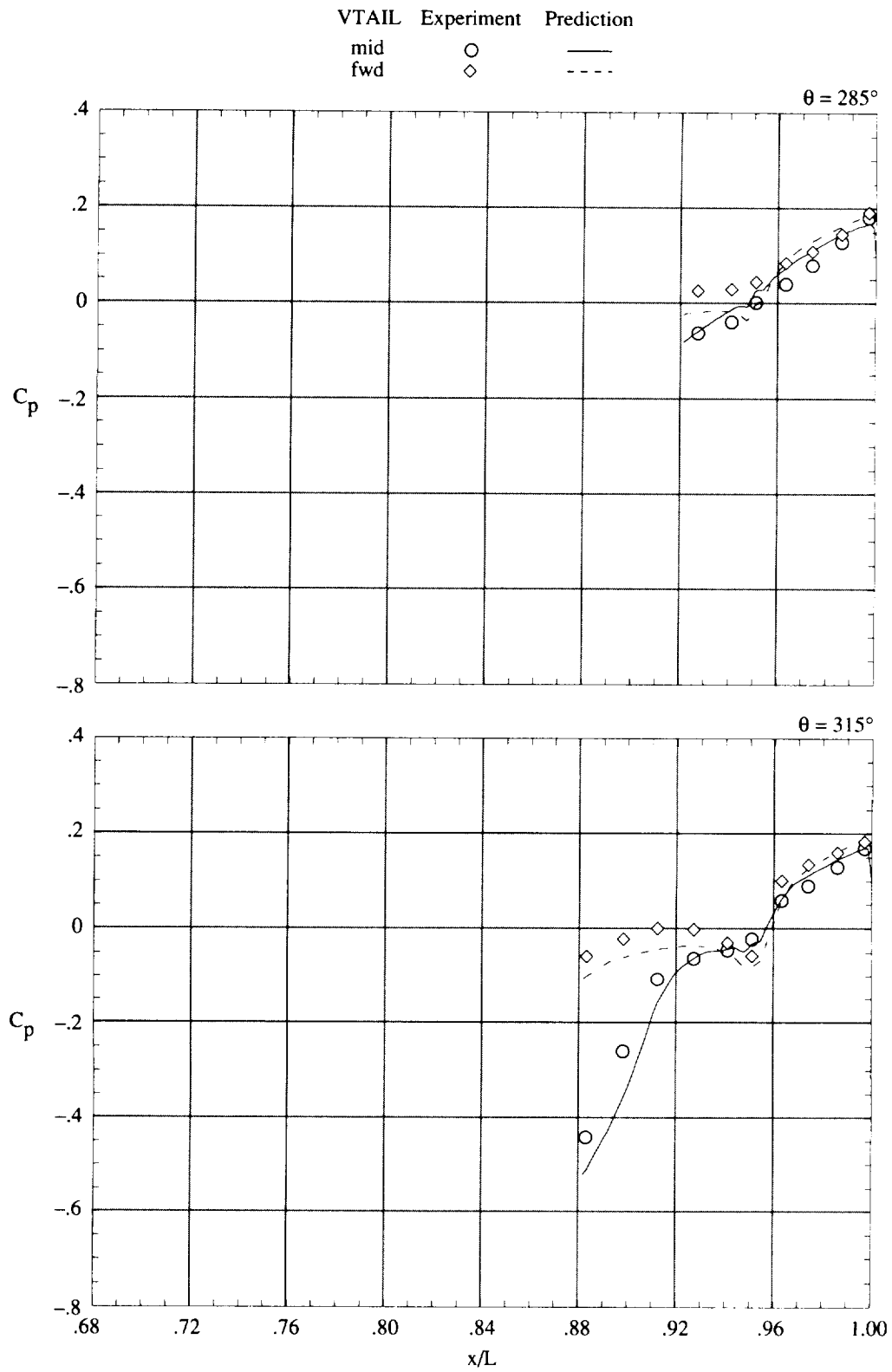
(b) $\theta = 90^\circ$ and 135° .

Figure 25. Continued.



(c) $\theta = 180^\circ$ and 225° .

Figure 25. Continued.



(d) $\theta = 285^\circ$ and 315° .

Figure 25. Concluded.

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